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PRESSURE DISTRIBUTION ON A VECTORED-THRUST V/STOL FIGHTER IN THE TRANSITION-SPEED RANGE

by

Raymond E. Mineck

Langley Directorate,
U.S. Army Air Mobility R&D Laboratory

and

Richard J. Margason

Langley Research Center
Hampton, Va. 23665

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16. Abstract

A wind-tunnel investigation has been conducted in the Langley V/STOL tunnel with a vectored-thrust V/STOL fighter configuration to obtain detailed pressure measurements on the body and on the wing in the transition-speed range. This investigation was a cooperative effort with the U.S. Air Force Flight Dynamics Laboratory. The vectored-thrust jet exhaust induced a region of negative pressure coefficients on the lower surface of the wing and on the bottom of the fuselage. The location of the jet exhaust relative to the wing was a major factor in determining the extent of the region of negative pressure coefficients.

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By Raymond E. Mineck
Langley Directorate, U.S. Army Air Mobility R&D Laboratory

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SUMMARY

A wind-tunnel investigation has been conducted in the Langley V/STOL tunnel with a vectored-thrust V/STOL fighter configuration to obtain detailed pressure measurements on the body and on the wing in the transition-speed range to verify aerodynamic prediction techniques developed by the Northrop Corporation under a U.S. Air Force contract. The model tested was equipped with one lift-jet and two vectored-thrust engine simulators. The vectored-thrust engine simulators could be tested at two alternate nozzle-exit locations and three nozzle-exit deflection angles (0^{o}, 45^{o}, and 90^{o}). Pressure data were obtained on the wing, fuselage, and engine nacelle through an angle-of-attack range from approximately 0^{o} to 20^{o} at four effective velocity ratios.

The jet exhaust induces a downwash on the wing as well as a region of negative pressure coefficients on the lower surface of the wing. The region is larger and the pressure coefficients are more negative for the front vectored-thrust nozzles than for the rear vectored-thrust nozzles. The jet exhaust also induces a region of negative pressure coefficients on the bottom of the fuselage. The induced pressure effects are largest at the smallest effective velocity ratio and at the location nearest to the jet. The effects decrease with increasing effective velocity ratio and increasing distance from the jet.

INTRODUCTION

The transition regime of VTOL aircraft covers the velocity range from hover (zero forward speed) to cruise (the speed necessary for wingborne flight). In the transition regime of VTOL aircraft, the jet exhaust effects tend to dominate the aerodynamic characteristics. To determine the aircraft performance, it is necessary to predict both the aerodynamic characteristics in forward flight and the effects of the pressure fields and velocity flow fields induced on the aircraft by the jet exhaust.

The Northrop Corporation, under a U.S. Air Force contract, has developed some methods for predicting the aerodynamic and jet-induced effects in transition flight for V/STOL aircraft. (See ref. 1.) The purpose of the present investigation, a cooperative effort with the U.S. Air Force Flight Dynamics Laboratory, was to obtain aerodynamic, flow-field, and pressure data to verify these prediction methods. The pressure data are presented in this report, and the aerodynamic and flow field data are presented in reference 2 without any comparisons with the prediction methods. Some comparisons of the prediction methods with the data may be found in Volume II of reference 1.

The model tested was a vectored-thrust V/STOL fighter configuration equipped with one lift-jet and two vectored-thrust engine simulators. Each vectored-thrust engine simulator was tested at two alternate nozzle-exit locations and up to three nozzle-exit deflection angles.

Pressure orifices were located on the wing, fuselage, and engine nacelle to obtain pressure data through an angle-of-attack range at four effective velocity ratios and through an effective-velocity-ratio range at two angles of attack. In addition, a limited amount of data was obtained through a range of sideslip angles at four effective velocity ratios. All data were obtained at Mach numbers ranging from 0 to 0.22.

SYMBOLS

The units used for the physical quantities defined in this paper are given in the International System of Units (SI) and parenthetically in the U.S. Customary Units. Conversion factors for the SI System are presented in reference 3.

Symbol notation within brackets is the computer notation in the machine pressure tabulations.

A_j effective jet-exit area (that is, sum of exit areas of devices producing thrust), cm^2 (in^2) (see fig. 1(a))

b wing span, 102.24 cm (40.25 in.)

c [C] local wing chord

C_p [CP] pressure coefficient, $\frac{p_l - p_\infty}{q_\infty}$

C_p' [CP'] hover pressure coefficient, $\frac{p_l - p_\infty}{T/2A_j}$

C_T thrust coefficient, $\frac{T}{q_\infty S}$

l [L] fuselage length, 128.14 cm (50.45 in.)

p_l local static pressure, N/m^2 (lbf/ft^2)

p_∞ tunnel free-stream static pressure, N/m^2 (lbf/ft^2)

q_j effective jet dynamic pressure, $\frac{T}{2A_j}$, N/m^2 (lbf/ft^2)

q_∞ [Q] tunnel free-stream dynamic pressure, N/m^2 (lbf/ft^2)

S wing area, 2090 cm^2 (2.25 ft^2)

T static thrust, N (lbf)

V_j effective jet velocity, m/sec (ft/sec)

V_∞ free-stream velocity, m/sec (ft/sec)

$\left(\frac{V_\infty}{V_j}\right)_e$ effective velocity ratio, $\sqrt{\frac{q_\infty}{T/2A_j}}$

X,Y,Z model body axes

x,y,z [X,Y] distance measured in body-axis system

y_{max} [YMAX] maximum half-width of the fuselage, 6.35 cm (2.50 in.)

α angle of attack, deg

β angle of sideslip, deg

ρ_∞ free-stream density, kg/m^3 (lbm/ft^3)

ρ_j jet density, kg/m^3 (lbm/ft^3)

η	fraction of semispan, $\frac{y}{b/2}$

Model component designations:

B	body
f	flap
J	lift jet
N_1	nacelles with front vectored-thrust nozzles, used with appropriate nozzle deflection angle in degrees
N_2	nacelles with rear vectored-thrust nozzles, used with appropriate nozzle deflection angle in degrees
plug	nacelle inlet closed with an elliptical plug
W	wing
0, 45, 90	nozzle deflection angles in degrees

MODEL AND APPARATUS

The model, presented in figures 1 and 2, was a vectored-thrust V/STOL fighter configuration equipped with one lift-jet and two vectored-thrust engine simulators. The lift-jet simulator, located within the fuselage, was a simple convergent nozzle without an external air intake. The lift-jet exit was located on the center line of the fuselage bottom 7.67 cm (3.02 in.) forward of the model moment reference center which was at the 25-percent-chord position of the mean aerodynamic chord. The lift jet exhausted 90° from the horizontal plane. The vectored-thrust engine simulators, mounted in removable fuselage-supported nacelles, were of the ejector type which induced inlet flow. The vectored-thrust jet exits could be set 2.67 cm (1.05 in.) forward of or 18.62 cm (7.33 in.) aft of the model moment reference center and could also be vectored in the X-Z plane at angles of 0°, 45°, and 90° from the horizontal plane.

The model was equipped with pressure orifices on the left wing (fig. 1(a) and table 1(a)), on the bottom of the fuselage (fig. 1(b) and table 1(b)), and along the center line of the top and bottom of the left nacelle (fig. 1(b) and table 1(c)). The pressure orifices

were located chordwise at four spanwise stations of the left wing, but the nacelle and the nacelle fairing covered up several of the orifices at the inboard station so that a detailed survey at this location was not always possible.

Two elliptical plugs (fig. 1(b)) were used to close the vectored-thrust engine inlets in some of the tests. The base diameter of the plug was 12.01 cm (4.73 in.) and the depth was 6.88 cm (2.71 in.). A 45° flap with a chord equal to 25 percent of the local wing chord was also used in some of the tests.

TEST AND PROCEDURES

The investigation was conducted in the Langley V/STOL tunnel which is a closed-return, atmospheric tunnel with a test section of 4.42 meters (14.50 ft) by 6.63 meters (21.75 ft). All testing was performed out of ground effect. Table 2 presents the test conditions for most of the tests.

The basic body-wing-nacelles configuration was tested with the vectored-thrust nozzle exits either forward of (N_1) or aft of (N_2) the model moment reference center. At each location, up to three nozzle-exit deflection angles were tested. In addition, tests were performed with the lift-jet simulator, with the flaps extended, and with the engine inlets closed by elliptical plugs.

The thrust of each lift-jet simulator was calibrated as a function of a reference pressure. The reference pressure for the lift jet was the difference between the lift-jet exit total pressure and free-stream static pressure. The reference pressure for the vectored-thrust engine simulators was the ejector plenum static pressure. A description of the calibration method appears in reference 4.

The static forces due to thrust were measured in the body-axis system. These forces were then used to determine the actual thrust direction in the X-Z plane. The nominal and actual thrust directions measured from the horizontal plane are shown in table 3.

The effective velocity ratio was determined from the measured thrust and an effective jet-exit area A_j. The effective jet-exit area was the sum of the exit areas of the thrust devices in use – the right and left vectored-thrust jets and/or the lift jet. The effective velocity ratio was determined as follows:

$$\left(\frac{V_\infty}{V_j}\right)_e = \sqrt{\frac{\rho_\infty V_\infty^2}{\rho_j V_j^2}} = \sqrt{\frac{q_\infty}{T/2A_j}} \qquad (1)$$

From equation (1), it can be seen that the effective velocity ratio is zero for hover and increases with increasing forward speed. The effective velocity ratio for the unpowered case (T = 0) is infinity.

From the definitions of thrust coefficient and effective velocity ratio, the relationship of the two parameters may be determined as follows:

$$C_T = \frac{2A_j}{S} \frac{1}{\left(\frac{V_\infty}{V_j}\right)_e^2} \tag{2}$$

The lift-jet configuration, the vectored-thrust-jet configuration, and the lift-jet—vectored-thrust-jet configuration all have different effective jet-exit areas. Figure 3 shows the different values of thrust coefficient which are associtated with each jet configuration for a given effective velocity ratio. Effective velocity ratio was chosen as the best parameter to present the data through the transition-speed range because it starts at zero for hover and increases toward a finite value at cruise.

The pressure data on the wing, fuselage, and nacelle have been referenced to the tunnel free-stream static pressure. In the transition-speed range, the pressure data have been expressed in coefficient form by nondimensionalizing the free-stream dynamic pressure, that is,

$$C_p = \frac{p_l - p_\infty}{q_\infty} \tag{3}$$

In hover, the pressure data have been nondimensionalized by the jet effective dynamic pressure and denoted by a prime; thus;

$$C_p' = \frac{p_l - p_\infty}{q_j} = \frac{p_l - p_\infty}{T/2A_j} \tag{4}$$

PRESENTATION OF DATA

The pressure data have been presented in both tabular and graphical form. The graphical format is intended to present only the trends and includes selected parts of the angle-of-attack and angle-of-sideslip data. The tabular format presents all the angle-of-attack and angle-of-sideslip data as well as the effective-velocity-ratio data. For convenience, the data have been arranged so that the data found in a figure of a given number may also be found in the table of the same number. An index to the pressure profile data (tables 4 to 54) is given on pages 17 to 23.

In general, each figure has four parts with part (a) presenting the unpowered data $\left[\left(V_\infty/V_j\right)_e = \infty\right]$ and the other parts presenting data for decreasing values of effective velocity ratio. Three different plot formats for the wing, fuselage, and nacelle are used to graphically present the data. For the wing, each plot represents the variation of the pressure coefficient across the wing chord. The x/c is the fraction of the local wing chord measured from the leading edge with the positive direction aft. Each horizontal row of plots shows the variation of the pressure profile with angle of attack or with angles of sideslip at a semispan station and each vertical row of plots shows the variation of the pressure profile with semispan station at an angle of attack or angle of sideslip.

For the fuselage, each plot shows the variation of the pressure coefficient across the bottom of the fuselage at a constant angle of attack. The x/l is the fraction of the fuselage length measured from the nose with the positive direction aft.

For the nacelle, each plot represents the variation of the pressure coefficient across the nacelle and wing at a constant angle of attack. The local wing chord was chosen as the reference length with the wing leading edge as the origin and the positive direction aft. At locations forward of the leading edge, the values of x/c are negative.

As an aid in identifying the various configurations, a system of letters and numbers is used to represent the various model components. The basic wing-body with the lift jet, but without the vectored-thrust nacelles, is BWJ; whereas, the wing-body with the front or rear vectored-thrust nacelles is BWN_1XX or BWN_2XX, respectively, with XX indicating the vectored-thrust nozzle-exit deflection angle (0^o, 45^o, or 90^o). The lift-jet—vectored-thrust-nacelles configuration is BWN_1XXJ or BWN_2XXJ.

All angle-of-attack and effective-velocity-ratio data are presented at 0^o sideslip and all sideslip data are presented at 0^o angle of attack. The results are presented as follows:

Configuration	Effect of angle of attack on pressure profile of –			Effect of sideslip angle on wing pressure profile	Effect of effective velocity ratio on pressure profile of –		Effect of thrust in hover on wing pressure profile
	Wing	Fuselage	Nacelle		Wing	Fuselage	
	Table & figure	Table & figure	Table & figure	Table & figure	Table	Table	Table
BWJ	4	17		27	30	39	46
BWJf	5						
BWN_10	6						
BWN_10f	7						
BWN_145	8	18			31	40	47
BWN_245	9	19			32	41	48
BWN_190	10	20	24	28	33	42	49
BWN_190f	11				34		50
BWN_290	12	21	25	29	35	43	51
BWN_290f	13				36		52
BWN_190J	14	22			37	44	53
BWN_290J	15	23			38	45	54
$BWN_190plug$	16		26				

Some of the data for the various locations are missing because of problems with the pressure orifices. Also, the ordinate scales used in figures 20(d) and 22 are different from those used in the rest of the fuselage plots because of the high pressure coefficients obtained.

DISCUSSION

Because of the large amount of data contained herein, no detailed analysis of the results is attempted. However, some of the more obvious trends are discussed. In the transition-speed range of VTOL aircraft, the position of the jet exhaust has a large effect on the pressures on the aircraft. The three jet-exhaust locations used in this investigation were for the lift jet, the front vectored-thrust jets, and the rear vectored-thrust jets. The unpowered case for each configuration, represented by $\left(V_\infty/V_j\right)_e = \infty$, was chosen as the baseline for determining the effects of jet thrust.

Effect of Angle of Attack on the Wing Pressures

The chordwise pressure distribution on the wing is presented at four spanwise stations through a range of angle of attack in figures 4 to 16. In general, without thrust at 0° angle of attack, the pressure coefficients on the upper and lower surfaces of the wing have similar distributions, but with thrust at 0° angle of attack, the pressure coefficients on the lower surface are more negative than on the upper surface. With thrust at 0° geometric angle of attack, the wing is operating at a negative local angle of attack because the jet exhaust is inducing a downwash.

In addition to inducing a downwash on the wing, the vectored-thrust jet exhaust induced a region of negative pressure coefficients on the lower surface of the wing. The low-pressure region is larger and the pressure coefficients are more negative for the front vectored-thrust nozzles than for the rear vectored-thrust nozzles. (See figs. 10(c) and 12(c).)

These induced effects are largest at the smallest effective velocity ratio and at the location nearest to the jet. The effects decrease with increasing effective velocity ratio and increasing distance from the jet.

To determine the effect of inlet flow into the vectored-thrust jets, the forward 90° nozzles were tested with the inlets closed by elliptical plugs (fig. 16). Examination of the force and moment data for the inlet open and the inlet closed (ref. 2) indicates that closing the inlets decreases the angle of attack approximately 2°. The pressure data for the inlets open (table 10(a)) and for the inlets closed (table 16) have been plotted, and the curves have been integrated to obtain the section lift coefficients for the different angles of attack.

A comparison showed that with the inlets closed, the wing must be operated at an angle of attack between 1° and 2° greater to obtain the same section lift coefficient obtained with the inlets open.

Effect of Angle of Attack on the Fuselage Pressures

The pressure distribution on the bottom of the fuselage is presented through an angle-of-attack range in figures 17 to 23. For all configurations without power, the pressures on the fuselage were very close to the free-stream static pressure. With thrust, the lift jet (figs. 17(b) to 17(d)) induced large negative pressure coefficients on the bottom of the fuselage with the peak negative pressure coefficients decreasing slightly with increasing angle of attack. Also, the blockage due to the jet caused a region of positive pressure coefficients upstream of the jet.

With thrust, the vectored-thrust jets (for example, fig. 18) induced a region of negative pressure coefficients which decreased with increasing angle of attack and increasing effective velocity ratio. The peak negative pressure coefficients are larger for the 90° vectored-thrust jets (figs. 20(b) to 20(d)) than for the 45° vectored-thrust jets (figs. 18(b) to 18(d)).

Effect of Angle of Attack on the Nacelle Pressure

The pressure distribution along the center line of the left vectored-thrust jet nacelle is presented through an angle-of-attack range in figures 24 to 26. For the vectored-thrust jets without thrust (figs. 24(a) and 25(a)), the pressure distribution shows the same trends as an airfoil would through a range of angle of attack with large negative pressure coefficients near the leading edge of the upper surface and positive pressure coefficients on the lower surface. The addition of thrust changes the pressure coefficients near the front of the nacelle.

To determine the effects of inlet flow, the front nozzles were closed with elliptical plugs (fig. 26). The plugs prevented the large negative pressure coefficients on the upper surface from forming up to the stall angle of attack.

Effect of Angle of Sideslip on Wing Pressures

The pressure distribution on the left wing is presented through an angle-of-sideslip range in figures 27 to 29. The pressure distribution on the wing changes with sideslip angle because of the change in effective chord and camber and because of the interference of the fuselage and nacelle with the flow over the wing. The change in pressure distribution produces an increase in lift on the left wing as the sideslip becomes more negative

and a decrease in lift as the sideslip becomes more positive. The effects are larger for the vectored-thrust configuration than for the lift-jet configuration.

CONCLUDING REMARKS

A wind-tunnel investigation has been conducted on a vectored-thrust V/STOL configuration in the Langley V/STOL tunnel. The location of the jet exhaust has a large effect on the pressures on the aircraft. The vectored-thrust jet exhaust induces a downwash on the wing as well as a region of negative pressure coefficients on the lower surface of the wing. The region is larger and the pressure coefficients are more negative for the front vectored-thrust nozzles than for the rear vectored-thrust nozzles. The jet exhaust also induces a region of negative pressure coefficients on the bottom of the fuselage. The induced pressure effects are largest at the smallest effective velocity ratio and at the location nearest to the jet. The effects decrease with increasing effective velocity ratio and increasing distance from the jet.

Langley Research Center,
National Aeronautics and Space Administration,
Hampton, Va., November 13, 1973.

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TABLE 1

LOCATION OF PRESSURE ORIFICES

(a) Wing

	Distance from fuselage center line, y		Distance from fuselage nose to wing leading edge		Local wing chord, c	
	cm	in.	cm	in.	cm	in.
$\eta = 0.250$	12.80	5.040	51.61	20.319	23.85	9.389
$\eta = 0.387$	19.79	7.792	52.81	20.793	21.98	8.655
$\eta = 0.524$	26.78	10.545	54.02	21.266	20.12	7.921
$\eta = 0.800$	40.77	16.050	56.42	22.212	16.39	6.453

x/c	$\eta = 0.250$		$\eta = 0.387$		$\eta = 0.524$		$\eta = 0.800$	
	x, cm	x, in.	x, cm	x, in.	x, cm	x, in.	x, cm	x, in.
0	0	0	0	0	0	0	0	0
.010	.239	.094	.221	.087				
.015					.302	.119		
.025	.597	.235	.549	.216	.503	.198	.409	.161
.050	1.191	.469	1.100	.433	1.006	.396	.820	.323
.100	2.385	.939	2.200	.866	2.012	.792	1.638	.645
.150	3.576	1.408	3.297	1.298	3.017	1.188	2.459	.968
.250	5.961	2.347	5.497	2.164	5.029	1.980	4.097	1.613
.400	9.540	3.756	8.793	3.462	8.047	3.168	6.556	2.581
.550	13.117	5.164	12.090	4.760	11.067	4.357	9.014	3.549
.700	16.693	6.572	15.390	6.059	14.084	5.545	11.473	4.517
.850	20.272	7.981	18.687	7.357	17.102	6.733	13.932	5.485
.950	22.657	8.920	20.884	8.222	19.114	7.525	15.570	6.130

TABLE 1.- Continued

LOCATION OF PRESSURE ORIFICES

(b) Fuselage

x/l	y = 0.000 cm (0.00 in.)		y = 3.175 cm (1.25 in.)		y = 5.080 cm (2.00 in.)	
	x, cm	x, in.	x, cm	x, in.	x, cm	x, in.
0.130	16.64	6.55				
.219	25.53	10.05				
.286	36.63	14.42				
.331	42.34	16.67				
.353	45.21	17.80	45.21	17.80	45.21	17.80
.375	48.11	18.94				
.397			50.93	20.05	50.93	20.05
.420	53.80	21.18	53.80	21.18	53.80	21.18
.442	56.64	22.30	56.64	22.30	56.64	22.30
.464	59.49	23.42	59.49	23.42	59.49	23.42
.487	62.36	24.55	62.36	24.55	62.36	24.55
.509	65.23	25.68	65.23	25.68	65.23	25.68
.531	68.07	26.80	68.07	26.80	68.07	26.80
.554	70.92	27.92				
.576	73.79	29.05				
.598	76.66	30.18	76.66	30.18	76.66	30.18
.620	79.50	31.30				
.643	82.35	32.42	82.35	32.42	82.35	32.42
.665	85.22	33.55	85.22	33.55	85.22	33.55
.687	88.09	34.68				
.710	90.93	35.80				
.732	93.78	36.92				

TABLE 1.- Concluded

LOCATION OF PRESSURE ORIFICES

(c) Nacelle

[Nacelle center line at $\eta = 0.250$]

x/c	Top		Bottom	
	x, cm	x, in.	x, cm	x, in.
*-0.620	36.83	14.50	36.83	14.50
*-.684	35.31	13.90	35.31	13.90
-.716	34.54	13.60	34.54	13.60
-.684	35.31	13.90	35.31	13.90
-.593	37.46	14.75		
-.501	39.67	15.62	39.67	15.62
-.380	42.54	16.75	42.54	16.75
-.268	45.21	17.80	45.21	17.80
-.148	48.06	18.92	48.06	18.92
-.029	50.93	20.05		
.060	53.04	20.88		

*Inside of nacelle inlet.

TABLE 2

TEST CONDITIONS FOR THE VECTORED-THRUST CONFIGURATION (N_1 AND N_2)

α, deg	β, deg	$\left(\frac{V_\infty}{V_j}\right)_e$	q_∞		T	
			N/m^2	lb/ft^2	N	lb
0 to 20	0	∞	2870	60.0	0	0
		0.3	2050	42.9	623	140
		.2	910	19.0	623	140
		.1	227	4.76	623	140
0	-12 to +12	∞	2870	60.0	0	0
		.3	2050	42.9	623	140
		.2	910	19.0	623	140
		.1	227	4.76	623	140
0 or 10	0	0 to 0.5	0 to 2870	0 to 60.0	623	140

TABLE 3

EXHAUST NOZZLE POSITION AND THRUST DIRECTION

Thrust device	Nominal direction, deg	Actual direction, deg	Exit center position			
			x station		z station	
			cm	in.	cm	in.
Lift jet	90	90.1	50.93	20.05	-7.82	-3.08
Front vectored nozzle	0	-1.2	49.71	19.57	2.11	.83
	45	40.1	51.51	20.28	.30	.12
	90	85.2	55.93	22.02	-4.11	-1.62
Rear vectored nozzle	45	40.0	72.80	28.66	.30	.12
	90	85.0	77.22	30.40	-4.11	-1.62

INDEX TO TABULATED PRESSURE PROFILE DATA

Table 11.- Effect of Angle of Attack on Pressure Profile on Wing – Forward Nozzles Deflected 90° With Flaps
(a) Effective velocity ratio = ∞
(b) Effective velocity ratio ≈ 0.3
(c) Effective velocity ratio ≈ 0.2
(d) Effective velocity ratio ≈ 0.1

Table 12.- Effect of Angle of Attack on Pressure Profile on Wing – Rear Nozzles Deflected 90°
(a) Effective velocity ratio = ∞
(b) Effective velocity ratio ≈ 0.3
(c) Effective velocity ratio ≈ 0.2
(d) Effective velocity ratio ≈ 0.1

Table 13.- Effect of Angle of Attack on Pressure Profile on Wing – Rear Nozzles Deflected 90° With Flaps
(a) Effective velocity ratio = ∞
(b) Effective velocity ratio ≈ 0.3
(c) Effective velocity ratio ≈ 0.2
(d) Effective velocity ratio ≈ 0.1

Table 14.- Effect of Angle of Attack on Pressure Profile on Wing – Forward Nozzles Deflected 90° With Lift Jet
(a) Effective velocity ratio ≈ 0.3
(b) Effective velocity ratio ≈ 0.2
(c) Effective velocity ratio ≈ 0.1

Table 15.- Effect of Angle of Attack on Pressure Profile on Wing – Rear Nozzles Deflected 90° With Lift Jet
(a) Effective velocity ratio = ∞
(b) Effective velocity ratio ≈ 0.3
(c) Effective velocity ratio ≈ 0.2
(d) Effective velocity ratio ≈ 0.1

Table 16.- Effect of Angle of Attack on Pressure Profile on Wing – Forward Nozzles Deflected 90° With Inlets Closed

Table 17.- Effect of Angle of Attack on Pressure Profile on Fuselage – Lift Jet Alone
(a) Effective velocity ratio = ∞
(b) Effective velocity ratio ≈ 0.3
(c) Effective velocity ratio ≈ 0.2
(d) Effective velocity ratio ≈ 0.1

Table 18.- Effect of Angle of Attack on Pressure Profile on Fuselage – Forward Nozzles Deflected 45°

(a) Effective velocity ratio = ∞

(b) Effective velocity ratio ≈ 0.3

(c) Effective velocity ratio ≈ 0.2

(d) Effective velocity ratio ≈ 0.1

Table 19.- Effect of Angle of Attack on Pressure Profile on Fuselage – Rear Nozzles Deflected 45°

(a) Effective velocity ratio = ∞

(b) Effective velocity ratio ≈ 0.3

(c) Effective velocity ratio ≈ 0.2

(d) Effective velocity ratio ≈ 0.1

Table 20.- Effect of Angle of Attack on Pressure Profile on Fuselage – Forward Nozzles Deflected 90°

(a) Effective velocity ratio = ∞

(b) Effective velocity ratio ≈ 0.3

(c) Effective velocity ratio ≈ 0.2

(d) Effective velocity ratio ≈ 0.1

Table 21.- Effect of Angle of Attack on Pressure Profile on Fuselage – Rear Nozzles Deflected 90°

(a) Effective velocity ratio = ∞

(b) Effective velocity ratio ≈ 0.3

(c) Effective velocity ratio ≈ 0.2

(d) Effective velocity ratio ≈ 0.1

Table 22.- Effect of Angle of Attack on Pressure Profile on Fuselage – Forward Nozzles Deflected 90° With Lift Jet

(a) Effective velocity ratio ≈ 0.3

(b) Effective velocity ratio ≈ 0.2

(c) Effective velocity ratio ≈ 0.1

Table 23.- Effect of Angle of Attack on Pressure Profile on Fuselage – Rear Nozzles Deflected 90° With Lift Jet

(a) Effective velocity ratio = ∞

(b) Effective velocity ratio ≈ 0.3

(c) Effective velocity ratio ≈ 0.2

(d) Effective velocity ratio ≈ 0.1

Table 24.- Effect of Angle of Attack on Pressure Profile on Nacelle – Forward Nozzles Deflected 90^{o}

(a) Effective velocity ratio = ∞

(b) Effective velocity ratio ≈ 0.3

(c) Effective velocity ratio ≈ 0.2

(d) Effective velocity ratio ≈ 0.1

Table 25.- Effect of Angle of Attack on Pressure Profile on Nacelle – Rear Nozzles Deflected 90^{o}

(a) Effective velocity ratio = ∞

(b) Effective velocity ratio ≈ 0.3

(c) Effective velocity ratio ≈ 0.2

(d) Effective velocity ratio ≈ 0.1

Table 26.- Effect of Angle of Attack on Pressure Profile on Nacelle – Forward Nozzles Deflected 90^{o} With Inlets Closed

Table 27.- Effect of Angle of Sideslip on Pressure Profile on Wing – Lift Jet Alone

(a) Effective velocity ratio = ∞

(b) Effective velocity ratio ≈ 0.3

(c) Effective velocity ratio ≈ 0.2

(d) Effective velocity ratio ≈ 0.1

Table 28.- Effect of Angle of Sideslip on Pressure Profile on Wing – Forward Nozzles Deflected 90^{o}

(a) Effective velocity ratio = ∞

(b) Effective velocity ratio ≈ 0.3

(c) Effective velocity ratio ≈ 0.2

(d) Effective velocity ratio ≈ 0.1

Table 29.- Effect of Angle of Sideslip on Pressure Profile on Wing – Rear Nozzles Deflected 90^{o}

(a) Effective velocity ratio = ∞

(b) Effective velocity ratio ≈ 0.3

(c) Effective velocity ratio ≈ 0.2

(d) Effective velocity ratio ≈ 0.1

Table 30.- Effect of Effective Velocity Ratio on Pressure Profile on Wing – Lift Jet Alone

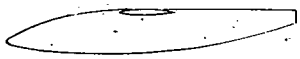
(a) $\alpha \approx 0^{o}$

(b) $\alpha \approx 10^{o}$

TABLE 4 (A)

PRESSURE PROFILE ON WING

LIFT JET ALONE



ALPHA = .00 DEG Q = 2843 N/SQ.M. (59.38 LBF/SQ.FT.) EFFECT. VELOC. RATIO = ∞

X/C	ETA = .250 CP		ETA = .387 CP		ETA = .524 CP		ETA = .800 CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000	.93		.56		.87		.98	
.010	-.02	.15	.49	-.11				
.015					-.18	-.25		
.025	-.404	-.280	-.205	-.291	-.191	-.266	-.190	-.316
.050	-.212	-.161	-.299	-.232	-.200	-.205	-.228	-.164
.100	-.231	-.254	-.194	-.248	-.221	-.264	-.181	-.263
.150	-.298	-.342	-.312	-.304	-.278	-.383	-.226	-.327
.250	-.286	-.343	-.328	-.360	-.315	-.352	-.292	-.316
.400	-.277	-.355	-.290	-.322	-.302	-.322	-.286	-.311
.550	-.194	-.255	-.189	-.232		-.237	-.197	-.235
.700	-.112	-.125	-.098	-.121	-.121	-.129	-.108	-.134
.850	-.024	-.040	-.018	-.037	-.005	-.026	.006	-.022
.950	.057	.053	.051	.063	.060	.061	.062	.070

ALPHA = 4.00 DEG Q = 2846 N/SQ.M. (59.44 LBF/SQ.FT.) EFFECT. VELOC. RATIO = ∞

X/C	ETA = .250 CP		ETA = .387 CP		ETA = .524 CP		ETA = .800 CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000	.67		.96		.79		.40	
.010	-1.39	.84	.54	.86				
.015					-1.17	.55		
.025	-1.271	.438	-1.260	.353	-1.110	.341	-.949	.414
.050	-.838	.227	-.756	.251	-.780	.259	-.747	.229
.100	-.575	.077	-.555	.116	-.619	.126	-.544	.082
.150	-.549	-.027	-.561	-.044	-.531	.002	-.525	-.022
.250	-.450	-.137	-.553	-.133	-.498	-.130	-.475	-.133
.400	-.399	-.212	-.422	-.191	-.439	-.159	-.385	-.180
.550	-.270	-.146	-.272	-.130		-.144	-.257	-.138
.700	-.154	-.087	-.155	-.069	-.153	-.081	-.125	-.096
.850	-.043	-.004	-.041	-.004	-.028	-.012	-.023	-.034
.950	.059	.061	.049	.066	.058	.066	.058	.067

ALPHA = 7.96 DEG Q = 2846 N/SQ.M. (59.44 LBF/SQ.FT.) EFFECT. VELOC. RATIO = ∞

X/C	ETA = .250 CP		ETA = .387 CP		ETA = .524 CP		ETA = .800 CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000	-.91		-.06		-.60		-1.68	
.010	-2.92	.98	.52	.98				
.015					-2.55	.96		
.025	-2.775	.834	-2.583	.745	-2.356	.802	-2.232	.841
.050	-1.322	.585	-1.470	.605	-1.359	.562	-1.250	.575
.100	-1.027	.396	-.998	.415	-1.052	.398	-.967	.356
.150	-.852	.220	-.923	.256	-.930	.272	-.796	.213
.250	-.641	.082	-.714	.101	-.703	.067	-.717	.053
.400	-.511	-.060	-.543	-.013	-.582	-.026	-.494	-.031
.550	-.330	-.047	-.364	-.042		-.027	-.335	-.055
.700	-.190	-.018	-.189	-.004	-.213	-.015	-.178	-.017
.850	-.059	.007	-.053	.013	-.065	.026	-.057	.017
.950	.044	.053	.046	.055	.033	.060	.029	.048

TABLE 4 (A) - CONTINUED

PRESSURE PROFILE ON WING

LIFT JET ALONE

ALPHA = 9.97 DEG Q = 2846 N/SQ.M. (59.44 LBF/SQ.FT.) EFFECT. VELOC. RATIO = ∞

	ETA = .250		ETA = .387		ETA = .524		ETA = .800	
X/C	CP		CP		CP		CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000	-2.21		-.53		-2.29		-3.01	
.010	-3.78	.92	.47	.91				
.015					-3.67	.98		
.025	-3.538	.944	-2.630	.933	-2.982	.914	-2.247	.901
.050	-1.635	.749	-1.775	.723	-1.678	.724	-1.673	.723
.100	-1.164	.540	-1.179	.509	-1.305	.498	-1.122	.475
.150	-.991	.332	-1.005	.327	-1.076	.366	-.963	.301
.250	-.735	.175	-.798	.174	-.819	.158	-.783	.141
.400	-.563	.023	-.599	.020	-.607	.068	-.563	.002
.550	-.361	-.010	-.379	.001		.020	-.364	-.025
.700	-.199	.018	-.206	.019	-.211	.025	-.233	-.008
.850	-.059	.026	-.063	.034	-.057	.034	-.067	.018
.950	.042	.058	-.033	.066	.038	.062	.018	.048

ALPHA = 12.01 DEG Q = 2842 N/SQ.M. (59.37 LBF/SQ.FT.) EFFECT. VELOC. RATIO = ∞

	ETA = .250		ETA = .387		ETA = .524		ETA = .800	
X/C	CP		CP		CP		CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000	-.85		.34		-.20		-.92	
.010	-1.37	.98	.44	.98				
.015					-1.11	.97		
.025	-1.519	.923	-1.231	.900	-1.140	.848	-1.170	.815
.050	-1.395	.739	-1.175	.717	-1.329	.685	-1.071	.655
.100	-1.465	.522	-1.192	.487	-1.149	.492	-1.097	.446
.150	-1.332	.356	-1.210	.335	-1.224	.339	-.997	.292
.250	-.983	.184	-1.034	.172	-.956	.166	-1.020	.115
.400	-.630	.025	-.721	.016	-.772	.018	-.788	-.008
.550	-.421	-.032	-.593	-.029		-.023	-.493	-.062
.700	-.277	-.024	-.399	-.035	-.431	-.050	-.337	-.060
.850	-.173	-.048	-.258	-.030	-.286	-.050	-.230	-.093
.950	-.119	-.070	-.199	-.036	-.212	-.109	-.140	-.042

ALPHA = 14.03 DEG Q = 2847 N/SQ.M. (59.46 LBF/SQ.FT.) EFFECT. VELOC. RATIO = ∞

	ETA = .250		ETA = .387		ETA = .524		ETA = .800	
X/C	CP		CP		CP		CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000	-.82		.42		.07		-.76	
.010	-1.44	.97	.44	.98				
.015					-.60	.94		
.025	-1.379	.938	-.759	.874	-.586	.843	-.793	.828
.050	-1.552	.770	-.642	.691	-.604	.676	-.818	.663
.100	-1.295	.526	-.711	.502	-.736	.489	-.799	.455
.150	-1.010	.378	-.722	.355	-.601	.352	-.844	.295
.250	-.688	.201	-.824	.191	-.722	.175	-.810	.131
.400	-.705	.007	-.794	.013	-.742	.009	-.846	-.014
.550	-.631	-.045	-.675	-.051		-.063	-.736	-.079
.700	-.483	-.061	-.617	-.082	-.617	-.087	-.681	-.106
.850	-.362	-.118	-.502	-.122	-.525	-.141	-.465	-.138
.950	-.322	-.170	-.405	-.225	-.451	-.233	-.469	-.183

TABLE 4 (A) - CONCLUDED

PRESSURE PROFILE ON WING

LIFT JET ALONE

ALPHA = 15.98 DEG Q = 2849 N/SQ.M. (59.50 LBF/SQ.FT.) EFFECT. VELOC. RATIO = ∞

	ETA = .250		ETA = .387		ETA = .524		ETA = .800	
X/C	CP		CP		CP		CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000	-.29		.50		.10		-.70	
.010	-.47	.96	.45	.98				
.015					-.49	.95		
.025	-.513	.906	-.466	.880	-.529	.847	-.794	.872
.050	-.672	.794	-.538	.728	-.550	.694	-.759	.688
.100	-1.262	.574	-.458	.524	-.511	.512	-.751	.500
.150	-.455	.413	-.617	.379	-.529	.351	-.808	.334
.250	-.511	.192	-.517	.172	-.522	.171	-.804	.157
.400	-.642	.007	-.586	.019	-.611	.006	-.838	-.008
.550	-.584	-.064	-.638	-.076		-.092	-.850	-.084
.700	-.594	-.131	-.554	-.107	-.633	-.145	-.788	-.129
.850	-.588	-.181	-.563	-.203	-.592	-.216	-.664	-.184
.950	-.493	-.300	-.483	-.280	-.565	-.372	-.520	-.272

ALPHA = 18.04 DEG Q = 2840 N/SQ.M. (59.31 LBF/SQ.FT.) EFFECT. VELOC. RATIO = ∞

	ETA = .250		ETA = .387		ETA = .524		ETA = .800	
X/C	CP		CP		CP		CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000	-.21		.42		-.06		-.76	
.010	-.46	.98	.46	.97				
.015					-.53	.97		
.025	-.485	.935	-.457	.899	-.523	.889	-.787	.887
.050	-.451	.782	-.450	.746	-.531	.759	-.745	.733
.100	-.444	.573	-.458	.556	-.513	.558	-.704	.536
.150	-.450	.436	-.454	.402	-.508	.405	-.699	.360
.250	-.454	.236	-.472	.219	-.522	.203	-.737	.174
.400	-.537	.022	-.539	.033	-.573	.029	-.754	.009
.550	-.565	-.074	-.562	-.076		-.075	-.795	-.090
.700	-.595	-.135	-.553	-.116	-.604	-.142	-.782	-.150
.850	-.554	-.237	-.552	-.248	-.612	-.256	-.746	-.249
.950	-.511	-.337	-.518	-.345	-.593	-.389	-.631	-.355

ALPHA = 20.05 DEG Q = 2849 N/SQ.M. (59.50 LBF/SQ.FT.) EFFECT. VELOC. RATIO = ∞

	ETA = .250		ETA = .387		ETA = .524		ETA = .800	
X/C	CP		CP		CP		CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000	-.32		.32		-.23		-.84	
.010	-1.35	.86	.46	.95				
.015					-.50	.98		
.025	-1.406	1.000	-.454	.937	-.506	.930	-.682	.919
.050	-1.517	.887	-.455	.804	-.495	.798	-.662	.784
.100	-1.223	.686	-.479	.604	-.499	.612	-.668	.572
.150	-.674	.524	-.441	.496	-.497	.464	-.664	.422
.250	-.499	.331	-.494	.297	-.507	.283	-.686	.227
.400	-.534	.096	-.535	.075	-.573	.084	-.735	.040
.550	-.561	-.021	-.547	-.041		-.032	-.760	-.073
.700	-.617	-.096	-.575	-.106	-.611	-.121	-.786	-.150
.850	-.631	-.213	-.550	-.244	-.609	-.243	-.733	-.269
.950	-.618	-.360	-.524	-.349	-.582	-.396	-.695	-.387

TABLE 4 (B)

PRESSURE PROFILE ON WING

LIFT JET ALONE

ALPHA = .02 DEG Q = 2844 N/SQ.M. (59.39 LBF/SQ.FT.) EFFECT. VELOC. RATIO = .318

X/C	ETA = .250 CP		ETA = .387 CP		ETA = .524 CP		ETA = .800 CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000	.92		.64		.92		.97	
.010	-.01	-.15	.23	-.11				
.015					-.14	-.29		
.025	-.389	-.397	-.224	-.219	-.228	-.311	-.223	-.303
.050	-.326	-.336	-.243	-.299	-.358	-.205	-.262	-.216
.100	-.239	-.353	-.244	-.304	-.253	-.282	-.211	-.218
.150	-.321	-.380	-.303	-.413	-.343	-.354	-.274	-.309
.250	-.316	-.435	-.323	-.405	-.326	-.344	-.314	-.328
.400	-.299	-.422	-.328	-.406	-.327	-.364	-.316	-.324
.550	-.206	-.333	-.220	-.292	-.419	-.265	-.219	-.238
.700	-.125	-.177	-.117	-.166	-.115	-.160	-.100	-.140
.850	-.046	-.074	-.043	-.061	-.024	-.063	-.015	-.043
.950	.041	.036	.032	.045	.037	.037	.043	.054

ALPHA = 4.04 DEG Q = 2847 N/SQ.M. (59.47 LBF/SQ.FT.) EFFECT. VELOC. RATIO = .319

X/C	ETA = .250 CP		ETA = .387 CP		ETA = .524 CP		ETA = .800 CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000	.76		.96		.65		.53	
.010	-1.18	.70	.25	.73				
.015					-1.36	.42		
.025	-1.233	.310	-1.274	.295	-1.182	.339	-1.085	.441
.050	-.825	.163	-.795	.213	-.807	.201	-.834	.279
.100	-.592	.013	-.606	.028	-.617	.072	-.603	.028
.150	-.577	-.128	-.597	-.055	-.558	-.065	-.495	-.042
.250	-.509	-.215	-.522	-.172	-.555	-.176	-.520	-.146
.400	-.412	-.271	-.438	-.244	-.439	-.222	-.406	-.196
.550	-.281	-.229	-.292	-.200	-.490	-.188	-.276	-.185
.700	-.169	-.125	-.166	-.118	-.174	-.101	-.147	-.112
.850	-.062	-.047	-.061	-.042	-.046	-.041	-.037	-.037
.950	.035	.033	.033	.044	.039	.031	.042	.040

ALPHA = 8.01 DEG Q = 2850 N/SQ.M. (59.52 LBF/SQ.FT.) EFFECT. VELOC. RATIO = .319

X/C	ETA = .250 CP		ETA = .387 CP		ETA = .524 CP		ETA = .800 CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000	-.59		.35		-.50		-1.57	
.010	-2.89	.98	.22	.97				
.015					-2.56	.94		
.025	-2.530	.800	-2.313	.737	-2.344	.744	-2.281	.750
.050	-1.244	.535	-1.413	.529	-1.349	.566	-1.399	.557
.100	-1.028	.313	-.995	.323	-1.022	.347	-1.012	.356
.150	-.884	.180	-.940	.154	-.881	.199	-.834	.203
.250	-.663	-.005	-.739	.006	-.708	.016	-.696	.045
.400	-.526	-.150	-.586	-.104	-.576	-.079	-.527	-.062
.550	-.351	-.141	-.370	-.113	-.538	-.087	-.351	-.082
.700	-.210	-.081	-.218	-.077	-.217	-.057	-.201	-.057
.850	-.074	-.023	-.080	-.015	-.069	-.010	-.074	-.019
.950	.030	.033	.018	.043	.030	.036	.011	.036

TABLE 4 (B) - CONTINUED

PRESSURE PROFILE ON WING

LIFT JET ALONE

ALPHA = 10.00 DEG Q = 2849 N/SQ.M. (59.50 LBF/SQ.FT.) EFFECT. VELOC. RATIO = .319

	ETA = .250		ETA = .387		ETA = .524		ETA = .800	
X/C	CP		CP		CP		CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000	-1.74		-.41		-2.03		-3.08	
.010	-3.86	.95	.15	.91				
.015					-3.60	.97		
.025	-3.440	.908	-2.764	.884	-2.893	.891	-2.420	.904
.050	-1.628	.654	-1.854	.653	-1.807	.731	-1.626	.728
.100	-1.240	.422	-1.249	.443	-1.236	.483	-1.231	.487
.150	-1.011	.275	-1.112	.279	-1.063	.315	-.972	.275
.250	-.783	.061	-.821	.113	-.847	.116	-.810	.097
.400	-.579	-.063	-.630	-.022	-.639	-.028	-.592	-.006
.550	-.381	-.096	-.406	-.072	-.579	-.042	-.380	-.051
.700	-.219	-.052	-.233	-.040	-.239	-.018	-.213	-.034
.850	-.081	-.025	-.085	-.018	-.082	.001	-.082	-.008
.950	.020	.032	.011	.038	.012	.030	.004	.024

ALPHA = 12.00 DEG Q = 2848 N/SQ.M. (59.48 LBF/SQ.FT.) EFFECT. VELOC. RATIO = .319

	ETA = .250		ETA = .387		ETA = .524		ETA = .800	
X/C	CP		CP		CP		CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000	-2.76		.07		-2.12		-1.47	
.010	-4.56	.81	.15	.90				
.015					-2.70	.97		
.025	-3.369	.960	-2.709	.924	-1.737	.933	-1.518	.866
.050	-1.833	.770	-2.025	.801	-2.086	.720	-1.283	.696
.100	-1.371	.510	-1.383	.484	-1.688	.526	-1.465	.470
.150	-1.128	.330	-1.158	.338	-1.359	.359	-1.325	.312
.250	-.861	.131	-.876	.161	-.901	.165	-.937	.123
.400	-.617	-.037	-.639	-.009	-.641	.007	-.736	-.004
.550	-.401	-.059	-.517	-.051	-.609	-.056	-.441	-.058
.700	-.247	-.054	-.234	-.054	-.238	-.034	-.260	-.044
.850	-.127	-.057	-.130	-.047	-.169	-.064	-.173	-.062
.950	-.045	-.015	-.080	.013	-.153	-.024	-.125	-.043

ALPHA = 13.96 DEG Q = 2850 N/SQ.M. (59.52 LBF/SQ.FT.) EFFECT. VELOC. RATIO = .319

	ETA = .250		ETA = .387		ETA = .524		ETA = .800	
X/C	CP		CP		CP		CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000	-.71		.33		-.25		-.96	
.010	-1.60	.94	.15	.97				
.015					-1.02	.96		
.025	-1.936	.951	-.793	.857	-.956	.853	-1.075	.846
.050	-.979	.708	-1.078	.679	-.939	.681	-1.027	.688
.100	-1.334	.478	-1.018	.458	-.982	.482	-1.010	.459
.150	-1.138	.303	-1.014	.303	-.908	.321	-1.093	.305
.250	-1.013	.109	-.937	.138	-.972	.134	-.974	.130
.400	-.805	-.069	-.787	-.045	-.788	-.025	-.854	-.026
.550	-.649	-.125	-.670	-.106	-.663	-.085	-.730	-.082
.700	-.483	-.107	-.522	-.105	-.554	-.094	-.632	-.102
.850	-.474	-.148	-.475	-.167	-.438	-.138	-.389	-.135
.950	-.277	-.150	-.311	-.224	-.362	-.225	-.304	-.213

TABLE 4 (B) - CONCLUDED

PRESSURE PROFILE ON WING

LIFT JET ALONE

ALPHA = 16.06 DEG Q = 2842 N/SQ.M. (59.36 LBF/SQ.FT.) EFFECT. VELOC. RATIO = .319

	ETA = .250		ETA = .387		ETA = .524		ETA = .800	
X/C	CP		CP		CP		CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000	-.14		.49		-.04		-.92	
.010	-.72	.97	.16	.98				
.015					-.57	.94		
.025	-.610	.884	-.667	.852	-.620	.832	-.893	.864
.050	-.512	.691	-.540	.689	-.639	.689	-.834	.706
.100	-1.042	.458	-.696	.455	-.647	.492	-.839	.505
.150	-.536	.303	-.734	.311	-.657	.334	-.838	.331
.250	-.805	.084	-.626	.131	-.659	.149	-.894	.140
.400	-.643	-.079	-.667	-.065	-.649	-.053	-.929	-.022
.550	-.703	-.145	-.663	-.158	-.681	-.135	-.852	-.100
.700	-.674	-.192	-.658	-.166	-.671	-.175	-.770	-.135
.850	-.591	-.246	-.553	-.236	-.586	-.228	-.612	-.204
.950	-.461	-.307	-.519	-.283	-.543	-.377	-.483	-.255

ALPHA = 18.03 DEG Q = 2844 N/SQ.M. (59.39 LBF/SQ.FT.) EFFECT. VELOC. RATIO = .319

	ETA = .250		ETA = .387		ETA = .524		ETA = .800	
X/C	CP		CP		CP		CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000	-.27		.47		-.09		-.93	
.010	-.53	.99	.19	.97				
.015					-.60	.96		
.025	-.564	.909	-.513	.868	-.579	.875	-.832	.885
.050	-.482	.731	-.513	.708	-.564	.728	-.799	.747
.100	-.559	.503	-.512	.493	-.549	.512	-.773	.520
.150	-.489	.350	-.516	.341	-.584	.359	-.772	.348
.250	-.490	.136	-.523	.153	-.580	.179	-.791	.167
.400	-.674	-.088	-.582	-.052	-.620	-.027	-.871	-.014
.550	-.618	-.178	-.586	-.154	-.682	-.124	-.865	-.111
.700	-.671	-.208	-.619	-.166	-.640	-.190	-.845	-.173
.850	-.586	-.288	-.584	-.298	-.643	-.293	-.778	-.246
.950	-.566	-.368	-.544	-.379	-.598	-.390	-.631	-.351

ALPHA = 20.00 DEG Q = 2847 N/SQ.M. (59.46 LBF/SQ.FT.) EFFECT. VELOC. RATIO = .318

	ETA = .250		ETA = .387		ETA = .524		ETA = .800	
X/C	CP		CP		CP		CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000	-.42		.25		-.21		-1.03	
.010	-.56	.96	.19	.95				
.015					-.56	.97		
.025	-.561	.944	-.525	.921	-.524	.909	-.769	.909
.050	-.534	.803	-.536	.757	-.529	.778	-.736	.775
.100	-.575	-.564	-.497	.571	-.545	.595	-.795	.563
.150	-.492	.406	-.535	.391	-.549	.419	-.774	.393
.250	-.680	.179	-.527	.193	-.557	.213	-.765	.197
.400	-.646	-.024	-.655	-.004	-.622	-.003	-.827	.011
.550	-.661	-.147	-.685	-.120	-.668	-.106	-.859	-.089
.700	-.671	-.196	-.627	-.170	-.639	-.180	-.844	-.177
.850	-.621	-.263	-.594	-.285	-.559	-.279	-.803	-.276
.950	-.602	-.389	-.586	-.386	-.618	-.423	-.734	-.374

TABLE 4 (c)

PRESSURE PROFILE ON WING

LIFT JET ALONE

ALPHA = -.02 DEG Q = 2746 N/SQ.M. (57.36 LBF/SQ.FT.) EFFECT. VELOC. RATIO = .200

	ETA = .250		ETA = .387		ETA = .524		ETA = .800	
X/C	CP		CP		CP		CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000	.78		.56		.86		.97	
.010	.01	-.16	.23	-.29				
.015					-.04	-.34		
.025	-.215	-.500	-.280	-.510	-.228	-.406	-.085	-.410
.050	-.203	-.355	-.270	-.412	-.233	-.341	-.276	-.190
.100	-.200	-.402	-.182	-.359	-.208	-.320	-.188	-.280
.150	-.286	-.438	-.260	-.422	-.277	-.387	-.252	-.334
.250	-.258	-.478	-.288	-.457	-.370	-.407	-.332	-.332
.400	-.280	-.450	-.310	-.432	-.318	-.398	-.295	-.345
.550	-.206	-.312	-.219	-.312	-.598	-.282	-.229	-.258
.700	-.117	-.174	-.113	-.171	-.120	-.149	-.098	-.145
.850	-.036	-.065	-.039	-.065	-.025	-.062	-.020	-.055
.950	.039	.036	.031	.041	.037	.039	.037	.048

ALPHA = 4.00 DEG Q = 2752 N/SQ.M. (57.47 LBF/SQ.FT.) EFFECT. VELOC. RATIO = .200

	ETA = .250		ETA = .387		ETA = .524		ETA = .800	
X/C	CP		CP		CP		CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000	.82		.98		.82		.54	
.010	-1.00	.75	.33	.62				
.015					-1.00	.45		
.025	-1.160	.241	-1.136	.317	-1.085	.347	-1.011	.326
.050	-.766	.182	-.732	.190	-.805	.138	-.782	.231
.100	-.607	-.106	-.588	-.011	-.611	.075	-.568	.065
.150	-.549	-.208	-.580	-.143	-.562	-.085	-.513	-.060
.250	-.494	-.246	-.527	-.207	-.499	-.204	-.510	-.165
.400	-.416	-.277	-.435	-.264	-.440	-.253	-.397	-.214
.550	-.279	-.206	-.285	-.203	-.473	-.199	-.282	-.178
.700	-.167	-.130	-.171	-.121	-.169	-.126	-.147	-.105
.850	-.062	-.043	-.067	-.036	-.050	-.046	-.042	-.039
.950	.031	.036	.025	.038	.029	.030	.039	.040

ALPHA = 8.04 DEG Q = 2749 N/SQ.M. (57.41 LBF/SQ.FT.) EFFECT. VELOC. RATIO = .200

	ETA = .250		ETA = .387		ETA = .524		ETA = .800	
X/C	CP		CP		CP		CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000	-.34		.32		-.46		-1.34	
.010	-2.72	.98	.37	.97				
.015					-2.42	.88		
.025	-2.276	.758	-2.462	.714	-2.505	.807	-2.213	.744
.050	-1.309	.526	-1.297	.595	-1.485	.485	-1.300	.542
.100	-1.042	.267	-.974	.266	-.998	.312	-.969	.328
.150	-.837	.079	-.912	.126	-.868	.140	-.854	.177
.250	-.666	-.048	-.711	-.026	-.704	.033	-.688	.006
.400	-.528	-.159	-.547	-.135	-.564	-.102	-.506	-.087
.550	-.355	-.131	-.373	-.131	-.423	-.101	-.348	-.095
.700	-.212	-.063	-.219	-.072	-.214	-.061	-.197	-.064
.850	-.079	-.019	-.083	-.017	-.065	-.019	-.074	-.018
.950	.028	.039	.016	.042	.026	.038	.013	.031

TABLE 4 (C) - CONTINUED

PRESSURE PROFILE ON WING

LIFT JET ALONE

ALPHA = 9.96 DEG Q = 2748 N/SQ.M. (57.39 LBF/SQ.FT.) EFFECT. VELOC. RATIO = .200

	ETA = .250		ETA = .387		ETA = .524		ETA = .800	
X/C	CP		CP		CP		CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000	-1.42		-.12		-1.23		-3.03	
.010	-3.93	.97	.38	.95				
.015					-3.59	.97		
.025	-3.070	.897	-2.838	.861	-2.990	.863	-2.346	.887
.050	-1.584	.642	-1.654	.698	-1.826	.701	-1.573	.678
.100	-1.127	.373	-1.189	.401	-1.284	.436	-1.178	.491
.150	-.956	.255	-1.040	.249	-1.025	.272	-1.032	.277
.250	-.761	.027	-.827	.086	-.810	.116	-.820	.075
.400	-.574	-.076	-.631	-.076	-.628	-.014	-.589	-.028
.550	-.388	-.098	-.399	-.078	-.398	-.056	-.384	-.038
.700	-.223	-.042	-.228	-.042	-.239	-.029	-.218	-.032
.850	-.084	-.009	-.085	-.009	-.081	-.000	-.089	-.006
.950	.022	.029	.007	.036	.016	.027	.001	.036

ALPHA = 11.98 DEG Q = 2749 N/SQ.M. (57.42 LBF/SQ.FT.) EFFECT. VELOC. RATIO = .199

	ETA = .250		ETA = .387		ETA = .524		ETA = .800	
X/C	CP		CP		CP		CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000	-3.00		-1.06		-3.05		-4.76	
.010	-4.69	.85	.28	.81				
.015					-4.54	.97		
.025	-3.385	.964	-3.118	.942	-3.256	.948	-2.867	.952
.050	-1.885	.767	-2.133	.744	-2.063	.767	-2.088	.776
.100	-1.377	.520	-1.380	.531	-1.455	.555	-1.402	.524
.150	-1.133	.377	-1.202	.365	-1.204	.385	-1.117	.366
.250	-.838	.143	-.901	.156	-.903	.167	-.885	.201
.400	-.627	-.045	-.676	.009	-.684	.028	-.628	.035
.550	-.405	-.048	-.433	-.031	-.399	-.011	-.410	-.024
.700	-.235	-.013	-.241	-.011	-.238	-.004	-.235	-.024
.850	-.090	-.005	-.088	-.001	-.087	.007	-.100	-.010
.950	.006	.025	-.003	.030	-.001	.021	-.011	.016

ALPHA = 14.02 DEG Q = 2748 N/SQ.M. (57.38 LBF/SQ.FT.) EFFECT. VELOC. RATIO = .199

	ETA = .250		ETA = .387		ETA = .524		ETA = .800	
X/C	CP		CP		CP		CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000	-.71		.18		-.44		-.91	
.010	-1.44	.97	.31	.96				
.015					-1.20	.96		
.025	-1.320	.950	-1.458	.893	-1.316	.855	-1.067	.859
.050	-1.176	.745	-1.291	.689	-1.334	.704	-1.028	.679
.100	-1.299	.480	-1.215	.474	-1.021	.491	-1.108	.470
.150	-1.175	.312	-1.140	.317	-1.054	.331	-1.023	.302
.250	-1.004	.117	-1.021	.129	-.942	.149	-1.024	.130
.400	-.793	-.042	-.773	-.039	-.778	-.021	-.848	-.023
.550	-.561	-.090	-.593	-.088	-.450	-.068	-.765	-.089
.700	-.453	-.087	-.395	-.084	-.501	-.083	-.517	-.099
.850	-.366	-.114	-.327	-.098	-.379	-.139	-.443	-.152
.950	-.262	-.134	-.258	-.199	-.286	-.226	-.325	-.143

TABLE 4 (C) - CONCLUDED
PRESSURE PROFILE ON WING

LIFT JET ALONE

ALPHA = 16.00 DEG Q = 2748 N/SQ.M. (57.39 LBF/SQ.FT.) EFFECT. VELOC. RATIO = .199

X/C	ETA = .250 CP		ETA = .387 CP		ETA = .524 CP		ETA = .800 CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000	-.46		.43		-.09		-.94	
.010	-.98	.97	.35	.97				
.015					-.74	.95		
.025	-.661	.919	-.823	.855	-.749	.854	-.875	.860
.050	-1.191	.720	-.725	.683	-.694	.698	-.883	.699
.100	-1.135	.486	-.666	.465	-.775	.492	-.871	.500
.150	-.944	.326	-.704	.351	-.802	.332	-.911	.322
.250	-.964	.128	-.775	.128	-.753	.136	-.968	.133
.400	-.791	-.048	-.817	-.042	-.733	-.028	-.946	-.030
.550	-.798	-.115	-.713	-.110	-.475	-.096	-.865	-.100
.700	-.710	-.117	-.654	-.134	-.702	-.135	-.769	-.124
.850	-.468	-.200	-.529	-.183	-.591	-.212	-.650	-.181
.950	-.431	-.253	-.464	-.242	-.479	-.291	-.517	-.246

ALPHA = 17.98 DEG Q = 2746 N/SQ.M. (57.35 LBF/SQ.FT.) EFFECT. VELOC. RATIO = .199

X/C	ETA = .250 CP		ETA = .387 CP		ETA = .524 CP		ETA = .800 CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000	-.36		.37		-.11		-1.12	
.010	-.70	.97	.36	.97				
.015					-.64	.96		
.025	-.631	.923	-.647	.881	-.652	.871	-.947	.886
.050	-.734	.734	-.576	.722	-.585	.719	-.945	.754
.100	-.588	.541	-.565	.517	-.601	.531	-.864	.544
.150	-.611	.337	-.587	.349	-.614	.373	-.884	.358
.250	-.636	.154	-.670	.160	-.614	.181	-.880	.186
.400	-.739	-.038	-.643	-.023	-.658	-.031	-.956	.007
.550	-.790	-.131	-.724	-.108	-.483	-.101	-.939	-.093
.700	-.726	-.152	-.739	-.146	-.719	-.165	-.900	-.144
.850	-.656	-.226	-.627	-.252	-.677	-.251	-.741	-.223
.950	-.553	-.317	-.553	-.326	-.572	-.396	-.592	-.297

ALPHA = 19.98 DEG Q = 2744 N/SQ.M. (57.32 LBF/SQ.FT.) EFFECT. VELOC. RATIO = .198

X/C	ETA = .250 CP		ETA = .387 CP		ETA = .524 CP		ETA = .800 CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000	-.52		.29		-.22		-1.18	
.010	-.62	.96	.37	.94				
.015					-.62	.97		
.025	-.660	.953	-.594	.914	-.594	.903	-.879	.918
.050	-.638	.821	-.559	.774	-.612	.769	-.876	.791
.100	-.611	.585	-.538	.558	-.598	.599	-.863	.589
.150	-.622	.425	-.594	.406	-.605	.428	-.916	.427
.250	-.665	.199	-.621	.191	-.652	.227	-.840	.221
.400	-.768	.002	-.700	.009	-.704	.006	-.952	.033
.550	-.746	-.110	-.682	-.101	-.496	-.090	-.957	-.077
.700	-.810	-.150	-.716	-.135	-.737	-.148	-.952	-.164
.850	-.711	-.229	-.667	-.265	-.706	-.263	-.893	-.248
.950	-.624	-.344	-.568	-.359	-.624	-.403	-.759	-.365

TABLE 4 (D)

PRESSURE PROFILE ON WING

LIFT JET ALONE

ALPHA = .02 DEG Q = 692 N/SQ.M. (14.46 LBF/SQ.FT.) EFFECT. VELOC. RATIO = .100

X/C	ETA = .250 CP		ETA = .387 CP		ETA = .524 CP		ETA = .800 CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000	.69		.09		.57		.92	
.010	.28	-.67	-1.31	-.74				
.015					.18	-.82		
.025	-.070	-.688	-.070	-.609	.021	-.549	-.022	-.502
.050	-.115	-.652	-.070	-.559	-.183	-.509	-.167	-.349
.100	-.106	-.479	-.170	-.413	-.188	-.420	-.147	-.356
.150	-.206	-.597	-.254	-.534	-.183	-.472	-.201	-.356
.250	-.247	-.505	-.227	-.434	-.238	-.408	-.247	-.372
.400	-.252	-.448	-.243	-.412	-.277	-.387	-.255	-.338
.550	-.173	-.316	-.175	-.329		-.299	-.186	-.268
.700	-.095	-.166	-.122	-.175	-.128	-.161	-.111	-.165
.850	-.016	-.067	-.028	-.071	-.014	-.059	-.005	-.048
.950	.042	.042	.036	.043	.047	.051	.050	.051

ALPHA = 3.96 DEG Q = 690 N/SQ.M. (14.41 LBF/SQ.FT.) EFFECT. VELOC. RATIO = .100

X/C	ETA = .250 CP		ETA = .387 CP		ETA = .524 CP		ETA = .800 CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000	.86		.85		.90		.70	
.010	-.60	.18	-1.24	.49				
.015					-.64	.17		
.025	-.705	-.091	-.749	.023	-.769	.018	-.867	.153
.050	-.518	-.194	-.546	.007	-.557	-.062	-.600	-.039
.100	-.427	-.183	-.400	-.190	-.416	-.139	-.427	-.053
.150	-.461	-.263	-.486	-.231	-.479	-.199	-.454	-.203
.250	-.416	-.331	-.418	-.322	-.409	-.276	-.447	-.226
.400	-.356	-.330	-.374	-.287	-.363	-.318	-.356	-.262
.550	-.248	-.285	-.252	-.253		-.222	-.254	-.216
.700	-.137	-.147	-.135	-.133	-.145	-.150	-.123	-.137
.850	-.051	-.047	-.040	-.037	-.034	-.050	-.040	-.055
.950	.032	.045	.049	.050	.038	.042	.025	.037

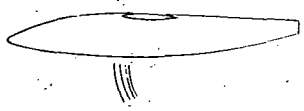
ALPHA = 7.99 DEG Q = 687 N/SQ.M. (14.35 LBF/SQ.FT.) EFFECT. VELOC. RATIO = .099

X/C	ETA = .250 CP		ETA = .387 CP		ETA = .524 CP		ETA = .800 CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000	.13		.64		.01		-.70	
.010	-1.80	.82	-1.23	.89				
.015					-1.94	.72		
.025	-1.662	.546	-1.632	.496	-1.989	.613	-1.719	.601
.050	-1.061	.360	-1.214	.367	-1.159	.326	-1.205	.413
.100	-.779	.186	-.788	.150	-.864	.260	-.845	.218
.150	-.708	-.018	-.738	.046	-.767	.017	-.699	.095
.250	-.614	-.136	-.628	-.105	-.651	-.081	-.625	-.049
.400	-.466	-.157	-.486	-.171	-.501	-.131	-.462	-.122
.550	-.304	-.171	-.324	-.139		-.127	-.333	-.136
.700	-.181	-.098	-.178	-.079	-.201	-.080	-.183	-.085
.850	-.063	-.023	-.064	-.019	-.063	-.028	-.077	-.027
.950	.035	.037	.021	.040	.021	.041	.006	.027

TABLE 4 (D) - CONTINUED

PRESSURE PROFILE ON WING

LIFT JET ALONE



ALPHA = 9.95 DEG. Q = 686 N/SQ.M. (14.33 LBF/SQ.FT.) EFFECT. VELOC. RATIO = .099

	ETA = .250		ETA = .387		ETA = .524		ETA = .800	
X/C	CP		CP		CP		CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000	-.53		.05		-.48		-2.21	
.010	-2.80	.93	-1.18	.93				
.015					-2.67	.90		
.025	-2.238	.762	-2.077	.746	-2.405	.739	-2.339	.771
.050	-1.361	.501	-1.409	.523	-1.388	.551	-1.281	.592
.100	-1.030	.312	-1.056	.317	-1.076	.319	-.994	.317
.150	-.842	.113	-.900	.145	-.861	.198	-.847	.179
.250	-.670	-.017	-.679	.057	-.734	.010	-.716	.022
.400	-.497	-.134	-.538	-.103	-.548	-.047	-.521	-.056
.550	-.335	-.079	-.355	-.103		-.085	-.341	-.100
.700	-.199	-.045	-.201	-.044	-.209	-.036	-.196	-.067
.850	-.065	-.001	-.061	.006	-.073	-.004	-.079	-.027
.950	.029	.038	.024	.045	.006	.035	.002	.015

ALPHA = 12.06 DEG Q = 688 N/SQ.M. (14.37 LBF/SQ.FT.) EFFECT. VELOC. RATIO = .099

	ETA = .250		ETA = .387		ETA = .524		ETA = .800	
X/C	CP		CP		CP		CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000	-1.89		-.33		-2.01		-3.53	
.010	-3.43	.91	-1.16	.84				
.015					-3.81	.93		
.025	-3.129	.869	-2.795	.825	-2.987	.862	-2.713	.887
.050	-1.603	.635	-1.774	.679	-1.747	.661	-1.637	.718
.100	-1.278	.377	-1.216	.480	-1.237	.491	-1.152	.473
.150	-.992	.269	-1.035	.306	-1.067	.281	-.969	.306
.250	-.802	.094	-.774	.109	-.809	.120	-.786	.104
.400	-.541	-.005	-.579	-.035	-.597	-.008	-.571	-.015
.550	-.354	-.079	-.364	-.052		-.021	-.384	-.050
.700	-.206	-.031	-.199	-.014	-.220	-.011	-.212	-.030
.850	-.072	-.001	-.073	-.012	-.073	-.018	-.086	-.010
.950	.017	.032	-.004	.031	.009	.011	-.007	.021

ALPHA = 14.00 DEG. Q = 689 N/SQ.M. (14.40 LBF/SQ.FT.) EFFECT. VELOC. RATIO = .099

	ETA = .250		ETA = .387		ETA = .524		ETA = .800	
X/C	CP		CP		CP		CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000	-1.91		-.73		-2.12		-1.98	
.010	-3.03	.89	-1.25	.82				
.015					-3.95	.94		
.025	-2.540	.894	-2.586	.896	-2.862	.903	-2.289	.889
.050	-1.777	.702	-1.770	.700	-1.909	.732	-1.628	.750
.100	-1.494	.490	-1.418	.508	-1.434	.522	-1.544	.508
.150	-1.256	.314	-1.233	.357	-1.194	.343	-1.297	.337
.250	-.840	.150	-.936	.157	-.874	.179	-.868	.171
.400	-.644	-.006	-.593	-.017	-.618	.023	-.586	.011
.550	-.357	-.067	-.389	-.043		-.023	-.385	-.038
.700	-.234	-.025	-.270	-.008	-.264	-.033	-.391	-.037
.850	-.102	-.017	-.129	-.025	-.103	-.032	-.188	-.061
.950	-.022	-.011	-.109	-.028	-.043	-.045	-.076	-.036

TABLE 4 (D) - CONCLUDED
PRESSURE PROFILE ON WING

LIFT JET ALONE

ALPHA = 15.98 DEG Q = 691 N/SQ.M. (14.42 LBF/SQ.FT.) EFFECT. VELOC. RATIO = .099

X/C	ETA = .250 CP		ETA = .387 CP		ETA = .524 CP		ETA = .800 CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000	-1.33		.04		-1.19		-1.15	
.010	-1.52	.89	-1.17	.89				
.015					-1.43	.95		
.025	-2.327	.896	-1.978	.880	-1.492	.862	-1.109	.857
.050	-1.232	.684	-1.376	.707	-1.289	.693	-1.282	.691
.100	-1.396	.488	-1.371	.472	-.988	.499	-1.036	.485
.150	-1.248	.312	-1.193	.337	-1.102	.337	-1.077	.335
.250	-1.050	.128	-.958	.153	-.913	.144	-1.086	.139
.400	-.687	-.023	-.748	.005	-.717	-.023	-.846	-.001
.550	-.551	-.073	-.613	-.057		-.064	-.684	-.066
.700	-.293	-.068	-.496	-.073	-.372	-.068	-.566	-.074
.850	-.311	-.098	-.286	-.109	-.374	-.131	-.352	-.106
.950	-.162	-.097	-.277	-.151	-.289	-.163	-.298	-.182

ALPHA = 17.98 DEG Q = 694 N/SQ.M. (14.49 LBF/SQ.FT.) EFFECT. VELOC. RATIO = .099

X/C	ETA = .250 CP		ETA = .387 CP		ETA = .524 CP		ETA = .800 CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000	-.45		.29		-.24		-.89	
.010	-1.10	.93	-1.10	.93				
.015					-.99	.95		
.025	-1.079	.881	-.934	.867	-.772	.851	-.811	.867
.050	-1.027	.722	-.847	.706	-.766	.715	-.875	.710
.100	-.904	.483	-1.063	.483	-.800	.501	-.888	.492
.150	-.999	.320	-.884	.324	-.718	.370	-.843	.327
.250	-.979	.146	-.954	.125	-.793	.154	-.943	.144
.400	-.756	-.038	-.705	-.048	-.757	-.016	-.940	-.012
.550	-.713	-.088	-.664	-.101		-.083	-.860	-.077
.700	-.597	-.110	-.640	-.125	-.650	-.118	-.822	-.111
.850	-.468	-.134	-.510	-.167	-.495	-.173	-.658	-.181
.950	-.398	-.190	-.468	-.239	-.425	-.258	-.508	-.301

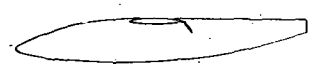
ALPHA = 20.05 DEG Q = 696 N/SQ.M. (14.54 LBF/SQ.FT.) EFFECT. VELOC. RATIO = .099

X/C	ETA = .250 CP		ETA = .387 CP		ETA = .524 CP		ETA = .800 CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000	-.48		.28		-.16		-.94	
.010	-.75	.94	-1.03	.94				
.015					-.68	.96		
.025	-.723	.924	-.692	.881	-.622	.874	-.807	.890
.050	-.769	.743	-.662	.741	-.585	.734	-.789	.743
.100	-.726	.510	-.637	.508	-.692	.517	-.766	.530
.150	-.825	.370	-.674	.372	-.615	.381	-.816	.374
.250	-.839	.144	-.597	.154	-.615	.172	-.818	.171
.400	-.794	-.028	-.653	-.010	-.671	-.000	-.855	.002
.550	-.779	-.075	-.731	-.093		-.099	-.856	-.077
.700	-.711	-.105	-.703	-.128	-.676	-.146	-.861	-.156
.850	-.662	-.181	-.595	-.222	-.635	-.231	-.757	-.230
.950	-.519	-.310	-.531	-.316	-.543	-.316	-.627	-.378

TABLE 5

PRESSURE PROFILE ON WING

LIFT JET ALONE WITH FLAPS



ALPHA = .00 DEG Q = 2762 N/SQ.M. (57.69 LBF/SQ.FT.) EFFECT. VELOC. RATIO = ∞

X/C	ETA = .250 CP		ETA = .387 CP		ETA = .524 CP		ETA = .800 CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000	.81		.98		.84		.77	
.010	-1.10	.70	.70	.67				
.015					-.84	.40		
.025	-1.054	.245	-.884	.297	-1.018	.228	-.646	.203
.050	-.704		-.641	.202	-.714	.250	-.498	.128
.100	-.534	.140	-.367	.143	-.506	.082	-.401	-.026
.150	-.555	-.006	-.567	.031	-.494	-.009	-.464	-.098
.250	-.533	.007	-.523	.006	-.531	-.013	-.465	-.171
.400	-.498	.048	-.501	.036	-.521	.002	-.433	-.182
.550	-.410	.170	-.420	.170		.115	-.312	-.113
.700	-.342	.354	-.346	.360	-.341	.322	-.182	-.065
.850	-.291	.559	-.295	.570	-.282		-.035	-.007
.950	-.234	.655	-.257	.643	-.266	.622	-.006	.058

ALPHA = 4.02 DEG Q = 2772 N/SQ.M. (57.90 LBF/SQ.FT.) EFFECT. VELOC. RATIO = ∞

X/C	ETA = .250 CP		ETA = .387 CP		ETA = .524 CP		ETA = .800 CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000	-.31		.32		-.25		-.57	
.010	-2.53	.98	.69	.97				
.015					-2.12	.88		
.025	-2.134	.759		.714	-2.113	.745	-1.601	.635
.050	-1.200	.611	-1.318	.615	-1.192	.539	-1.158	.469
.100	-.991	.377	-.965	.402	-.942	.374	-.808	.271
.150	-.858	.254	-.885	.284	-.892	.271	-.757	
.250	-.708	.191	-.728	.197	-.748	.167	-.686	-.010
.400	-.609	.158	-.653	.163	-.663	.131	-.528	-.051
.550	-.489	.267	-.525	.228		.219	-.390	-.038
.700	-.398	.411	-.409	.411	-.401	.370	-.253	.009
.850	-.327	.597	-.335	.601	-.316	.557	-.131	.005
.950	-.250	.678	-.264	.677	-.277	.670	-.039	.032

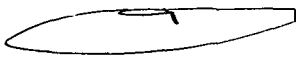
ALPHA = 8.00 DEG Q = 2847 N/SQ.M. (59.46 LBF/SQ.FT.) EFFECT. VELOC. RATIO = ∞

X/C	ETA = .250 CP		ETA = .387 CP		ETA = .524 CP		ETA = .800 CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000	-3.09		-1.19		-2.72		-3.76	
.010	-3.08	.83	.61	.80				
.015					-4.56	.96		
.025	-3.418	.976	-3.329	.964	-2.926	.950	-2.438	.930
.050	-1.907	.843	-2.052	.831	-2.127	.802	-1.820	.747
.100	-1.362	.613	-1.449	.629	-1.419	.636	-1.239	.516
.150	-1.167	.495	-1.224	.480	-1.233	.464	-1.075	.331
.250	-.921	.360	-.969	.373	-.965	.354	-.893	.189
.400	-.754	.289	-.807	.281	-.805	.276	-.659	.059
.550	-.569	.339	-.608	.328		.299	-.476	.031
.700	-.457	.474	-.464	.466	-.463	.406	-.299	.029
.850	-.359	.629	-.374	.653	-.352	.587	-.165	.018
.950	-.265	.699	-.295	.695	-.296	.663	-.057	.008

TABLE 5 - CONTINUED

PRESSURE PROFILE ON WING

LIFT JET ALONE WITH FLAPS



ALPHA = 10.04 DEG Q = 2847 N/SQ.M. (59.46 LBF/SQ.FT.) EFFECT. VELOC. RATIO = ∞

	ETA = .250		ETA = .387		ETA = .524		ETA = .800	
X/C	CP		CP		CP		CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000	-1.20		.08		-.49		-1.37	
.010	-1.54	.93	.58	.93				
.015					-1.22	.97		
.025	-1.477	.978	-1.278	.932	-1.100	.904	-1.401	.875
.050	-1.536	.820	-1.287	.791	-1.158	.773	-1.515	.723
.100	-1.447	.645	-1.246	.624	-1.187	.610	-1.394	.499
.150	-1.515	.526	-1.274	.502	-1.192	.492	-1.410	.363
.250	-1.312	.403	-1.275	.390	-1.237	.354	-1.388	.199
.400	-1.084	.330	-1.176	.319	-1.161	.279	-1.102	.082
.550	-.827	.370	-.928	.364		.321	-.802	.032
.700	-.634	.499	-.642	.494	-.809	.419	-.556	.013
.850	-.459	.644	-.622	.645	-.596	.600	-.328	-.016
.950	-.391	.716	-.467	.702	-.572	.662	-.199	-.055

ALPHA = 12.03 DEG Q = 2848 N/SQ.M. (59.49 LBF/SQ.FT.) EFFECT. VELOC. RATIO = ∞

	ETA = .250		ETA = .387		ETA = .524		ETA = .800	
X/C	CP		CP		CP		CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000	-1.63		.15		-.29		-1.18	
.010	-2.13	.86	.56	.94				
.015					-.81	.97		
.025	-2.157	.994	-.862	.941	-.806	.914	-1.112	.882
.050	-2.205	.881	-.751	.821	-.881	.777	-1.136	.738
.100	-1.538	.697	-.770	.645	-.772	.642	-1.110	.532
.150		.570	-.803	.538	-.795	.521	-1.139	.396
.250	-.817	.437	-.896	.419	-.882	.380	-1.139	.225
.400	-1.010	.357	-.922	.340	-.955	.304	-1.151	.118
.550	-.955	.387	-.954	.378		.335	-1.081	.035
.700	-.936	.502	-.882	.495	-.843	.408	-.877	.001
.850	-.772	.654	-.770	.650	-.778	.600	-.590	-.038
.950	-.678	.722	-.721	.709	-.714	.666	-.418	-.131

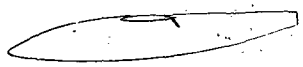
ALPHA = 14.01 DEG Q = 2845 N/SQ.M. (59.42 LBF/SQ.FT.) EFFECT. VELOC. RATIO = ∞

	ETA = .250		ETA = .387		ETA = .524		ETA = .800	
X/C	CP		CP		CP		CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000	-1.67		.12				-1.16	
.010	-1.85	.82	.54	.91				
.015					-.72	.97		
.025	-1.885	.998	-.763	.959	-.732	.929	-1.035	.897
.050	-1.955	.894	-.729	.853	-.698	.822	-1.051	.755
.100	-1.606	.732	-.753	.694	-.699	.679	-.997	.572
.150	-.920	.599	-.731	.576	-.707	.550	-1.002	.426
.250	-.819	.478	-.760	.448	-.756	.419	-1.093	
.400	-.793	.375	-.843	.367	-.816	.326	-1.055	.112
.550	-.851	.406	-.890	.395		.352	-1.090	.033
.700	-.858	.494	-.847	.491	-.811	.414	-.946	-.011
.850	-.810	.656	-.801	.654	-.764	.603	-.746	-.104
.950	-.714	.720	-.753	.712	-.724	.665	-.572	-.220

TABLE 5 - CONCLUDED

PRESSURE PROFILE ON WING

LIFT JET ALONE WITH FLAPS



ALPHA = 16.05 DEG Q = 2845 N/SQ.M. (59.42 LBF/SQ.FT.) EFFECT. VELOC. RATIO = ∞

X/C	ETA = .250 CP		ETA = .387 CP		ETA = .524 CP		ETA = .800 CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000	-1.74		-.10		-.48		-1.08	
.010	-1.77	.76	.51	.87				
.015					-.72	.96		
.025	-1.804	1.000	-.752	.977	-.736	.960	-.831	.918
.050	-1.904	.935	-.731	.882	-.741	.863	-.811	.786
.100	-1.531	.780	-.782	.735	-.731	.727	-.813	.589
.150	-.947	.647	-.742	.627	-.716	.606	-.806	.455
.250	-.774		-.790	.493	-.731	.461	-.792	.290
.400	-.823	.423	-.811	.396	-.786	.356	-.900	
.550	-.832	.434	-.833	.415		.387	-.968	.031
.700	-.837	.494	-.827	.484	-.798	.433	-.921	-.041
.850	-.798		-.787	.682	-.760	.608	-.916	-.162
.950	-.751	.721	-.776	.708	-.791	.664	-.758	-.312

ALPHA = 18.00 DEG Q = 2849 N/SQ.M. (59.51 LBF/SQ.FT.) EFFECT. VELOC. RATIO = ∞

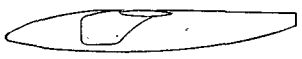
X/C	ETA = .250 CP		ETA = .387 CP		ETA = .524 CP		ETA = .800 CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000	-.81		-.11		-.48		-.92	
.010	-.81	.85	.49	.82				
.015					-.67	.96		
.025	-.622	.999	-.717	.985	-.632	.963	-.743	.921
.050	-.662	.934		.912	-.645	.879	-.682	
.100	-.677	.797	-.619	.761	-.721	.749	-.705	.638
.150	-.633	.672	-.660	.651	-.611	.636	-.728	.475
.250	-.800	.551	-.557	.549	-.672	.496	-.693	.302
.400	-.766	.439	-.857	.421	-.734	.390	-.704	.145
.550	-.837	.455	-.788	.445		.388	-.782	.038
.700	-.675	.503	-.715	.509	-.661	.450	-.791	-.047
.850	-.745	.668	-.780	.665	-.723	.619	-.764	-.190
.950	-.755	.734	-.887	.717	-.709	.666	-.774	-.372

ALPHA = 20.01 DEG Q = 2848 N/SQ.M. (59.49 LBF/SQ.FT.) EFFECT. VELOC. RATIO = ∞

X/C	ETA = .250 CP		ETA = .387 CP		ETA = .524 CP		ETA = .800 CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000	-.74		-.14		-.58		-.98	
.010	-.59	.80	.46	.80				
.015					-.61	.95		
.025	-.664	1.008	-.629	.990	-.617	.970	-.714	.940
.050	-.585	.966	-.632	.936	-.597	.908	-.738	.857
.100	-.651	.854	-.607	.806	-.621	.794	-.718	.684
.150	-.594	.727	-.588	.709	-.585	.696	-.631	.536
.250	-.634	.589	-.807	.580	-.777	.530	-.697	.373
.400	-.650	.486	-.823	.466	-.645	.425	-.718	.192
.550	-.643	.472	-.650	.464		.436	-.840	.075
.700	-.741	.532	-.841	.524	-.663	.476	-.863	-.017
.850	-.707	.691	-.726	.682	-.699	.623	-.778	
.950	-.748	.720	-.752	.724	-.724	.669	-.779	-.363

TABLE 6 (A)

PRESSURE PROFILE ON WING



FORWARD NOZZLES DEFLECTED 0 DEGREES

ALPHA = -.02 DEG Q = 2841 N/SQ.M. (59.33 LBF/SQ.FT.) EFFECT. VELOC. RATIO = ∞

	ETA = .250		ETA = .387		ETA = .524		ETA = .800	
X/C	CP		CP		CP		CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			.443		.866		.987	
.010			.729	-.253				
.015					-.016	-.375		
.025			-.245	-.629	-.095	-.470	-.137	-.346
.050			-.271	-.409	-.211	-.339	-.221	-.293
.100			-.323	-.434	-.190	-.314	-.226	-.241
.150			-.276	-.413	-.253	-.325	-.227	-.280
.250	-.241		-.310	-.307	-.336	-.356	-.293	-.330
.400	-.299		-.314	-.229	-.307	-.324	-.288	-.304
.550	-.218	-.065	-.229	-.112	-.230	-.214	-.205	-.239
.700	-.152	-.047	-.123	-.043	-.118	-.084	-.100	-.115
.850	-.088	-.020	-.058	-.019	-.013	-.010	-.005	-.026
.950	-.041	.002	.005	.040	.057	.070	.063	.070

ALPHA = 3.98 DEG Q = 2846 N/SQ.M. (59.44 LBF/SQ.FT.) EFFECT. VELOC. RATIO = ∞

	ETA = .250		ETA = .387		ETA = .524		ETA = .800	
X/C	CP		CP		CP		CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			1.00		.889		.601	
.010			.793	.590				
.015					-.963	.370		
.025			-1.025	.233	-1.172	.301	-1.145	.414
.050			-.767	.154	-.790	.163	-.728	.237
.100			-.546	-.029	-.617	.060	-.590	.060
.150			-.541	-.047	-.585	-.063	-.502	-.031
.250	-.329		-.512	-.169	-.501	-.142	-.483	-.146
.400	-.358		-.436	-.139	-.424	-.177	-.389	-.171
.550	-.257	-.015	-.282	-.070	-.300	-.121	-.260	-.154
.700	-.162	.006	-.165	-.005	-.158	-.068	-.131	-.096
.850	-.073	.023	-.064	.026	-.030	.011	-.022	-.024
.950	-.002	.039	.023	.072	.062	.067	.059	.065

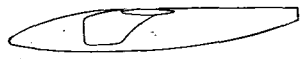
ALPHA = 7.98 DEG Q = 2847 N/SQ.M. (59.47 LBF/SQ.FT.) EFFECT. VELOC. RATIO = ∞

	ETA = .250		ETA = .387		ETA = .524		ETA = .800	
X/C	CP		CP		CP		CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			.536		-.512		-1.543	
.010			.678	1.0				
.015					-2.827	.957		
.025			-2.199	.755	-2.219	.769	-2.468	.749
.050			-1.160	.502	-1.462	.564	-1.430	.593
.100			-.908	.329	-1.061	.384	-.927	.367
.150			-.800	.228	-.864	.184	-.886	.194
.250	-.427		-.661	.055	-.704	.078	-.692	.062
.400	-.415		-.515	-.034	-.547	-.024	-.528	-.078
.550	-.294	.023	-.329	-.010	-.362	-.046	-.354	-.057
.700	-.175	.042	-.189	.029	-.197	-.013	-.193	-.045
.850	-.064	.050	-.059	.054	-.053	.032	-.068	-.003
.950	.017	.057	.037	.083	.048	.061	.017	.035

TABLE 6 (A) - CONTINUED

PRESSURE PROFILE ON WING

FORWARD NOZZLES DEFLECTED 0 DEGREES



ALPHA = 9.99 DEG Q = 2847 N/SQ.M. (59.46 LBF/SQ.FT.) EFFECT. VELOC. RATIO = ∞

X/C	ETA = .250 CP		ETA = .387 CP		ETA = .524 CP		ETA = .800 CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			-.090		-1.758		-3.216	
.010			.449	.993				
.015					-3.637	.999		
.025			-2.597	.894	-3.136	.906	-2.325	.887
.050			-1.660	.713	-1.768	.714	-1.662	.719
.100			-1.107	.493	-1.224	.489	-1.196	.452
.150			-.962	.319	-1.035	.346	-.969	.287
.250	-.463		-.738	.156	-.793	.152	-.790	.126
.400	-.443		-.554	.031	-.607	.022	-.570	.011
.550	-.304	.038	-.357	.022	-.385	.007	-.370	-.029
.700	-.172	.056	-.158	.038	-.210	.024	-.206	-.013
.850	-.059	.058	-.053	.059	-.056	.045	-.071	.015
.950	.014	.063	.043	.088	.042	.069	.019	.046

ALPHA = 11.99 DEG Q = 2849 N/SC.M. (59.49 LBF/SQ.FT.) EFFECT. VELOC. RATIO = ∞

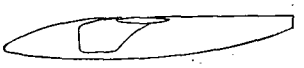
X/C	ETA = .250 CP		ETA = .387 CP		ETA = .524 CP		ETA = .800 CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			.314		-.313		-.991	
.010			.430	.987				
.015					-1.199	.974		
.025			-1.161	.924	-1.137	.857	-1.270	.841
.050			-1.786	.747	-1.323	.694	-1.197	.669
.100			-1.267	.512	-.985	.486	-1.080	.453
.150			-1.421	.352	-1.126	.331	-1.105	.300
.250	-.597		-.960	.165	-.989	.160	-.970	.125
.400	-.524		-.695	.030	-.730	.013	-.683	-.004
.550	-.389	.039	-.510	.015	-.519	-.016	-.539	-.057
.700	-.223	.036	-.359	.027	-.339	-.023	-.384	-.060
.850	-.116	.044	-.215	.019	-.252	-.038	-.211	-.071
.950	-.009	.053	-.042	.004	-.194	-.071	-.135	-.074

ALPHA = 15.96 DEG Q = 2846 N/SQ.M. (59.44 LBF/SQ.FT.) EFFECT. VELOC. RATIO = ∞

X/C	ETA = .250 CP		ETA = .387 CP		ETA = .524 CP		ETA = .800 CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			.438		.132		-.685	
.010			.529	1.0				
.015					-.439	.973		
.025			-.494	.938	-.457	.874	-.745	.872
.050			-.479	.789	-.422	.717	-.709	.692
.100			-.536	.602	-.408	.537	-.742	.493
.150			-.459	.395	-.448	.368	-.723	.326
.250	-.480		-.521	.211	-.468	.181	-.712	.155
.400	-.548		-.619	.009	-.538	.018	-.781	-.003
.550	-.618	-.041	-.637	-.032	-.573	-.049	-.760	-.085
.700	-.633	-.040	-.609	-.060	-.599	-.094	-.737	-.133
.850	-.573	-.094	-.575	-.139	-.544	-.181	-.653	-.191
.950	-.401	-.168	-.524	-.237	-.467	-.282	-.542	-.263

TABLE 6 (A) - CONCLUDED
PRESSURE PROFILE ON WING

FORWARD NOZZLES DEFLECTED 0 DEGREES



ALPHA = 19.97 DEG Q = 2840 N/SC.M. (59.31 LBF/SQ.FT.) EFFECT. VELOC. RATIO = ∞

X/C	ETA = .250 CP		ETA = .387 CP		ETA = .524 CP		ETA = .800 CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			.121		-.114		-.793	
.010			.454	.926				
.015					-.438	1.0		
.025			-.474	1.00	-.425	.963	-.602	.932
.050			-.465	.910	-.417	.829	-.611	.796
.100			-.469	.715	-.428	.648	-.566	.590
.150			-.491	.545	-.426	.489	-.609	.425
.250	-.482		-.509	.280	-.443	.284	-.596	.236
.400	-.545		-.539	.061	-.527	.085	-.659	.049
.550	-.593	-.044	-.577	-.016	-.550	-.006	-.680	-.066
.700	-.611	-.066	-.593	-.056	-.559	-.079	-.730	-.156
.850	-.641	-.140	-.585	-.171	-.538	-.192	-.724	-.267
.950	-.629	-.283	-.559	-.316	-.477	-.306	-.691	-.403

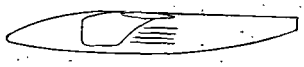
ALPHA = -.03 DEG Q = 2840 N/SC.M. (59.32 LBF/SQ.FT.) EFFECT. VELOC. RATIO = ∞

X/C	ETA = .250 CP		ETA = .387 CP		ETA = .524 CP		ETA = .800 CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			.315		.842		.992	
.010			.523	-.306				
.015					.016	-.507		
.025			-.170	-.588	-.282	-.441	-.148	-.257
.050			-.217	-.335	-.187	-.295	-.201	-.271
.100			-.261	-.388	-.230	-.279	-.213	-.219
.150			-.274	-.472	-.288	-.360	-.257	-.302
.250	-.232		-.313	-.332	-.356	-.364	-.285	-.332
.400	-.302		-.319	-.227	-.309	-.307	-.299	-.299
.550	-.226	-.059	-.220	-.112	-.213	-.215	-.208	-.225
.700	-.148	-.047	-.126	-.034	-.121	-.080	-.091	-.108
.850	-.080	-.007	-.049	-.014	-.011	-.009	.004	-.016
.950	-.023	.003	.010	.045	.058	.074	.069	.078

TABLE 6 (B)

PRESSURE PROFILE ON WING

FORWARD NOZZLES DEFLECTED 0 DEGREES



ALPHA = -.02 DEG Q = 1166 N/SQ.M. (24.35 LBF/SQ.FT.) EFFECT. VELOC. RATIO = .298

	ETA = .250		ETA = .387		ETA = .524		ETA = .800	
X/C	CP		CP		CP		CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			.143		.501		1.031	
.010			.574	-.958				
.015					.297	-.481		
.025			.114	-.944	.103	-.717	.091	-.213
.050			.012	-.704	-.023	-.508	-.136	-.181
.100			-.054	-.639	-.058	-.415	-.085	-.216
.150			-.144	-.669	-.058	-.359	-.136	-.332
.250	-.096		-.197	-.532	-.190	-.342	-.197	-.268
.400	-.166		-.174	-.337	-.168	-.295	-.181	-.274
.550	-.082	-.180	-.101	-.099	-.095	-.192	-.109	-.177
.700	-.011	-.121	-.019	-.003	-.027	-.048	-.046	-.071
.850	.070	.033	.076	.088	.091	.053	.069	.035
.950	.158	.160	.150	.177	.143	.134	.134	.123

ALPHA = 4.01 DEG Q = 1168 N/SQ.M. (24.40 LBF/SQ.FT.) EFFECT. VELOC. RATIO = .298

	ETA = .250		ETA = .387		ETA = .524		ETA = .800	
X/C	CP		CP		CP		CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			1.046		1.048		.746	
.010			.921	.256				
.015					-.628	.350		
.025			-.749	.117	-.817	.228	-.823	.396
.050			-.483	.119	-.537	.152	-.577	.223
.100			-.385	-.152	-.431	.002	-.403	.115
.150			-.384	-.200	-.413	-.049	-.370	.042
.250	-.204		-.351	-.267	-.376	-.128	-.378	-.076
.400	-.259		-.301	-.193	-.305	-.142	-.297	-.122
.550	-.149	-.238	-.163	-.031	-.181	-.067	-.160	-.084
.700	-.060	-.173	-.060	.058	-.053	.012	-.037	-.020
.850	.042	-.003	.052	.114	.061	.086	.065	.058
.950	.135	.114	.143	.175	.145	.153	.137	.146

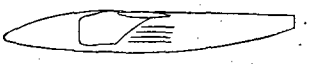
ALPHA = 7.99 DEG Q = 1169 N/SQ.M. (24.41 LBF/SQ.FT.) EFFECT. VELOC. RATIO = .297

	ETA = .250		ETA = .387		ETA = .524		ETA = .800	
X/C	CP		CP		CP		CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			.864		-.050		-1.667	
.010			.926	1.069				
.015					-2.011	.903		
.025			-1.646	.748	-2.305	.814	-1.974	.816
.050			-1.034	.549	-1.069	.616	-1.017	.607
.100			-.702	.303	-.806	.364	-.873	.418
.150			-.631	.147	-.714	.256	-.683	.248
.250	-.287		-.508	.002	-.575	.075	-.582	.087
.400	-.302		-.377	-.038	-.416	.016	-.413	.027
.550	-.186	-.251	-.208	.068	-.243	.036	-.240	.007
.700	-.070	-.181	-.082	.125	-.099	.087	-.098	.043
.850	.042	-.002	.047	.154	.044	.115	.023	.083
.950	.122	.102	.147	.188	.136	.156	.108	.132

TABLE 6 (B) - CONTINUED

PRESSURE PROFILE ON WING

FORWARD NOZZLES DEFLECTED 0 DEGREES



ALPHA = 9.99 DEG Q = 1172 N/SQ.M. (24.48 LBF/SQ.FT.) EFFECT. VELOC. RATIO = .298

	ETA = .250		ETA = .387		ETA = .524		ETA = .800	
X/C	CP		CP		CP		CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			.251		-1.266		-2.575	
.010			.726	1.066				
.015					-3.071	1.049		
.025			-2.705	.982	-3.036	.922	-2.254	.934
.050			-1.408	.715	-1.527	.763	-1.447	.728
.100			-.867	.466	-1.058	.584	-1.042	.520
.150			-.774	.293	-.895	.371	-.847	.375
.250	-.339		-.603	.102	-.640	.212	-.668	.177
.400	-.344		-.433	.033	-.475	.100	-.462	.069
.550	-.209	-.300	-.245	.107	-.274	.093	-.268	.035
.700	-.087	-.163	-.104	.138	-.113	.105	-.105	.054
.850	.025	-.025	.037	.151	.040	.130	.021	.086
.950	.102	.085	.129	.176	.127	.151	.104	.123

ALPHA = 11.98 DEG Q = 1167 N/SQ.M. (24.38 LBF/SQ.FT.) EFFECT. VELOC. RATIO = .297

	ETA = .250		ETA = .387		ETA = .524		ETA = .800	
X/C	CP		CP		CP		CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			-.720		-2.476		-1.955	
.010			.523	.940				
.015					-4.493	1.048		
.025			-2.439	1.062	-2.972	1.040	-1.451	1.036
.050			-1.683	.868	-1.922	.903	-1.816	.858
.100			-1.055	.680	-1.274	.650	-1.243	.638
.150			-.949	.426	-1.034	.484	-.983	.445
.250	-.369		-.654	.230	-.779	.315	-.744	.275
.400	-.364		-.483	.125	-.542	.168	-.518	.145
.550	-.226	-.274	-.268	.143	-.293	.132	-.291	.092
.700	-.086	-.168	-.103	.176	-.115	.144	-.116	.092
.850	.027	-.015	.042	.164	.035	.137	-.135	.100
.950	.111	.089	.121	.165	.108	.133	-.005	.118

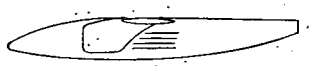
ALPHA = 15.97 DEG Q = 1166 N/SQ.M. (24.35 LBF/SQ.FT.) EFFECT. VELOC. RATIO = .297

	ETA = .250		ETA = .387		ETA = .524		ETA = .800	
X/C	CP		CP		CP		CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			.416		.014		-.822	
.010			.682	1.044				
.015					-.592	1.061		
.025			-.480	1.024	-.523	.966	-.792	.966
.050			-.457	.889	-.591	.834	-.790	.796
.100			-.540	.678	-.538	.639	-.752	.606
.150			-.459	.492	-.499	.474	-.755	.450
.250	-.463		-.480	.238	-.600	.292	-.826	.261
.400	-.558		-.692	.052	-.553	.128	-.797	.098
.550	-.540	-.392	-.567	.051	-.662	.056	-.752	.027
.700	-.497	-.323	-.577	.055	-.569	.039	-.653	-.020
.850	-.442	-.209	-.506	-.034	-.466	-.080	-.492	-.089
.950	-.286	-.242	-.400	-.169	-.383	-.164	-.438	-.163

TABLE 6 (B) - CONCLUDED

PRESSURE PROFILE ON WING

FORWARD NOZZLES DEFLECTED 0 DEGREES



ALPHA = 20.02 DEG Q = 1169 N/SQ.M. (24.42 LBF/SQ.FT.) EFFECT. VELOC. RATIO = .297

	ETA = .250		ETA = .387		ETA = .524		ETA = .800	
X/C	CP		CP		CP		CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			.175		-.262		-.976	
.010			.615	.971				
.015					-.453	1.076		
.025			-.385	1.072	-.449	1.041	-.709	1.023
.050			-.382	.975	-.445	.925	-.696	.903
.100			-.389	.794	-.441	.771	-.677	.708
.150			-.413	.606	-.444	.606	-.664	.557
.250	-.418		-.437	.351	-.463	.387	-.707	.337
.400	-.473		-.473	.142	-.521	.218	-.726	.147
.550	-.489	-.428	-.510	.095	-.551	.099	-.755	.038
.700	-.564	-.375	-.521	.040	-.567	.006	-.763	-.038
.850	-.563	-.305	-.506	-.107	-.547	-.119	-.731	-.156
.950	-.592	-.393	-.472	-.230	-.513	-.271	-.640	-.294

ALPHA = -.01 DEG Q = 1167 N/SQ.M. (24.37 LBF/SQ.FT.) EFFECT. VELOC. RATIO = .297

	ETA = .250		ETA = .387		ETA = .524		ETA = .800	
X/C	CP		CP		CP		CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			.323		.685		1.037	
.010			.704	-.734				
.015					.288	-.579		
.025			.185	-.903	.101	-.517	-.012	-.390
.050			-.086	-.611	.051	-.505	-.028	-.309
.100			.011	-.602	-.039	-.412	-.104	-.205
.150			-.139	-.622	-.086	-.418	-.125	-.301
.250	-.097		-.140	-.546	-.147	-.375	-.182	-.283
.400	-.135		-.152	-.310	-.164	-.298	-.160	-.219
.550	-.069	-.150	-.096	-.107	-.084	-.167	-.087	-.149
.700	.007	-.109	-.003	.020	-.019	-.051	-.015	-.049
.850	.091	.021	.101	.107	.115	.065	.096	.056
.950	.175	.182	.164	.193	.154	.162	.146	.150

TABLE 7 (A)

PRESSURE PROFILE ON WING

FORWARD NOZZLES DEFLECTED 0 DEGREES WITH FLAPS

ALPHA = -.02 DEG Q = 2838 N/SQ.M. (59.28 LBF/SQ.FT.) EFFECT. VELOC. RATIO = ∞

	ETA = .250		ETA = .387		ETA = .524		ETA = .800	
X/C	CP		CP		CP		CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			.954		.979		.914	
.010			.555	.334				
.015					-.553	.102		
.025			-.653	-.050	-.728	.141	-.542	.072
.050			-.474	-.094	-.558	.003	-.498	-.031
.100			-.433	-.148	-.497	-.038	-.431	-.106
.150			-.471	-.168	-.520	-.106	-.403	-.160
.250	-.373		-.472	-.136	-.453	-.120	-.433	-.195
.400	-.438		-.482	-.034	-.474	-.064	-.398	-.195
.550	-.384	.105	-.404	.105	-.391	.090	-.310	-.147
.700	-.337	.143	-.326	.218	-.309	.260	-.165	-.084
.850	-.323	.178	-.279	.353	-.221	.507	-.072	.005
.950	-.330	.164	-.237	.471	-.182	.629	.006	.061

ALPHA = 4.00 DEG Q = 2851 N/SQ.M. (59.55 LBF/SQ.FT.) EFFECT. VELOC. RATIO = ∞

	ETA = .250		ETA = .387		ETA = .524		ETA = .800	
X/C	CP		CP		CP		CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			.821		.196		-.420	
.010			.524	.969				
.015					-2.045	.772		
.025			-1.508	.566	-1.633	.653	-1.511	.571
.050			-1.083	.424	-1.114	.456	-1.195	.429
.100			-.823	.277	-.886	.318	-.714	.199
.150			-.713	.167	-.845	.229	-.756	.134
.250	-.490		-.677	.102	-.706	.049	-.664	-.015
.400	-.522		-.598	.073	-.622	.078	-.515	-.086
.550	-.440	.187	-.475	.166	-.469	.172	-.369	-.050
.700	-.377	.210	-.381	.298	-.357	.338	-.238	-.019
.850	-.352	.250	-.300	.450	-.257	.551	-.108	-.001
.950	-.363	.220	-.227	.579	-.192	.669	-.016	.034

ALPHA = 8.00 DEG Q = 2844 N/SQ.M. (59.40 LBF/SQ.FT.) EFFECT. VELOC. RATIO = ∞

	ETA = .250		ETA = .387		ETA = .524		ETA = .800	
X/C	CP		CP		CP		CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			-.283		-2.042		-3.321	
.010			.287	.976				
.015					-3.609	.987		
.025			-2.749	.931	-3.274	.930	-2.627	.916
.050			-1.742	.817	-1.917	.774	-1.779	.693
.100			-1.201	.573	-1.366	.579	-1.203	.507
.150			-1.077	.430	-1.170	.450	-1.018	.356
.250	-.605		-.898	.283	-.940	.270	-.865	.170
.400	-.606		-.726	.210	-.767	.202	-.662	.036
.550	-.500	.244	-.552	.260	-.559	.252	-.465	.010
.700	-.427	.266	-.441	.349	-.419	.336	-.290	.017
.850	-.373	.309	-.335	.524	-.290	.585	-.148	.017
.950	-.393	.266	-.243	.646	-.223	.682	-.048	.025

TABLE 7 (A) – CONTINUED

PRESSURE PROFILE ON WING

FORWARD NOZZLES DEFLECTED 0 DEGREES WITH FLAPS

ALPHA = 10.01 DEG Q = 2845 N/SQ.M. (59.43 LBF/SQ.FT.) EFFECT. VELOC. RATIO = ∞

	ETA = .250		ETA = .387		ETA = .524		ETA = .800	
X/C	CP		CP		CP		CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			-.085		-.534		-1.362	
.010			.246	.941				
.015					-1.324	.994		
.025			-1.553	.956	-1.283	.915	-1.496	.874
.050			-1.386	.829	-1.411	.770	-1.492	.706
.100			-1.576	.640	-1.444	.594	-1.606	.495
.150			-1.371	.487	-1.313	.450	-1.407	.371
.250	-.776		-1.410	.333	-1.370	.313	-1.371	.193
.400	-.764		-.973	.255	-1.091	.241	-.987	.065
.550	-.549	.263	-.667	.286	-.682	.288	-.600	.022
.700	-.486	.291	-.601	.374	-.648	.403	-.387	.011
.850	-.357	.328	-.401	.546	-.460	.601	-.263	-.022
.950	-.311	.312	-.348	.662	-.405	.689	-.166	-.050

ALPHA = 12.46 DEG Q = 2846 N/SQ.M. (59.44 LBF/SQ.FT.) EFFECT. VELOC. RATIO = ∞

	ETA = .250		ETA = .387		ETA = .524		ETA = .800	
X/C	CP		CP		CP		CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			.232		-.263		-1.111	
.010			.233	.968				
.015					-.734	.986		
.025			-.708	.972	-.817	.915	-1.086	.878
.050			-.754	.872	-.799	.781	-1.073	.719
.100			-.783	.665	-.793	.621	-1.153	.521
.150			-.736	.542	-.835	.496	-1.107	.378
.250	-.670		-.811	.365	-.770	.353	-1.165	.217
.400	-.825		-.807	.262	-.908	.268	-1.187	.078
.550	-.829	.277	-.893	.294	-.941	.306	-1.069	.019
.700	-.809	.307	-.894	.388	-.867	.407	-.819	-.012
.850	-.673	.333	-.719	.568	-.748	.599	-.659	-.061
.950	-.543	.324	-.668	.671	-.643	.687	-.441	-.131

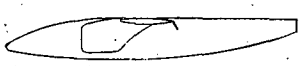
ALPHA = 16.00 DEG Q = 2842 N/SQ.M. (59.36 LBF/SQ.FT.) EFFECT. VELOC. RATIO = ∞

	ETA = .250		ETA = .387		ETA = .524		ETA = .800	
X/C	CP		CP		CP		CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			-.175		-.424		-.990	
.010			.132	.843				
.015					-.714	.992		
.025			-.712	1.002	-.723	.972	-.802	.900
.050			-.707	.918	-.699	.857	-.822	.771
.100			-.734	.763	-.719	.709	-.811	.592
.150			-.726	.651	-.712	.587	-.790	.444
.250	-.725		-.743	.449	-.700	.432	-.793	.278
.400	-.792		-.759	.318	-.743	.317	-.890	.116
.550	-.768	.289	-.788	.329	-.729	.334	-.979	.021
.700	-.777	.318	-.784	.409	-.734	.406	-.971	-.050
.850	-.754	.355	-.752	.581	-.765	.605	-.878	-.148
.950	-.690	.359	-.756	.666	-.796	.694	-.777	-.322

TABLE 7 (A) - CONCLUDED

PRESSURE PROFILE ON WING

FORWARD NOZZLES DEFLECTED 0 DEGREES WITH FLAPS



ALPHA = 19.99 DEG Q = 2838 N/SQ.M. (59.28 LBF/SQ.FT.) EFFECT. VELOC. RATIO = ∞

	ETA = .250		ETA = .387		ETA = .524		ETA = .800	
X/C	CP		CP		CP		CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			-.435		-.660		-.942	
.010			.024	.699				
.015					-.685	.953		
.025			-.707	.991	-.692	1.000	-.690	.944
.050			-.680	.882	-.675	.933	-.695	.856
.100			-.705	.865	-.692	.800	-.691	.679
.150			-.690	.753	-.681	.673	-.690	.536
.250	-.690		-.717	.550	-.682	.552	-.687	.361
.400	-.746		-.757	.384	-.710	.390	-.751	.189
.550	-.729	.313	-.745	.330	-.729	.396	-.737	.071
.700	-.738	.345	-.740	.439	-.758	.469	-.762	-.021
.850	-.739	.386	-.767	.596	-.738	.630	-.747	-.174
.950	-.708	.396	-.769	.677	-.787	.710	-.731	-.365

ALPHA = .01 DEG Q = 2840 N/SQ.M. (59.32 LBF/SQ.FT.) EFFECT. VELOC. RATIO = ∞

	ETA = .250		ETA = .387		ETA = .524		ETA = .800	
X/C	CP		CP		CP		CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			.942		.976		.907	
.010			.297	.221				
.015					-.579	.266		
.025			-.627	-.009	-.677	.018	-.555	.005
.050			-.496	-.088	-.530	.026	-.466	-.039
.100			-.439	-.174	-.468	-.008	-.378	-.100
.150			-.473	-.174	-.440	-.126	-.409	-.100
.250	-.378		-.467	-.126	-.489	-.125	-.431	-.204
.400	-.439		-.472	-.035	-.459	-.066	-.398	-.204
.550	-.374	.122	-.411	.100	-.379	.086	-.313	-.146
.700	-.335	.151	-.318	.207	-.301	.272	-.165	-.085
.850	-.319	.179	-.282	.365	-.219	.516	-.075	.004
.950	-.322	.175	-.238	.467	-.184	.624	.007	.062

TABLE 7 (B)

PRESSURE PROFILE ON WING

FORWARD NOZZLES DEFLECTED 0 DEGREES WITH FLAPS

ALPHA = .00 DEG Q = 1169 N/SQ.M. (24.41 LBF/SQ.FT.) EFFECT. VELOC. RATIO = .297

	ETA = .250		ETA = .387		ETA = .524		ETA = .800	
X/C	CP		CP		CP		CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			.978		.896		.727	
.010			.013	.539				
.015					-.957	.404		
.025			-.851	.225	-.929	.351	-.666	.285
.050			-.553	.175	-.628	.217	-.541	.134
.100			-.503	.110	-.635	.146	-.552	.067
.150			-.602	-.041	-.630	.079	-.437	-.034
.250	-.422		-.576	.012	-.611	.057	-.472	-.094
.400	-.539		-.587	.278	-.602	.180	-.410	-.063
.550	-.498	.966	-.527	.548	-.504	.362	-.314	-.048
.700	-.495	1.498	-.511	.606	-.476	.562	-.182	.001
.850	-.561	1.404	-.579	.431	-.516	.560	-.080	.048
.950	-.678	2.811	-.645	.549	-.706	.446	.013	.097

ALPHA = 4.01 DEG Q = 1163 N/SQ.M. (24.29 LBF/SQ.FT.) EFFECT. VELOC. RATIO = .296

	ETA = .250		ETA = .387		ETA = .524		ETA = .800	
X/C	CP		CP		CP		CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			.414		-.342		-1.284	
.010			.271	1.016				
.015					-2.481	.949		
.025			-2.125	.758	-2.067	.789	-2.328	.713
.050			-1.335	.568	-1.385	.651	-1.152	.506
.100			-.903	.462	-1.001	.460	-.900	.346
.150			-.917	.313	-.956	.369	-.841	.190
.250	-.550		-.797	.219	-.864	.258	-.712	.081
.400	-.647		-.734	.354	-.767	.252	-.575	.002
.550	-.600	.965	-.647	.587	-.639	.397	-.420	.013
.700	-.567	1.394	-.600	.686	-.569	.566	-.281	.017
.850	-.591	1.401	-.640	.478	-.604	.608	-.149	.023
.950	-.684	2.654	-.700	.618	-.756	.497	-.038	.029

ALPHA = 7.99 DEG Q = 1171 N/SQ.M. (24.45 LBF/SQ.FT.) EFFECT. VELOC. RATIO = .298

	ETA = .250		ETA = .387		ETA = .524		ETA = .800	
X/C	CP		CP		CP		CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			-1.097		-3.574		-4.292	
.010			-.107	.813				
.015					-4.877	.965		
.025			-3.028	.991	-3.821	.988	-2.686	.958
.050			-2.058	.876	-2.222	.879	-1.910	.813
.100			-1.424	.677	-1.520	.715	-1.416	.602
.150			-1.232	.580	-1.355	.571	-1.126	.404
.250	-.707		-1.009	.441	-1.084	.431	-.945	.237
.400	-.763		-.905	.464	-.908	.370	-.734	.131
.550	-.665	.910	-.736	.626	-.757	.437	-.528	.070
.700	-.624	1.292	-.674	.743	-.658	.574	-.350	.065
.850	-.644	1.603	-.703	.632	-.659	.636	-.190	.035
.950	-.734	2.492	-.741	.669	-.813	.557	-.079	.009

TABLE 7 (B) - CONTINUED

PRESSURE PROFILE ON WING

FORWARD NOZZLES DEFLECTED 0 DEGREES WITH FLAPS

ALPHA = 10.01 DEG Q = 1173 N/SQ.M. (24.50 LBF/SQ.FT.) EFFECT. VELOC. RATIO = .298

	ETA = .250		ETA = .387		ETA = .524		ETA = .800	
X/C	CP		CP		CP		CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			-.643		-1.217		-2.215	
.010			-.233	.774				
.015					-1.752	.989		
.025			-2.161	1.010	-1.622	.974	-1.879	.936
.050			-1.900	.938	-1.643	.879	-1.988	.799
.100			-2.082	.760	-1.702	.720	-1.936	.602
.150			-1.921	.640	-1.677	.598	-1.988	.435
.250	-.860		-1.713	.488	-1.724	.475	-1.836	.286
.400	-.905		-1.272	.511	-1.356	.428	-1.349	.153
.550	-.779	.943	-.891	.634	-1.099	.464	-.738	.086
.700	-.712	1.272	-.768	.753	-.818	.591	-.479	.063
.850	-.692	1.689	-.728	.711	-.715	.667	-.298	.009
.950	-.731	2.472	-.804	.740	-.753	.536	-.180	-.037

ALPHA = 12.00 DEG Q = 1162 N/SQ.M. (24.26 LBF/SQ.FT.) EFFECT. VELOC. RATIO = .297

	ETA = .250		ETA = .387		ETA = .524		ETA = .800	
X/C	CP		CP		CP		CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			-.608		-1.130		-2.269	
.010			-.301	.766				
.015					-1.356	.960		
.025			-1.508	.994	-1.389	.980	-1.871	.943
.050			-1.366	.941	-1.357	.899	-1.808	.827
.100			-1.470	.778	-1.341	.740	-1.813	.642
.150			-1.341	.676	-1.470	.633	-1.869	.497
.250	-1.158		-1.292	.543	-1.517	.511	-1.881	.324
.400	-1.264		-1.455	.528	-1.497	.448	-1.801	.186
.550	-.953	.900	-1.359	.660	-1.451	.485	-1.320	.095
.700	-.780	1.214	-1.039	.757	-1.185	.597	-.844	.066
.850	-1.004	1.619	-.937	.702	-1.140	.673	-.372	.006
.950	-.898	2.520	-.906	.772	-.893	.603	-.208	-.032

ALPHA = 16.01 DEG Q = 1167 N/SQ.M. (24.37 LBF/SQ.FT.) EFFECT. VELOC. RATIO = .297

	ETA = .250		ETA = .387		ETA = .524		ETA = .800	
X/C	CP		CP		CP		CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			-.766		-1.198		-2.036	
.010			-.305	.596				
.015					-1.214	.896		
.025			-1.096	.973	-1.196	.984	-1.659	.941
.050			-1.130	.977	-1.206	.960	-1.700	.864
.100			-1.106	.868	-1.176	.826	-1.609	.696
.150			-1.146	.779	-1.236	.744	-1.615	.563
.250	-1.092		-1.192	.643	-1.306	.600	-1.690	.391
.400	-1.220		-1.240	.607	-1.355	.523	-1.722	.232
.550	-1.290	.949	-1.291	.689	-1.391	.522	-1.711	.128
.700	-1.308	1.313	-1.287	.796	-1.372	.615	-1.480	.061
.850	-1.251	1.531	-1.220	.747	-1.227	.677	-1.090	-.030
.950	-1.094	2.551	-1.173	.806	-1.259	.565	-.666	-.194

TABLE 7 (B) - CONCLUDED

PRESSURE PROFILE ON WING

FORWARD NOZZLES DEFLECTED 0 DEGREES WITH FLAPS

ALPHA = 20.00 DEG Q = 1172 N/SQ.M. (24.48 LBF/SQ.FT.) EFFECT. VELOC. RATIO = .298

	ETA = .250		ETA = .387		ETA = .524		ETA = .800	
X/C	CP		CP		CP		CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			-1.199		-1.357		-1.671	
.010			-.469	.295				
.015					-1.152	.792		
.025			-1.179	.876	-1.161	.947	-1.515	.923
.050			-1.226	.978	-1.159	.978	-1.529	.911
.100			-1.207	.940	-1.159	.896	-1.501	.759
.150			-1.207	.860	-1.185	.811	-1.519	.630
.250	-1.233		-1.241	.713	-1.246	.685	-1.519	.464
.400	-1.283		-1.299	.664	-1.336	.565	-1.591	.281
.550	-1.304	.967	-1.314	.709	-1.322	.566	-1.622	.142
.700	-1.340	1.285	-1.329	.801	-1.316	.614	-1.625	.014
.850	-1.367	1.515	-1.317	.743	-1.310	.664	-1.396	-.164
.950	-1.255	2.416	-1.360	.795	-1.389	.567	-1.174	-.432

ALPHA = .00 DEG Q = 1168 N/SQ.M. (24.39 LBF/SQ.FT.) EFFECT. VELOC. RATIO = .298

	ETA = .250		ETA = .387		ETA = .524		ETA = .800	
X/C	CP		CP		CP		CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			.972		.864		.795	
.010			.412	.486				
.015					-1.121	.280		
.025			-.720	.231	-.729	.301	-.762	.214
.050			-.824	.211	-.745	.150	-.557	.207
.100			-.513	.049	-.612	.175	-.516	-.025
.150			-.621	-.013	-.580	.073	-.462	-.056
.250	-.441		-.623	-.024	-.644	.062	-.503	-.119
.400	-.537		-.617	.247	-.605	.150	-.436	-.110
.550	-.527	.924	-.536	.527	-.528	.325	-.352	-.075
.700	-.532	1.393	-.524	.578	-.506	.522	-.219	-.011
.850	-.578	1.286	-.599	.357	-.558	.510	-.103	.028
.950	-.697	2.799	-.698	.514	-.723	.433	-.014	.063

TABLE 8 (A)

PRESSURE PROFILE ON WING

FORWARD NOZZLES DEFLECTED 45 DEGREES

ALPHA = 0.00 DEG Q = 2853 N/SQ.M. (59.60 LBF/SQ.FT.) EFFECT. VELOC. RATIO = ∞

X/C	ETA = .250 CP		ETA = .387 CP		ETA = .524 CP		ETA = .800 CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			.594		.852		.959	
.010			.281	-.316				
.015					-.195	-.236		
.025			-.278	-.473	-.316	-.448	-.230	-.270
.050			-.354	-.336	-.381	-.285	-.262	-.278
.100			-.340	-.391	-.331	-.313	-.335	-.246
.150			-.360	-.453	-.309	-.366	-.266	-.309
.250	-.287		-.381	-.397	-.390	-.365	-.338	-.355
.400	-.352		-.374	-.355	-.356	-.369	-.323	-.314
.550	-.281	-.278	-.282	-.297	-.267	-.307	-.232	-.270
.700	-.220	-.311	-.188	-.239	-.176	-.197	-.130	-.151
.850	-.183	-.292	-.133	-.227	-.059	-.098	-.030	-.056
.950	-.196	-.262	-.086	-.123	.005	-.005	.033	.041

ALPHA = 4.04 DEG Q = 2845 N/SQ.M. (59.41 LBF/SQ.FT.) EFFECT. VELOC. RATIO = ∞

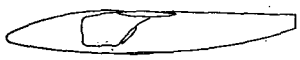
X/C	ETA = .250 CP		ETA = .387 CP		ETA = .524 CP		ETA = .800 CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			.989		.815		.385	
.010			.328	.684				
.015					-1.135	.531		
.025			-1.093	.486	-1.107	.343	-1.128	.466
.050			-.797	.231	-.850	.199	-.830	.190
.100			-.593	-.008	-.663	.059	-.600	.066
.150			-.591	-.162	-.624	-.078	-.571	-.060
.250	-.384		-.537	-.264	-.530	-.199	-.511	-.166
.400	-.406		-.466	-.292	-.470	-.243	-.426	-.222
.550	-.311	-.272	-.320	-.261	-.319	-.208	-.295	-.180
.700	-.227	-.309	-.212	-.219	-.193	-.160	-.157	-.133
.850	-.161	-.283	-.117	-.153	-.064	-.067	-.051	-.059
.950	-.153	-.221	-.043	-.049	.021	.009	.033	.041

ALPHA = 7.98 DEG Q = 2846 N/SQ.M. (59.45 LBF/SQ.FT.) EFFECT. VELOC. RATIO = ∞

X/C	ETA = .250 CP		ETA = .387 CP		ETA = .524 CP		ETA = .800 CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			.499		-.700		-1.493	
.010			.349					
.015					-2.574	.901		
.025			-2.220	.718	-2.395	.778	-2.370	.730
.050			-1.240	.515	-1.536	.548	-1.352	.572
.100			-.921	.313	-1.037	.347	-1.012	.319
.150			-.825	.132	-.916	.188	-.854	.173
.250	-.452		-.678	-.071	-.767	.029	-.726	.020
.400	-.440		-.539	-.218	-.574	-.119	-.534	-.081
.550	-.319	-.297	-.361	-.219	-.377	-.124	-.362	-.092
.700	-.207	-.329	-.219	-.188	-.217	-.077	-.204	-.061
.850	-.108	-.251	-.094	-.116	-.074	-.050	-.081	-.028
.950	-.067	-.142	-.005	-.014	.022	.015	.005	.018

TABLE 8 (A) - CONTINUED

PRESSURE PROFILE ON WING



FORWARD NOZZLES DEFLECTED 45 DEGREES

ALPHA = 9.98 DEG Q = 2847 N/SQ.M. (59.46 LBF/SQ.FT.) EFFECT. VELOC. RATIO = ∞

X/C	ETA = .250 CP		ETA = .387 CP		ETA = .524 CP		ETA = .800 CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			-.197		-2.039		-2.959	
.010			.300	.976				
.015					-3.677	.979		
.025			-2.556	.876	-2.959	.897	-2.397	.876
.050			-1.677	.668	-1.795	.704	-1.687	.685
.100			-1.086	.430	-1.292	.503	-1.228	.438
.150			-.963	.276	-1.118	.305	-.973	.284
.250	-.482		-.749	.012	-.809	.094	-.820	.100
.400	-.456		-.578	-.169	-.613	-.041	-.587	-.011
.550	-.317	-.308	-.368	-.194	-.405	-.060	-.389	-.062
.700	-.189	-.336	-.213	-.153	-.228	-.059	-.217	-.037
.850	-.090	-.216	-.081	-.088	-.079	-.016	-.088	-.011
.950	-.023	-.097	.012	.004	.010	.026	.000	.028

ALPHA = 12.01 DEG Q = 2846 N/SQ.M. (59.43 LBF/SQ.FT.) EFFECT. VELOC. RATIO = ∞

X/C	ETA = .250 CP		ETA = .387 CP		ETA = .524 CP		ETA = .800 CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			-1.126		-3.106		-4.569	
.010			.189	.834				
.015					-3.944	.967		
.025			-3.063	.976	-3.450	.965	-3.022	.961
.050			-1.920	.796	-2.186	.853	-2.195	.801
.100			-1.252	.570	-1.484	.582	-1.411	.553
.150			-1.061	.407	-1.254	.409	-1.113	.370
.250	-.519		-.793	.095	-.912	.187	-.900	.233
.400	-.474		-.588	-.137	-.659	.017	-.646	.034
.550	-.328	-.322	-.372	-.171	-.418	-.021	-.409	-.027
.700	-.192	-.333	-.210	-.130	-.230	-.028	-.229	-.017
.850	-.082	-.178	-.073	-.061	-.080	-.015	-.095	-.014
.950	-.017	-.045	.008	.004	.000	.016	-.016	.005

ALPHA = 13.98 DEG Q = 2846 N/SQ.M. (59.43 LBF/SQ.FT.) EFFECT. VELOC. RATIO = ∞

X/C	ETA = .250 CP		ETA = .387 CP		ETA = .524 CP		ETA = .800 CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			.553		.167		-.672	
.010			.219	1.000				
.015					-.507	.929		
.025			-.539	.874	-.543	.810	-.849	.819
.050			-.629	.689	-.526	.644	-.807	.638
.100			-.653	.463	-.596	.462	-.852	.436
.150			-.548	.289	-.526	.313	-.778	.282
.250	-.571		-.619	.016	-.649	.114	-.845	.095
.400	-.643		-.708	-.190	-.723	-.072	-.862	-.047
.550	-.590	-.369	-.662	-.239	-.726	-.136	-.791	-.119
.700	-.542	-.395	-.663	-.196	-.628	-.149	-.750	-.138
.850	-.472	-.228	-.606	-.197	-.540	-.210	-.624	-.178
.950	-.307	-.117	-.434	-.247	-.443	-.282	-.438	-.221

TABLE 8 (A) - CONCLUDED

PRESSURE PROFILE ON WING

FORWARD NOZZLES DEFLECTED 45 DEGREES

ALPHA = 16.02 DEG Q = 2842 N/SQ.M. (59.35 LBF/SQ.FT.) EFFECT. VELOC. RATIO = ∞

	ETA = .250		ETA = .387		ETA = .524		ETA = .800	
X/C	CP		CP		CP		CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			.374		.077		-.711	
.010			.222	.986				
.015					-.476	.951		
.025			-.503	.910	-.501	.843	-.749	.851
.050			-.516	.741	-.440	.683	-.800	.691
.100			-.498	.528	-.475	.490	-.758	.460
.150			-.495	.331	-.493	.331	-.750	.303
.250	-.499		-.554	.032	-.483	.118	-.776	.111
.400	-.595		-.566	-.209	-.558	-.060	-.785	-.041
.550	-.633	-.408	-.698	-.255	-.641	-.135	-.843	-.117
.700	-.657	-.398	-.681	-.238	-.634	-.174	-.790	-.163
.850	-.629	-.241	-.648	-.247	-.582	-.244	-.694	-.237
.950	-.447	-.198	-.580	-.299	-.501	-.326	-.589	-.353

ALPHA = 17.99 DEG Q = 2843 N/SQ.M. (59.37 LBF/SQ.FT.) EFFECT. VELOC. RATIO = ∞

	ETA = .250		ETA = .387		ETA = .524		ETA = .800	
X/C	CP		CP		CP		CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			.238		-.077		-.698	
.010			.222	.942				
.015					-.447	.973		
.025			-.510	.960	-.463	.900	-.673	.878
.050			-.505	.816	-.437	.739	-.673	.715
.100			-.504	.601	-.463	.579	-.645	.494
.150			-.499	.398	-.455	.395	-.628	.350
.250	-.525		-.538	.081	-.476	.166	-.651	.154
.400	-.585		-.621	-.192	-.549	-.024	-.702	-.016
.550	-.627	-.444	-.648	-.237	-.586	-.122	-.735	-.118
.700	-.665	-.417	-.672	-.222	-.600	-.164	-.709	-.182
.850	-.657	-.251	-.638	-.240	-.561	-.245	-.721	-.287
.950	-.598	-.248	-.597	-.323	-.508	-.338	-.691	-.398

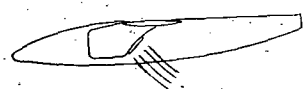
ALPHA = 20.00 DEG Q = 2846 N/SQ.M. (59.45 LBF/SQ.FT.) EFFECT. VELOC. RATIO = ∞

	ETA = .250		ETA = .387		ETA = .524		ETA = .800	
X/C	CP		CP		CP		CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			.094		-.224		-.739	
.010			.210	.893				
.015					-.433	.990		
.025			-.494	.981	-.441	.940	-.607	.900
.050			-.503	.881	-.457	.819	-.598	.758
.100			-.520	.646	-.441	.619	-.618	.553
.150			-.496	.452	-.456	.445	-.633	.400
.250	-.527		-.534	.136	-.471	.224	-.621	.215
.400	-.580		-.600	-.175	-.539	.021	-.651	.021
.550	-.610	-.451	-.613	-.234	-.575	-.089	-.718	-.107
.700	-.687	-.446	-.641	-.219	-.592	-.146	-.719	-.187
.850	-.684	-.255	-.638	-.251	-.564	-.240	-.715	-.307
.950	-.655	-.289	-.601	-.332	-.499	-.346	-.716	-.436

TABLE 8 (B)

PRESSURE PROFILE ON WING

FORWARD NOZZLES DEFLECTED 45 DEGREES



ALPHA = -.01 DEG Q = 2034 N/SQ.M. (42.48 LBF/SQ.FT.) EFFECT. VELOC. RATIO = .296

	ETA = .250		ETA = .387		ETA = .524		ETA = .800	
X/C	CP		CP		CP		CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			-.332		.448		.926	
.010			.445	-1.333				
.015					.094	-1.113		
.025			-.084	-.914	-.094	-1.032	-.081	-.597
.050			-.096	-1.126	-.057	-.755	-.163	-.456
.100			-.194	-1.085	-.190	-.782	-.223	-.433
.150			-.303	-1.178	-.273	-.820	-.266	-.479
.250	-.266		-.333	-1.332	-.338	-.928	-.329	-.497
.400	-.346		-.381	-1.484	-.352	-.915	-.331	-.458
.550	-.283	-1.948	-.286	-1.451	-.287	-.795	-.244	-.366
.700	-.224	-1.945	-.237	-1.164	-.227	-.588	-.151	-.247
.850	-.227	-1.688	-.205	-.749	-.119	-.391	-.061	-.132
.950	-.338	-1.072	-.210	-.412	-.100	-.194	-.010	-.014

ALPHA = 4.00 DEG Q = 2028 N/SQ.M. (42.36 LBF/SQ.FT.) EFFECT. VELOC. RATIO = .295

	ETA = .250		ETA = .387		ETA = .524		ETA = .800	
X/C	CP		CP		CP		CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			.842		.968		.608	
.010			.537	.162				
.015					-.810	.079		
.025			-.777	-.419	-.877	-.127	-1.014	.195
.050			-.626	-.391	-.673	-.207	-.710	-.034
.100			-.500	-.637	-.613	-.379	-.568	-.077
.150			-.548	-.869	-.568	-.515	-.540	-.174
.250	-.373		-.505	-1.156	-.536	-.697	-.537	-.301
.400	-.414		-.466	-1.430	-.480	-.772	-.440	-.338
.550	-.322	-1.958	-.350	-1.399	-.360	-.722	-.303	-.290
.700	-.252	-1.985	-.278	-1.121	-.251	-.576	-.184	-.202
.850	-.233	-1.687	-.234	-.745	-.162	-.361	-.084	-.117
.950	-.300	-.798	-.222	-.384	-.116	-.192	-.013	-.013

ALPHA = 8.04 DEG Q = 2039 N/SQ.M. (42.58 LBF/SQ.FT.) EFFECT. VELOC. RATIO = .295

	ETA = .250		ETA = .387		ETA = .524		ETA = .800	
X/C	CP		CP		CP		CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			.876		.380		-1.026	
.010			.725	.888				
.015					-2.267	.586		
.025			-1.600	.297	-2.065	.439	-2.042	.671
.050			-1.251	.097	-1.264	.231	-1.274	.460
.100			-.855	-.199	-.950	-.032	-.996	.223
.150			-.818	-.581	-.875	-.238	-.876	.092
.250	-.465		-.678	-.951	-.722	-.490	-.766	-.098
.400	-.481		-.571	-1.376	-.627	-.648	-.552	-.192
.550	-.369	-2.035	-.424	-1.369	-.432	-.630	-.382	-.196
.700	-.275	-2.023	-.307	-1.086	-.301	-.537	-.233	-.140
.850	-.220	-1.390	-.235	-.720	-.182	-.354	-.116	-.093
.950	-.253	-.635	-.227	-.383	-.114	-.200	-.026	-.021

TABLE 8 (B) - CONTINUED

PRESSURE PROFILE ON WING

FORWARD NOZZLES DEFLECTED 45 DEGREES

ALPHA = 9.99 DEG Q = 2030 N/SQ.M. (42.40 LBF/SQ.FT.) EFFECT. VELOC. RATIO = .294

	ETA = .250		ETA = .387		ETA = .524		ETA = .800	
X/C	CP		CP		CP		CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			.484		-.969		-2.542	
.010			.802	.980				
.015					-2.821	.875		
.025			-2.309	.554	-2.752	.658	-2.515	.810
.050			-1.471	.297	-1.565	.408	-1.716	.605
.100			-1.011	-.094	-1.214	.127	-1.192	.388
.150			-.960	-.407	-1.042	-.098	-1.008	.173
.250	-.514		-.772	-.887	-.843	-.387	-.833	.023
.400	-.525		-.634	-1.356	-.669	-.569	-.622	-.125
.550	-.394	-2.069	-.455	-1.352	-.475	-.589	-.418	-.152
.700	-.294	-2.096	-.331	-1.077	-.323	-.507	-.255	-.122
.850	-.230	-1.161	-.259	-.705	-.189	-.340	-.129	-.083
.950	-.245	-.530	-.227	-.386	-.121	-.195	-.032	-.021

ALPHA = 12.01 DEG Q = 2036 N/SQ.M. (42.53 LBF/SQ.FT.) EFFECT. VELOC. RATIO = .294

	ETA = .250		ETA = .387		ETA = .524		ETA = .800	
X/C	CP		CP		CP		CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			-.312		-1.922		-4.332	
.010			.758	1.001				
.015					-4.211	.984		
.025			-3.024	.767	-3.336	.860	-3.029	.944
.050			-1.880	.481	-2.084	.592	-2.077	.747
.100			-1.287	.093	-1.494	.276	-1.443	.480
.150			-1.046	-.235	-1.223	.046	-1.135	.298
.250	-.566		-.854	-.746	-.933	-.245	-.944	.109
.400	-.559		-.676	-1.332	-.734	-.484	-.682	-.049
.550	-.413	-2.115	-.481	-1.327	-.505	-.540	-.455	-.118
.700	-.314	-2.071	-.346	-1.061	-.332	-.483	-.281	-.091
.850	-.235	-.939	-.257	-.699	-.196	-.341	-.142	-.070
.950	-.230	-.468	-.229	-.387	-.120	-.197	-.055	-.030

ALPHA = 14.01 DEG Q = 2038 N/SQ.M. (42.57 LBF/SQ.FT.) EFFECT. VELOC. RATIO = .294

	ETA = .250		ETA = .387		ETA = .524		ETA = .800	
X/C	CP		CP		CP		CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			-.542		-2.581		-1.454	
.010			.670	.928				
.015					-4.062	1.001		
.025			-3.016	.885	-3.098	.900	-1.109	.893
.050			-1.979	.610	-1.941	.667	-1.416	.703
.100			-1.333	.189	-1.514	.361	-1.302	.463
.150			-1.146	-.145	-1.296	.110	-1.406	.294
.250	-.593		-.879	-.721	-.992	-.212	-1.233	.102
.400	-.568		-.672	-1.319	-.743	-.489	-.928	-.060
.550	-.421	-2.173	-.519	-1.374	-.527	-.553	-.643	-.118
.700	-.314	-2.168	-.360	-1.097	-.380	-.497	-.433	-.122
.850	-.233	-.896	-.258	-.730	-.318	-.401	-.307	-.126
.950	-.222	-.407	-.225	-.410	-.232	-.317	-.360	-.139

TABLE 8 (B) - CONCLUDED
PRESSURE PROFILE ON WING

FORWARD NOZZLES DEFLECTED 45 DEGREES

ALPHA = 15.99 DEG Q = 2039 N/SQ.M. (42.59 LBF/SQ.FT.) EFFECT. VELOC. RATIO = .297

X/C	ETA = .250 CP		ETA = .387 CP		ETA = .524 CP		ETA = .800 CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			.135		-.383		-1.063	
.010			.684	.994				
.015					-1.385	.963		
.025			-1.253	.831	-1.036	.836	-1.049	.877
.050			-1.531	.576	-1.021	.613	-.964	.710
.100			-1.287	.192	-1.094	.326	-1.021	.488
.150			-1.391	-.158	-1.139	.091	-1.014	.303
.250	-.694		-1.103	-.739	-1.029	-.219	-1.090	.110
.400	-.721		-.867	-1.338	-.868	-.502	-1.026	-.056
.550	-.523	-2.256	-.669	-1.388	-.805	-.570	-.942	-.134
.700	-.394	-2.123	-.640	-1.158	-.673	-.554	-.748	-.154
.850	-.379	-1.107	-.492	-.824	-.540	-.487	-.641	-.200
.950	-.268	-.493	-.370	-.571	-.458	-.403	-.444	-.247

ALPHA = 18.04 DEG Q = 2036 N/SQ.M. (42.53 LBF/SQ.FT.) EFFECT. VELOC. RATIO = .297

X/C	ETA = .250 CP		ETA = .387 CP		ETA = .524 CP		ETA = .800 CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			.389		-.082		-1.216	
.010			.688	1.001				
.015					-.763	.965		
.025			-.795	.855	-.700	.834	-.946	.917
.050			-.605	.603	-.694	.616	-.912	.748
.100			-.752	.206	-.726	.343	-.929	.522
.150			-.717	-.150	-.785	.096	-.938	.359
.250	-.684		-.852	-.743	-.734	-.200	-.931	.154
.400	-.734		-.948	-1.392	-.783	-.503	-1.010	-.042
.550	-.761	-2.309	-.838	-1.457	-.831	-.613	-.962	-.145
.700	-.725	-2.305	-.750	-1.231	-.762	-.616	-.902	-.188
.850	-.540	-1.405	-.673	-.958	-.651	-.535	-.763	-.247
.950	-.401	-.506	-.572	-.602	-.539	-.524	-.644	-.356

ALPHA = 20.01 DEG Q = 2036 N/SQ.M. (42.52 LBF/SQ.FT.) EFFECT. VELOC. RATIO = .297

X/C	ETA = .250 CP		ETA = .387 CP		ETA = .524 CP		ETA = .800 CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			.317		-.141		-1.199	
.010			.683	.983				
.015					-.630	.977		
.025			-.592	.864	-.599	.856	-.876	.926
.050			-.630	.640	-.570	.665	-.871	.791
.100			-.593	.255	-.599	.373	-.836	.562
.150			-.651	-.116	-.648	.136	-.883	.405
.250	-.609		-.614	-.728	-.604	-.189	-.841	.186
.400	-.667		-.744	-1.389	-.729	-.487	-.914	-.019
.550	-.758	-2.286	-.736	-1.521	-.765	-.613	-.994	-.122
.700	-.791	-2.462	-.773	-1.357	-.769	-.640	-.938	-.193
.350	-.770	-1.486	-.717	-1.011	-.703	-.622	-.868	-.294
.950	-.679	-.758	-.614	-.748	-.628	-.594	-.785	-.417

TABLE 8 (C)

PRESSURE PROFILE ON WING

FORWARD NOZZLES DEFLECTED 45 DEGREES

ALPHA = -.01 DEG Q = 912 N/SQ.M. (19.05 LBF/SQ.FT.) EFFECT. VELOC. RATIO = .198

	ETA = .250		ETA = .387		ETA = .524		ETA = .800	
X/C	CP		CP		CP		CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			-.729		-.087		.927	
.010			.266	-2.009				
.015					.468	-1.807		
.025			.191	-1.874	.264	-1.550	-.025	-.667
.050			-.035	-1.487	.041	-1.123	-.056	-.582
.100			-.113	-1.505	-.099	-1.006	-.085	-.501
.150			-.218	-1.667	-.227	-1.004	-.254	-.534
.250	-.221		-.275	-1.809	-.251	-1.133	-.289	-.570
.400	-.285		-.301	-2.140	-.299	-1.134	-.288	-.532
.550	-.228	-3.065	-.259	-1.883	-.242	-.958	-.219	-.398
.700	-.182	-2.615	-.209	-1.386	-.200	-.727	-.171	-.285
.850	-.143	-.855	-.197	-.810	-.143	-.470	-.062	-.149
.950	-.172	-.377	-.208	-.445	-.105	-.251	-.012	-.027

ALPHA = 4.03 DEG Q = 906 N/SQ.M. (18.92 LBF/SQ.FT.) EFFECT. VELOC. RATIO = .197

	ETA = .250		ETA = .387		ETA = .524		ETA = .800	
X/C	CP		CP		CP		CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			.699		.988		.764	
.010			.294	-.306				
.015					-.375	-.316		
.025			-.468	-.767	-.579	-.455	-.671	.106
.050			-.506	-.711	-.565	-.508	-.642	-.028
.100			-.443	-.979	-.462	-.563	-.488	-.155
.150			-.475	-1.290	-.495	-.762	-.483	-.261
.250	-.315		-.438	-1.590	-.447	-.879	-.457	-.347
.400	-.390		-.426	-2.022	-.458	-.992	-.401	-.390
.550	-.298	-3.074	-.341	-1.857	-.344	-.888	-.302	-.335
.700	-.216	-2.588	-.244	-1.349	-.260	-.725	-.176	-.248
.850	-.171	-.699	-.217	-.777	-.155	-.448	-.086	-.138
.950	-.177	-.288	-.229	-.442	-.139	-.245	-.022	-.027

ALPHA = 8.03 DEG Q = 908 N/SQ.M. (18.97 LBF/SQ.FT.) EFFECT. VELOC. RATIO = .197

	ETA = .250		ETA = .387		ETA = .524		ETA = .800	
X/C	CP		CP		CP		CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			.986		.712		-.828	
.010			.355	.674				
.015					-1.633	.554		
.025			-1.571	.069	-1.626	.154	-1.708	.651
.050			-1.071	-.136	-1.192	.053	-1.279	.337
.100			-.762	-.554	-.835	-.192	-.927	.190
.150			-.740	-.892	-.776	-.431	-.800	.016
.250	-.436		-.643	-1.367	-.683	-.681	-.689	-.166
.400	-.456		-.544	-1.925	-.581	-.872	-.539	-.243
.550	-.354	-3.088	-.410	-1.790	-.433	-.824	-.377	-.258
.700	-.258	-2.382	-.301	-1.315	-.307	-.694	-.230	-.207
.850	-.200	-.402	-.264	-.771	-.210	-.460	-.122	-.132
.950	-.174	-.217	-.242	-.433	-.166	-.270	-.040	-.055

TABLE 8 (C) - CONTINUED

PRESSURE PROFILE ON WING

FORWARD NOZZLES DEFLECTED 45 DEGREES

ALPHA = 9.98 DEG Q = 909 N/SQ.M. (18.99 LBF/SQ.FT.) EFFECT. VELOC. RATIO = .197

	ETA = .250		ETA = .387		ETA = .524		ETA = .800	
X/C	CP		CP		CP		CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			.823		-.057		-1.948	
.010			.393	.849				
.015					-2.270	.750		
.025			-1.960	.341	-2.137	.487	-2.813	.747
.050			-1.334	.116	-1.494	.196	-1.477	.532
.100			-.922	-.348	-1.064	-.016	-1.100	.281
.150			-.896	-.691	-.977	-.298	-1.016	.113
.250	-.485		-.738	-1.275	-.795	-.546	-.806	-.077
.400	-.511		-.599	-1.850	-.646	-.797	-.585	-.204
.550	-.379	-3.061	-.448	-1.744	-.475	-.778	-.422	-.197
.700	-.294	-2.363	-.334	-1.288	-.332	-.673	-.262	-.180
.850	-.217	-.331	-.270	-.763	-.220	-.469	-.143	-.125
.950	-.192	-.185	-.259	-.452	-.165	-.274	-.066	-.059

ALPHA = 12.03 DEG Q = 910 N/SQ.M. (19.00 LBF/SQ.FT.) EFFECT. VELOC. RATIO = .197

	ETA = .250		ETA = .387		ETA = .524		ETA = .800	
X/C	CP		CP		CP		CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			.313		-.766		-4.272	
.010			.381	.992				
.015					-3.541	.936		
.025			-2.480	.661	-3.124	.658	-2.698	.902
.050			-1.564	.284	-1.765	.481	-1.844	.692
.100			-1.136	-.135	-1.349	.185	-1.351	.417
.150			-1.018	-.534	-1.167	-.115	-1.121	.213
.250	-.539		-.830	-1.157	-.883	-.474	-.890	.011
.400	-.535		-.653	-1.813	-.724	-.718	-.671	-.125
.550	-.413	-3.027	-.484	-1.714	-.514	-.724	-.457	-.170
.700	-.305	-2.223	-.356	-1.280	-.355	-.653	-.281	-.158
.850	-.230	-.304	-.286	-.761	-.238	-.477	-.154	-.113
.950	-.185	-.214	-.271	-.451	-.174	-.282	-.070	-.065

ALPHA = 14.02 DEG Q = 913 N/SQ.M. (19.07 LBF/SQ.FT.) EFFECT. VELOC. RATIO = .198

	ETA = .250		ETA = .387		ETA = .524		ETA = .800	
X/C	CP		CP		CP		CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			-.218		-2.459		-5.761	
.010			.357	1.000				
.015					-4.239	1.000		
.025			-3.431	.800	-3.512	.855	-3.164	.993
.050			-1.887	.472	-2.056	.586	-2.202	.805
.100			-1.277	.058	-1.550	.245	-1.555	.529
.150			-1.181	-.366	-1.276	-.044	-1.264	.365
.250	-.584		-.886	-1.051	-1.012	-.401	-1.003	.113
.400	-.570		-.690	-1.742	-.779	-.665	-.707	-.056
.550	-.427	-3.031	-.496	-1.689	-.539	-.714	-.483	-.125
.700	-.313	-2.009	-.367	-1.263	-.370	-.645	-.305	-.121
.850	-.232	-.210	-.292	-.747	-.242	-.488	-.175	-.108
.950	-.201	-.201	-.258	-.447	-.181	-.297	-.089	-.076

TABLE 8 (C) - CONCLUDED
PRESSURE PROFILE ON WING

FORWARD NOZZLES DEFLECTED 45 DEGREES

ALPHA = 16.01 DEG Q = 909 N/SQ.M. (18.98 LBF/SQ.FT.) EFFECT. VELOC. RATIO = .197

	ETA = .250		ETA = .387		ETA = .524		ETA = .800	
X/C	CP		CP		CP		CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			-1.012		-1.409		-1.222	
.010			.283	.975				
.015					-3.772	1.0		
.025			-2.073	.907	-2.539	.900	-1.055	.925
.050			-2.023	.613	-1.874	.632	-1.166	.743
.100			-1.402	.138	-1.575	.306	-1.055	.505
.150			-1.210	-.317	-1.279	.015	-1.360	.320
.250	-.591		-.903	-1.003	-1.069	-.345	-1.265	.107
.400	-.594		-.711	-1.793	-.861	-.657	-1.069	-.085
.550	-.429	-3.063	-.511	-1.720	-.701	-.749	-.875	-.141
.700	-.329	-2.010	-.500	-1.331	-.542	-.676	-.458	-.159
.850	-.249	-.252	-.416	-.806	-.401	-.546	-.605	-.161
.950	-.210	-.180	-.312	-.475	-.304	-.410	-.483	-.224

ALPHA = 18.00 DEG Q = 909 N/SQ.M. (18.98 LBF/SQ.FT.) EFFECT. VELOC. RATIO = .197

	ETA = .250		ETA = .387		ETA = .524		ETA = .800	
X/C	CP		CP		CP		CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			.284		-.434		-1.138	
.010			.298	1.0				
.015					-1.190	.998		
.025			-1.311	.846	-1.170	.832	-.947	.940
.050			-1.327	.577	-.907	.600	-.968	.766
.100			-1.377	.106	-1.086	.281	-.917	.515
.150			-1.401	-.328	-1.162	-.000	-.921	.357
.250	-.772		-1.034	-1.046	-1.068	-.339	-1.047	.123
.400	-.672		-.954	-1.797	-.986	-.678	-1.088	-.074
.550	-.535	-3.166	-.817	-1.810	-.847	-.764	-1.019	-.165
.700	-.432	-2.193	-.634	-1.406	-.732	-.728	-.901	-.188
.850	-.363	-.315	-.513	-.882	-.566	-.627	-.790	-.266
.950	-.260	-.217	-.427	-.557	-.481	-.500	-.641	-.344

ALPHA = 19.39 DEG Q = 910 N/SQ.M. (19.01 LBF/SQ.FT.) EFFECT. VELOC. RATIO = .197

	ETA = .250		ETA = .387		ETA = .524		ETA = .800	
X/C	CP		CP		CP		CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			.273		-.151		-1.148	
.010			.304	1.0				
.015					-.847	.981		
.025			-1.010	.835	-.888	.841	-.923	.944
.050			-.920	.572	-.781	.612	-.902	.787
.100			-1.023	.086	-.828	.320	-.951	.545
.150			-.961	-.326	-.954	.010	-.918	.368
.250	-.814		-1.015	-1.032	-.882	-.332	-1.008	.135
.400	-.841		-.950	-1.831	-.876	-.688	-.989	-.066
.550	-.735	-3.258	-.908	-1.836	-.866	-.786	-1.061	-.172
.700	-.665	-2.176	-.811	-1.470	-.794	-.757	-.978	-.215
.850	-.477	-.309	-.628	-.948	-.668	-.683	-.853	-.274
.950	-.323	-.252	-.521	-.628	-.503	-.542	-.797	-.415

TABLE 8 (D)

PRESSURE PROFILE ON WING

FORWARD NOZZLES DEFLECTED 45 DEGREES

ALPHA = .02 DEG Q = 225 N/SQ.M. (4.69 LBF/SQ.FT.) EFFECT. VELOC. RATIO = .098

X/C	ETA = .250 CP		ETA = .387 CP		ETA = .524 CP		ETA = .800 CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			-4.059		-2.104		.609	
.010			-1.908	-4.620				
.015					1.0	-3.134		
.025			.777	-3.316	.819	-2.714	.546	-.856
.050			.525	-2.616	.546	-1.830	.405	-.800
.100			.286	-2.468	.342	-1.564	.265	-.562
.150			.167	-2.735	.216	-1.522	.125	-.597
.250	.034		.034	-3.075	.090	-1.488	.027	-.575
.400	-.004		-.024	-3.530	-.017	-1.383	-.031	-.469
.550	.034	-2.491	.003	-2.232	-.004	-1.077	.013	-.344
.700	.057	-.520	.027	-1.186	.017	-.741	.071	-.218
.850	.091	-.092	.006	-.534	.040	-.391	.115	-.038
.950	.078	-.007	-.045	-.276	-.007	-.157	.135	.084

ALPHA = 4.03 DEG Q = 226 N/SQ.M. (4.72 LBF/SQ.FT.) EFFECT. VELOC. RATIO = .098

X/C	ETA = .250 CP		ETA = .387 CP		ETA = .524 CP		ETA = .800 CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			-1.630		.239		1.103	
.010			-1.734	-2.230				
.015					.511	-1.623		
.025			.288	-2.167	.329	-1.644	-.159	-.228
.050			.037	-1.916	.162	-1.149	-.131	-.221
.100			.051	-2.083	.085	-1.135	-.082	-.270
.150			-.054	-2.369	-.082	-1.198	-.145	-.361
.250	-.054		-.082	-2.985	-.032	-1.288	-.166	-.345
.400	-.078		-.112	-3.498	-.119	-1.271	-.139	-.352
.550	-.038	-2.248	-.041	-2.160	-.078	-1.014	-.044	-.251
.700	.027	-.298	-.014	-1.156	-.031	-.720	.003	-.166
.850	.081	-.034	-.014	-.501	.020	-.372	.081	-.055
.950	.101	.077	-.034	-.217	.020	-.129	.138	.114

ALPHA = 8.03 DEG Q = 224 N/SQ.M. (4.69 LBF/SQ.FT.) EFFECT. VELOC. RATIO = .098

X/C	ETA = .250 CP		ETA = .387 CP		ETA = .524 CP		ETA = .800 CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			.692		1.0		.748	
.010			-1.477	-.684				
.015					-.480	-.305		
.025			-.326	-.916	-.579	-.396	-.740	.362
.050			-.298	-1.077	-.298	-.312	-.593	.187
.100			-.228	-1.421	-.312	-.586	-.480	.130
.150			-.298	-1.969	-.326	-.853	-.340	-.045
.250	-.122		-.270	-2.721	-.291	-1.077	-.368	-.165
.400	-.148		-.230	-3.272	-.264	-1.135	-.284	-.216
.550	-.084	-2.081	-.148	-2.160	-.141	-.928	-.141	-.175
.700	-.002	-.284	-.080	-1.118	-.084	-.720	-.019	-.101
.850	.039	-.087	-.036	-.519	.008	-.376	.076	-.026
.950	.080	.070	-.043	-.220	.001	-.121	.117	.046

TABLE 8 (D) - CONTINUED

PRESSURE PROFILE ON WING

FORWARD NOZZLES DEFLECTED 45 DEGREES

ALPHA = 9.87 DEG Q = 228 N/SQ.M. (4.75 LBF/SQ.FT.) EFFECT. VELOC. RATIO = .099

X/C	ETA = .250 CP		ETA = .387 CP		ETA = .524 CP		ETA = .800 CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000							-.211	
.010			-1.222	-.017				
.015					-.778	.184		
.025			-.744	-.501	-.820	-.017	-1.305	.648
.050			-.550	-.702	-.633	-.176	-.827	.413
.100			-.363	-1.201	-.481	-.488	-.612	.233
.150			-.418	-1.672	-.453	-.723	-.522	.053
.250	-.183		-.356	-2.552	-.418	-.948	-.460	-.122
.400	-.203		-.280	-3.200	-.314	-1.059	-.330	-.179
.550	-.122	-2.039	-.193	-2.129	-.223	-.934	-.176	-.176
.700	-.042	-.196	-.079	-1.116	-.102	-.730	-.062	-.095
.850	.032	-.059	-.048	-.538	-.022	-.411	.032	-.012
.950	.059	.052	-.059	-.236	-.025	-.146	.079	.032

ALPHA = 11.97 DEG Q = 225 N/SQ.M. (4.69 LBF/SQ.FT.) EFFECT. VELOC. RATIO = .098

X/C	ETA = .250 CP		ETA = .387 CP		ETA = .524 CP		ETA = .800 CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000					.824		-1.321	
.010			-1.069	.571				
.015					-1.405	.403		
.025			-1.048	-.172	-1.195	.221	-1.924	.852
.050			-.802	-.403	-.964	.053	-1.307	.648
.100			-.522	-.964	-.725	-.382	-.760	.340
.150			-.473	-1.447	-.620	-.564	-.718	.158
.250	-.228		-.445	-2.354	-.508	-.835	-.585	-.006
.400	-.268		-.441	-3.118	-.397	-.995	-.427	-.145
.550	-.159	-2.078	-.223	-2.034	-.264	-.896	-.217	-.149
.700	-.064	-.230	-.074	-1.100	-.145	-.702	-.098	-.098
.850	.008	-.030	-.020	-.539	-.060	-.427	-.006	-.013
.950	.025	.042	-.101	-.257	-.040	-.176	.045	.055

ALPHA = 13.99 DEG Q = 225 N/SQ.M. (4.70 LBF/SQ.FT.) EFFECT. VELOC. RATIO = .098

X/C	ETA = .250 CP		ETA = .387 CP		ETA = .524 CP		ETA = .800 CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000					.280		-2.596	
.010			-.889	.945				
.015					-1.959	.763		
.025			-1.490	.238	-1.882	.420	-2.813	.980
.050			-1.315	-.259	-1.553	.273	-1.476	.721
.100			-.707	-.693	-.833	-.119	-1.036	.448
.150			-.630	-1.224	-.805	-.441	-.868	.273
.250	-.287		-.532	-2.235	-.602	-.736	-.700	.034
.400	-.278		-.420	-3.103	-.468	-.970	-.488	-.112
.550	-.166	-2.092	-.264	-2.079	-.308	-.871	-.285	-.125
.700	-.091	-.196	-.152	-1.132	-.196	-.691	-.166	-.105
.850	-.135	-.051	-.112	-.556	-.081	-.447	-.051	-.037
.950	.033	.055	-.122	-.274	-.064	-.200	.011	.034

TABLE 8 (D) - CONCLUDED

PRESSURE PROFILE ON WING

FORWARD NOZZLES DEFLECTED 45 DEGREES

ALPHA = 16.00 DEG Q = 227 N/SQ.M. (4.74 LBF/SQ.FT.) EFFECT. VELOC. RATIO = .098

	ETA = .250		ETA = .387		ETA = .524		ETA = .800	
X/C	CP		CP		CP		CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			.856		-.096		-4.077	
.010			-.679					
.015					-2.785			
.025			-2.173	.453	-2.576	.780	-2.882	
.050			-1.506	.127	-1.492	.509	-1.729	.842
.100			-.825	-.408	-1.089	.036	-1.263	.648
.150			-.784	-1.006	-.916	-.242	-1.013	.384
.250	-.304		-.589	-2.106	-.686	-.611	-.756	.140
.400	-.321		-.439	-2.935	-.520	-.880	-.510	-.001
.550	-.203	-1.951	-.254	-2.022	-.314	-.826	-.291	-.075
.700	-.096	-.186	-.156	-1.096	-.183	-.671	-.139	-.065
.850	-.005	.029	-.106	-.540	-.082	-.422	-.025	-.018
.950	.039	.076	-.099	-.240	-.065	-.173	.036	.039

ALPHA = 18.04 DEG Q = 227 N/SQ.M. (4.74 LBF/SQ.FT.) EFFECT. VELOC. RATIO = .099

	ETA = .250		ETA = .387		ETA = .524		ETA = .800	
X/C	CP		CP		CP		CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			.268		-1.341		-1.279	
.010			-.523					
.015					-3.228			
.025			-2.694	.622	-3.152	.892	-1.390	
.050			-1.501	.351	-1.716	.622	-1.105	.872
.100			-1.022	-.294	-1.223	.199	-1.133	.587
.150			-.863	-.911	-.953	-.134	-1.189	.421
.250	-.342		-.620	-1.934	-.717	-.515	-1.084	.218
.400	-.287		-.438	-2.929	-.522	-.801	-.673	.036
.550	-.179	-1.954	-.263	-1.964	-.297	-.791	-.404	-.038
.700	-.061	-.162	-.139	-1.097	-.236	-.636	-.303	-.055
.850	.009	.130	-.085	-.522	-.061	-.424	-.162	-.048
.950	.060	.103	-.068	-.219	-.058	-.202	-.018	-.018

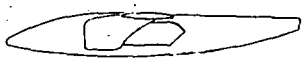
ALPHA = 20.02 DEG Q = 231 N/SQ.M. (4.83 LBF/SQ.FT.) EFFECT. VELOC. RATIO = .099

	ETA = .250		ETA = .387		ETA = .524		ETA = .800	
X/C	CP		CP		CP		CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			-.016		-1.126		-.990	
.010			-.370					
.015					-3.222			
.025			-3.304	.875	-2.283	.957	-.833	
.050			-1.589	.508	-1.371	.644	-.813	.909
.100			-.990	-.105	-1.146	.242	-.888	.664
.150			-.867	-.731	-.983	-.064	-1.146	.453
.250	-.336		-.609	-1.824	-.786	-.468	-.881	.225
.400	-.320		-.448	-2.886	-.699	-.801	-.890	.047
.550	-.194	-1.788	-.283	-2.002	-.478	-.821	-.762	-.029
.700	-.115	-.135	-.221	-1.131	-.353	-.663	-.636	-.066
.850	.017	.027	-.142	-.544	-.270	-.491	-.458	-.069
.950	.079	.129	-.161	-.241	-.198	-.277	-.349	-.043

TABLE 9 (A)

PRESSURE PROFILE ON WING

REAR NOZZLES DEFLECTED 45 DEGREES



ALPHA = .01 DEG Q = 2844 N/SQ.M. (59.40 LBF/SQ.FT.) EFFECT. VELOC. RATIO = ∞

	ETA = .250		ETA = .387		ETA = .524		ETA = .800	
X/C	CP		CP		CP		CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			.736		.883		.965	
.010			.380	-.441				
.015					-.074	-.357		
.025			-.216	-.554	-.267	-.403	-.326	-.344
.050			-.250	-.445	-.208	-.327	-.223	-.233
.100			-.260	-.428	-.235	-.279	-.159	-.279
.150			-.328	-.486	-.281	-.398	-.253	-.305
.250	-.264		-.331	-.439	-.361	-.402	-.316	-.366
.400	-.329		-.350	-.361	-.346	-.361	-.315	-.361
.550	-.256	-.278	-.278	-.270	-.254	-.277	-.238	-.264
.700	-.189	-.199	-.176	-.212	-.168	-.154	-.136	-.143
.850	-.136	-.278	-.094	-.203	-.041	-.076	-.020	-.046
.950	-.109	-.106	-.043	-.079	.015	.015	.034	.050

ALPHA = 3.98 DEG Q = 2840 N/SC.M. (59.32 LBF/SQ.FT.) EFFECT. VELOC. RATIO = ∞

	ETA = .250		ETA = .387		ETA = .524		ETA = .800	
X/C	CP		CP		CP		CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			.980		.790		.298	
.010			.420	.743				
.015					-1.141	.488		
.025			-1.036	.213	-1.261	.319	-1.014	.361
.050			-.829	.254	-.845	.161	-.879	.175
.100			-.588	-.002	-.649	.042	-.578	.068
.150			-.591	-.187	-.569	-.068	-.529	-.046
.250	-.364		-.513	-.229	-.543	-.166	-.537	-.140
.400	-.405		-.454	-.224	-.472	-.206	-.414	-.212
.550	-.301	-.175	-.305	-.176	-.311	-.182	-.284	-.175
.700	-.212	-.181	-.198	-.127	-.193	-.117	-.156	-.108
.850	-.138	-.389	-.110	-.191	-.063	-.067	-.050	-.048
.950	-.086	-.163	-.034	-.090	.018	.015	.036	.046

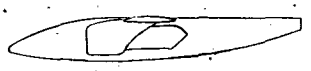
ALPHA = 8.02 DEG Q = 2841 N/SQ.M. (59.34 LBF/SQ.FT.) EFFECT. VELOC. RATIO = ∞

	ETA = .250		ETA = .387		ETA = .524		ETA = .800	
X/C	CP		CP		CP		CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			.401		-.691		-2.021	
.010			.407	.999				
.015					-2.538	.942		
.025			-2.382	.749	-2.513	.768	-2.395	.762
.050			-1.343	.551	-1.525	.583	-1.381	.533
.100			-.939	.335	-1.093	.349	-1.019	.338
.150			-.873	-.150	-.913	.195	-.821	.166
.250	-.458		-.692	-.003	-.740	.073	-.743	.039
.400	-.445		-.557	-.078	-.592	-.051	-.548	-.068
.550	-.327	-.102	-.379	-.061	-.391	-.073	-.362	-.095
.700	-.226	-.134	-.239	-.062	-.238	-.054	-.207	-.051
.850	-.136	-.435	-.121	-.173	-.092	-.021	-.086	-.016
.950	-.079	-.276	-.043	-.095	-.006	.019	-.003	.015

TABLE 9 (A) - CONTINUED

PRESSURE PROFILE ON WING

REAR NOZZLES DEFLECTED 45 DEGREES



ALPHA = 10.06 DEG Q = 2846 N/SQ.M. (59.45 LBF/SQ.FT.) EFFECT. VELOC. RATIO = ∞

X/C	ETA = .250 CP		ETA = .387 CP		ETA = .524 CP		ETA = .800 CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			-.365		-1.998		-3.719	
.010			.324	.919				
.015					-4.019	.977		
.025			-2.731	.895	-2.752	.903	-2.490	.904
.050			-1.723	.718	-1.805	.722	-1.807	.717
.100			-1.180	.471	-1.286	.524	-1.228	.452
.150			-.965	.314	-1.149	.326	-1.062	.294
.250	-.509		-.791	.132	-.846	.138	-.835	.136
.400	-.485		-.603	-.041	-.641	.014	-.600	-.010
.550	-.356	-.059	-.406	.005	-.422	-.002	-.404	-.032
.700	-.238	-.127	-.250	-.020	-.248	-.021	-.225	-.030
.850	-.144	-.426	-.130	-.150	-.098	-.014	-.096	-.009
.950	-.096	-.289	-.050	-.089	-.009	.019	-.007	.022

ALPHA = 12.02 DEG Q = 2847 N/SQ.M. (59.46 LBF/SQ.FT.) EFFECT. VELOC. RATIO = ∞

X/C	ETA = .250 CP		ETA = .387 CP		ETA = .524 CP		ETA = .800 CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			.378		-.191		-.836	
.010			.339	.994				
.015					-1.181	.937		
.025			-.947	.879	-.922	.841	-1.003	.825
.050			-.900	.717	-.762	.679	-1.087	.636
.100			-1.023	.508	-.969	.485	-1.143	.425
.150			-.755	.336	-1.077	.313	-1.168	.271
.250	-.646		-.944	.120	-1.002	.142	-1.020	.098
.400	-.549		-.839	-.028	-.857	-.001	-.888	-.030
.550	-.469	-.073	-.672	-.038	-.689	-.047	-.576	-.082
.700	-.309	-.135	-.366	-.080	-.500	-.076	-.419	-.096
.850	-.267	-.464	-.369	-.243	-.289	-.149	-.391	-.087
.950	-.155	-.322	-.285	-.207	-.300	-.136	-.283	-.127

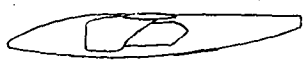
ALPHA = 14.01 DEG Q = 2843 N/SQ.M. (59.38 LBF/SQ.FT.) EFFECT. VELOC. RATIO = ∞

X/C	ETA = .250 CP		ETA = .387 CP		ETA = .524 CP		ETA = .800 CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			.487		.049		-.803	
.010			.360	.999				
.015					-.622	.944		
.025			-.518	.882	-.537	.830	-.895	.830
.050			-.527	.734	-.630	.686	-.856	.642
.100			-.578	.525	-.519	.487	-.900	.432
.150			-.499	.355	-.557	.325	-.916	.303
.250	-.561		-.575	.130	-.670	.159	-.852	.115
.400	-.611		-.658	-.055	-.645	-.002	-.916	-.038
.550	-.613	-.096	-.700	-.063	-.739	-.081	-.849	-.103
.700	-.613	-.173	-.672	-.132	-.668	-.128	-.747	-.143
.850	-.588	-.521	-.618	-.321	-.566	-.219	-.560	-.188
.950	-.438	-.450	-.502	-.352	-.516	-.312	-.453	-.232

TABLE 9 (A) - CONCLUDED

PRESSURE PROFILE ON WING

REAR NOZZLES DEFLECTED 45 DEGREES



ALPHA = 15.99 DEG Q = 2845 N/SQ.M. (59.42 LBF/SQ.FT.) EFFECT. VELOC. RATIO = ∞

	ETA = .250		ETA = .387		ETA = .524		ETA = .800	
X/C	CP		CP		CP		CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			.353		-.025		-.915	
.010			.362	.979				
.015					-.496	.965		
.025			-.495	.920	-.496	.866	-.804	.858
.050			-.469	.756	-.485	.747	-.828	.719
.100			-.480	.585	-.488	.538	-.804	.507
.150			-.481	.388	-.472	.403	-.789	.331
.250	-.485		-.512	.175	-.564	.205	-.830	.154
.400	-.566		-.601	-.006	-.612	.012	-.845	-.024
.550	-.626	-.105	-.620	-.049	-.620	-.069	-.858	-.103
.700	-.656	-.205	-.648	-.134	-.665	-.141	-.798	-.152
.850	-.660	-.567	-.618	-.368	-.621	-.256	-.682	-.222
.950	-.599	-.507	-.553	-.411	-.552	-.360	-.598	-.297

ALPHA = 18.00 DEG Q = 2844 N/SQ.M. (59.40 LBF/SQ.FT.) EFFECT. VELOC. RATIO = ∞

	ETA = .250		ETA = .387		ETA = .524		ETA = .800	
X/C	CP		CP		CP		CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			.118		-.154		-.921	
.010			.334	.950				
.015					-.463	.975		
.025			-.503	.964	-.495	.913	-.756	.886
.050			-.501	.842	-.475	.778	-.734	.740
.100			-.485	.641	-.477	.586	-.745	.539
.150			-.498	.467	-.487	.450	-.728	.388
.250	-.509		-.528	.219	-.511	.246	-.757	.191
.400	-.588		-.555	.018	-.575	.049	-.767	-.003
.550	-.605	-.103	-.621	-.032	-.617	-.064	-.829	-.104
.700	-.691	-.236	-.656	-.146	-.628	-.135	-.829	-.172
.850	-.699	-.621	-.635	-.398	-.602	-.262	-.787	-.281
.950	-.655	-.554	-.590	-.456	-.557	-.387	-.716	-.379

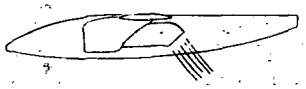
ALPHA = 20.03 DEG Q = 2843 N/SQ.M. (59.38 LBF/SQ.FT.) EFFECT. VELOC. RATIO = ∞

	ETA = .250		ETA = .387		ETA = .524		ETA = .800	
X/C	CP		CP		CP		CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			-.022		-.299		-.897	
.010			.314	.839				
.015					-.483	.979		
.025			-.457	.977	-.492	.949	-.699	.906
.050			-.491	.906	-.488	.821	-.687	.777
.100			-.495	.713	-.485	.664	-.705	.590
.150			-.512	.561	-.498	.507	-.691	.423
.250	-.526		-.506	.305	-.511	.314	-.731	.222
.400	-.628		-.576	.062	-.579	.086	-.746	.033
.550	-.632	-.076	-.619	.001	-.609	-.036	-.794	-.097
.700	-.681	-.234	-.654	-.123	-.644	-.120	-.785	-.177
.850	-.719	-.659	-.629	-.404	-.627	-.257	-.779	-.306
.950	-.706	-.584	-.599	-.473	-.558	-.381	-.761	-.444

TABLE 9 (B)

PRESSURE PROFILE ON WING

REAR NOZZLES DEFLECTED 45 DEGREES



ALPHA = .00 DEG Q = 2032 N/SQ.M. (42.44 LBF/SQ.FT.) EFFECT. VELOC. RATIO = .291

X/C	ETA = .250 CP		ETA = .387 CP		ETA = .524 CP		ETA = .800 CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			.593		.938		.972	
.010			-.036	-.029				
.015					-.449	-.290		
.025			-.401	-.360	-.412	-.284	-.347	-.275
.050			-.365	-.409	-.327	-.314	-.304	-.146
.100			-.388	-.437	-.350	-.198	-.362	-.258
.150			-.394	-.474	-.371	-.366	-.335	-.301
.250	-.316		-.417	-.464	-.425	-.406	-.383	-.357
.400	-.403		-.431	-.438	-.418	-.392	-.357	-.323
.550	-.369	-.282	-.348	-.353	-.335	-.340	-.297	-.272
.700	-.303	-.315	-.266	-.359	-.226	-.262	-.156	-.185
.850	-.280	-.822	-.222	-.479	-.144	-.196	-.064	-.092
.950	-.295	-.489	-.196	-.317	-.088	-.103	-.003	.008

ALPHA = 4.01 DEG Q = 2038 N/SQ.M. (42.57 LBF/SQ.FT.) EFFECT. VELOC. RATIO = .292

X/C	ETA = .250 CP		ETA = .387 CP		ETA = .524 CP		ETA = .800 CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			.953		.524		.073	
.010			.015	.858				
.015					-1.539	.538		
.025			-1.262	.367	-1.455	.381	-1.323	.439
.050			-1.095	.216	-1.049	.267	-.900	.260
.100			-.759	.039	-.763	.093	-.671	.110
.150			-.665	-.175	-.699	-.052	-.599	-.042
.250	-.453		-.611	-.215	-.674	-.156	-.586	-.169
.400	-.511		-.556	-.320	-.570	-.237	-.488	-.216
.550	-.420	-.234	-.435	-.283	-.413	-.242	-.345	-.206
.700	-.353	-.369	-.339	-.315	-.306	-.206	-.211	-.145
.850	-.317	-.917	-.264	-.498	-.179	-.194	-.104	-.085
.950	-.319	-.592	-.228	-.363	-.109	-.120	-.022	-.010

ALPHA = 7.99 DEG Q = 2039 N/SQ.M. (42.59 LBF/SQ.FT.) EFFECT. VELOC. RATIO = .292

X/C	ETA = .250 CP		ETA = .387 CP		ETA = .524 CP		ETA = .800 CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			.025		-1.685		-2.600	
.010			-.021	.990				
.015					-3.425	.950		
.025			-2.787	.800	-3.082	.852	-2.666	.826
.050			-1.551	.604	-1.705	.636	-1.499	.637
.100			-1.095	.393	-1.259	.402	-1.138	.377
.150			-1.018	.198	-1.060	.254	-.956	.185
.250	-.564		-.818	-.025	-.880	.047	-.819	.023
.400	-.605		-.714	-.143	-.715	-.095	-.627	-.088
.550	-.489	-.201	-.514	-.177	-.517	-.155	-.442	-.118
.700	-.401	-.338	-.397	-.260	-.362	-.170	-.284	-.096
.850	-.354	-.960	-.305	-.490	-.219	-.176	-.148	-.066
.950	-.348	-.712	-.253	-.379	-.141	-.131	-.057	-.023

TABLE 9 (B) - CONTINUED

PRESSURE PROFILE ON WING

REAR NOZZLES DEFLECTED 45 DEGREES

ALPHA = 10.05 DEG Q = 2033 N/SQ.M. (42.46 LBF/SQ.FT.) EFFECT. VELOC. RATIO = .292

X/C	ETA = .250 CP		ETA = .387 CP		ETA = .524 CP		ETA = .800 CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			-.847		-2.894		-4.110	
.010			-.158	.858				
.015					-4.791	.951		
.025			-2.771	.944	-2.953	.939	-2.850	.929
.050			-2.019	.772	-2.119	.782	-1.972	.752
.100			-1.374	.547	-1.520	.544	-1.379	.490
.150			-1.164	.366	-1.301	.359	-1.154	.317
.250	-.627		-.940	.125	-.989	.155	-.954	.121
.400	-.634		-.754	-.082	-.786	-.011	-.711	-.033
.550	-.521	-.187	-.549	-.116	-.556	-.089	-.482	-.083
.700	-.429	-.365	-.417	-.238	-.379	-.139	-.309	-.082
.850	-.373	-.972	-.319	-.485	-.238	-.168	-.173	-.071
.950	-.371	-.756	-.266	-.389	-.149	-.146	-.079	-.048

ALPHA = 11.99 DEG Q = 2036 N/SQ.M. (42.52 LBF/SQ.FT.) EFFECT. VELOC. RATIO = .292

X/C	ETA = .250 CP		ETA = .387 CP		ETA = .524 CP		ETA = .800 CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			.049		-.503		-1.335	
.010			-.201	.943				
.015					-1.187	.963		
.025			-1.346	.926	-1.269	.875	-1.385	.859
.050			-1.133	.777	-1.244	.718	-1.371	.683
.100			-1.068	.541	-1.031	.512	-1.331	.453
.150			-1.183	.359	-1.259	.352	-1.520	.303
.250	-.761		-1.318	.109	-1.098	.151	-1.301	.112
.400	-.790		-1.018	-.089	-1.065	-.043	-1.100	-.039
.550	-.705	-.155	-.803	-.149	-.935	-.133	-.822	-.107
.700	-.491	-.340	-.603	-.287	-.636	-.209	-.607	-.131
.850	-.423	-1.002	-.478	-.549	-.555	-.281	-.415	-.151
.950	-.403	-.795	-.472	-.500	-.398	-.366	-.285	-.167

ALPHA = 13.99 DEG Q = 2037 N/SQ.M. (42.54 LBF/SQ.FT.) EFFECT. VELOC. RATIO = .292

X/C	ETA = .250 CP		ETA = .387 CP		ETA = .524 CP		ETA = .800 CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			.179		-.332		-1.260	
.010			-.165	.961				
.015					-.871	.971		
.025			-.767	.948	-.804	.890	-1.250	.878
.050			-.752	.798	-.833	.752	-1.161	.721
.100			-.749	.570	-.866	.541	-1.211	.503
.150			-.783	.357	-.847	.379	-1.237	.338
.250	-.716		-.836	.134	-.933	.172	-1.265	.144
.400	-.812		-.918	-.085	-.977	-.031	-1.263	-.028
.550	-.846	-.163	-.901	-.165	-.990	-.159	-1.156	-.111
.700	-.787	-.390	-.867	-.351	-.929	-.249	-.966	-.142
.850	-.734	-1.070	-.785	-.665	-.830	-.376	-.659	-.192
.950	-.660	-.915	-.684	-.660	-.783	-.471	-.441	-.241

TABLE 9 (B) - CONCLUDED

PRESSURE PROFILE ON WING

REAR NOZZLES DEFLECTED 45 DEGREES

ALPHA = 15.99 DEG Q = 2034 N/SQ.M. (42.48 LBF/SQ.FT.) EFFECT. VELOC. RATIO = .291

	ETA = .250		ETA = .387		ETA = .524		ETA = .800	
X/C	CP		CP		CP		CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			.041		-.450		-1.321	
.010			-.186	.897				
.015					-.792	.967		
.025			-.722	.955	-.799	.919	-1.141	.894
.050			-.710	.832	-.785	.784	-1.103	.741
.100			-.720	.626	-.776	.578	-1.116	.538
.150			-.731	.463	-.799	.412	-1.132	.358
.250	-.708		-.754	.174	-.839	.204	-1.133	.167
.400	-.759		-.794	-.064	-.888	-.022	-1.205	-.019
.550	-.806	-.155	-.851	-.174	-.870	-.162	-1.146	-.126
.700	-.812	-.436	-.855	-.380	-.941	-.283	-1.100	-.188
.850	-.831	-1.173	-.825	-.754	-.880	-.457	-.898	-.250
.950	-.787	-1.023	-.754	-.749	-.831	-.597	-.750	-.376

ALPHA = 18.03 DEG Q = 2032 N/SQ.M. (42.45 LBF/SQ.FT.) EFFECT. VELOC. RATIO = .291

	ETA = .250		ETA = .387		ETA = .524		ETA = .800	
X/C	CP		CP		CP		CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			-.117		-.574		-1.300	
.010			-.205	.821				
.015					-.806	.975		
.025			-.701	.975	-.795	.958	-1.037	.917
.050			-.699	.885	-.778	.827	-1.027	.790
.100			-.706	.688	-.775	.644	-1.079	.585
.150			-.723	.525	-.792	.487	-1.039	.419
.250	-.716		-.747	.234	-.801	.253	-1.061	.202
.400	-.774		-.796	-.022	-.871	.016	-1.110	.003
.550	-.818	-.162	-.828	-.163	-.895	-.158	-1.150	-.122
.700	-.830	-.461	-.840	-.382	-.923	-.292	-1.131	-.206
.850	-.849	-1.241	-.838	-.773	-.914	-.472	-1.017	-.341
.950	-.847	-1.088	-.775	-.795	-.874	-.643	-.914	-.470

ALPHA = 19.99 DEG Q = 2038 N/SQ.M. (42.56 LBF/SQ.FT.) EFFECT. VELOC. RATIO = .291

	ETA = .250		ETA = .387		ETA = .524		ETA = .800	
X/C	CP		CP		CP		CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			-.310		-.735		-1.402	
.010			-.241	.744				
.015					-.809	.938		
.025			-.713	.976	-.822	.959	-1.072	.924
.050			-.718	.917	-.807	.867	-1.039	.812
.100			-.737	.764	-.816	.699	-1.086	.618
.150			-.737	.597	-.812	.562	-1.054	.457
.250	-.725		-.763	.318	-.840	.318	-1.058	.243
.400	-.810		-.811	.022	-.894	.061	-1.121	.031
.550	-.842	-.125	-.865	-.130	-.943	-.135	-1.109	-.116
.700	-.851	-.446	-.881	-.382	-.965	-.285	-1.102	-.227
.850	-.906	-1.286	-.868	-.811	-.939	-.484	-1.081	-.382
.950	-.912	-1.132	-.802	-.810	-.899	-.657	-.987	-.550

TABLE 9 (C)

PRESSURE PROFILE ON WING

REAR NOZZLES DEFLECTED 45 DEGREES

ALPHA = .01 DEG Q = 910 N/SQ.M. (19.00 LBF/SQ.FT.) EFFECT. VELOC. RATIO = .194

X/C	ETA = .250 CP		ETA = .387 CP		ETA = .524 CP		ETA = .800 CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			.601		.762		.923	
.010			-2.213	-.194				
.015					-.463	-.421		
.025			-.342	-.508	-.409	-.494	-.283	-.070
.050			-.352	-.418	-.286	-.418	-.397	-.234
.100			-.369	-.491	-.383	-.324	-.309	-.226
.150			-.461	-.532	-.390	-.404	-.345	-.340
.250	-.323		-.446	-.555	-.375	-.456	-.397	-.356
.400	-.412		-.476	-.493	-.442	-.423	-.360	-.380
.550	-.366	-.337	-.389	-.431	-.367	-.391	-.293	-.305
.700	-.318	-.392	-.327	-.446	-.281	-.332	-.205	-.212
.850	-.315	-.945	-.266	-.574	-.178	-.259	-.093	-.129
.950	-.345	-.615	-.254	-.400	-.127	-.156	-.036	-.035

ALPHA = 4.02 DEG Q = 908 N/SQ.M. (18.97 LBF/SQ.FT.) EFFECT. VELOC. RATIO = .194

X/C	ETA = .250 CP		ETA = .387 CP		ETA = .524 CP		ETA = .800 CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			.929		.716		.169	
.010			-2.027	.641				
.015					-1.241	.495		
.025			-1.189	.343	-1.562	.339	-1.206	.395
.050			-.957	.162	-1.080	.180	-.913	.256
.100			-.655	-.054	-.710	.057	-.722	.006
.150			-.705	-.094	-.733	-.046	-.667	-.089
.250	-.483		-.629	-.334	-.670	-.203	-.648	-.209
.400	-.549		-.601	-.376	-.583	-.323	-.516	-.255
.550	-.464	-.328	-.465	-.353	-.475	-.310	-.388	-.234
.700	-.398	-.473	-.377	-.416	-.332	-.296	-.264	-.186
.850	-.385	-1.030	-.318	-.592	-.225	-.269	-.148	-.132
.950	-.392	-.752	-.285	-.447	-.175	-.203	-.073	-.062

ALPHA = 8.00 DEG Q = 906 N/SQ.M. (18.92 LBF/SQ.FT.) EFFECT. VELOC. RATIO = .194

X/C	ETA = .250 CP		ETA = .387 CP		ETA = .524 CP		ETA = .800 CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			.090		-1.011		-2.345	
.010			-1.884	.934				
.015					-3.141	.906		
.025			-2.522	.800	-2.924	.786	-2.962	.783
.050			-1.562	.569	-1.670	.602	-1.553	.532
.100			-1.150	.376	-1.238	.377	-1.188	.296
.150			-1.009	.125	-1.118	.179	-.990	.163
.250	-.598		-.852	-.068	-.974	-.018	-.882	-.026
.400	-.641		-.734	-.245	-.765	-.169	-.671	-.141
.550	-.536	-.310	-.559	-.260	-.557	-.222	-.490	-.186
.700	-.460	-.453	-.452	-.367	-.415	-.242	-.329	-.156
.850	-.420	-1.089	-.369	-.597	-.289	-.255	-.208	-.135
.950	-.438	-.829	-.335	-.465	-.220	-.198	-.124	-.085

TABLE 9 (C) - CONTINUED

PRESSURE PROFILE ON WING

REAR NOZZLES DEFLECTED 45 DEGREES

ALPHA = 10.01 DEG Q = 909 N/SQ.M. (18.99 LBF/SQ.FT.) EFFECT. VELOC. RATIO = .194

X/C	ETA = .250 CP		ETA = .387 CP		ETA = .524 CP		ETA = .800 CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			-1.312		-2.995		-4.543	
.010			-1.841	.830				
.015					-4.205	.915		
.025			-2.867	.904	-3.094	.892	-2.796	.863
.050			-1.940	.702	-2.113	.740	-2.020	.690
.100			-1.356	.482	-1.517	.482	-1.434	.445
.150			-1.214	.287	-1.295	.305	-1.170	.261
.250	-.662		-.969	.037	-1.051	.079	-.980	.078
.400	-.681		-.794	-.155	-.831	-.098	-.756	-.078
.550	-.577	-.279	-.622	-.219	-.604	-.189	-.529	-.145
.700	-.486	-.419	-.482	-.361	-.448	-.234	-.367	-.146
.850	-.445	-1.128	-.397	-.609	-.306	-.261	-.236	-.136
.950	-.461	-.881	-.348	-.511	-.241	-.230	-.145	-.110

ALPHA = 11.97 DEG Q = 907 N/SQ.M. (18.94 LBF/SQ.FT.) EFFECT. VELOC. RATIO = .194

X/C	ETA = .250 CP		ETA = .387 CP		ETA = .524 CP		ETA = .800 CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			-.127		-.751		-1.684	
.010			-1.856	.864				
.015					-1.444	.923		
.025			-1.606	.881	-1.078	.843	-1.590	.813
.050			-1.648	.735	-1.319	.679	-1.580	.643
.100			-1.390	.490	-1.736	.493	-1.484	.424
.150			-1.630	.307	-1.248	.299	-1.721	.243
.250	-.812		-1.430	.053	-1.357	.101	-1.429	.061
.400	-.801		-1.115	-.164	-1.080	-.102	-1.057	-.090
.550	-.684	-.241	-.878	-.240	-.903	-.211	-.853	-.170
.700	-.585	-.436	-.625	-.396	-.609	-.278	-.517	-.199
.850	-.490	-1.176	-.474	-.673	-.520	-.345	-.365	-.231
.950	-.481	-.920	-.435	-.586	-.668	-.399	-.324	-.216

ALPHA = 14.00 DEG Q = 912 N/SQ.M. (19.05 LBF/SQ.FT.) EFFECT. VELOC. RATIO = .194

X/C	ETA = .250 CP		ETA = .387 CP		ETA = .524 CP		ETA = .800 CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			.068		-.499		-1.511	
.010			-1.929	.859				
.015					-.986	.911		
.025			-.972	.842	-.991	.845	-1.324	.835
.050			-1.012	.750	-.975	.686	-1.411	.654
.100			-.977	.541	-.972	.493	-1.329	.439
.150			-1.115	.336	-1.010	.320	-1.357	.265
.250	-.846		-1.181	.086	-1.133	.108	-1.399	.070
.400	-.938		-1.149	-.156	-1.126	-.090	-1.361	-.084
.550	-.896	-.250	-.994	-.242	-1.077	-.247	-1.245	-.176
.700	-.826	-.475	-.952	-.442	-1.002	-.315	-1.034	-.208
.850	-.675	-1.245	-.824	-.737	-.903	-.430	-.742	-.272
.950	-.639	-1.017	-.665	-.727	-.834	-.610	-.536	-.267

TABLE 9 (C) - CONCLUDED

PRESSURE PROFILE ON WING

REAR NOZZLES DEFLECTED 45 DEGREES

ALPHA = 16.01 DEG Q = 907 N/SQ.M. (18.95 LBF/SQ.FT.) EFFECT. VELOC. RATIO = .194

	ETA = .250		ETA = .387		ETA = .524		ETA = .800	
X/C	CP		CP		CP		CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			-.052		-.555		-1.521	
.010			-1.957	.831				
.015					-.957	.914		
.025			-.823	.899	-.948	.866	-1.330	.841
.050			-.840	.791	-.906	.744	-1.304	.699
.100			-.821	.572	-.915	.532	-1.262	.490
.150			-.861	.400	-.934	.364	-1.267	.317
.250	-.811		-.873	.105	-.936	.148	-1.281	.057
.400	-.860		-.942	-.129	-1.073	-.105	-1.347	-.089
.550	-.967	-.252	-.982	-.269	-1.057	-.265	-1.322	-.189
.700	-.950	-.540	-.955	-.488	-1.035	-.389	-1.183	-.259
.850	-.888	-1.375	-.921	-.871	-1.000	-.557	-.949	-.336
.950	-.834	-1.180	-.817	-.821	-.908	-.695	-.737	-.419

ALPHA = 17.97 DEG Q = 912 N/SQ.M. (19.05 LBF/SQ.FT.) EFFECT. VELOC. RATIO = .194

	ETA = .250		ETA = .387		ETA = .524		ETA = .800	
X/C	CP		CP		CP		CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			-.190		-.710		-1.547	
.010			-1.889	.751				
.015					-.906	.910		
.025			-.838	.914	-.895	.895	-1.208	.864
.050			-.814	.822	-.899	.776	-1.189	.741
.100			-.821	.634	-.916	.592	-1.193	.535
.150			-.830	.468	-.918	.423	-1.194	.363
.250	-.821		-.859	.157	-.928	.179	-1.224	.141
.400	-.920		-.919	-.116	-.945	-.085	-1.271	-.072
.550	-.918	-.237	-.940	-.262	-1.004	-.250	-1.275	-.191
.700	-.922	-.538	-.947	-.510	-1.039	-.394	-1.219	-.275
.850	-.936	-1.438	-.955	-.915	-1.011	-.597	-1.114	-.407
.950	-.919	-1.239	-.868	-.895	-.945	-.746	-.964	-.507

ALPHA = 20.01 DEG Q = 912 N/SQ.M. (19.05 LBF/SQ.FT.) EFFECT. VELOC. RATIO = .194

	ETA = .250		ETA = .387		ETA = .524		ETA = .800	
X/C	CP		CP		CP		CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			-.473		-.892		-1.537	
.010			-1.939	.645				
.015					-.932	.875		
.025			-.822	.907	-.927	.904	-1.219	.868
.050			-.861	.864	-.913	.818	-1.219	.778
.100			-.847	.693	-.922	.622	-1.167	.564
.150			-.870	.524	-.936	.465	-1.164	.388
.250	-.841		-.892	.221	-.965	.241	-1.184	.178
.400	-.937		-.939	-.072	-1.023	-.033	-1.231	-.054
.550	-.932	-.244	-.964	-.251	-1.069	-.227	-1.270	-.187
.700	-.979	-.537	-.992	-.525	-1.077	-.412	-1.304	-.307
.850	-1.061	-1.509	-.982	-.968	-1.073	-.608	-1.172	-.465
.950	-.979	-1.299	-.931	-.937	-1.013	-.778	-1.073	-.627

TABLE 9 (D)

PRESSURE PROFILE ON WING

REAR NOZZLES DEFLECTED 45 DEGREES

ALPHA = 0.00 DEG Q = 226 N/SQ.M. (4.73 LBF/SQ.FT.) EFFECT. VELOC. RATIO = .097

X/C	ETA = .250 CP		ETA = .387 CP		ETA = .524 CP		ETA = .800 CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			.217		.440		.628	
.010			-8.025	-.611				
.015					-.166	-.758		
.025			-.382	-.959	-.368	-.834	-.465	-.737
.050			-.458	-.918	-.451	-.737	-.416	-.479
.100			-.430	-.737	-.458	-.716	-.479	-.618
.150			-.521	-.855	-.500	-.751	-.500	-.681
.250	-.521		-.556	-.923	-.549	-.741	-.535	-.680
.400	-.613		-.606	-.839	-.620	-.751	-.559	-.643
.550	-.566	-.670	-.586	-.772	-.556	-.704	-.492	-.579
.700	-.573	-.741	-.542	-.788	-.488	-.654	-.431	-.522
.850	-.589	-1.396	-.522	-.913	-.434	-.586	-.357	-.424
.950	-.657	-1.038	-.542	-.768	-.404	-.471	-.303	-.306

ALPHA = 4.06 DEG Q = 229 N/SQ.M. (4.78 LBF/SQ.FT.) EFFECT. VELOC. RATIO = .097

X/C	ETA = .250 CP		ETA = .387 CP		ETA = .524 CP		ETA = .800 CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			.625		.487		.129	
.010			-7.243	.260				
.015					-1.394	.205		
.025			-1.242	.012	-1.366	-.043	-1.215	.094
.050			-.932	-.202	-1.049	-.126	-.925	-.236
.100			-.774	-.450	-.856	-.312	-.815	-.278
.150			-.808	-.484	-.829	-.457	-.843	-.381
.250	-.643		-.781	-.673	-.815	-.527	-.836	-.510
.400	-.704		-.744	-.704	-.740	-.623	-.734	-.543
.550	-.650	-.657	-.673	-.677	-.667	-.617	-.620	-.496
.700	-.617	-.824	-.600	-.730	-.567	-.590	-.476	-.440
.850	-.623	-1.405	-.583	-.857	-.483	-.523	-.406	-.376
.950	-.690	-1.121	-.583	-.760	-.463	-.473	-.346	-.346

ALPHA = 7.99 DEG Q = 228 N/SQ.M. (4.77 LBF/SQ.FT.) EFFECT. VELOC. RATIO = .097

X/C	ETA = .250 CP		ETA = .387 CP		ETA = .524 CP		ETA = .800 CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			.065		-.771		-1.606	
.010			-6.616	.630				
.015					-2.972	.568		
.025			-2.192	.423	-2.510	.396	-2.165	.389
.050			-1.647	.203	-1.764	.209	-1.744	.223
.100			-1.136	-.067	-1.233	-.018	-1.116	-.053
.150			-1.095	-.239	-1.157	-.143	-1.088	-.170
.250	-.743		-.978	-.440	-1.012	-.353	-1.005	-.329
.400	-.791		-.861	-.570	-.888	-.487	-.841	-.446
.550	-.714	-.707	-.754	-.594	-.751	-.527	-.681	-.453
.700	-.684	-.888	-.671	-.701	-.640	-.550	-.553	-.430
.850	-.674	-1.503	-.627	-.881	-.537	-.533	-.440	-.390
.950	-.748	-1.156	-.634	-.758	-.490	-.470	-.380	-.349

TABLE 9 (D) - CONTINUED

PRESSURE PROFILE ON WING

REAR NOZZLES DEFLECTED 45 DEGREES

ALPHA = 10.04 DEG Q = 227 N/SQ.M. (4.74 LBF/SQ.FT.) EFFECT. VELOC. RATIO = .097

	ETA = .250		ETA = .387		ETA = .524		ETA = .800	
X/C	CP		CP		CP		CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			-.570		-2.488		-3.850	
.010			-6.157	.584				
.015					-3.627			
.025			-3.641	.535	-3.370	.515	-3.419	.508
.050			-1.765	.369	-1.959	.348	-1.883	.327
.100			-1.348	.098	-1.515	.125	-1.459	.077
.150			-1.285	-.069	-1.369	-.055	-1.292	-.097
.250	-.806		-1.077	-.368	-1.158	-.264	-1.202	-.250
.400	-.877		-.954	-.527	-.995	-.453	-.927	-.399
.550	-.796	-.729	-.813	-.581	-.816	-.527	-.749	-.439
.700	-.722	-.917	-.725	-.709	-.695	-.567	-.594	-.436
.850	-.719	-1.551	-.682	-.904	-.584	-.607	-.496	-.426
.950	-.786	-1.224	-.675	-.776	-.523	-.513	-.419	-.392

ALPHA = 12.02 DEG Q = 228 N/SQ.M. (4.75 LBF/SQ.FT.) EFFECT. VELOC. RATIO = .097

	ETA = .250		ETA = .387		ETA = .524		ETA = .800	
X/C	CP		CP		CP		CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			-1.077		-3.543		-4.692	
.010			-5.745	.433				
.015					-5.073	.599		
.025			-2.559	.613	-3.723		-3.363	.571
.050			-2.102	.446	-2.324	.453	-2.213	.426
.100			-1.590	.211	-1.984	.225	-1.666	.169
.150			-1.403	.017	-1.881	.017	-1.444	-.025
.250	-.869		-1.500	-.274	-1.437	-.214	-1.202	-.214
.400	-.922		-1.026	-.496	-1.053	-.415	-1.006	-.375
.550	-.828	-.737	-.852	-.586	-.872	-.523	-.798	-.442
.700	-.764	-.925	-.764	-.717	-.721	-.583	-.650	-.455
.850	-.741	-1.570	-.697	-.939	-.613	-.596	-.539	-.455
.950	-.815	-1.214	-.697	-.811	-.573	-.549	-.455	-.442

ALPHA = 14.04 DEG Q = 228 N/SQ.M. (4.75 LBF/SQ.FT.) EFFECT. VELOC. RATIO = .097

	ETA = .250		ETA = .387		ETA = .524		ETA = .800	
X/C	CP		CP		CP		CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			-.596		-1.025		-1.884	
.010			-5.375	.477				
.015					-1.337	.574		
.025			-1.676	.554	-1.586	.505	-1.732	.491
.050			-1.669	.429	-1.503	.360	-1.663	.332
.100			-1.649	.214	-1.392	.152	-1.669	.124
.150			-1.545	.006	-1.573	.013	-1.683	-.049
.250	-1.095		-1.427	-.272	-1.545	-.231	-1.621	-.235
.400	-1.014		-1.343	-.493	-1.410	-.436	-1.487	-.406
.550	-.930	-.708	-1.040	-.604	-1.084	-.537	-1.218	-.480
.700	-.940	-.923	-1.003	-.765	-.993	-.631	-1.158	-.503
.850	-.802	-1.604	-.802	-1.000	-.997	-.705	-.701	-.547
.950	-.856	-1.212	-.742	-.859	-.772	-.675	-.668	-.534

TABLE 9 (D) - CONCLUDED

PRESSURE PROFILE ON WING

REAR NOZZLES DEFLECTED 45 DEGREES

ALPHA = 16.00 DEG Q = 225 N/SQ.M. (4.69 LBF/SQ.FT.) EFFECT. VELOC. RATIO = .096

	ETA = .250		ETA = .387		ETA = .524		ETA = .800	
X/C	CP		CP		CP		CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			-.402		-.858		-1.846	
.010			-5.064	.460				
.015					-1.271	.544		
.025			-1.257	.544	-1.327	.488	-1.594	.474
.050			-1.257	.418	-1.299	.362	-1.650	.334
.100			-1.398	.208	-1.257	.159	-1.538	.138
.150			-1.285	.019	-1.264	-.045	-1.587	-.059
.250	-1.201		-1.447	-.271	-1.334	-.237	-1.594	-.237
.400	-1.253		-1.358	-.509	-1.331	-.448	-1.586	-.410
.550	-1.219	-.702	-1.290	-.618	-1.345	-.590	-1.501	-.495
.700	-1.141	-.988	-1.171	-.808	-1.283	-.702	-1.134	-.560
.850	-1.018	-1.688	-1.069	-1.134	-1.192	-.815	-.988	-.594
.950	-.920	-1.260	-.951	-.998	-1.039	-.855	-.957	-.679

ALPHA = 18.01 DEG Q = 226 N/SQ.M. (4.72 LBF/SQ.FT.) EFFECT. VELOC. RATIO = .096

	ETA = .250		ETA = .387		ETA = .524		ETA = .800	
X/C	CP		CP		CP		CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			-.510		-.935		-1.891	
.010			-4.736	.418				
.015					-1.235	.501		
.025			-1.159	.508	-1.277	.466	-1.514	.459
.050			-1.103	.404	-1.200	.348	-1.570	.334
.100			-1.110	.194	-1.179	.201	-1.584	.118
.150			-1.159	.027	-1.207	.013	-1.584	-.064
.250	-1.131		-1.193	-.245	-1.263	-.225	-1.500	-.231
.400	-1.158		-1.232	-.519	-1.242	-.451	-1.590	-.427
.550	-1.222	-.759	-1.286	-.644	-1.310	-.617	-1.550	-.529
.700	-1.252	-1.002	-1.245	-.867	-1.300	-.749	-1.489	-.613
.850	-1.296	-1.763	-1.171	-1.195	-1.259	-.874	-1.293	-.722
.950	-1.137	-1.300	-1.120	-1.090	-1.134	-1.009	-1.171	-.833

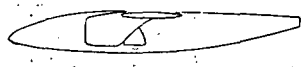
ALPHA = 20.00 DEG Q = 226 N/SC.M. (4.73 LBF/SQ.FT.) EFFECT. VELOC. RATIO = .097

	ETA = .250		ETA = .387		ETA = .524		ETA = .800	
X/C	CP		CP		CP		CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			-.657		-1.067		-1.763	
.010			-4.416	.290				
.015					-1.172	.499		
.025			-1.116	.506	-1.172	.499	-1.478	.471
.050			-1.109	.450	-1.227	.388	-1.443	.360
.100			-1.151	.262	-1.179	.200	-1.450	.151
.150			-1.095	.095	-1.172	.074	-1.450	-.009
.250	-1.158		-1.151	-.196	-1.186	-.182	-1.450	-.226
.400	-1.174		-1.188	-.506	-1.258	-.445	-1.468	-.412
.550	-1.292	-.732	-1.252	-.675	-1.285	-.614	-1.538	-.550
.700	-1.312	-1.022	-1.255	-.901	-1.302	-.769	-1.481	-.648
.850	-1.245	-1.862	-1.231	-1.242	-1.258	-.941	-1.390	-.789
.950	-1.235	-1.414	-1.167	-1.157	-1.221	-1.059	-1.248	-.931

TABLE 10 (A)

PRESSURE PROFILE ON WING

FORWARD NOZZLES DEFLECTED 90 DEGREES



ALPHA = .02 DEG Q = 2840 N/SQ.M. (59.31 LBF/SQ.FT.) EFFECT. VELOC. RATIO = ∞

X/C	ETA = .250 CP		ETA = .387 CP		ETA = .524 CP		ETA = .800 CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			.825		.938		.968	
.010			.517	.045				
.015					-.334	-.009		
.025			-.700	-.272	-.603	-.249	-.387	-.204
.050			-.414	-.312	-.406	-.226	-.366	-.236
.100			-.362	-.371	-.403	-.278	-.277	-.236
.150			-.473	-.483	-.394	-.334	-.355	-.315
.250	-.312		-.409	-.509	-.409	-.406	-.366	-.337
.400	-.403		-.427	-.432	-.410	-.409	-.358	-.354
.550	-.323	-.353	-.335	-.351	-.308	-.352	-.260	-.271
.700	-.278	-.391	-.232	-.298	-.199	-.274	-.135	-.194
.850	-.260	-.411	-.179	-.305	-.094	-.144	-.043	-.071
.950	-.299	-.427	-.151	-.196	-.025	-.048	.017	.026

ALPHA = 4.04 DEG Q = 2851 N/SQ.M. (59.55 LBF/SQ.FT.) EFFECT. VELOC. RATIO = ∞

X/C	ETA = .250 CP		ETA = .387 CP		ETA = .524 CP		ETA = .800 CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			.962		.608		-.063	
.010			.524	.835				
.015					-1.510	.614		
.025			-1.299	.409	-1.383	.440	-1.180	.460
.050			-.932	.176	-1.046	.247	-1.052	.232
.100			-.651	.017	-.697	.099	-.648	.108
.150			-.688	-.153	-.694	-.064	-.590	-.036
.250	-.405		-.583	-.309	-.587	-.200	-.581	-.133
.400	-.455		-.505	-.339	-.512	-.296	-.463	-.211
.550	-.366	-.337	-.375	-.315	-.359	-.264	-.319	-.190
.700	-.290	-.362	-.267	-.274	-.232	-.219	-.181	-.134
.850	-.254	-.388	-.175	-.247	-.108	-.144	-.073	-.074
.950	-.250	-.355	-.106	-.131	-.022	-.049	.018	.021

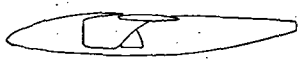
ALPHA = 8.03 DEG Q = 2849 N/SQ.M. (59.50 LBF/SQ.FT.) EFFECT. VELOC. RATIO = ∞

X/C	ETA = .250 CP		ETA = .387 CP		ETA = .524 CP		ETA = .800 CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			.183		-1.229		-2.083	
.010			.494	1.00				
.015					-2.904	.939		
.025			-2.542	.808	-3.184	.777	-2.460	.790
.050			-1.497	.638	-1.534	.645	-1.489	.570
.100			-1.006	.395	-1.145	.396	-1.089	.350
.150			-.927	.161	-1.005	.221	-.897	.234
.250	-.490		-.731	-.117	-.791	-.028	-.759	.035
.400	-.492		-.592	-.312	-.655	-.171	-.585	-.079
.550	-.382	-.307	-.415	-.353	-.439	-.196	-.395	-.109
.700	-.280	-.366	-.277	-.299	-.270	-.172	-.232	-.083
.850	-.201	-.298	-.153	-.215	-.121	-.125	-.107	-.037
.950	-.170	-.237	-.068	-.110	-.034	-.050	-.019	-.001

TABLE 10 (A) - CONTINUED

PRESSURE PROFILE ON WING

FORWARD NOZZLES DEFLECTED 90 DEGREES



ALPHA = 10.04 DEG Q = 2845 N/SQ.M. (59.43 LBF/SQ.FT.) EFFECT. VELOC. RATIO = ∞

	ETA = .250		ETA = .387		ETA = .524		ETA = .800	
X/C	CP		CP		CP		CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			-.600		-3.149		-3.482	
.010			.416	.961				
.015					-4.487	.986		
.025			-2.732	.943	-2.933	.929	-2.723	.912
.050			-1.877	.738	-2.036	.719	-1.836	.734
.100			-1.227	.477	-1.392	.491	-1.260	.468
.150			-1.086	.276	-1.174	.307	-1.063	.310
.250	-.535		-.811	-.001	-.888	.098	-.880	.115
.400	-.518		-.651	-.247	-.677	-.097	-.621	-.032
.550	-.392	-.299	-.440	-.303	-.453	-.150	-.412	-.070
.700	-.277	-.356	-.284	-.275	-.276	-.145	-.247	-.053
.850	-.194	-.279	-.155	-.206	-.128	-.110	-.112	-.031
.950	-.159	-.223	-.065	-.098	-.043	-.052	-.023	.004

ALPHA = 12.02 DEG Q = 2843 N/SQ.M. (59.37 LBF/SQ.FT.) EFFECT. VELOC. RATIO = ∞

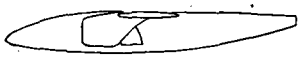
	ETA = .250		ETA = .387		ETA = .524		ETA = .800	
X/C	CP		CP		CP		CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			.340		-.265		-.824	
.010			.439	.957				
.015					-.838	.944		
.025			-.965	.882	-.915	.838	-1.035	.828
.050			-.881	.736	-.860	.661	-1.055	.646
.100			-.858	.490	-1.022	.447	-1.279	.415
.150			-.856	.267	-.669	.270	-1.096	.258
.250	-.624		-.922	-.024	-1.029	.063	-1.021	.084
.400	-.586		-.877	-.290	-.912	-.127	-.830	-.056
.550	-.539	-.323	-.694	-.337	-.755	-.217	-.777	-.124
.700	-.421	-.374	-.479	-.313	-.485	-.201	-.504	-.130
.850	-.254	-.304	-.376	-.297	-.468	-.232	-.378	-.122
.950	-.239	-.255	-.320	-.217	-.333	-.234	-.316	-.158

ALPHA = 14.03 DEG Q = 2843 N/SQ.M. (59.39 LBF/SQ.FT.) EFFECT. VELOC. RATIO = ∞

	ETA = .250		ETA = .387		ETA = .524		ETA = .800	
X/C	CP		CP		CP		CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			.422		.062		-.742	
.010			.460	1.00				
.015					-.561	.947		
.025			-.564	.904	-.580	.820	-.835	.827
.050			-.542	.718	-.612	.670	-.878	.654
.100			-.546	.473	-.523	.470	-.872	.445
.150			-.590	.292	-.625	.281	-.814	.283
.250	-.536		-.585	-.019	-.757	.064	-.883	.094
.400	-.614		-.716	-.299	-.607	-.130	-.888	-.068
.550	-.668	-.358	-.770	-.369	-.700	-.223	-.848	-.147
.700	-.661	-.387	-.697	-.349	-.686	-.237	-.765	-.165
.850	-.512	-.371	-.605	-.362	-.603	-.310	-.609	-.213
.950	-.416	-.314	-.500	-.373	-.491	-.362	-.528	-.267

TABLE 10 (A) - CONCLUDED
PRESSURE PROFILE ON WING

FORWARD NOZZLES DEFLECTED 90 DEGREES



ALPHA = 16.02 DEG Q = 2844 N/SQ.M. (59.41 LBF/SQ.FT.) EFFECT. VELOC. RATIO = ∞

	ETA = .250		ETA = .387		ETA = .524		ETA = .800	
X/C	CP		CP		CP		CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			.325		.040		-.776	
.010			.477	.978				
.015					-.501	.970		
.025			-.496	.927	-.492	.861	-.755	.855
.050			-.505	.755	-.494	.723	-.762	.682
.100			-.508	.534	-.470	.494	-.786	.499
.150			-.540	.313	-.505	.316	-.791	.310
.250	-.535		-.551	-.001	-.525	.103	-.790	.120
.400	-.602		-.614	-.302	-.621	-.125	-.851	-.043
.550	-.660	-.384	-.694	-.381	-.613	-.219	-.858	-.136
.700	-.721	-.411	-.696	-.373	-.661	-.269	-.798	-.188
.850	-.658	-.393	-.636	-.393	-.606	-.322	-.689	-.282
.950	-.485	-.347	-.565	-.403	-.536	-.390	-.641	-.319

ALPHA = 18.02 DEG Q = 2845 N/SQ.M. (59.42 LBF/SQ.FT.) EFFECT. VELOC. RATIO = ∞

	ETA = .250		ETA = .387		ETA = .524		ETA = .800	
X/C	CP		CP		CP		CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			.199		-.032		-.860	
.010			.476	.936				
.015					-.475	.985		
.025			-.523	.969	-.472	.902	-.699	.880
.050			-.514	.844	-.465	.756	-.721	.736
.100			-.523	.617	-.465	.558	-.717	.515
.150			-.526	.387	-.481	.369	-.728	.366
.250	-.556		-.570	.046	-.513	.129	-.729	.139
.400	-.639		-.657	-.291	-.564	-.098	-.749	-.033
.550	-.699	-.401	-.673	-.370	-.590	-.227	-.799	-.135
.700	-.716	-.413	-.669	-.361	-.629	-.263	-.822	-.196
.850	-.673	-.395	-.656	-.381	-.595	-.332	-.789	-.298
.950	-.565	-.396	-.582	-.401	-.520	-.400	-.754	-.399

ALPHA = 20.04 DEG Q = 2850 N/SQ.M. (59.52 LBF/SQ.FT.) EFFECT. VELOC. RATIO = ∞

	ETA = .250		ETA = .387		ETA = .524		ETA = .800	
X/C	CP		CP		CP		CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			-.042		-.161		-.761	
.010			.465	.886				
.015					-.476	1.004		
.025			-.511	.996	-.473	.948	-.631	.908
.050			-.554	.893	-.478	.825	-.628	.784
.100			-.547	.674	-.475	.610	-.602	.562
.150			-.559	.453	-.478	.432	-.637	.400
.250	-.577		-.607	.060	-.531	.185	-.620	.195
.400	-.641		-.645	-.283	-.559	-.072	-.710	.016
.550	-.710	-.394	-.687	-.315	-.623	-.174	-.729	-.118
.700	-.750	-.429	-.690	-.345	-.611	-.248	-.717	-.203
.850	-.685	-.407	-.640	-.400	-.582	-.309	-.727	-.315
.950	-.595	-.408	-.632	-.419	-.523	-.389	-.744	-.459

TABLE LO (B)

PRESSURE PROFILE ON WING

FORWARD NOZZLES DEFLECTED 90 DEGREES

ALPHA = .01 DEG Q = 2029 N/SQ.M. (42.38 LBF/SQ.FT.) EFFECT. VELOC. RATIO = .302

X/C	ETA = .250 CP		ETA = .387 CP		ETA = .524 CP		ETA = .800 CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			.655		.943		.937	
.010			.283	-.247				
.015					-.361	-.570		
.025			-.693	-.786	-.559	-.564	-.311	-.319
.050			-.459	-.654	-.521	-.501	-.506	-.210
.100			-.469	-.862	-.463	-.639	-.420	-.338
.150			-.537	-1.068	-.514	-.832	-.498	-.443
.250	-.400		-.536	-1.259	-.517	-.911	-.484	-.491
.400	-.492		-.515	-1.302	-.507	-.990	-.452	-.520
.550	-.431	-1.507	-.438	-1.268	-.412	-.889	-.348	-.437
.700	-.398	-1.717	-.352	-1.149	-.305	-.736	-.221	-.343
.850	-.431	-1.583	-.325	-.890	-.226	-.523	-.142	-.221
.950	-.629	-1.529	-.332	-.530	-.202	-.315	-.083	-.102

ALPHA = 4.01 DEG Q = 2036 N/SQ.M. (42.52 LBF/SQ.FT.) EFFECT. VELOC. RATIO = .302

X/C	ETA = .250 CP		ETA = .387 CP		ETA = .524 CP		ETA = .800 CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			.990		.758		.184	
.010			.301	.621				
.015					-1.736	.372		
.025			-1.444	.097	-1.478	.132	-1.355	.311
.050			-1.137	-.128	-1.287	.023	-1.239	.279
.100			-.849	-.349	-.870	-.142	-.758	-.019
.150			-.800	-.696	-.789	-.422	-.770	-.168
.250	-.491		-.742	-1.010	-.791	-.663	-.697	-.249
.400	-.580		-.645	-1.278	-.675	-.867	-.574	-.405
.550	-.485	-1.506	-.506	-1.248	-.509	-.848	-.438	-.381
.700	-.441	-1.552	-.407	-1.115	-.380	-.705	-.293	-.311
.850	-.437	-1.560	-.358	-.856	-.278	-.530	-.175	-.231
.950	-.576	-1.351	-.332	-.541	-.216	-.335	-.099	-.122

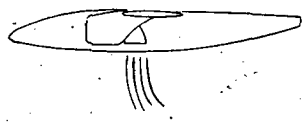
ALPHA = 8.03 DEG Q = 2037 N/SQ.M. (42.54 LBF/SQ.FT.) EFFECT. VELOC. RATIO = .302

X/C	ETA = .250 CP		ETA = .387 CP		ETA = .524 CP		ETA = .800 CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			.317		-1.338		-2.375	
.010			.327	1.0				
.015					-3.893	.912		
.025			-3.048	.733	-3.333	.778	-2.678	.849
.050			-1.714	.472	-1.922	.450	-1.754	.576
.100			-1.199	-.039	-1.363	.139	-1.282	.292
.150			-1.131	-.314	-1.188	-.121	-1.082	.075
.250	-.608		-.929	-.828	-1.007	-.448	-.956	-.120
.400	-.660		-.790	-1.156	-.787	-.735	-.716	-.276
.550	-.553	-1.486	-.580	-1.169	-.594	-.768	-.517	-.302
.700	-.464	-1.490	-.472	-1.083	-.448	-.696	-.348	-.261
.850	-.450	-1.390	-.351	-.848	-.323	-.556	-.216	-.224
.950	-.528	-1.150	-.346	-.554	-.249	-.361	-.130	-.141

TABLE 10 (B) - CONTINUED

PRESSURE PROFILE ON WING

FORWARD NOZZLES DEFLECTED 90 DEGREES



ALPHA = 10.02 DEG Q = 2040 N/SQ.M. (42.62 LBF/SQ.FT.) EFFECT. VELOC. RATIO = .302

	ETA = .250		ETA = .387		ETA = .524		ETA = .800	
X/C	CP		CP		CP		CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			-.799		-2.945		-4.344	
.010			.300	.920				
.015					-5.245	1.001		
.025			-3.383	.907	-3.692	.910	-3.260	.926
.050			-2.287	.681	-2.441	.629	-2.172	.665
.100			-1.445	.219	-1.746	.291	-1.527	.387
.150			-1.295	-.146	-1.462	.029	-1.309	.205
.250	-.683		-1.030	-.713	-1.106	-.323	-1.071	-.029
.400	-.708		-.852	-1.113	-.875	-.612	-.787	-.218
.550	-.592	-1.537	-.632	-1.190	-.645	-.726	-.561	-.268
.700	-.492	-1.482	-.494	-1.033	-.484	-.676	-.389	-.244
.850	-.456	-1.350	-.410	-.859	-.335	-.557	-.243	-.222
.950	-.526	-1.120	-.371	-.561	-.275	-.388	-.158	-.155

ALPHA = 12.02 DEG Q = 2035 N/SQ.M. (42.50 LBF/SQ.FT.) EFFECT. VELOC. RATIO = .301

	ETA = .250		ETA = .387		ETA = .524		ETA = .800	
X/C	CP		CP		CP		CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			-2.349		-4.492		-6.242	
.010			.191	.783				
.015					-4.915	.961		
.025			-3.935	.993	-4.422	.983	-3.640	.975
.050			-2.580	.769	-2.848	.761	-2.507	.784
.100			-1.721	.391	-1.910	.471	-1.726	.529
.150			-1.414	.014	-1.624	.116	-1.439	.313
.250	-.714		-1.121	-.595	-1.213	-.219	-1.147	.044
.400	-.737		-.887	-1.102	-.943	-.572	-.843	-.171
.550	-.601	-1.507	-.673	-1.190	-.687	-.693	-.605	-.241
.700	-.526	-1.551	-.528	-1.096	-.500	-.685	-.409	-.235
.850	-.474	-1.401	-.422	-.873	-.354	-.572	-.271	-.221
.950	-.509	-1.059	-.370	-.587	-.288	-.419	-.181	-.167

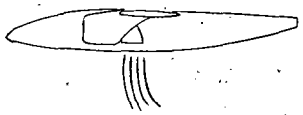
ALPHA = 14.01 DEG Q = 2040 N/SQ.M. (42.60 LBF/SQ.FT.) EFFECT. VELOC. RATIO = .301

	ETA = .250		ETA = .387		ETA = .524		ETA = .800	
X/C	CP		CP		CP		CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			.147		-.352		-1.476	
.010			.232	.983				
.015					-1.046	.977		
.025			-1.232	.918	-1.053	.850	-1.315	.881
.050			-1.101	.690	-1.254	.633	-1.369	.700
.100			-1.182	.325	-.973	.344	-1.346	.423
.150			-.983	-.022	-1.106	.068	-1.340	.218
.250	-.915		-1.221	-.608	-1.177	-.300	-1.314	-.014
.400	-.972		-1.128	-1.112	-1.111	-.634	-1.283	-.220
.550	-.892	-1.466	-1.038	-1.203	-1.095	-.776	-1.201	-.311
.700	-.789	-1.418	-.959	-1.134	-.977	-.781	-.910	-.347
.850	-.672	-1.349	-.615	-.965	-.854	-.746	-.643	-.346
.950	-.608	-1.052	-.585	-.698	-.672	-.714	-.598	-.343

TABLE 10 (B) - CONCLUDED

PRESSURE PROFILE ON WING

FORWARD NOZZLES DEFLECTED 90 DEGREES



ALPHA = 16.04 DEG Q = 2044 N/SQ.M. (42.69 LBF/SQ.FT.) EFFECT. VELOC. RATIO = .302

	ETA = .250		ETA = .387		ETA = .524		ETA = .800	
X/C	CP		CP		CP		CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			-.103		-.370		-1.508	
.010			.269	.978				
.015					-.861	.978		
.025			-.866	.923	-.872	.889	-1.220	.896
.050			-.749	.743	-.815	.667	-1.217	.717
.100			-.819	.379	-.829	.397	-1.216	.458
.150			-.816	.040	-.909	.121	-1.233	.256
.250	-.780		-.896	-.562	-.909	-.246	-1.216	.020
.400	-.934		-.955	-1.060	-.999	-.622	-1.275	-.211
.550	-.958	-1.401	-.961	-1.205	-1.010	-.823	-1.225	-.328
.700	-.936	-1.359	-.982	-1.170	-.997	-.858	-1.121	-.384
.850	-.815	-1.372	-.912	-1.038	-.923	-.857	-.917	-.416
.950	-.742	-1.007	-.783	-.882	-.797	-.810	-.800	-.486

ALPHA = 18.03 DEG Q = 2029 N/SQ.M. (42.37 LBF/SQ.FT.) EFFECT. VELOC. RATIO = .302

	ETA = .250		ETA = .387		ETA = .524		ETA = .800	
X/C	CP		CP		CP		CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			-.013		-.401		-1.427	
.010			.270	.916				
.015					-.828	.995		
.025			-.731	.982	-.807	.923	-1.159	.921
.050			-.766	.791	-.773	.738	-1.108	.748
.100			-.745	.421	-.800	.457	-1.113	.503
.150			-.769	.156	-.793	.200	-1.081	.325
.250	-.767		-.745	-.492	-.825	-.195	-1.076	.059
.400	-.884		-.867	-.981	-.941	-.592	-1.166	-.195
.550	-.912	-1.299	-.895	-1.224	-.949	-.805	-1.134	-.332
.700	-.974	-1.328	-.951	-1.187	-.957	-.840	-1.140	-.411
.850	-.974	-1.273	-.894	-1.070	-.946	-.878	-1.070	-.521
.950	-.867	-1.053	-.785	-.894	-.890	-.845	-.917	-.616

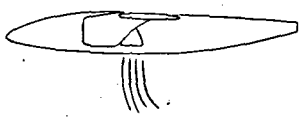
ALPHA = 20.02 DEG Q = 2037 N/SC.M. (42.54 LBF/SQ.FT.) EFFECT. VELOC. RATIO = .302

	ETA = .250		ETA = .387		ETA = .524		ETA = .800	
X/C	CP		CP		CP		CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			-.228		-.545		-1.456	
.010			.206	.823				
.015					-.749	.988		
.025			-.746	.994	-.744	.958	-1.004	.925
.050			-.718	.826	-.748	.780	-.977	.785
.100			-.717	.514	-.753	.493	-1.019	.572
.150			-.731	.199	-.776	.244	-1.011	.360
.250	-.707		-.747	-.297	-.807	-.144	-1.039	.094
.400	-.867		-.831	-.948	-.844	-.531	-1.094	-.161
.550	-.905	-1.264	-.919	-1.115	-.930	-.741	-1.100	-.335
.700	-.971	-1.328	-.932	-1.096	-.979	-.847	-1.153	-.431
.850	-.967	-1.190	-.913	-1.025	-.931	-.890	-1.114	-.567
.950	-.906	-1.100	-.813	-.934	-.881	-.877	-1.031	-.692

TABLE 10 (C)

PRESSURE PROFILE ON WING

FORWARD NOZZLES DEFLECTED 90 DEGREES



ALPHA = .00 DEG Q = 913 N/SQ.M. (19.06 LBF/SQ.FT.) EFFECT. VELOC. RATIO = .203

	ETA = .250		ETA = .387		ETA = .524		ETA = .800	
X/C	CP		CP		CP		CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			-.152		.191		.898	
.010			-.451	-1.520				
.015					.158	-1.603		
.025			-.163	-1.669	-.023	-1.534	-.144	-.701
.050			-.253	-1.223	-.170	-1.183	-.171	-.717
.100			-.349	-1.380	-.287	-1.171	-.332	-.596
.150			-.387	-1.489	-.339	-1.311	-.304	-.746
.250	-.354		-.465	-1.546	-.463	-1.381	-.437	-.780
.400	-.465		-.498	-1.892	-.473	-1.398	-.420	-.701
.550	-.413	-3.704	-.432	-2.100	-.401	-1.280	-.336	-.630
.700	-.385	-4.107	-.391	-2.016	-.348	-1.043	-.267	-.473
.850	-.445	-3.020	-.391	-1.485	-.291	-.735	-.172	-.305
.950	-.697	-2.297	-.443	-.810	-.276	-.451	-.128	-.164

ALPHA = 4.00 DEG Q = 904 N/SQ.M. (18.87 LBF/SQ.FT.) EFFECT. VELOC. RATIO = .202

	ETA = .250		ETA = .387		ETA = .524		ETA = .800	
X/C	CP		CP		CP		CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			.651		.943		.819	
.010			-.517	-.046				
.015					-.766	-.330		
.025			-.880	-.697	-.894	-.740	-.885	.024
.050			-.765	-.742	-.712	-.618	-.829	-.189
.100			-.569	-1.011	-.662	-.784	-.627	-.296
.150			-.655	-1.208	-.596	-.986	-.639	-.426
.250	-.453		-.603	-1.527	-.644	-1.146	-.608	-.533
.400	-.523		-.593	-1.867	-.605	-1.306	-.545	-.593
.550	-.449	-3.602	-.473	-2.118	-.501	-1.215	-.410	-.530
.700	-.402	-3.837	-.403	-1.958	-.381	-1.051	-.284	-.448
.850	-.428	-2.956	-.401	-1.370	-.302	-.708	-.190	-.303
.950	-.594	-1.986	-.443	-.803	-.281	-.454	-.132	-.158

ALPHA = 7.98 DEG Q = 904 N/SQ.M. (18.88 LBF/SQ.FT.) EFFECT. VELOC. RATIO = .201

	ETA = .250		ETA = .387		ETA = .524		ETA = .800	
X/C	CP		CP		CP		CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			.940		.682		-1.018	
.010			-.469	.743				
.015					-1.921	.469		
.025			-1.806	.080	-1.830	.139	-1.693	.499
.050			-1.321	-.159	-1.443	-.106	-1.475	.204
.100			-.952	-.560	-1.081	-.389	-1.008	.007
.150			-.927	-.884	-.926	-.690	-.887	-.195
.250	-.532		-.788	-1.327	-.826	-.914	-.819	-.344
.400	-.576		-.679	-1.769	-.719	-1.135	-.648	-.457
.550	-.479	-3.472	-.536	-2.043	-.550	-1.136	-.479	-.457
.700	-.408	-3.585	-.439	-1.845	-.418	-.969	-.328	-.385
.850	-.408	-2.902	-.405	-1.328	-.320	-.691	-.226	-.294
.950	-.547	-2.056	-.416	-.740	-.287	-.439	-.150	-.190

TABLE 10 (C) - CONTINUED

PRESSURE PROFILE ON WING

FORWARD NOZZLES DEFLECTED 90 DEGREES

ALPHA = 10.02 DEG Q = 905 N/SQ.M. (18.90 LBF/SQ.FT.) EFFECT. VELOC. RATIO = .202

X/C	ETA = .250 CP		ETA = .387 CP		ETA = .524 CP		ETA = .800 CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			.788		.175		-1.802	
.010			-.374	.917				
.015					-2.856	.666		
.025			-2.344	.424	-2.405	.473	-2.788	.760
.050			-1.527	.053	-1.661	.105	-1.638	.461
.100			-1.083	-.456	-1.191	-.243	-1.245	.185
.150			-1.053	-.672	-1.130	-.523	-1.062	-.008
.250	-.588		-.861	-1.240	-.926	-.829	-.874	-.235
.400	-.607		-.725	-1.789	-.748	-1.113	-.701	-.379
.550	-.495	-3.525	-.557	-1.994	-.561	-1.048	-.502	-.401
.700	-.405	-3.268	-.447	-1.807	-.425	-.943	-.346	-.359
.850	-.406	-2.668	-.394	-1.216	-.318	-.677	-.235	-.262
.950	-.507	-1.524	-.403	-.659	-.271	-.417	-.160	-.180

ALPHA = 12.00 DEG Q = 907 N/SQ.M. (18.94 LBF/SQ.FT.) EFFECT. VELOC. RATIO = .202

X/C	ETA = .250 CP		ETA = .387 CP		ETA = .524 CP		ETA = .800 CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			.274		-1.219		-3.417	
.010			-.254	1.020				
.015					-3.577	.883		
.025			-3.071	.582	-2.917	.641	-2.941	.871
.050			-1.806	.241	-2.002	.328	-1.988	.624
.100			-1.238	-.301	-1.384	-.026	-1.450	.332
.150			-1.151	-.685	-1.184	-.350	-1.215	.076
.250	-.593		-.930	-1.215	-1.022	-.707	-1.001	-.151
.400	-.623		-.765	-1.818	-.809	-1.038	-.762	-.325
.550	-.502	-3.352	-.589	-1.973	-.589	-1.046	-.546	-.361
.700	-.399	-3.005	-.462	-1.631	-.447	-.926	-.370	-.328
.850	-.376	-2.455	-.397	-1.111	-.313	-.671	-.247	-.251
.950	-.461	-1.412	-.396	-.637	-.268	-.426	-.170	-.179

ALPHA = 13.98 DEG Q = 908 N/SQ.M. (18.96 LBF/SQ.FT.) EFFECT. VELOC. RATIO = .202

X/C	ETA = .250 CP		ETA = .387 CP		ETA = .524 CP		ETA = .800 CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			-.372		-2.492		-5.145	
.010			-.214	1.000				
.015					-4.602	1.000		
.025			-3.656	.800	-3.913	.819	-3.333	.977
.050			-2.166	.460	-2.377	.507	-2.320	.792
.100			-1.490	-.054	-1.730	.108	-1.667	.446
.150			-1.256	-.490	-1.414	-.215	-1.501	.229
.250	-.643		-1.028	-1.146	-1.112	-.626	-1.099	-.058
.400	-.654		-.809	-1.745	-.863	-.946	-.813	-.266
.550	-.514	-3.003	-.603	-1.910	-.636	-.986	-.577	-.297
.700	-.403	-2.904	-.464	-1.567	-.467	-.887	-.381	-.294
.850	-.352	-2.099	-.396	-1.082	-.331	-.679	-.251	-.274
.950	-.424	-1.112	-.385	-.606	-.276	-.420	-.175	-.191

TABLE 10 (C) - CONCLUDED

PRESSURE PROFILE ON WING

FORWARD NOZZLES DEFLECTED 90 DEGREES

ALPHA = 16.01 DEG Q = 911 N/SQ.M. (19.03 LBF/SQ.FT.) EFFECT. VELOC. RATIO = .202

	ETA = .250		ETA = .387		ETA = .524		ETA = .800	
X/C	CP		CP		CP		CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			-.857		-2.892		-1.620	
.010			-.261	.930				
.015					-5.213	0.000		
.025			-2.485	.866	-3.526	.857	-1.496	.904
.050			-2.383	.533	-2.267	.566	-1.622	.672
.100			-1.606	.018	-1.707	.170	-1.420	.404
.150			-1.477	-.401	-1.449	-.157	-1.400	.182
.250	-.646		-1.042	-1.062	-1.143	-.584	-1.423	-.054
.400	-.653		-.796	-1.727	-.963	-.956	-1.130	-.257
.550	-.510	-2.972	-.596	-1.882	-.683	-1.003	-.937	-.344
.700	-.404	-2.640	-.463	-1.583	-.508	-.904	-.626	-.326
.850	-.352	-2.066	-.450	-1.068	-.424	-.744	-.429	-.310
.950	-.410	-1.029	-.390	-.626	-.387	-.525	-.403	-.341

ALPHA = 17.99 DEG Q = 915 N/SQ.M. (19.11 LBF/SQ.FT.) EFFECT. VELOC. RATIO = .202

	ETA = .250		ETA = .387		ETA = .524		ETA = .800	
X/C	CP		CP		CP		CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			.181		-.663		-1.219	
.010			-.196	.994				
.015					-1.328	.960		
.025			-1.405	.823	-1.157	.792	-1.059	.898
.050			-1.547	.517	-1.533	.503	-1.140	.689
.100			-1.336	.066	-1.359	.136	-1.164	.426
.150			-1.516	-.393	-1.054	-.141	-1.185	.221
.250	-.763		-1.268	-1.012	-1.216	-.586	-1.206	-.038
.400	-.759		-.970	-1.718	-1.142	-.927	-1.160	-.249
.550	-.586	-2.860	-.908	-1.917	-.899	-1.023	-1.042	-.346
.700	-.515	-2.644	-.633	-1.573	-.791	-.917	-1.008	-.379
.850	-.414	-2.063	-.594	-1.117	-.697	-.808	-.835	-.384
.950	-.436	-1.048	-.517	-.628	-.580	-.612	-.739	-.430

ALPHA = 20.02 DEG Q = 909 N/SQ.M. (18.99 LBF/SQ.FT.) EFFECT. VELOC. RATIO = .202

	ETA = .250		ETA = .387		ETA = .524		ETA = .800	
X/C	CP		CP		CP		CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			.300		-.086		-1.179	
.010			-.156	1.007				
.015					-.846	.950		
.025			-.787	.824	-.780	.768	-1.040	.917
.050			-.768	.551	-.768	.538	-.995	.737
.100			-.858	.099	-.903	.181	-.969	.444
.150			-1.003	-.296	-.724	-.151	-.943	.253
.250	-.882		-.867	-1.001	-.867	-.508	-1.049	.002
.400	-.874		-1.003	-1.617	-1.001	-.915	-1.134	-.250
.550	-.918	-2.877	-.985	-1.833	-.990	-1.039	-1.169	-.368
.700	-.877	-2.763	-.917	-1.637	-.874	-.981	-1.054	-.431
.850	-.678	-2.279	-.829	-1.253	-.756	-.840	-.986	-.522
.950	-.583	-1.254	-.659	-.759	-.665	-.723	-.808	-.572

TABLE 10 (D) -

PRESSURE PROFILE ON WING

FORWARD NOZZLES DEFLECTED 90 DEGREES

ALPHA = -.01 DEG Q = 224 N/SQ.M. (4.68 LBF/SQ.FT.) EFFECT. VELOC. RATIO = .100

X/C	ETA = .250 CP		ETA = .387 CP		ETA = .524 CP		ETA = .800 CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			-2.376		-2.095		.234	
.010			-3.213	-3.741				
.015					.860	-3.649		
.025			.572	-3.115	.706	-2.693	.473	-1.567
.050			.291	-2.102	.403	-2.144	.234	-1.454
.100			.079	-1.961	.164	-1.898	.108	-1.159
.150			-.068	-2.024	.016	-1.806	.016	-1.138
.250	-.181		-.181	-1.890	-.132	-1.723	-.153	-1.007
.400	-.297		-.267	-2.105	-.243	-1.549	-.219	-.901
.550	-.280	-3.114	-.253	-2.193	-.215	-1.368	-.185	-.744
.700	-.263	-3.881	-.253	-1.890	-.212	-1.082	-.130	-.522
.850	-.369	-3.435	-.311	-1.481	-.198	-.734	-.086	-.311
.950	-.679	-2.585	-.458	-.901	-.256	-.437	-.052	-.161

ALPHA = 4.02 DEG Q = 225 N/SQ.M. (4.69 LBF/SQ.FT.) EFFECT. VELOC. RATIO = .100

X/C	ETA = .250 CP		ETA = .387 CP		ETA = .524 CP		ETA = .800 CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			-1.310		-.363			
.010			-2.901	-1.891				
.015					.534	-2.018		
.025			.170	-1.702	.205	-1.990	.072	-.994
.050			-.069	-1.639	-.005	-1.520	-.013	-.875
.100			-.057	-1.534	-.062	-1.394	-.132	-.798
.150			-.237	-1.793	-.167	-1.576	-.195	-.791
.250	-.209		-.272	-1.948	-.251	-1.533	-.272	-.864
.400	-.282		-.303	-2.162	-.316	-1.560	-.289	-.785
.550	-.235	-4.727	-.259	-2.515	-.255	-1.421	-.204	-.626
.700	-.225	-5.305	-.232	-2.454	-.215	-1.183	-.133	-.514
.850	-.276	-3.919	-.272	-1.784	-.181	-.721	-.075	-.282
.950	-.544	-2.447	-.344	-.911	-.201	-.418	-.048	-.109

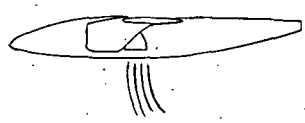
ALPHA = 8.00 DEG Q = 223 N/SQ.M. (4.65 LBF/SQ.FT.) EFFECT. VELOC. RATIO = .100

X/C	ETA = .250 CP		ETA = .387 CP		ETA = .524 CP		ETA = .800 CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			.412		.879			
.010			-2.650	-.621				
.015					-.041	-1.285		
.025			-.255	-1.243	-.295	-1.314	-.621	-.338
.050			-.288	-.988	-.309	-.974	-.430	-.422
.100			-.295	-1.292	-.323	-1.137	-.323	-.458
.150			-.373	-1.441	-.352	-1.285	-.394	-.571
.250	-.224		-.380	-1.686	-.359	-1.460	-.415	-.675
.400	-.298		-.329	-2.334	-.366	-1.566	-.346	-.692
.550	-.229	-6.044	-.264	-2.811	-.298	-1.433	-.240	-.596
.700	-.188	-6.232	-.233	-2.440	-.229	-1.186	-.171	-.445
.850	-.243	-4.230	-.243	-1.583	-.181	-.723	-.041	-.284
.950	-.430	-2.355	-.315	-.912	-.181	-.366	-.003	-.054

TABLE 10 (D) - CONTINUED

PRESSURE PROFILE ON WING

FORWARD NOZZLES DEFLECTED 90 DEGREES



ALPHA = 10.01 DEG Q = 222 N/SQ.M. (4.65 LBF/SQ.FT.) EFFECT. VELOC. RATIO = .100

X/C	ETA = .250 CP UPPER	ETA = .250 CP LOWER	ETA = .387 CP UPPER	ETA = .387 CP LOWER	ETA = .524 CP UPPER	ETA = .524 CP LOWER	ETA = .800 CP UPPER	ETA = .800 CP LOWER
.000			.913				.963	
.010			-2.304	-.291				
.015					-.539	-.617		
.025			-.681	-.681	-.504	-.646	-.738	.084
.050			-.532	-.596	-.547	-.773	-.617	-.079
.100			-.447	-.844	-.412	-.901	-.454	-.306
.150			-.433	-1.255	-.454	-1.156	-.461	-.426
.250	-.242		-.405	-1.557	-.412	-1.286	-.447	-.541
.400	-.286		-.365	-2.230	-.389	-1.464	-.362	-.606
.550	-.214	-5.747	-.286	-2.745	-.266	-1.358	-.252	-.534
.700	-.166	-6.265	-.225	-2.498	-.204	-1.190	-.115	-.389
.850	-.183	-3.570	-.225	-1.557	-.125	-.699	-.012	-.256
.950	-.451	-2.048	-.321	-.791	-.139	-.345	.016	-.070

ALPHA = 11.99 DEG Q = 224 N/SQ.M. (4.68 LBF/SQ.FT.) EFFECT. VELOC. RATIO = .100

X/C	ETA = .250 CP UPPER	ETA = .250 CP LOWER	ETA = .387 CP UPPER	ETA = .387 CP LOWER	ETA = .524 CP UPPER	ETA = .524 CP LOWER	ETA = .800 CP UPPER	ETA = .800 CP LOWER
.000							.721	
.010			-1.903	.327				
.015					-.925	-.039		
.025			-.967	-.201	-.911	-.391	-1.171	.383
.050			-.686	-.405	-.693	-.489	-.862	.186
.100			-.503	-.728	-.566	-.721	-.616	-.074
.150			-.517	-1.066	-.503	-.897	-.573	-.250
.250	-.222		-.412	-1.476	-.440	-1.179	-.503	-.395
.400	-.258		-.371	-2.117	-.395	-1.370	-.371	-.494
.550	-.190	-5.534	-.248	-2.789	-.279	-1.302	-.211	-.426
.700	-.143	-6.250	-.166	-2.448	-.163	-1.067	-.088	-.330
.850	-.160	-3.972	-.173	-1.486	-.098	-.623	.004	-.200
.950	-.340	-1.756	-.238	-.719	-.102	-.303	.055	-.085

ALPHA = 14.02 DEG Q = 224 N/SQ.M. (4.69 LBF/SQ.FT.) EFFECT. VELOC. RATIO = .100

X/C	ETA = .250 CP UPPER	ETA = .250 CP LOWER	ETA = .387 CP UPPER	ETA = .387 CP LOWER	ETA = .524 CP UPPER	ETA = .524 CP LOWER	ETA = .800 CP UPPER	ETA = .800 CP LOWER
.000							-.125	
.010			-1.550	.809				
.015					-1.417	.297		
.025			-1.269	.156	-1.438	-.090	-1.543	.648
.050			-.862	-.195	-1.017	-.272	-1.305	.360
.100			-.658	-.658	-.708	-.518	-.757	.142
.150			-.602	-.722	-.672	-.806	-.672	-.118
.250	-.237		-.462	-1.467	-.525	-1.082	-.560	-.286
.400	-.272		-.405	-2.287	-.442	-1.348	-.415	-.415
.550	-.190	-5.620	-.241	-2.832	-.303	-1.273	-.241	-.422
.700	-.122	-6.131	-.177	-2.509	-.163	-1.069	-.105	-.275
.850	-.146	-4.112	-.180	-1.498	-.088	-.643	.007	-.163
.950	-.337	-1.494	-.221	-.752	-.088	-.292	.058	-.092

TABLE 10 (D) - CONCLUDED

PRESSURE PROFILE ON WING

FORWARD NOZZLES DEFLECTED 90 DEGREES

ALPHA = 15.98 DEG Q = 224 N/SQ.M. (4.68 LBF/SQ.FT.) EFFECT. VELOC. RATIO = .100

	ETA = .250		ETA = .387		ETA = .524		ETA = .800	
X/C	CP		CP		CP		CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000					-.796		-1.068	
.010			-1.237	1.127				
.015					-2.039	.642		
.025			-1.722	.388	-1.772	.276	-1.997	.888
.050			-1.195	.079	-1.251	.008	-1.736	.578
.100			-.787	-.386	-.920	-.259	-.977	.255
.150			-.660	-.751	-.780	-.561	-.815	-.020
.250	-.252		-.540	-1.289	-.597	-1.040	-.667	-.221
.400	-.276		-.416	-2.329	-.501	-1.268	-.467	-.334
.550	-.143	-6.752	-.255	-2.789	-.310	-1.210	-.255	-.327
.700	-.092	-6.329	-.180	-2.479	-.180	-.982	-.112	-.235
.850	-.068	-3.004	-.153	-1.493	-.095	-.644	-.010	-.112
.950	-.286	-1.575	-.225	-.736	-.078	-.269	.048	-.064

ALPHA = 18.00 DEG Q = 226 N/SQ.M. (4.72 LBF/SQ.FT.) EFFECT. VELOC. RATIO = .101

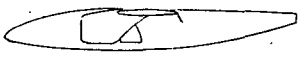
	ETA = .250		ETA = .387		ETA = .524		ETA = .800	
X/C	CP		CP		CP		CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			.990		.515		-1.954	
.010			-.914	1.192				
.015					-2.358	.899		
.025			-1.940	.487	-2.142	.634	-2.749	
.050			-1.466	.299	-1.723	.160	-1.556	.746
.100			-.866	-.156	-.963	-.140	-1.068	.390
.150			-.789	-.656	-.859	-.503	-.900	.139
.250	-.266		-.587	-1.242	-.649	-.833	-.726	-.076
.400	-.292		-.454	-2.219	-.512	-1.215	-.468	-.268
.550	-.174	-7.131	-.262	-2.865	-.309	-1.174	-.275	-.285
.700	-.089	-6.543	-.157	-2.469	-.164	-.985	-.116	-.191
.850	-.052	-2.587	-.137	-1.249	-.086	-.620	-.005	-.062
.950	-.258	-.992	-.201	-.556	-.059	-.245	.063	-.022

ALPHA = 19.99 DEG Q = 227 N/SC.M. (4.74 LBF/SQ.FT.) EFFECT. VELOC. RATIO = .101

	ETA = .250		ETA = .387		ETA = .524		ETA = .800	
X/C	CP		CP		CP		CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			.726		-.356		-3.367	
.010			-.668					
.015					-2.923	1.018		
.025			-2.535	.844	-2.715	.712	-3.576	
.050			-1.550	.532	-1.730	.532	-1.758	.851
.100			-1.008	-.037	-1.175	.081	-1.217	.546
.150			-.870	-.550	-1.064	-.308	-.932	.254
.250	-.273		-.613	-1.116	-.738	-.746	-.724	.027
.400	-.276		-.447	-2.192	-.531	-1.123	-.545	-.141
.550	-.161	-6.773	-.272	-2.717	-.313	-1.099	-.316	-.205
.700	-.064	-6.840	-.161	-2.206	-.168	-.988	-.141	-.138
.850	-.050	-2.418	-.121	-1.210	-.084	-.625	-.033	-.040
.950	-.205	-1.056	-.161	-.605	-.057	-.222	.027	.010

TABLE 11 (A)
PRESSURE PROFILE ON WING

FORWARD NOZZLES DEFLECTED 90 DEGREES WITH FLAPS



ALPHA = -.02 DEG Q = 2844 N/SQ.M. (59.39 LBF/SQ.FT.) EFFECT. VELOC. RATIO = ∞

	ETA = .250		ETA = .387		ETA = .524		ETA = .800	
X/C	CP		CP		CP		CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			.943		.930		.911	
.010			.837	.181				
.015					-.641	.117		
.025			-.833	-.060	-.796	.012	-.569	.018
.050			-.575	-.081	-.633	.087	-.458	-.036
.100			-.531	-.164	-.477	-.061	-.431	-.099
.150			-.544	-.304	-.531	-.171	-.416	-.178
.250	-.394		-.518	-.319	-.521	-.235	-.493	-.268
.400	-.475		-.534	-.307	-.522	-.237	-.437	-.252
.550	-.413	-.392	-.432	-.235	-.424	-.104	-.332	-.201
.700	-.357	-.373	-.343	-.097	-.340	.140	-.191	-.131
.850	-.344	-.353	-.308	.232	-.272	.434	-.094	-.041
.950	-.369	-.246	-.254	.407	-.259	.589	-.023	.026

ALPHA = 4.00 DEG Q = 2849 N/SQ.M. (59.51 LBF/SQ.FT.) EFFECT. VELOC. RATIO = ∞

	ETA = .250		ETA = .387		ETA = .524		ETA = .800	
X/C	CP		CP		CP		CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			.770		.090		-.396	
.010			.819	.925				
.015					-2.203	.813		
.025			-1.649	.587	-1.901	.607	-1.490	.610
.050			-1.149	.431	-1.282	.409	-1.304	.361
.100			-.813	.216	-.962	.227	-.851	.230
.150			-.815	.022	-.892	.063	-.754	.053
.250	-.510		-.718	-.154	-.783	.000	-.695	-.064
.400	-.564		-.631	-.204	-.662	-.080	-.551	-.121
.550	-.474	-.332	-.506	-.158	-.505	.007	-.401	-.118
.700	-.409	-.339	-.403	-.013	-.402	.200	-.253	-.076
.850	-.365	-.303	-.331	.297	-.313	.462	-.141	-.040
.950	-.365	-.125	-.257	.458	-.272	.605	-.051	-.003

ALPHA = 7.98 DEG Q = 2842 N/SQ.M. (59.37 LBF/SQ.FT.) EFFECT. VELOC. RATIO = ∞

	ETA = .250		ETA = .387		ETA = .524		ETA = .800	
X/C	CP		CP		CP		CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			-.735		-2.647		-3.110	
.010			.693	.923				
.015					-4.567	.979		
.025			-2.740	.898	-2.894	.916	-2.685	.893
.050			-1.829	.743	-1.990	.738	-1.740	.709
.100			-1.255	.511	-1.431	.518	-1.270	.445
.150			-1.081	.315	-1.219	.366	-1.047	.320
.250	-.616		-.899	.070	-.974	.151	-.887	.111
.400	-.632		-.746	-.109	-.785	.043	-.677	-.008
.550	-.531	-.270	-.577	-.088	-.583	.091	-.473	-.047
.700	-.438	-.226	-.449	.047	-.447	.237	-.302	-.035
.850	-.377	-.209	-.349	.323	-.335	.502	-.164	-.029
.950	-.351	-.037	-.264	.483	-.281	.637	-.067	-.018

TABLE 11 (A) - CONTINUED
PRESSURE PROFILE ON WING

FORWARD NOZZLES DEFLECTED 90 DEGREES WITH FLAPS

ALPHA = 9.99 DEG Q = 2840 N/SQ.M. (59.32 LBF/SQ.FT.) EFFECT. VELOC. RATIO = ∞

	ETA = .250		ETA = .387		ETA = .524		ETA = .800	
X/C	CP		CP		CP		CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			-.106		-.505		-1.241	
.010			.636	.957				
.015					-1.261	.972		
.025			-1.457	.941	-1.335	.884	-1.589	.859
.050			-1.317	.773	-1.298	.722	-1.579	.679
.100			-1.345	.536	-1.200	.526	-1.611	.460
.150			-1.189	.375	-1.506	.370	-1.489	.310
.250	-.744		-1.377	.124	-1.387	.189	-1.421	.136
.400	-.797		-.953	-.094	-1.079	.078	-.989	.011
.550	-.603	-.243	-.622	-.075	-.970	.103	-.633	-.036
.700	-.462	-.228	-.587	.062	-.676	.247	-.449	-.040
.850	-.365	-.120	-.449	.337	-.516	.513	-.411	-.060
.950	-.302	.056	-.388	.491	-.416	.638	-.176	-.075

ALPHA = 12.01 DEG Q = 2847 N/SQ.M. (59.47 LBF/SQ.FT.) EFFECT. VELOC. RATIO = ∞

	ETA = .250		ETA = .387		ETA = .524		ETA = .800	
X/C	CP		CP		CP		CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			.223		-.248		-1.127	
.010			.612	.959				
.015					-.844	.969		
.025			-.797	.933	-.842	.891	-1.219	.857
.050			-.779	.780	-.741	.731	-1.172	.713
.100			-.882	.584	-.853	.559	-1.126	.488
.150			-.774	.406	-.822	.387	-1.221	.330
.250	-.823		-.876	.126	-1.031	.201	-1.269	.187
.400	-.860		-.869	-.094	-1.022	.085	-1.183	.029
.550	-.879	-.263	-.906	-.097	-.914	.127	-1.031	-.029
.700	-.799	-.226	-.886	.046	-.901	.245	-.820	-.041
.850	-.686	-.132	-.740	.332	-.770	.514	-.606	-.096
.950	-.499	.045	-.673	.484	-.667	.644	-.403	-.159

ALPHA = 13.98 DEG Q = 2849 N/SQ.M. (59.50 LBF/SQ.FT.) EFFECT. VELOC. RATIO = ∞

	ETA = .250		ETA = .387		ETA = .524		ETA = .800	
X/C	CP		CP		CP		CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			.158		-.233		-1.223	
.010			.577	.938				
.015					-.708	.975		
.025			-.688	.952	-.696	.910	-1.061	.878
.050			-.687	.829	-.700	.762	-.918	.724
.100			-.709	.628	-.657	.581	-.987	.510
.150			-.697	.418	-.693	.426	-.958	.363
.250	-.672		-.732	.140	-.704	.235	-1.083	.189
.400	-.759		-.779	-.107	-.801	.110	-1.183	.034
.550	-.788	-.257	-.742	-.057	-.770	.125	-1.070	-.034
.700	-.777	-.238	-.774	.056	-.730	.235	-.926	-.065
.850	-.714	-.131	-.718	.326	-.724	.513	-.678	-.141
.950	-.585	.037	-.727	.479	-.726	.645	-.552	-.238

TABLE 11 (A) - CONCLUDED

PRESSURE PROFILE ON WING

FORWARD NOZZLES DEFLECTED 90 DEGREES WITH FLAPS

ALPHA = 15.98 DEG Q = 2839 N/SQ.M. (59.29 LBF/SQ.FT.) EFFECT. VELOC. RATIO = ∞

	ETA = .250		ETA = .387		ETA = .524		ETA = .800	
X/C	CP		CP		CP		CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			.019		-.412		-.869	
.010			.539	.858				
.015					-.670	.985		
.025			-.705	.980	-.685	.946	-.756	.873
.050			-.685	.860	-.677	.813	-.768	.742
.100			-.667	.657	-.666	.616	-.768	.535
.150			-.672	.501	-.673	.451	-.735	.392
.250	-.710		-.717	.166	-.722	.284	-.770	.218
.400	-.747		-.705	-.100	-.736	.140	-.789	.065
.550	-.775	-.266	-.729	-.107	-.713	.140	-.823	-.043
.700	-.726	-.240	-.711	.043	-.725	.263	-.857	-.105
.850	-.678	-.178	-.732	.311	-.704	.528	-.811	-.214
.950	-.627	.066	-.756	.469	-.758	.655	-.751	-.360

ALPHA = 17.99 DEG Q = 2842 N/SQ.M. (59.36 LBF/SQ.FT.) EFFECT. VELOC. RATIO = ∞

	ETA = .250		ETA = .387		ETA = .524		ETA = .800	
X/C	CP		CP		CP		CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			-.166		-.514		-.869	
.010			-.467	.789				
.015					-.666	.970		
.025			-.673	.983	-.687	.956	-.682	.900
.050			-.671	.893	-.679	.851	-.693	.792
.100			-.683	.721	-.675	.679	-.684	.587
.150			-.695	.541	-.680	.523	-.680	.440
.250	-.708		-.701	.232	-.691	.312	-.739	.251
.400	-.731		-.710	-.063	-.735	.180	-.707	.085
.550	-.730	-.255	-.711	-.064	-.733	.174	-.743	-.016
.700	-.730	-.228	-.741	.053	-.722	.264	-.770	-.090
.850	-.710	-.116	-.738	.311	-.746	.529	-.729	-.227
.950	-.675	.074	-.747	.466	-.769	.649	-.709	-.374

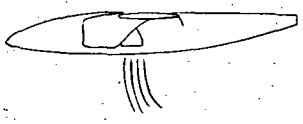
ALPHA = 19.98 DEG Q = 2843 N/SQ.M. (59.38 LBF/SQ.FT.) EFFECT. VELOC. RATIO = ∞

	ETA = .250		ETA = .387		ETA = .524		ETA = .800	
X/C	CP		CP		CP		CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			-.378		-.596		-.900	
.010			.416	.701				
.015					-.673	.958		
.025			-.666	.983	-.668	.979	-.679	.919
.050			-.693	.938	-.665	.896	-.679	.819
.100			-.677	.778	-.670	.734	-.690	.635
.150			-.705	.598	-.697	.582	-.670	.503
.250	-.725		-.695	.297	-.698	.393	-.696	.306
.400	-.753		-.723	.026	-.721	.220	-.707	.143
.550	-.741	-.232	-.731	-.047	-.732	.217	-.757	.011
.700	-.731	-.263	-.730	.068	-.751	.301	-.755	-.082
.850	-.706	-.103	-.750	.318	-.764	.543	-.723	-.210
.950	-.703	.073	-.775	.467	-.763	.665	-.697	-.378

TABLE 11 (B)

PRESSURE PROFILE ON WING

FORWARD NOZZLES DEFLECTED 90 DEGREES WITH FLAPS



ALPHA = -.02 DEG Q = 2038 N/SC.M. (42.57 LBF/SQ.FT.) EFFECT. VELOC. RATIO = .302

	ETA = .250		ETA = .387		ETA = .524		ETA = .800	
X/C	CP		CP		CP		CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			.855		.955		.848	
.010			.717	-.181				
.015					-.717	-.191		
.025			-.812	-.443	-1.012	-.386	-.684	-.100
.050			-.672	-.534	-.541	-.414	-.787	-.148
.100			-.589	-.725	-.610	-.464	-.544	-.257
.150			-.635	-.918	-.581	-.567	-.520	-.367
.250	-.458		-.581	-1.065	-.563	-.811	-.528	-.367
.400	-.546		-.614	-1.282	-.617	-.775	-.493	-.434
.550	-.519	-1.978	-.530	-1.229	-.528	-.605	-.393	-.376
.700	-.466	-1.966	-.468	-.861	-.462	-.288	-.278	-.285
.850	-.471	-1.559	-.463	-.307	-.430	.149	-.194	-.196
.950	-.537	-.955	-.452	-.037	-.476	.368	-.119	-.097

ALPHA = 3.96 DEG Q = 2042 N/SQ.M. (42.64 LBF/SQ.FT.) EFFECT. VELOC. RATIO = .303

	ETA = .250		ETA = .387		ETA = .524		ETA = .800	
X/C	CP		CP		CP		CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			.857		.331		-.457	
.010			.708	.754				
.015					-2.055	.687		
.025			-1.751	.304	-1.741	.457	-1.740	.469
.050			-1.190	.034	-1.348	.112	-1.264	.248
.100			-.963	-.294	-1.035	-.128	-.962	.061
.150			-.948	-.540	-.935	-.334	-.870	-.068
.250	-.572		-.855	-.870	-.833	-.547	-.755	-.245
.400	-.646		-.745	-1.169	-.766	-.646	-.641	-.301
.550	-.571	-1.764	-.617	-1.107	-.625	-.516	-.489	-.298
.700	-.523	-1.757	-.529	-.775	-.542	-.227	-.346	-.239
.850	-.493	-1.325	-.508	-.231	-.490	.191	-.230	-.190
.950	-.520	-.717	-.476	.032	-.496	.403	-.144	-.122

ALPHA = 8.01 DEG Q = 2039 N/SC.M. (42.59 LBF/SQ.FT.) EFFECT. VELOC. RATIO = .302

	ETA = .250		ETA = .387		ETA = .524		ETA = .800	
X/C	CP		CP		CP		CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			-.037		-1.637		-3.574	
.010			.660	1.012				
.015					-4.371	.938		
.025			-3.244	.752	-3.567	.817	-2.889	.872
.050			-1.964	.524	-2.133	.568	-1.901	.647
.100			-1.383	.113	-1.557	.259	-1.411	.327
.150			-1.234	-.214	-1.334	.007	-1.175	.166
.250	-.684		-1.033	-.636	-1.102	-.304	-1.036	-.040
.400	-.740		-.861	-1.036	-.917	-.495	-.790	-.184
.550	-.640	-1.677	-.690	-1.043	-.713	-.429	-.569	-.225
.700	-.551	-1.662	-.576	-.708	-.591	-.179	-.412	-.194
.850	-.513	-1.200	-.526	-.191	-.521	.206	-.268	-.179
.950	-.518	-.568	-.490	.069	-.510	.432	-.165	-.134

TABLE 11 (B) - CONTINUED

PRESSURE PROFILE ON WING

FORWARD NOZZLES DEFLECTED 90 DEGREES WITH FLAPS

ALPHA = 10.00 DEG Q = 2039 N/SQ.M. (42.60 LBF/SQ.FT.) EFFECT. VELOC. RATIO = .302

X/C	ETA = .250 CP		ETA = .387 CP		ETA = .524 CP		ETA = .800 CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			-1.508		-3.320		-5.230	
.010			.532	.902				
.015					-5.422	.997		
.025			-3.865	.943	-3.753	.950	-3.447	.958
.050			-2.355	.685	-2.653	.720	-2.263	.791
.100			-1.637	.294	-1.862	.387	-1.630	.489
.150			-1.403	-.040	-1.529	.122	-1.379	.272
.250	-.760		-1.105	-.493	-1.227	-.171	-1.129	.047
.400	-.788		-.927	-.982	-.974	-.402	-.850	-.122
.550	-.672	-1.620	-.738	-1.001	-.756	-.390	-.613	-.177
.700	-.571	-1.618	-.603	-.674	-.614	-.179	-.420	-.171
.850	-.506	-1.239	-.536	-.177	-.528	.225	-.274	-.169
.950	-.523	-.539	-.500	.075	-.521	.449	-.183	-.150

ALPHA = 11.99 DEG Q = 2039 N/SQ.M. (42.59 LBF/SQ.FT.) EFFECT. VELOC. RATIO = .302

X/C	ETA = .250 CP		ETA = .387 CP		ETA = .524 CP		ETA = .800 CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			-.173		-.830		-1.955	
.010			.503	.962				
.015					-1.574	1.003		
.025			-1.778	.947	-1.395	.885	-1.756	.919
.050			-1.323	.715	-1.583	.667	-1.759	.732
.100			-1.614	.369	-1.422	.385	-1.837	.485
.150			-1.744	.043	-1.537	.139	-1.694	.296
.250	-1.021		-1.592	-.475	-1.527	-.160	-1.666	.091
.400	-1.012		-1.412	-.936	-1.426	-.364	-1.433	-.104
.550	-.796	-1.628	-1.160	-.941	-1.215	-.331	-1.088	-.166
.700	-.668	-1.585	-.866	-.627	-.887	-.117	-.716	-.177
.850	-.533	-1.075	-.663	-.134	-.793	.270	-.505	-.193
.950	-.487	-.501	-.740	.099	-.703	.460	-.346	-.198

ALPHA = 14.05 DEG Q = 2037 N/SQ.M. (42.55 LBF/SQ.FT.) EFFECT. VELOC. RATIO = .302

X/C	ETA = .250 CP		ETA = .387 CP		ETA = .524 CP		ETA = .800 CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			.029		-.523		-1.952	
.010			.478	.966				
.015					-1.136	1.005		
.025			-1.099	.959	-1.056	.904	-1.375	.932
.050			-.985	.744	-1.116	.709	-1.529	.775
.100			-1.147	.400	-1.183	.433	-1.540	.529
.150			-1.060	.062	-1.095	.187	-1.569	.342
.250	-1.056		-1.088	-.428	-1.223	-.105	-1.470	.125
.400	-1.024		-1.305	-.914	-1.244	-.357	-1.547	-.078
.550	-1.033	-1.531	-1.179	-.931	-1.243	-.312	-1.328	-.158
.700	-.982	-1.606	-1.099	-.626	-1.116	-.128	-1.098	-.195
.850	-.829	-1.035	-.961	-.135	-.925	.278	-.719	-.242
.950	-.641	-.490	-.830	.104	-.846	.487	-.659	-.281

TABLE 11 (B) - CONCLUDED

PRESSURE PROFILE ON WING

FORWARD NOZZLES DEFLECTED 90 DEGREES WITH FLAPS

ALPHA = 15.99 DEG Q = 2045 N/SQ.M. (42.72 LBF/SQ.FT.) EFFECT. VELOC. RATIO = .302

	ETA = .250		ETA = .387		ETA = .524		ETA = .800	
X/C	CP		CP		CP		CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			-.036		-.490		-1.519	
.010			.454	.970				
.015					-.910			
.025			-.937	.976	-.934	.925	-1.229	.938
.050			-.890	.788	-.897	.751	-1.289	.788
.100			-.873	.443	-.915	.484	-1.319	.556
.150			-.910	.174	-.902	.220	-1.226	.350
.250	-.882		-.897	-.380	-.925	-.061	-1.244	.134
.400	-.947		-.947	-.890	-.971	-.332	-1.278	-.066
.550	-.997	-1.514	-.984	-.894	-.984	-.305	-1.238	-.176
.700	-1.012	-1.536	-1.019	-.604	-1.000	-.111	-1.146	-.229
.850	-.865	-1.036	-.966	-.127	-.899	.274	-1.026	-.343
.950	-.747	-.458	-.946	.111	-.943	.488	-.834	-.391

ALPHA = 17.96 DEG Q = 2037 N/SQ.M. (42.55 LBF/SQ.FT.) EFFECT. VELOC. RATIO = .301

	ETA = .250		ETA = .387		ETA = .524		ETA = .800	
X/C	CP		CP		CP		CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			-.215		-.617		-1.303	
.010			.318	.875				
.015					-.921			
.025			-.909	.989	-.896	.969	-1.007	.943
.050			-.878	.868	-.875	.804	-1.040	.804
.100			-.870	.529	-.863	.526	-1.089	.583
.150			-.883	.184	-.891	.290	-1.069	.372
.250	-.906		-.907	-.352	-.871	-.029	-1.055	.161
.400	-.934		-.906	-.860	-.933	-.307	-1.027	-.060
.550	-.948	-1.553	-.947	-.947	-.928	-.311	-1.111	-.187
.700	-.921	-1.508	-.935	-.634	-.919	-.127	-1.122	-.265
.850	-.920	-1.084	-.954	-.158	-.949	.266	-1.065	-.384
.950	-.818	-.493	-.966	.113	-.971	.491	-.970	-.543

ALPHA = 19.98 DEG Q = 2038 N/SQ.M. (42.57 LBF/SQ.FT.) EFFECT. VELOC. RATIO = .301

	ETA = .250		ETA = .387		ETA = .524		ETA = .800	
X/C	CP		CP		CP		CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			-.361		-.766		-1.252	
.010			.231	.815				
.015					-.842			
.025			-.875	1.010	-.866	.983	-.914	.953
.050			-.876	.899	-.870	.832	-.931	.828
.100			-.874	.588	-.849	.586	-.902	.609
.150			-.890	.260	-.878	.345	-.974	.432
.250	-.860		-.857	-.336	-.891	.007	-.934	.190
.400	-.944		-.941	-.891	-.925	-.284	-.964	-.030
.550	-.944	-1.521	-.948	-.967	-.929	-.303	-1.047	-.165
.700	-.948	-1.546	-.955	-.666	-.946	-.136	-1.004	-.253
.850	-.929	-1.053	-.941	-.177	-.908	.267	-.991	-.406
.950	-.869	-.509	-.956	.098	-.944	.484	-.909	-.594

TABLE 11 (C) -
PRESSURE PROFILE ON WING

FORWARD NOZZLES DEFLECTED 90 DEGREES WITH FLAPS

ALPHA = -.02 DEG Q = 909 N/SQ.M. (18.98 LBF/SQ.FT.) EFFECT. VELOC. RATIO = .201

	ETA = .250		ETA = .387		ETA = .524		ETA = .800	
X/C	CP		CP		CP		CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			.273		.656		1.027	
.010			-.550	-1.053				
.015					.006	-1.036		
.025			-.303	-1.239	-.306	-1.130	-.309	-.542
.050			-.348	-1.069	-.297	-.847	-.287	-.446
.100			-.351	-1.204	-.372	-.941	-.325	-.386
.150			-.504	-1.345	-.434	-1.000	-.355	-.576
.250	-.401		-.479	-1.555	-.533	-1.186	-.464	-.613
.400	-.515		-.549	-1.877	-.522	-1.151	-.439	-.597
.550	-.464	-4.379	-.478	-1.977	-.468	-.934	-.350	-.516
.700	-.440	-3.591	-.443	-1.385	-.445	-.502	-.277	-.364
.850	-.457	-1.883	-.459	-.659	-.449	-.018	-.132	-.212
.950	-.540	-.917	-.480	-.299	-.520	.229	-.095	-.090

ALPHA = 3.90 DEG Q = 912 N/SQ.M. (19.05 LBF/SQ.FT.) EFFECT. VELOC. RATIO = .201

	ETA = .250		ETA = .387		ETA = .524		ETA = .800	
X/C	CP		CP		CP		CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			.896		.960		.399	
.010			-.451	.191				
.015					-1.120	-.033		
.025			-1.080	-.465	-1.261	-.310	-1.329	.331
.050			-.883	-.544	-.933	-.394	-.931	-.018
.100			-.771	-.771	-.750	-.498	-.769	-.258
.150			-.757	-1.140	-.776	-.747	-.722	-.332
.250	-.519		-.760	-1.357	-.741	-.938	-.695	-.431
.400	-.614		-.681	-1.831	-.702	-1.068	-.579	-.468
.550	-.566	-4.074	-.570	-1.857	-.569	-.872	-.465	-.426
.700	-.497	-3.332	-.526	-1.335	-.500	-.487	-.333	-.344
.850	-.493	-1.819	-.517	-.587	-.488	-.012	-.228	-.258
.950	-.546	-.776	-.524	-.259	-.550	.229	-.148	-.141

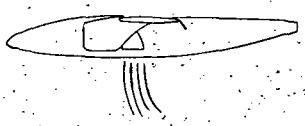
ALPHA = 8.02 DEG Q = 904 N/SQ.M. (18.87 LBF/SQ.FT.) EFFECT. VELOC. RATIO = .200

	ETA = .250		ETA = .387		ETA = .524		ETA = .800	
X/C	CP		CP		CP		CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			.808		.222		-1.196	
.010			-.388	.830				
.015					-2.636	.646		
.025			-2.080	.314	-2.251	.374	-2.563	.660
.050			-1.541	-.087	-1.666	.182	-1.693	.429
.100			-1.140	-.411	-1.241	-.254	-1.231	.112
.150			-1.044	-.758	-1.143	-.467	-1.067	-.061
.250	-.624		-.904	-1.162	-.969	-.741	-.939	-.207
.400	-.699		-.809	-1.744	-.842	-.893	-.748	-.359
.550	-.603	-3.901	-.649	-1.827	-.679	-.747	-.537	-.370
.700	-.530	-3.008	-.576	-1.237	-.577	-.424	-.399	-.309
.850	-.512	-1.324	-.544	-.508	-.529	.029	-.278	-.252
.950	-.527	-.512	-.532	-.202	-.562	.276	-.195	-.179

TABLE 11 (C) - CONTINUED

PRESSURE PROFILE ON WING

FORWARD NOZZLES DEFLECTED 90 DEGREES WITH FLAPS



ALPHA = 9.97 DEG Q = 913 N/SC.M. (19.06 LBF/SQ.FT.) EFFECT. VELOC. RATIO = .202

X/C	ETA = .250 CP		ETA = .387 CP		ETA = .524 CP		ETA = .800 CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			.064		-.910		-3.294	
.010			-1.146	.705				
.015					-3.639	.561		
.025			-3.166	.223	-3.270	.285	-3.368	.542
.050			-2.025	-.086	-2.372	.018	-2.210	.299
.100			-1.640	-.540	-1.830	-.367	-1.709	-.053
.150			-1.476	-.840	-1.562	-.606	-1.490	-.221
.250	-.968		-1.301	-1.313	-1.370	-.887	-1.334	-.442
.400	-1.026		-1.166	-1.987	-1.197	-1.087	-1.092	-.596
.550	-.922	-4.188	-.989	-2.002	-1.019	-1.004	-.869	-.620
.700	-.825	-3.282	-.889	-1.464	-.891	-.717	-.712	-.574
.850	-.791	-1.530	-.839	-.771	-.823	-.245	-.586	-.532
.950	-.795	-.729	-.830	-.459	-.841	-.007	-.499	-.472

ALPHA = 11.97 DEG Q = 912 N/SC.M. (19.04 LBF/SQ.FT.) EFFECT. VELOC. RATIO = .202

X/C	ETA = .250 CP		ETA = .387 CP		ETA = .524 CP		ETA = .800 CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			-.740		-2.041		-5.236	
.010			-1.127	.706				
.015					-4.557	.671		
.025			-3.999	.447	-4.317	.507	-3.655	.613
.050			-2.330	.103	-2.669	.106	-2.574	.419
.100			-1.762	-.379	-1.974	-.182	-1.952	.132
.150			-1.630	-.747	-1.801	-.509	-1.715	-.110
.250	-1.020		-1.387	-1.256	-1.475	-.788	-1.426	-.354
.400	-1.053		-1.157	-1.907	-1.248	-1.036	-1.139	-.522
.550	-.933	-4.238	-1.017	-1.956	-1.044	-.947	-.931	-.569
.700	-.852	-3.140	-.856	-1.442	-.904	-.729	-.744	-.556
.850	-.795	-1.347	-.841	-.723	-.832	-.232	-.608	-.532
.950	-.778	-.686	-.823	-.455	-.849	-.002	-.516	-.482

ALPHA = 13.97 DEG Q = 910 N/SC.M. (19.00 LBF/SQ.FT.) EFFECT. VELOC. RATIO = .202

X/C	ETA = .250 CP		ETA = .387 CP		ETA = .524 CP		ETA = .800 CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			-1.566		-3.051		-2.225	
.010			-1.166	.557				
.015					-5.718	.703		
.025			-3.926	.582	-4.148	.550	-2.257	.604
.050			-2.852	.273	-2.928	.299	-2.243	.440
.100			-2.018	-.168	-2.312	-.071	-2.288	.166
.150			-2.033	-.573	-2.028	-.369	-2.243	-.085
.250	-1.093		-1.507	-1.154	-1.607	-.714	-1.682	-.309
.400	-1.153		-1.271	-1.832	-1.358	-.953	-1.625	-.497
.550	-.972	-4.208	-1.070	-1.894	-1.110	-.897	-1.263	-.554
.700	-.869	-3.130	-.955	-1.351	-1.038	-.693	-.872	-.558
.850	-.804	-1.209	-.882	-.685	-.910	-.201	-.781	-.554
.950	-.771	-.563	-.836	-.416	-.970	.003	-.633	-.574

TABLE 11 (C) - CONCLUDED

PRESSURE PROFILE ON WING

FORWARD NOZZLES DEFLECTED 90 DEGREES WITH FLAPS

ALPHA = 15.99 DEG Q = 911 N/SQ.M. (19.04 LBF/SQ.FT.) EFFECT. VELOC. RATIO = .202

	ETA = .250		ETA = .387		ETA = .524		ETA = .800	
X/C	CP		CP		CP		CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			-.310		-.953		-2.124	
.010			-1.062	.670				
.015					-2.146	.665		
.025			-2.148	.549	-1.589	.502	-1.745	.604
.050			-1.987	.281	-1.883	.253	-1.838	.416
.100			-1.747	-.137	-1.700	-.079	-1.876	.160
.150			-1.805	-.523	-1.958	-.360	-1.866	-.039
.250	-1.325		-1.828	-1.109	-1.742	-.710	-1.925	-.266
.400	-1.378		-1.596	-1.762	-1.644	-.941	-1.911	-.473
.550	-1.102	-4.015	-1.437	-1.834	-1.486	-.871	-1.556	-.551
.700	-1.015	-2.935	-1.246	-1.328	-1.279	-.662	-1.412	-.550
.850	-.876	-1.114	-1.010	-.664	-1.080	-.185	-1.023	-.596
.950	-.763	-.547	-.944	-.367	-1.112	.050	-.981	-.596

ALPHA = 18.03 DEG Q = 911 N/SQ.M. (19.03 LBF/SQ.FT.) EFFECT. VELOC. RATIO = .202

	ETA = .250		ETA = .387		ETA = .524		ETA = .800	
X/C	CP		CP		CP		CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			-.181		-.760		-1.890	
.010			-.982	.667				
.015					-1.412	.641		
.025			-1.350	.552	-1.455	.502	-1.718	.609
.050			-1.563	.295	-1.428	.262	-1.710	.442
.100			-1.308	-.115	-1.295	-.039	-1.602	.193
.150			-1.412	-.509	-1.338	-.331	-1.739	-.001
.250	-1.355		-1.390	-1.086	-1.423	-.657	-1.661	-.251
.400	-1.396		-1.469	-1.759	-1.598	-.911	-1.777	-.457
.550	-1.415	-3.886	-1.464	-1.831	-1.471	-.868	-1.819	-.555
.700	-1.248	-2.855	-1.403	-1.288	-1.404	-.666	-1.606	-.599
.850	-1.024	-1.107	-1.241	-.645	-1.251	-.197	-1.361	-.644
.950	-.925	-.583	-1.095	-.386	-1.179	.026	-1.100	-.698

ALPHA = 20.01 DEG Q = 908 N/SQ.M. (18.95 LBF/SQ.FT.) EFFECT. VELOC. RATIO = .201

	ETA = .250		ETA = .387		ETA = .524		ETA = .800	
X/C	CP		CP		CP		CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			-.362		-.845		-1.890	
.010			-.933	.598				
.015					-1.373	.673		
.025			-1.331	.565	-1.320	.544	-1.602	.619
.050			-1.324	.346	-1.336	.305	-1.487	.461
.100			-1.289	-.083	-1.287	.013	-1.511	.225
.150			-1.357	-.449	-1.305	-.268	-1.393	-.016
.250	-1.334		-1.341	-1.061	-1.345	-.647	-1.503	-.234
.400	-1.420		-1.372	-1.741	-1.378	-.894	-1.662	-.451
.550	-1.449	-3.569	-1.348	-1.762	-1.377	-.863	-1.683	-.572
.700	-1.389	-2.807	-1.379	-1.252	-1.387	-.663	-1.592	-.626
.850	-1.220	-1.143	-1.348	-.676	-1.324	-.190	-1.515	-.744
.950	-1.098	-.576	-1.364	-.384	-1.349	.035	-1.340	-.862

TABLE 12 (D)

PRESSURE PROFILE ON WING

FORWARD NOZZLES DEFLECTED 90 DEGREES WITH FLAPS

ALPHA = -.01 DEG Q = 230 N/SQ.M. (4.79 LBF/SQ.FT.) EFFECT. VELOC. RATIO = .102

X/C	ETA = .250 CP		ETA = .387 CP		ETA = .524 CP		ETA = .800 CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			-2.218		-1.497		.137	
.010			-4.834	-2.019				
.015					.645	-2.966		
.025			.199	-2.774	.363	-2.547	.171	-1.696
.050			.137	-2.232	.075	-2.087	-.021	-1.291
.100			-.220	-1.902	-.062	-1.641	-.110	-1.078
.150			-.213	-1.833	-.172	-1.765	-.186	-1.113
.250	-.351		-.364	-1.941	-.351	-1.655	-.323	-1.013
.400	-.447		-.447	-2.134	-.434	-1.512	-.377	-.923
.550	-.427	-4.870	-.430	-2.447	-.440	-1.262	-.347	-.803
.700	-.427	-4.573	-.434	-2.018	-.434	-.736	-.274	-.563
.850	-.493	-3.179	-.500	-1.282	-.480	-.320	-.227	-.404
.950	-.647	-1.468	-.550	-.883	-.587	-.034	-.197	-.277

ALPHA = 3.94 DEG Q = 228 N/SQ.M. (4.75 LBF/SQ.FT.) EFFECT. VELOC. RATIO = .101

X/C	ETA = .250 CP		ETA = .387 CP		ETA = .524 CP		ETA = .800 CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			-.292		.012		.940	
.010			-4.703	-1.615				
.015					.151	-1.878		
.025			-.161	-1.788	-.043	-1.802	-.251	-.985
.050			-.237	-1.470	-.244	-1.393	-.251	-.729
.100			-.375	-1.546	-.285	-1.421	-.306	-.763
.150			-.459	-1.656	-.368	-1.483	-.375	-.777
.250	-.403		-.465	-1.724	-.486	-1.496	-.445	-.891
.400	-.465		-.529	-2.375	-.509	-1.529	-.462	-.804
.550	-.445	-5.765	-.475	-2.764	-.455	-1.277	-.375	-.683
.700	-.408	-6.138	-.452	-2.160	-.445	-.734	-.297	-.562
.850	-.452	-2.932	-.485	-1.167	-.469	-.244	-.240	-.344
.950	-.690	-1.113	-.532	-.667	-.549	.055	-.176	-.200

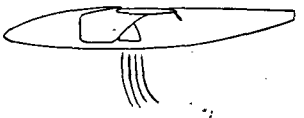
ALPHA = 8.01 DEG Q = 223 N/SQ.M. (4.65 LBF/SQ.FT.) EFFECT. VELOC. RATIO = .100

X/C	ETA = .250 CP		ETA = .387 CP		ETA = .524 CP		ETA = .800 CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			.666		.900		.794	
.010			-4.467	-.175				
.015					-.444	-.769		
.025			-.805	-.783	-.614	-.833	-.995	-.104
.050			-.670	-1.031	-.790	-.868	-.868	-.324
.100			-.649	-1.222	-.656	-1.052	-.691	-.437
.150			-.649	-1.427	-.677	-1.215	-.677	-.493
.250	-.436		-.628	-1.635	-.628	-1.305	-.677	-.675
.400	-.575		-.627	-2.444	-.644	-1.446	-.575	-.723
.550	-.503	-7.074	-.538	-2.738	-.558	-1.186	-.455	-.627
.700	-.452	-6.869	-.503	-2.032	-.497	-.726	-.370	-.490
.850	-.473	-2.269	-.500	-1.048	-.476	-.202	-.229	-.387
.950	-.531	-1.227	-.534	-.593	-.510	.120	-.181	-.215

TABLE 11 (D) - CONTINUED

PRESSURE PROFILE ON WING

FORWARD NOZZLES DEFLECTED 90 DEGREES WITH FLAPS



ALPHA = 10.01 DEG Q = 231 N/SC.M. (4.83 LBF/SQ.FT.) EFFECT. VELOC. RATIO = .102

	ETA = .250		ETA = .387		ETA = .524		ETA = .800	
X/C	CP		CP		CP		CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			.857		.945		.278	
.010			-3.956	-.076				
.015					-1.083	-.273		
.025			-1.274	-.614	-1.077	-.457	-1.362	.067
.050			-.920	-.702	-.913	-.620	-1.049	-.076
.100			-.757	-1.056	-.791	-.927	-.832	-.348
.150			-.757	-1.165	-.804	-1.090	-.797	-.484
.250	-.546		-.729	-1.776	-.729	-1.328	-.729	-.585
.400	-.575		-.677	-2.433	-.687	-1.394	-.648	-.658
.550	-.499	-7.016	-.579	-2.802	-.608	-1.172	-.499	-.582
.700	-.450	-6.604	-.503	-2.182	-.516	-.783	-.351	-.473
.850	-.473	-1.997	-.503	-1.021	-.466	-.192	-.272	-.374
.950	-.542	-.908	-.536	-.569	-.513	.085	-.206	-.249

ALPHA = 12.02 DEG Q = 227 N/SQ.M. (4.74 LBF/SQ.FT.) EFFECT. VELOC. RATIO = .101

	ETA = .250		ETA = .387		ETA = .524		ETA = .800	
X/C	CP		CP		CP		CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			.959		.910		-.283	
.010			-3.800	.348				
.015					-1.427	.015		
.025			-1.532	-.179	-1.615	-.227	-1.733	.307
.050			-1.275	-.463	-1.219	-.345	-1.427	.119
.100			-1.011	-.831	-1.032	-.713	-1.032	-.165
.150			-.900	-1.101	-.879	-.949	-.900	-.394
.250	-.560		-.762	-1.449	-.852	-1.173	-.831	-.524
.400	-.608		-.679	-2.182	-.722	-1.321	-.689	-.625
.550	-.531	-8.483	-.558	-2.679	-.595	-1.139	-.514	-.564
.700	-.460	-6.825	-.457	-2.007	-.487	-.786	-.379	-.470
.850	-.457	-2.249	-.463	-1.015	-.447	-.178	-.278	-.373
.950	-.507	-.816	-.490	-.581	-.463	.115	-.208	-.299

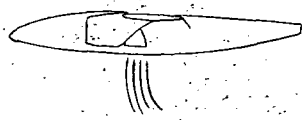
ALPHA = 14.01 DEG Q = 229 N/SQ.M. (4.79 LBF/SQ.FT.) EFFECT. VELOC. RATIO = .102

	ETA = .250		ETA = .387		ETA = .524		ETA = .800	
X/C	CP		CP		CP		CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			.886		.570		-1.157	
.010			-3.564	.707				
.015					-2.037	.384		
.025			-2.119	.054	-1.892	-.015	-2.140	.453
.050			-1.638	-.118	-1.549	-.139	-1.886	.219
.100			-1.060	-.661	-1.150	-.469	-1.177	-.043
.150			-.943	-.923	-1.060	-.847	-1.081	-.283
.250	-.586		-.847	-1.362	-.937	-1.059	-.930	-.442
.400	-.642		-.732	-2.339	-.782	-1.236	-.739	-.519
.550	-.529	-7.890	-.589	-2.539	-.612	-1.062	-.552	-.506
.700	-.476	-6.980	-.506	-1.976	-.522	-.799	-.416	-.442
.850	-.439	-2.069	-.476	-.932	-.462	-.166	-.309	-.352
.950	-.476	-.652	-.482	-.449	-.482	.098	-.229	-.319

TABLE 11 (D) - CONCLUDED

PRESSURE PROFILE ON WING

FORWARD NOZZLES DEFLECTED 90 DEGREES WITH FLAPS



ALPHA = 16.01 DEG Q = 227 N/SQ.M. (4.74 LBF/SQ.FT.) EFFECT. VELOC. RATIO = .101

	ETA = .250		ETA = .387		ETA = .524		ETA = .800	
X/C	CP		CP		CP		CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			.384		-.032		-2.474	
.010			-3.604	.828				
.015					-2.952	.551		
.025			-2.592	.329	-2.640	.273	-2.980	.620
.050			-1.884	.024	-2.113	.003	-1.912	.391
.100			-1.274	-.469	-1.440	-.476	-1.523	.030
.150			-1.218	-.816	-1.315	-.774	-1.274	-.178
.250	-.698		-1.003	-1.479	-1.093	-1.073	-1.086	-.417
.400	-.740		-.878	-2.239	-.938	-1.254	-.878	-.548
.550	-.619	-7.747	-.709	-2.700	-.760	-1.137	-.652	-.575
.700	-.525	-6.933	-.602	-2.024	-.609	-.938	-.508	-.471
.850	-.521	-1.950	-.562	-1.032	-.568	-.249	-.400	-.397
.950	-.535	-.760	-.528	-.609	-.541	.057	-.340	-.330

ALPHA = 18.01 DEG Q = 231 N/SQ.M. (4.82 LBF/SQ.FT.) EFFECT. VELOC. RATIO = .102

	ETA = .250		ETA = .387		ETA = .524		ETA = .800	
X/C	CP		CP		CP		CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			.037		-.995		-4.266	
.010			-3.426					
.015					-3.399	.672		
.025			-3.454	.474	-3.378	.440	-3.898	.781
.050			-1.951	.166	-2.108	.187	-2.402	.487
.100			-1.473	-.305	-1.698	-.223	-1.678	.153
.150			-1.282	-.694	-1.411	-.551	-1.377	-.080
.250	-.735		-1.063	-1.213	-1.179	-.858	-1.166	-.213
.400	-.719		-.839	-2.199	-.915	-1.054	-.852	-.425
.550	-.590	-7.894	-.660	-2.557	-.716	-.971	-.647	-.504
.700	-.504	-5.265	-.551	-1.885	-.590	-.776	-.481	-.441
.850	-.445	-1.494	-.501	-.839	-.504	-.190	-.372	-.349
.950	-.448	-.564	-.504	-.461	-.491	.098	-.286	-.279

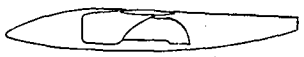
ALPHA = 20.06 DEG Q = 229 N/SQ.M. (4.78 LBF/SQ.FT.) EFFECT. VELOC. RATIO = .101

	ETA = .250		ETA = .387		ETA = .524		ETA = .800	
X/C	CP		CP		CP		CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			-.657		-2.136		-5.305	
.010			-3.108	.833				
.015					-4.699	.826		
.025			-4.051	.709	-4.417	.626	-3.693	.819
.050			-2.233	.378	-2.329	.275	-2.426	.530
.100			-1.578	-.097	-1.847	-.145	-1.819	.220
.150			-1.392	-.531	-1.558	-.449	-1.992	.034
.250	-.745		-1.117	-1.176	-1.255	-.826	-1.454	-.275
.400	-.729		-.903	-2.178	-.969	-1.050	-.943	-.462
.550	-.602	-8.206	-.696	-2.686	-.756	-1.009	-.699	-.492
.700	-.532	-6.469	-.585	-1.971	-.595	-.742	-.525	-.455
.850	-.485	-1.514	-.525	-.923	-.522	-.175	-.412	-.378
.950	-.492	-.555	-.495	-.529	-.499	.113	-.348	-.365

TABLE 12 (A)

PRESSURE PROFILE ON WING

REAR NOZZLES DEFLECTED 90 DEGREES



ALPHA = -.00 DEG Q = 2845 N/SQ.M. (59.42 LBF/SQ.FT.) EFFECT. VELOC. RATIO = ∞

	ETA = .250		ETA = .387		ETA = .524		ETA = .800	
X/C	CP		CP		CP		CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			.680		.932		.961	
.010			.515	.096				
.015					-.153	-.367		
.025			-.331	-.517	-.286	-.341	-.339	-.210
.050			-.376	-.352	-.412	-.270	-.280	-.051
.100			-.290	-.389	-.295	-.305	-.304	-.266
.150			-.359	-.447	-.356	-.340	-.331	-.279
.250	-.281		-.383	-.405	-.368	-.314	-.354	-.340
.400	-.343		-.367	-.330	-.376	-.334	-.329	-.315
.550	-.275	-.221	-.287	-.235	-.279	-.251	-.239	-.246
.700	-.204	-.126	-.183	-.177	-.175	-.158	-.133	-.161
.850	-.146	-.281	-.117	-.221	-.070	-.091	-.035	-.055
.950	-.107	-.088	-.063	-.110	-.008	-.008	.030	.043

ALPHA = 3.99 DEG Q = 2843 N/SQ.M. (59.38 LBF/SQ.FT.) EFFECT. VELOC. RATIO = ∞

	ETA = .250		ETA = .387		ETA = .524		ETA = .800	
X/C	CP		CP		CP		CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			.984		.639		.138	
.010			.493	.729				
.015					-1.381	.602		
.025			-1.244	.315	-1.358	.370	-1.077	.357
.050			-.812	.198	-.955	.259	-.879	.257
.100			-.603	.023	-.709	.130	-.625	.114
.150			-.617	-.104	-.652	-.005	-.584	-.046
.250	-.378		-.560	-.210	-.590	-.129	-.573	-.143
.400	-.430		-.475	-.198	-.497	-.170	-.450	-.194
.550	-.325	-.100	-.338	-.149	-.349	-.160	-.315	-.170
.700	-.233	-.086	-.234	-.148	-.224	-.120	-.176	-.109
.850	-.150	-.338	-.139	-.236	-.094	-.087	-.063	-.060
.950	-.088	-.150	-.066	-.139	-.009	-.018	.022	.032

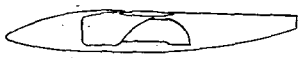
ALPHA = 8.05 DEG Q = 2846 N/SQ.M. (59.45 LBF/SQ.FT.) EFFECT. VELOC. RATIO = ∞

	ETA = .250		ETA = .387		ETA = .524		ETA = .800	
X/C	CP		CP		CP		CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			.265		-1.021		-2.003	
.010			.428	.998				
.015					-2.922	.934		
.025			-2.488	.827	-3.074	.818	-2.561	.840
.050			-1.436	.592	-1.647	.605	-1.358	.609
.100			-1.085	.376	-1.132	.407	-1.066	.400
.150			-.915	.213	-1.005	.272	-.912	.229
.250	-.485		-.748	.096	-.802	.103	-.748	.024
.400	-.492		-.590	-.058	-.627	-.032	-.569	-.063
.550	-.368	-.031	-.408	-.040	-.423	-.046	-.388	-.078
.700	-.258	-.071	-.272	-.082	-.267	-.046	-.232	-.058
.850	-.163	-.382	-.155	-.214	-.126	-.045	-.106	-.016
.950	-.100	-.250	-.077	-.136	-.034	.004	-.019	.018

TABLE 12 (A) - CONTINUED

PRESSURE PROFILE ON WING

REAR NOZZLES DEFLECTED 90 DEGREES



ALPHA = 10.00 DEG Q = 2848 N/SC.M. (59.48 LBF/SQ.FT.) EFFECT. VELOC. RATIO = ∞

X/C	ETA = .250 CP		ETA = .387 CP		ETA = .524 CP		ETA = .800 CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			-.557		-2.371		-3.865	
.010			.314	.939				
.015					-4.468	.977		
.025			-2.836	.938	-2.753	.919	-2.781	.908
.050			-1.825	.726	-2.136	.758	-1.784	.736
.100			-1.203	.555	-1.395	.519	-1.275	.497
.150			-1.035	.374	-1.171	.369	-1.061	.306
.250	-.537		-.830	.154	-.902	.226	-.869	.121
.400	-.516		-.635	.006	-.680	.045	-.631	-.005
.550	-.383	-.000	-.432	.023	-.453	-.006	-.420	-.043
.700	-.269	-.055	-.282	-.047	-.278	-.021	-.247	-.037
.850	-.169	-.372	-.159	-.156	-.125	-.037	-.112	-.020
.950	-.109	-.274	-.084	-.134	-.036	-.013	-.023	.007

ALPHA = 11.99 DEG Q = 2843 N/SC.M. (59.37 LBF/SQ.FT.) EFFECT. VELOC. RATIO = ∞

X/C	ETA = .250 CP		ETA = .387 CP		ETA = .524 CP		ETA = .800 CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			.330		-.099		-.940	
.010			.307	.986				
.015					-.730	.955		
.025			-.856	.895	-.678	.839	-.959	.815
.050			-.822	.735	-.875	.668	-1.006	.640
.100			-.854	.527	-.880	.493	-1.054	.432
.150			-.862	.353	-.695	.330	-1.029	.274
.250	-.655		-.858	.148	-.884	.162	-1.048	.102
.400	-.683		-.837	-.013	-.888	.014	-.916	-.040
.550	-.553	-.009	-.800	-.022	-.762	-.068	-.828	-.096
.700	-.471	-.076	-.614	-.092	-.629	-.110	-.595	-.107
.850	-.273	-.448	-.489	-.286	-.558	-.144	-.362	-.129
.950	-.184	-.338	-.416	-.274	-.459	-.248	-.393	-.152

ALPHA = 14.04 DEG Q = 2843 N/SC.M. (59.39 LBF/SQ.FT.) EFFECT. VELOC. RATIO = ∞

X/C	ETA = .250 CP		ETA = .387 CP		ETA = .524 CP		ETA = .800 CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			.389		.001		-.813	
.010			.302	.983				
.015					-.572	.948		
.025			-.528	.917	-.583	.850	-.866	.820
.050			-.537	.765	-.578	.721	-.869	.659
.100			-.545	.561	-.537	.507	-.875	.455
.150			-.553	.390	-.561	.351	-.885	.291
.250	-.551		-.550	.173	-.610	.188	-.875	.110
.400	-.633		-.613	-.007	-.646	.014	-.934	-.036
.550	-.654	-.019	-.666	-.041	-.722	-.080	-.899	-.119
.700	-.678	-.105	-.720	-.140	-.687	-.145	-.799	-.158
.850	-.673	-.518	-.651	-.371	-.646	-.255	-.695	-.213
.950	-.550	-.420	-.566	-.433	-.579	-.375	-.567	-.288

TABLE 12 (A) - CONCLUDED

PRESSURE PROFILE ON WING

REAR NOZZLES DEFLECTED 90 DEGREES

ALPHA = 16.01 DEG Q = 2848 N/SQ.M. (59.48 LBF/SQ.FT.) EFFECT. VELOC. RATIO = ∞

X/C	ETA = .250 CP		ETA = .387 CP		ETA = .524 CP		ETA = .800 CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			.260		-.111		-.960	
.010			.291	.959				
.015					-.517	.958		
.025			-.544	.952	-.531	.910	-.916	.863
.050			-.544	.835	-.508	.756	-.876	.711
.100			-.542	.618	-.524	.574	-.858	.516
.150			-.536	.478	-.520	.411	-.819	.325
.250	-.555		-.560	.227	-.552	.236	-.845	.160
.400	-.624		-.626	.028	-.649	.040	-.922	-.020
.550	-.676	-.005	-.655	-.017	-.663	-.053	-.892	-.117
.700	-.701	-.131	-.691	-.146	-.664	-.137	-.850	-.172
.850	-.723	-.576	-.670	-.416	-.647	-.265	-.809	-.277
.950	-.670	-.484	-.640	-.463	-.601	-.391	-.617	-.348

ALPHA = 18.02 DEG Q = 2845 N/SQ.M. (59.42 LBF/SQ.FT.) EFFECT. VELOC. RATIO = ∞

X/C	ETA = .250 CP		ETA = .387 CP		ETA = .524 CP		ETA = .800 CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			.088		-.274		-.938	
.010			.262	.887				
.015					-.530	.977		
.025			-.564	.986	-.531	.940	-.784	.896
.050			-.562	.889	-.526	.827	-.795	.759
.100			-.556	.698	-.535	.630	-.752	.537
.150			-.566	.546	-.546	.478	-.783	.390
.250	-.591		-.577	.315	-.559	.267	-.791	.186
.400	-.642		-.647	.075	-.618	.074	-.823	.009
.550	-.682	-.003	-.671	.006	-.686	-.039	-.857	-.100
.700	-.725	-.133	-.710	-.131	-.701	-.134	-.864	-.181
.850	-.753	-.610	-.726	-.409	-.665	-.274	-.842	-.303
.950	-.676	-.511	-.663	-.488	-.607	-.411	-.800	-.445

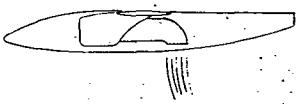
ALPHA = 20.02 DEG Q = 2847 N/SQ.M. (59.46 LBF/SQ.FT.) EFFECT. VELOC. RATIO = ∞

X/C	ETA = .250 CP		ETA = .387 CP		ETA = .524 CP		ETA = .800 CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			-.190		-.371		-1.018	
.010			.214	.791				
.015					-.535	.971		
.025			-.574	.983	-.553	.966	-.756	.913
.050			-.572	.924	-.524	.862	-.748	.792
.100			-.604	.770	-.549	.693	-.753	.609
.150			-.589	.644	-.546	.549	-.762	.446
.250	-.586		-.596	.374	-.571	.339	-.778	.237
.400	-.665		-.662	.127	-.636	.147	-.817	.055
.550	-.722	.040	-.719	.033	-.681	-.006	-.870	-.082
.700	-.753	-.137	-.729	-.113	-.683	-.112	-.843	-.185
.850	-.781	-.650	-.716	-.414	-.677	-.261	-.854	-.320
.950	-.745	-.555	-.690	-.511	-.622	-.413	-.829	-.476

TABLE 12 (B)

PRESSURE PROFILE ON WING

REAR NOZZLES DEFLECTED 90 DEGREES



ALPHA = .00 DEG Q = 2033 N/SQ.M. (42.46 LBF/SQ.FT.) EFFECT. VELOC. RATIO = .294

X/C	ETA = .250 CP		ETA = .387 CP		ETA = .524 CP		ETA = .800 CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			.840		.971		.815	
.010			-.012	.256				
.015					-.846	.221		
.025			-.664	.081	-.650	-.143	-.667	.002
.050			-.537	-.122	-.594	-.050	-.519	-.080
.100			-.512	-.262	-.495	-.188	-.439	-.167
.150			-.536	-.353	-.559	-.264	-.468	-.261
.250	-.383		-.491	-.395	-.518	-.332	-.457	-.326
.400	-.483		-.521	-.407	-.503	-.405	-.439	-.347
.550	-.427	-.226	-.418	-.398	-.408	-.376	-.339	-.295
.700	-.384	-.341	-.358	-.474	-.302	-.366	-.205	-.223
.850	-.373	-1.038	-.314	-.663	-.229	-.332	-.126	-.140
.950	-.391	-.641	-.297	-.486	-.176	-.227	-.054	-.045

ALPHA = 4.02 DEG Q = 2028 N/SQ.M. (42.36 LBF/SQ.FT.) EFFECT. VELOC. RATIO = .294

X/C	ETA = .250 CP		ETA = .387 CP		ETA = .524 CP		ETA = .800 CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			.829		-.180		-.687	
.010			.073	.932				
.015					-2.065	.792		
.025			-1.702	.601	-1.905	.605	-1.793	.595
.050			-1.159	.348	-1.215	.368	-1.219	.361
.100			-.816	.182	-1.050	.222	-.870	.194
.150			-.815	.017	-.888	.013	-.738	.022
.250	-.502		-.708	-.192	-.754	-.128	-.714	-.131
.400	-.581		-.658	-.271	-.662	-.215	-.561	-.209
.550	-.491	-.139	-.510	-.311	-.502	-.290	-.415	-.229
.700	-.426	-.308	-.422	-.434	-.383	-.311	-.272	-.174
.850	-.388	-1.088	-.350	-.668	-.265	-.316	-.165	-.130
.950	-.401	-.725	-.322	-.499	-.207	-.243	-.084	-.066

ALPHA = 8.04 DEG Q = 2035 N/SQ.M. (42.51 LBF/SQ.FT.) EFFECT. VELOC. RATIO = .294

X/C	ETA = .250 CP		ETA = .387 CP		ETA = .524 CP		ETA = .800 CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			-.845		-2.508		-3.551	
.010			-.016	.853				
.015					-4.209	.983		
.025			-2.854	.934	-2.907	.939	-2.524	.874
.050			-1.983	.756	-2.132	.735	-1.853	.702
.100			-1.314	.494	-1.505	.531	-1.347	.448
.150			-1.166	.345	-1.210	.313	-1.150	.263
.250	-.633		-.936	.097	-1.001	.125	-.953	.068
.400	-.659		-.774	-.100	-.798	-.072	-.703	-.076
.550	-.558	-.115	-.598	-.188	-.600	-.199	-.513	-.126
.700	-.469	-.318	-.472	-.363	-.437	-.246	-.330	-.124
.850	-.426	-1.133	-.390	-.650	-.300	-.281	-.201	-.117
.950	-.431	-.839	-.344	-.513	-.224	-.233	-.109	-.075

TABLE 12 (B) - CONTINUED

PRESSURE PROFILE ON WING

REAR NOZZLES DEFLECTED 90 DEGREES

ALPHA = 10.02 DEG Q = 2037 N/SQ.M. (42.55 LBF/SQ.FT.) EFFECT. VELOC. RATIO = .294

	ETA = .250		ETA = .387		ETA = .524		ETA = .800	
X/C	CP		CP		CP		CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			-2.006		-4.265		-1.627	
.010			-.200	.588				
.015					-4.343	.980		
.025			-3.480	.997	-3.838	.913	-1.514	.873
.050			-2.299	.884	-2.493	.740	-1.762	.691
.100			-1.577	.620	-1.708	.532	-1.803	.455
.150			-1.319	.458	-1.480	.363	-1.698	.288
.250	-.681		-1.034	.214	-1.094	.210	-1.322	.093
.400	-.714		-.835	-.029	-.866	-.017	-.907	-.067
.550	-.576	-.078	-.634	-.151	-.637	-.128	-.596	-.141
.700	-.491	-.315	-.494	-.338	-.464	-.218	-.444	-.158
.850	-.440	-1.141	-.403	-.654	-.319	-.333	-.371	-.163
.950	-.442	-.882	-.355	-.540	-.240	-.353	-.318	-.158

ALPHA = 12.02 DEG Q = 2040 N/SQ.M. (42.61 LBF/SQ.FT.) EFFECT. VELOC. RATIO = .294

	ETA = .250		ETA = .387		ETA = .524		ETA = .800	
X/C	CP		CP		CP		CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			-.106		-.519		-1.465	
.010			-.202	.920				
.015					-.946	.979		
.025			-1.073	.964	-.963	.916	-1.344	.885
.050			-.856	.842	-.927	.757	-1.335	.736
.100			-.983	.609	-.986	.570	-1.350	.489
.150			-1.042	.460	-1.034	.404	-1.385	.313
.250	-.843		-1.141	.184	-1.052	.204	-1.410	.129
.400	-.908		-1.159	-.035	-1.130	-.026	-1.275	-.051
.550	-.872	-.035	-1.009	-.171	-1.054	-.170	-1.173	-.140
.700	-.800	-.316	-.883	-.389	-.998	-.287	-.910	-.177
.850	-.722	-1.204	-.743	-.752	-.888	-.401	-.638	-.217
.950	-.624	-1.000	-.713	-.720	-.754	-.512	-.503	-.285

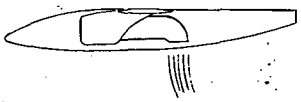
ALPHA = 14.01 DEG Q = 2039 N/SQ.M. (42.58 LBF/SQ.FT.) EFFECT. VELOC. RATIO = .294

	ETA = .250		ETA = .387		ETA = .524		ETA = .800	
X/C	CP		CP		CP		CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			-.074		-.550		-1.471	
.010			-.205	.879				
.015					-.867	.977		
.025			-.806	.965	-.900	.948	-1.279	.897
.050			-.778	.872	-.852	.814	-1.205	.745
.100			-.777	.672	-.865	.627	-1.198	.533
.150			-.817	.497	-.879	.464	-1.238	.358
.250	-.805		-.833	.250	-.951	.233	-1.214	.162
.400	-.879		-.883	-.007	-.985	-.004	-1.296	-.042
.550	-.901	-.016	-.927	-.174	-1.016	-.178	-1.258	-.149
.700	-.921	-.312	-.963	-.423	-1.020	-.316	-1.125	-.221
.850	-.884	-1.294	-.911	-.824	-.946	-.506	-.856	-.285
.950	-.721	-1.087	-.811	-.816	-.865	-.674	-.715	-.389

TABLE L2 (B) , CONCLUDED

PRESSURE PROFILE ON WING

REAR NOZZLES DEFLECTED 90 DEGREES



ALPHA = 16.01 DEG Q = 2036 N/SQ.M. (42.53 LBF/SQ.FT.) EFFECT. VELOC. RATIO = .294

X/C	ETA = .250 CP		ETA = .387 CP		ETA = .524 CP		ETA = .800 CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			-.274		-.658		-1.453	
.010			-.221	.756				
.015					-.852	.971		
.025			-.762	.993	-.842	.965	-1.175	.911
.050			-.753	.919	-.845	.859	-1.160	.788
.100			-.764	.758	-.827	.670	-1.135	.563
.150			-.786	.593	-.843	.536	-1.136	.399
.250	-.775		-.817	.340	-.870	.282	-1.159	.186
.400	-.834		-.895	.045	-.930	.035	-1.147	-.022
.550	-.876	.042	-.882	-.142	-.963	-.154	-1.219	-.152
.700	-.927	-.322	-.907	-.423	-.967	-.315	-1.168	-.242
.850	-.953	-1.344	-.909	-.862	-.971	-.525	-1.043	-.375
.950	-.941	-1.160	-.826	-.852	-.905	-.692	-.919	-.517

ALPHA = 18.01 DEG Q = 2039 N/SQ.M. (42.59 LBF/SQ.FT.) EFFECT. VELOC. RATIO = .293

X/C	ETA = .250 CP		ETA = .387 CP		ETA = .524 CP		ETA = .800 CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			-.476		-.805		-1.461	
.010			-.255	.707				
.015					-.851	.939		
.025			-.770		-.832	.975	-1.114	.923
.050			-.774	.952	-.835	.885	-1.094	.827
.100			-.764	.792	-.859	.724	-1.091	.621
.150			-.771	.640	-.871	.571	-1.101	.464
.250	-.797		-.794	.399	-.883	.343	-1.111	.233
.400	-.846		-.869	.094	-.956	.074	-1.145	.003
.550	-.884	.046	-.923	-.112	-.980	-.119	-1.199	-.145
.700	-.946	-.300	-.940	-.407	-.988	-.294	-1.175	-.260
.850	-.968	-1.379	-.926	-.863	-.989	-.528	-1.137	-.420
.950	-.982	-1.197	-.854	-.885	-.923	-.712	-1.029	-.604

ALPHA = 20.00 DEG Q = 2036 N/SQ.M. (42.52 LBF/SQ.FT.) EFFECT. VELOC. RATIO = .293

X/C	ETA = .250 CP		ETA = .387 CP		ETA = .524 CP		ETA = .800 CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			-.652		-.915		-1.284	
.010			-.299	.530				
.015					-.844	.907		
.025			-.776	.958	-.839	.978	-1.098	.925
.050			-.787	.978	-.833	.931	-1.088	.850
.100			-.813	.864	-.838	.780	-1.119	.669
.150			-.790	.711	-.871	.640	-1.092	.495
.250	-.802		-.809	.441	-.861	.396	-1.083	.282
.400	-.846		-.933	.136	-.961	.140	-1.112	.053
.550	-.899	-.008	-.945	-.087	-1.018	-.079	-1.137	-.122
.700	-.922	-.392	-.989	-.401	-1.035	-.263	-1.161	-.261
.850	-.976	-1.426	-.933	-.895	-1.041	-.505	-1.128	-.447
.950	-1.003	-1.307	-.883	-.891	-.955	-.707	-1.075	-.639

TABLE 12 (C)

PRESSURE PROFILE ON WING

REAR NOZZLES DEFLECTED 90 DEGREES

ALPHA = .01 DEG Q = 911 N/SQ.M. (19.02 LBF/SQ.FT.) EFFECT. VELOC. RATIO = .196

X/C	ETA = .250 CP		ETA = .387 CP		ETA = .524 CP		ETA = .800 CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			.939		.939		.890	
.010			-1.015	.262				
.015					-.622	.004		
.025			-.702	-.165	-.843	-.231	-.646	-.044
.050			-.530	-.319	-.584	-.210	-.568	-.155
.100			-.520	-.357	-.584	-.266	-.437	-.243
.150			-.525	-.463	-.553	-.373	-.445	-.362
.250	-.432		-.544	-.555	-.580	-.477	-.499	-.408
.400	-.554		-.609	-.584	-.550	-.530	-.476	-.460
.550	-.485	-.403	-.518	-.551	-.462	-.549	-.389	-.406
.700	-.454	-.596	-.442	-.663	-.390	-.523	-.281	-.325
.850	-.486	-1.386	-.393	-.866	-.298	-.481	-.180	-.236
.950	-.544	-.947	-.358	-.658	-.276	-.359	-.125	-.130

ALPHA = 4.00 DEG Q = 912 N/SQ.M. (19.04 LBF/SQ.FT.) EFFECT. VELOC. RATIO = .196

X/C	ETA = .250 CP		ETA = .387 CP		ETA = .524 CP		ETA = .800 CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			.793		.178		-.397	
.010			-.880	.854				
.015					-1.923	.696		
.025			-1.730	.468	-1.754	.466	-1.583	.537
.050			-1.272	.207	-1.339	.290	-1.282	.250
.100			-.861	-.043	-.997	.088	-.824	.045
.150			-.921	-.102	-.918	-.085	-.762	-.152
.250	-.550		-.805	-.387	-.816	-.276	-.767	-.228
.400	-.640		-.734	-.489	-.703	-.409	-.623	-.342
.550	-.561	-.405	-.604	-.529	-.600	-.481	-.480	-.343
.700	-.507	-.670	-.523	-.685	-.478	-.501	-.347	-.289
.850	-.515	-1.474	-.488	-.947	-.380	-.481	-.247	-.237
.950	-.571	-1.081	-.475	-.743	-.329	-.419	-.169	-.169

ALPHA = 8.02 DEG Q = 906 N/SQ.M. (18.93 LBF/SQ.FT.) EFFECT. VELOC. RATIO = .196

X/C	ETA = .250 CP		ETA = .387 CP		ETA = .524 CP		ETA = .800 CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			-.474		-1.865		-3.287	
.010			-.838	.918				
.015					-3.977	.941		
.025			-3.416	.868	-3.346	.842	-2.931	.796
.050			-2.016	.668	-2.032	.626	-1.895	.588
.100			-1.361	.430	-1.531	.409	-1.368	.332
.150			-1.194	.207	-1.453	.198	-1.130	.145
.250	-.685		-.978	-.080	-1.079	-.055	-.985	-.042
.400	-.753		-.842	-.336	-.896	-.256	-.767	-.209
.550	-.652	-.427	-.674	-.443	-.688	-.340	-.585	-.269
.700	-.581	-.695	-.580	-.636	-.538	-.454	-.422	-.256
.850	-.567	-1.537	-.513	-.930	-.420	-.473	-.300	-.234
.950	-.616	-1.187	-.497	-.784	-.372	-.404	-.211	-.178

TABLE 12 (C) - CONTINUED

PRESSURE PROFILE ON WING

REAR NOZZLES DEFLECTED 90 DEGREES

ALPHA = 10.00 DEG Q = 904 N/SQ.M. (18.89 LBF/SQ.FT.) EFFECT. VELOC. RATIO = .195

X/C	ETA = .250 CP		ETA = .387 CP		ETA = .524 CP		ETA = .800 CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			-1.545		-3.952		-4.754	
.010			-.872	.704				
.015					-5.407	.913		
.025			-3.494	.943	-3.599	.931	-3.235	.913
.050			-2.297	.777	-2.463	.755	-2.264	.727
.100			-1.571	.516	-1.782	.485	-1.651	.448
.150			-1.346	.323	-1.501	.337	-1.378	.246
.250	-.756		-1.106	.006	-1.209	.038	-1.123	.042
.400	-.798		-.918	-.282	-1.170	-.206	-.844	-.135
.550	-.682	-.401	-.723	-.406	-.737	-.344	-.619	-.222
.700	-.601	-.722	-.605	-.623	-.579	-.436	-.463	-.242
.850	-.582	-1.605	-.544	-.949	-.451	-.479	-.316	-.240
.950	-.635	-1.219	-.520	-.808	-.394	-.405	-.237	-.205

ALPHA = 12.02 DEG Q = 905 N/SQ.M. (18.90 LBF/SQ.FT.) EFFECT. VELOC. RATIO = .196

X/C	ETA = .250 CP		ETA = .387 CP		ETA = .524 CP		ETA = .800 CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			-.397		-.844		-1.940	
.010			-.829	.839				
.015					-1.445	.935		
.025			-1.665	.923	-1.339	.873	-1.861	.841
.050			-1.261	.778	-1.487	.691	-1.760	.669
.100			-1.187	.526	-1.332	.476	-1.658	.420
.150			-1.605	.333	-1.412	.282	-1.734	.247
.250	-1.006		-1.581	.052	-1.564	.041	-1.625	.022
.400	-.895		-1.250	-.258	-1.296	-.196	-1.318	-.169
.550	-.785	-.392	-.896	-.418	-1.158	-.373	-1.006	-.253
.700	-.726	-.692	-.768	-.653	-.878	-.471	-.665	-.279
.850	-.641	-1.617	-.645	-.995	-.734	-.590	-.481	-.308
.950	-.674	-1.228	-.685	-.849	-.688	-.573	-.381	-.316

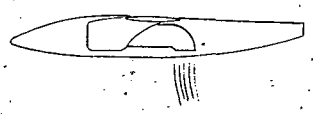
ALPHA = 14.00 DEG Q = 908 N/SQ.M. (18.96 LBF/SQ.FT.) EFFECT. VELOC. RATIO = .196

X/C	ETA = .250 CP		ETA = .387 CP		ETA = .524 CP		ETA = .800 CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			-.189		-.746		-1.803	
.010			-.756	.839				
.015					-1.152	.928		
.025			-1.060	.933	-1.189	.879	-1.604	.851
.050			-1.019	.801	-1.105	.712	-1.564	.700
.100			-1.012	.554	-1.145	.518	-1.520	.480
.150			-1.147	.367	-1.168	.353	-1.527	.268
.250	-1.067		-1.223	.083	-1.355	.088	-1.571	.063
.400	-1.069		-1.247	-.244	-1.279	-.210	-1.556	-.142
.550	-1.057	-.332	-1.196	-.422	-1.277	-.385	-1.386	-.248
.700	-.963	-.708	-1.087	-.693	-1.147	-.501	-1.162	-.304
.850	-.878	-1.662	-.986	-1.090	-1.051	-.685	-.873	-.342
.950	-.762	-1.288	-.857	-.974	-.891	-.724	-.543	-.382

TABLE 12 (C) - CONCLUDED

PRESSURE PROFILE ON WING

REAR NUZZLES DEFLECTED 90 DEGREES



ALPHA = 15.98 DEG Q = 910 N/SQ.M. (19.00 LBF/SQ.FT.) EFFECT. VELOC. RATIO = .197

	ETA = .250		ETA = .387		ETA = .524		ETA = .800	
X/C	CP		CP		CP		CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			-.279		-.792		-1.823	
.010			-.752	.783				
.015					-1.043	.933		
.025			-.965	.939	-1.064	.904	-1.452	.869
.050			-.974	.842	-1.059	.783	-1.449	.738
.100			-.964	.611	-1.054	.571	-1.398	.505
.150			-.988	.443	-1.066	.384	-1.378	.317
.250	-.931		-.979	.121	-1.102	.114	-1.438	.092
.400	-1.031		-1.029	-.210	-1.153	-.171	-1.527	-.135
.550	-1.079	-.325	-1.061	-.418	-1.213	-.389	-1.460	-.262
.700	-1.084	-.710	-1.141	-.718	-1.204	-.546	-1.353	-.331
.850	-1.067	-1.757	-1.068	-1.165	-1.105	-.738	-1.018	-.413
.950	-.992	-1.381	-.956	-1.039	-1.042	-.890	-.872	-.540

ALPHA = 18.03 DEG Q = 913 N/SQ.M. (19.08 LBF/SQ.FT.) EFFECT. VELOC. RATIO = .197

	ETA = .250		ETA = .387		ETA = .524		ETA = .800	
X/C	CP		CP		CP		CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			-.505		-.956		-1.678	
.010			-.800	.664				
.015					-1.038	.916		
.025			-.911	.942	-1.026	.925	-1.364	.882
.050			-.857	.882	-1.011	.806	-1.287	.764
.100			-.919	.688	-1.006	.618	-1.357	.547
.150			-.954	.505	-1.021	.442	-1.297	.364
.250	-.964		-.940	.183	-1.044	.188	-1.307	.142
.400	-.980		-1.021	-.172	-1.128	-.141	-1.366	-.108
.550	-1.045	-.317	-1.076	-.400	-1.110	-.355	-1.465	-.261
.700	-1.017	-.748	-1.083	-.717	-1.174	-.557	-1.330	-.372
.850	-1.051	-1.847	-1.043	-1.190	-1.154	-.772	-1.245	-.494
.950	-1.043	-1.468	-.973	-1.083	-1.087	-.904	-1.069	-.667

ALPHA = 19.99 DEG Q = 910 N/SQ.M. (19.00 LBF/SQ.FT.) EFFECT. VELOC. RATIO = .196

	ETA = .250		ETA = .387		ETA = .524		ETA = .800	
X/C	CP		CP		CP		CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			-.621		-1.020		-1.636	
.010			-1.011	.530				
.015					-1.004	.868		
.025			-.905	.920	-1.013	.927	-1.298	.878
.050			-.912	.897	-.978	.849	-1.293	.788
.100			-.909	.745	-.987	.686	-1.250	.580
.150			-.924	.573	-1.014	.520	-1.266	.406
.250	-.954		-.973	.255	-1.046	.237	-1.278	.170
.400	-1.006		-1.054	-.103	-1.120	-.086	-1.292	-.075
.550	-1.051	-.297	-1.074	-.357	-1.135	-.338	-1.371	-.262
.700	-1.126	-.739	-1.116	-.713	-1.165	-.550	-1.311	-.389
.850	-1.167	-1.865	-1.068	-1.193	-1.157	-.776	-1.259	-.556
.950	-1.111	-1.499	-1.006	-1.115	-1.112	-.935	-1.199	-.739

TABLE 12 (D)

PRESSURE PROFILE ON WING

REAR NOZZLES DEFLECTED 90 DEGREES

ALPHA = .00 DEG Q = 226 N/SQ.M. (4.72 LBF/SQ.FT.) EFFECT. VELOC. RATIO = .098

X/C	ETA = .250 CP		ETA = .387 CP		ETA = .524 CP		ETA = .800 CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			.346		.632		.743	
.010			-6.055	-.540				
.015					-.470	-1.028		
.025			-.707	-.777	-.533	-.805	-.442	-.700
.050			-.540	-.644	-.540	-.721	-.526	-.707
.100			-.556	-.833	-.561	-.777	-.554	-.679
.150			-.616	-.930	-.630	-.805	-.526	-.728
.250	-.582		-.630	-.963	-.658	-.895	-.644	-.794
.400	-.655		-.682	-.987	-.672	-.950	-.635	-.784
.550	-.622	-.791	-.625	-.929	-.615	-.912	-.534	-.740
.700	-.615	-.865	-.605	-.970	-.574	-.848	-.473	-.655
.850	-.642	-1.507	-.608	-1.075	-.513	-.791	-.409	-.568
.950	-.733	-1.135	-.622	-.990	-.507	-.649	-.355	-.409

ALPHA = 3.99 DEG Q = 226 N/SQ.M. (4.72 LBF/SQ.FT.) EFFECT. VELOC. RATIO = .098

X/C	ETA = .250 CP		ETA = .387 CP		ETA = .524 CP		ETA = .800 CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			.750		.611		.227	
.010			-5.560	.255				
.015					-1.376	.109		
.025			-1.376	-.003	-1.460	-.100	-1.376	.081
.050			-1.258	-.247	-1.112	-.219	-1.091	-.247
.100			-.881	-.407	-.944	-.351	-.909	-.351
.150			-.902	-.540	-.937	-.498	-.874	-.491
.250	-.651		-.826	-.760	-.840	-.628	-.833	-.541
.400	-.689		-.777	-.811	-.797	-.753	-.723	-.618
.550	-.645	-.737	-.696	-.835	-.686	-.777	-.632	-.601
.700	-.649	-.936	-.639	-.933	-.595	-.784	-.486	-.541
.850	-.666	-1.575	-.639	-1.037	-.520	-.696	-.412	-.466
.950	-.764	-1.291	-.689	-.956	-.517	-.679	-.372	-.409

ALPHA = 7.98 DEG Q = 224 N/SQ.M. (4.67 LBF/SQ.FT.) EFFECT. VELOC. RATIO = .097

X/C	ETA = .250 CP		ETA = .387 CP		ETA = .524 CP		ETA = .800 CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			.339		-.443		-1.690	
.010			-5.143	.776				
.015					-2.712	.656		
.025			-2.360	.473	-2.170	.395	-2.261	.367
.050			-1.733	.219	-1.930	.142	-1.867	.113
.100			-1.162	.001	-1.275	-.105	-1.113	-.035
.150			-1.099	-.274	-1.162	-.246	-1.063	-.218
.250	-.739		-.986	-.540	-1.035	-.461	-.986	-.383
.400	-.806		-.892	-.721	-.943	-.656	-.840	-.543
.550	-.724	-.810	-.769	-.775	-.789	-.724	-.690	-.560
.700	-.697	-1.018	-.711	-.888	-.676	-.731	-.567	-.516
.850	-.707	-1.660	-.687	-1.056	-.598	-.683	-.482	-.458
.950	-.813	-1.380	-.711	-.936	-.574	-.639	-.420	-.413

TABLE 12 (D) - CONTINUED
PRESSURE PROFILE ON WING

REAR NOZZLES DEFLECTED 90 DEGREES

ALPHA = 10.04 DEG Q = 225 N/SQ.M. (4.70 LBF/SQ.FT.) EFFECT. VELOC. RATIO = .098

	ETA = .250		ETA = .387		ETA = .524		ETA = .800	
X/C	CP		CP		CP		CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			-.447		-1.700		-2.434	
.010			-4.827	.742				
.015					-3.316	.700		
.025			-2.889	.582	-3.078	.561	-3.225	.554
.050			-1.770	.379	-2.000	.344	-1.790	.316
.100			-1.441	.050	-1.637	.120	-1.497	.050
.150			-1.308	-.160	-1.378	-.125	-1.315	-.160
.250	-.825		-1.126	-.417	-1.175	-.353	-1.133	-.370
.400	-.845		-.970	-.655	-1.011	-.567	-.933	-.482
.550	-.787	-.824	-.824	-.750	-.845	-.709	-.763	-.526
.700	-.729	-1.052	-.740	-.892	-.723	-.723	-.614	-.519
.850	-.743	-1.713	-.709	-1.075	-.628	-.699	-.499	-.461
.950	-.852	-1.445	-.736	-.943	-.601	-.621	-.438	-.400

ALPHA = 11.97 DEG Q = 227 N/SQ.M. (4.74 LBF/SQ.FT.) EFFECT. VELOC. RATIO = .098

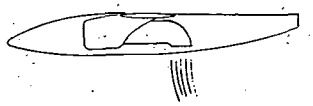
	ETA = .250		ETA = .387		ETA = .524		ETA = .800	
X/C	CP		CP		CP		CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			-1.390		-2.992		-4.394	
.010			-4.657	.552				
.015					-4.844	.705		
.025			-4.102	.691	-4.262	.663	-3.728	.635
.050			-2.181	.504	-2.500	.379	-2.292	.414
.100			-1.633	.212	-1.952	.122	-1.730	.122
.150			-1.487	.004	-1.598	-.023	-1.508	-.086
.250	-.911		-1.217	-.377	-1.328	-.280	-1.300	-.270
.400	-.939		-1.067	-.646	-1.117	-.532	-1.006	-.471
.550	-.848	-.851	-.858	-.737	-.925	-.687	-.814	-.542
.700	-.787	-1.104	-.798	-.909	-.774	-.747	-.673	-.539
.850	-.801	-1.766	-.754	-1.120	-.673	-.734	-.549	-.508
.950	-.892	-1.487	-.784	-.966	-.643	-.636	-.471	-.465

ALPHA = 14.01 DEG Q = 226 N/SQ.M. (4.72 LBF/SQ.FT.) EFFECT. VELOC. RATIO = .098

	ETA = .250		ETA = .387		ETA = .524		ETA = .800	
X/C	CP		CP		CP		CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			-.864		-1.456		-2.377	
.010			-4.441	.607				
.015					-1.979			
.025			-2.112	.670	-2.035	.621	-2.258	.600
.050			-2.321	.503	-2.112	.433	-2.223	.419
.100			-2.258	.217	-2.244	.182	-2.105	.147
.150			-2.210	-.041	-2.119	-.027	-2.119	-.062
.250	-1.045		-1.777	-.336	-1.701	-.309	-1.763	-.272
.400	-1.032		-1.134	-.640	-1.201	-.576	-1.205	-.465
.550	-.928	-.847	-.854	-.745	-.941	-.711	-1.009	-.563
.700	-.823	-1.151	-.803	-.934	-.826	-.779	-.847	-.576
.850	-.820	-1.800	-.765	-1.134	-.813	-.809	-.664	-.576
.950	-.911	-1.496	-.769	-.985	-.711	-.715	-.586	-.482

TABLE 12 (D) - CONCLUDED
PRESSURE PROFILE ON WING

REAR NOZZLES DEFLECTED 90 DEGREES



ALPHA = 16.03 DEG Q = 224 N/SQ.M. (4.68 LBF/SQ.FT.) EFFECT. VELOC. RATIO = .097

	ETA = .250		ETA = .387		ETA = .524		ETA = .800	
X/C	CP		CP		CP		CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			-.651		-1.052		-2.079	
.010			-4.238	.545				
.015					-1.615	.685		
.025			-1.495	0.000	-1.657	.601	-1.826	.594
.050			-1.776	.552	-1.516	.446	-1.903	.425
.100			-1.798	.263	-1.516	.200	-1.784	.151
.150			-1.657	.031	-1.565	-.011	-1.812	-.046
.250	-1.207		-1.734	-.307	-1.551	-.270	-1.854	-.263
.400	-1.248		-1.552	-.614	-1.586	-.553	-1.650	-.481
.550	-1.030	-.853	-1.241	-.740	-1.425	-.716	-1.405	-.590
.700	-1.095	-1.119	-1.132	-.938	-1.159	-.822	-1.132	-.617
.850	-.924	-1.794	-.989	-1.173	-.989	-.839	-.917	-.634
.950	-.941	-1.473	-.818	-.996	-.917	-.859	-.805	-.631

ALPHA = 18.00 DEG Q = 225 N/SQ.M. (4.70 LBF/SQ.FT.) EFFECT. VELOC. RATIO = .098

	ETA = .250		ETA = .387		ETA = .524		ETA = .800	
X/C	CP		CP		CP		CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			-.460		-1.027		-2.056	
.010			-3.980	.554				
.015					-1.405	.666		
.025			-1.286	0.000	-1.307	.645	-1.825	.617
.050			-1.251	.554	-1.363	.463	-1.755	.463
.100			-1.314	.316	-1.349	.225	-1.692	.183
.150			-1.356	.036	-1.454	.036	-1.797	-.027
.250	-1.293		-1.384	-.249	-1.475	-.256	-1.818	-.226
.400	-1.321		-1.511	-.585	-1.464	-.548	-1.711	-.470
.550	-1.304	-.799	-1.385	-.751	-1.497	-.721	-1.613	-.595
.700	-1.155	-1.114	-1.355	-.985	-1.399	-.853	-1.389	-.629
.850	-1.053	-1.793	-1.209	-1.263	-1.260	-.982	-1.189	-.700
.950	-1.148	-1.494	-1.080	-1.074	-1.172	-.982	-.890	-.694

ALPHA = 20.00 DEG Q = 226 N/SQ.M. (4.71 LBF/SQ.FT.) EFFECT. VELOC. RATIO = .098

	ETA = .250		ETA = .387		ETA = .524		ETA = .800	
X/C	CP		CP		CP		CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			-.587		-1.013		-2.054	
.010			-3.821	.460				
.015					-1.328			
.025			-1.188	.635	-1.258	.635	-1.691	.614
.050			-1.160	.565	-1.272	.474	-1.656	.432
.100			-1.188	.321	-1.328	.251	-1.572	.209
.150			-1.251	.132	-1.258	.055	-1.656	-.001
.250	-1.174		-1.188	-.241	-1.335	-.241	-1.677	-.245
.400	-1.230		-1.359	-.580	-1.420	-.543	-1.640	-.465
.550	-1.382	-.847	-1.406	-.807	-1.392	-.763	-1.731	-.604
.700	-1.315	-1.155	-1.359	-1.017	-1.392	-.915	-1.558	-.685
.850	-1.264	-1.863	-1.291	-1.348	-1.332	-1.057	-1.315	-.797
.950	-1.155	-1.555	-1.155	-1.162	-1.230	-1.135	-1.152	-.858

TABLE 13 (A)

PRESSURE PROFILE ON WING

REAR NOZZLES DEFLECTED 90 DEGREES WITH FLAPS

ALPHA = .00 DEG Q = 2845 N/SQ.M. (59.42 LBF/SQ.FT.) EFFECT. VELOC. RATIO = ∞

X/C	ETA = .250 CP		ETA = .387 CP		ETA = .524 CP		ETA = .800 CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			.943		.953		.838	
.010			.534	.551				
.015					-.779	.290		
.025			-.908	.073	-.550	.167	-.505	.058
.050			-.508	.030	-.589	.029	-.639	.001
.100			-.522	-.069	-.538	.027	-.445	-.004
.150			-.547	-.109	-.531	-.061	-.494	-.111
.250	-.397		-.514	-.099	-.543	-.111	-.460	-.157
.400	-.484		-.534	-.079	-.509	-.054	-.414	-.192
.550	-.427	.033	-.437	.007	-.434	.078	-.338	-.132
.700	-.380	.119	-.363	.260	-.345	.284	-.189	-.098
.850	-.357	.187	-.319	.441	-.286	.507	-.097	-.010
.950	-.359	.150	-.255	.570	-.264	.623	-.015	.048

ALPHA = 3.99 DEG Q = 2842 N/SQ.M. (59.37 LBF/SQ.FT.) EFFECT. VELOC. RATIO = ∞

X/C	ETA = .250 CP		ETA = .387 CP		ETA = .524 CP		ETA = .800 CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			.720		-.337		-.671	
.010			.574	.974				
.015					-2.303	.855		
.025			-2.013	.637	-1.983	.723	-1.577	.633
.050			-1.154	.459	-1.179	.497	-1.108	.483
.100			-.873	.307	-.960	.429	-.719	.293
.150			-.879	.182	-.870	.221	-.803	.120
.250	-.513		-.722	.111	-.780	.113	-.684	.008
.400	-.575		-.656	.133	-.667	.103	-.543	-.099
.550	-.497	.159	-.517	.203	-.518	.182	-.430	-.050
.700	-.423	.255	-.421	.351	-.414	.351	-.261	-.036
.850	-.381	.204	-.353	.494	-.328	.542	-.142	-.010
.950	-.342	.214	-.270	.609	-.286	.650	-.050	.014

ALPHA = 8.00 DEG Q = 2839 N/SQ.M. (59.29 LBF/SQ.FT.) EFFECT. VELOC. RATIO = ∞

X/C	ETA = .250 CP		ETA = .387 CP		ETA = .524 CP		ETA = .800 CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			-.863		-2.997		-3.633	
.010			.504	.878				
.015					-4.089	.973		
.025			-2.933	.934	-3.123	.964	-2.463	.901
.050			-1.905	.818	-2.059	.824	-1.824	.742
.100			-1.334	.631	-1.466	.627	-1.341	.509
.150			-1.163	.527	-1.258	.468	-1.081	.337
.250	-.648		-.948	.328	-1.010	.306	-.926	.166
.400	-.667		-.796	.240	-.823	.239	-.706	.059
.550	-.568	.266	-.608	.325	-.630	.281	-.501	.010
.700	-.472	.285	-.482	.403	-.479	.382	-.327	.009
.850	-.411	.233	-.378	.519	-.378	.566	-.179	-.004
.950	-.357	.254	-.300	.636	-.314	.672	-.073	-.006

TABLE 13 (A) - CONTINUED

PRESSURE PROFILE ON WING

REAR NOZZLES DEFLECTED 90 DEGREES WITH FLAPS

ALPHA = 10.03 DEG Q = 2849 N/SQ.M. (59.50 LBF/SQ.FT.) EFFECT. VELOC. RATIO = ∞

	ETA = .250		ETA = .387		ETA = .524		ETA = .800	
X/C	CP		CP		CP		CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			-.047		-.528		-1.418	
.010			.485	.921				
.015					-1.289	.979		
.025			-1.491	.960	-1.135	.921	-1.469	.871
.050			-1.410	.842	-1.006	.777	-1.457	.706
.100			-1.257	.666	-1.177	.607	-1.453	.511
.150			-1.383	.527	-1.071	.480	-1.445	.365
.250	-.843		-1.328	.358	-1.369	.359	-1.452	.192
.400	-.813		-1.235	.274	-1.162	.274	-1.303	.069
.550	-.728	.290	-.992	.352	-1.017	.313	-.748	.020
.700	-.573	.308	-.847	.424	-.906	.394	-.583	-.004
.850	-.489	.269	-.502	.540	-.570	.584	-.366	-.039
.950	-.318	.310	-.456	.646	-.581	.676	-.211	-.063

ALPHA = 11.99 DEG Q = 2845 N/SQ.M. (59.42 LBF/SQ.FT.) EFFECT. VELOC. RATIO = ∞

	ETA = .250		ETA = .387		ETA = .524		ETA = .800	
X/C	CP		CP		CP		CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			.107		-.393		-1.274	
.010			.504	.951				
.015					-.848	.981		
.025			-.746	.971	-.849	.921	-1.199	.880
.050			-.877	.853	-.951	.804	-1.150	.732
.100			-.904	.697	-.785	.632	-1.155	.539
.150			-.834	.559	-.824	.509	-1.236	.387
.250	-.746		-.864	.393	-1.001	.372	-1.282	.212
.400	-.866		-.930	.300	-.919	.300	-1.228	.087
.550	-.907	.312	-.952	.359	-1.041	.317	-1.142	.025
.700	-.868	.333	-.883	.433	-.941	.392	-.858	-.004
.850	-.739	.277	-.765	.548	-.829	.586	-.663	-.101
.950	-.526	.301	-.709	.650	-.739	.671	-.454	-.129

ALPHA = 14.01 DEG Q = 2841 N/SQ.M. (59.34 LBF/SQ.FT.) EFFECT. VELOC. RATIO = ∞

	ETA = .250		ETA = .387		ETA = .524		ETA = .800	
X/C	CP		CP		CP		CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			-.059		-.492		-1.086	
.010			.499	.841				
.015					-.735	.976		
.025			-.758	.988	-.765	.947	-.908	.889
.050			-.737	.900	-.710	.840	-.949	.754
.100			-.761	.734	-.719	.669	-.975	.558
.150			-.788	.610	-.734	.561	-1.053	.408
.250	-.734		-.801	.434	-.744	.409	-.985	.240
.400	-.799		-.799	.325	-.813	.319	-.992	.089
.550	-.858	.314	-.826	.383	-.826	.321	-1.110	.010
.700	-.855	.319	-.835	.447	-.816	.403	-.998	-.047
.850	-.751	.268	-.755	.549	-.772	.589	-.823	-.142
.950	-.651	.277	-.770	.650	-.768	.671	-.758	-.270

TABLE 13 (A) - CONCLUDED

PRESSURE PROFILE ON WING

REAR NOZZLES DEFLECTED 90 DEGREES WITH FLAPS

ALPHA = 15.98 DEG Q = 2842 N/SQ.M. (59.36 LBF/SQ.FT.) EFFECT. VELOC. RATIO = ∞

	ETA = .250		ETA = .387		ETA = .524		ETA = .800	
X/C	CP		CP		CP		CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			-.212		-.563		-.954	
.010			.478	.803				
.015					-.710	.969		
.025			-.746		-.695	.963	-.798	.897
.050			-.709	.931	-.738	.874	-.787	.774
.100			-.748	.792	-.710	.731	-.795	.579
.150			-.707	.664	-.700	.591	-.748	.436
.250	-.733		-.771	.476	-.727	.458	-.776	.280
.400	-.792		-.817	.349	-.773	.344	-.838	.108
.550	-.791	.321	-.812	.401	-.788	.350	-.898	.007
.700	-.797	.316	-.820	.442	-.771	.415	-.946	-.067
.850	-.770	.268	-.784	.545	-.747	.590	-.811	-.199
.950	-.703	.281	-.770	.647	-.758	.676	-.790	-.368

ALPHA = 18.00 DEG Q = 2842 N/SQ.M. (59.37 LBF/SQ.FT.) EFFECT. VELOC. RATIO = ∞

	ETA = .250		ETA = .387		ETA = .524		ETA = .800	
X/C	CP		CP		CP		CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			-.310		-.673		-1.018	
.010			.444	.710				
.015					-.738	.941		
.025			-.739	.983	-.714	.974	-.730	.915
.050			-.754	.952	-.693	.908	-.720	.808
.100			-.696	.840	-.719	.758	-.758	.641
.150			-.719	.720	-.723	.668	-.749	.485
.250	-.746		-.724	.538	-.743	.518	-.747	.310
.400	-.758		-.785	.382	-.761	.382	-.834	.137
.550	-.762	.343	-.751	.424	-.772	.376	-.802	.023
.700	-.744	.336	-.757	.459	-.765	.430	-.841	-.061
.850	-.768	.276	-.799	.548	-.807	.604	-.820	-.205
.950	-.717	.261	-.820	.643	-.802	.682	-.761	-.390

ALPHA = 20.04 DEG Q = 2844 N/SQ.M. (59.39 LBF/SQ.FT.) EFFECT. VELOC. RATIO = ∞

	ETA = .250		ETA = .387		ETA = .524		ETA = .800	
X/C	CP		CP		CP		CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			-.530		-.752		-.805	
.010			.394	.584				
.015					-.701	.922		
.025			-.719	.947	-.693	.979	-.702	.929
.050			-.718	.974	-.735	.943	-.692	.839
.100			-.702	.883	-.701	.823	-.700	.675
.150			-.712	.770	-.731	.709	-.721	.531
.250	-.742		-.727	.584	-.712	.553	-.703	.367
.400	-.768		-.731	.428	-.715	.416	-.737	.172
.550	-.767	.372	-.755	.444	-.797	.400	-.737	.067
.700	-.750	.347	-.758	.478	-.805	.464	-.783	-.033
.850	-.786	.287	-.798	.536	-.806	.609	-.768	-.192
.950	-.733	.273	-.803	.650	-.796	.691	-.765	-.384

TABLE 13 (B)

PRESSURE PROFILE ON WING

REAR NOZZLES DEFLECTED 90 DEGREES WITH FLAPS

ALPHA = .00 DEG Q = 2040 N/SQ.M. (42.62 LBF/SQ.FT.) EFFECT. VELOC. RATIO = .295

	ETA = .250		ETA = .387		ETA = .524		ETA = .800	
X/C	CP UPPER	CP LOWER	CP UPPER	CP LOWER	CP UPPER	CP LOWER	CP UPPER	CP LOWER
.000			.913		.712		.537	
.010			.219	.846				
.015					-1.430	.602		
.025			-1.391	.469	-1.310	.484	-1.127	.392
.050			-.990	.252	-1.005	.277	-.823	.175
.100			-.754	.084	-.801	.156	-.673	.052
.150			-.789	.003	-.796	.089	-.608	-.065
.250	-.550		-.764	-.046	-.785	-.048	-.631	-.186
.400	-.676		-.752	-.015	-.721	-.064	-.574	-.226
.550	-.645	.134	-.672	.087	-.648	.020	-.465	-.210
.700	-.648	.226	-.641	.176	-.600	.175	-.348	-.168
.850	-.720	.078	-.665	.312	-.606	.397	-.232	-.129
.950	-.886	.062	-.690	.443	-.687	.517	-.150	-.075

ALPHA = 4.02 DEG Q = 2029 N/SQ.M. (42.37 LBF/SQ.FT.) EFFECT. VELOC. RATIO = .293

	ETA = .250		ETA = .387		ETA = .524		ETA = .800	
X/C	CP UPPER	CP LOWER	CP UPPER	CP LOWER	CP UPPER	CP LOWER	CP UPPER	CP LOWER
.000			.169		-1.502		-2.014	
.010			.198	.984				
.015					-3.515	.952		
.025			-2.919	.865	-3.163	.852	-2.323	.752
.050			-1.665	.665	-1.676	.675	-1.450	.559
.100			-1.253	.437	-1.314	.442	-1.130	.298
.150			-1.144	.306	-1.148	.294	-1.113	.152
.250	-.702		-1.003	.174	-.996	.122	-.901	-.015
.400	-.794		-.916	.094	-.918	.056	-.740	-.120
.550	-.742	.214	-.774	.142	-.784	.085	-.557	-.130
.700	-.723	.256	-.724	.188	-.712	.189	-.414	-.152
.850	-.773	.080	-.730	.329	-.693	.414	-.296	-.149
.950	-.892	.010	-.743	.466	-.753	.527	-.194	-.135

ALPHA = 8.02 DEG Q = 2033 N/SQ.M. (42.46 LBF/SQ.FT.) EFFECT. VELOC. RATIO = .293

	ETA = .250		ETA = .387		ETA = .524		ETA = .800	
X/C	CP UPPER	CP LOWER	CP UPPER	CP LOWER	CP UPPER	CP LOWER	CP UPPER	CP LOWER
.000			-.728		-1.315		-2.396	
.010			.149	.773				
.015					-2.069	.953		
.025			-2.330	.969	-2.052	.926	-2.377	.885
.050			-2.397	.859	-1.896	.814	-2.576	.718
.100			-2.252	.662	-2.082	.606	-2.365	.480
.150			-2.248	.550	-1.931	.467	-2.158	.323
.250	-.927		-1.814	.361	-1.900	.311	-1.300	.123
.400	-.991		-1.084	.223	-1.212	.176	-1.136	-.026
.550	-.871	.307	-.877	.239	-1.213	.193	-.721	-.095
.700	-.825	.305	-.813	.269	-.790	.255	-.548	-.134
.850	-.811	.111	-.779	.355	-.797	.448	-.415	-.171
.950	-.871	.035	-.783	.497	-.808	.544	-.298	-.188

TABLE 13 (B) - CONTINUED

PRESSURE PROFILE ON WING

REAR NOZZLES DEFLECTED 90 DEGREES WITH FLAPS

ALPHA = 10.04 DEG Q = 2037 N/SC.M. (42.54 LBF/SQ.FT.) EFFECT. VELOC. RATIO = .293

	ETA = .250		ETA = .387		ETA = .524		ETA = .800	
X/C	CP		CP		CP		CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			-.914		-1.342		-2.451	
.010			.074	.656				
.015					-1.643	.942		
.025			-1.983	.978	-1.713	.955	-2.131	.904
.050			-2.065	.922	-1.713	.842	-2.087	.772
.100			-2.024	.756	-1.686	.685	-2.109	.548
.150			-1.963	.635	-1.775	.545	-2.106	.380
.250	-1.193		-1.853	.433	-1.813	.370	-2.147	.196
.400	-1.203		-1.819	.282	-1.792	.242	-1.596	.025
.550	-1.091	.352	-1.420	.298	-1.615	.220	-1.086	-.064
.700	-.909	.339	-1.055	.301	-1.239	.288	-.708	-.110
.850	-.886	.128	-.952	.381	-1.112	.471	-.465	-.169
.950	-.886	.081	-.880	.491	-.960	.546	-.325	-.193

ALPHA = 12.00 DEG Q = 2036 N/SC.M. (42.52 LBF/SQ.FT.) EFFECT. VELOC. RATIO = .293

	ETA = .250		ETA = .387		ETA = .524		ETA = .800	
X/C	CP		CP		CP		CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			-.821		-1.439		-2.513	
.010			.022	.594				
.015					-1.515	.911		
.025			-1.586	.965	-1.552	.957	-2.001	.909
.050			-1.666	.930	-1.539	.880	-2.063	.806
.100			-1.541	.793	-1.556	.726	-2.011	.600
.150			-1.594	.680	-1.546	.600	-2.021	.425
.250	-1.416		-1.733	.489	-1.624	.414	-2.161	.241
.400	-1.486		-1.748	.330	-1.733	.274	-2.054	.060
.550	-1.443	.400	-1.659	.310	-1.681	.248	-1.631	-.049
.700	-1.206	.393	-1.488	.317	-1.530	.301	-1.055	-.106
.850	-1.022	.178	-1.229	.395	-1.403	.483	-.583	-.175
.950	-.884	.136	-1.042	.531	-1.165	.563	-.406	-.203

ALPHA = 14.00 DEG Q = 2037 N/SQ.M. (42.54 LBF/SQ.FT.) EFFECT. VELOC. RATIO = .292

	ETA = .250		ETA = .387		ETA = .524		ETA = .800	
X/C	CP		CP		CP		CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			-.847		-1.454		-2.311	
.010			-.043	.551				
.015					-1.424	.872		
.025			-1.332	.954	-1.433	.954	-1.911	.904
.050			-1.319	.951	-1.438	.910	-1.946	.812
.100			-1.337	.834	-1.420	.761	-1.853	.626
.150			-1.303	.718	-1.458	.634	-1.932	.466
.250	-1.266		-1.377	.523	-1.467	.454	-1.904	.263
.400	-1.408		-1.526	.365	-1.652	.304	-1.946	.073
.550	-1.478	.424	-1.565	.337	-1.602	.263	-1.881	-.045
.700	-1.463	.416	-1.462	.335	-1.583	.310	-1.433	-.118
.850	-1.225	.202	-1.394	.398	-1.412	.489	-1.047	-.204
.950	-1.136	.160	-1.267	.542	-1.301	.564	-.589	-.278

TABLE 13 (B) - CONCLUDED

PRESSURE PROFILE ON WING

REAR NOZZLES DEFLECTED 90 DEGREES WITH FLAPS

ALPHA = 16.00 DEG Q = 2033 N/SC.M. (42.47 LBF/SQ.FT.) EFFECT. VELOC. RATIO = .292

X/C	ETA = .250 CP		ETA = .387 CP		ETA = .524 CP		ETA = .800 CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			-.952		-1.368		-2.101	
.010			-.114	.430				
.015					-1.321	.828		
.025			-1.251	.924	-1.278	.938	-1.749	.897
.050			-1.221	.958	-1.290	.921	-1.751	.842
.100			-1.239	.872	-1.331	.790	-1.674	.661
.150			-1.240	.757	-1.328	.667	-1.711	.498
.250	-1.237		-1.290	.565	-1.373	.489	-1.712	.292
.400	-1.335		-1.375	.405	-1.476	.334	-1.800	.102
.550	-1.446	.419	-1.396	.340	-1.477	.286	-1.804	-.039
.700	-1.456	.404	-1.432	.343	-1.436	.318	-1.665	-.142
.850	-1.381	.186	-1.390	.399	-1.373	.489	-1.287	-.272
.950	-1.194	.157	-1.327	.541	-1.343	.562	-1.069	-.453

ALPHA = 17.99 DEG Q = 2037 N/SQ.M. (42.54 LBF/SQ.FT.) EFFECT. VELOC. RATIO = .294

X/C	ETA = .250 CP		ETA = .387 CP		ETA = .524 CP		ETA = .800 CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			-1.169		-1.414		-1.758	
.010			-.180	.227				
.015					-1.227	.778		
.025			-1.258	.890	-1.246	.928	-1.565	.882
.050			-1.251	.962	-1.233	.947	-1.622	.850
.100			-1.263	.909	-1.246	.838	-1.580	.689
.150			-1.263	.802	-1.271	.724	-1.620	.536
.250	-1.284		-1.297	.613	-1.265	.546	-1.609	.339
.400	-1.347		-1.335	.427	-1.359	.367	-1.665	.125
.550	-1.376	.459	-1.365	.390	-1.420	.320	-1.728	-.025
.700	-1.415	.430	-1.354	.364	-1.370	.346	-1.669	-.151
.850	-1.368	.189	-1.328	.409	-1.341	.497	-1.507	-.347
.950	-1.255	.177	-1.336	.554	-1.366	.578	-1.267	-.554

ALPHA = 20.01 DEG Q = 2039 N/SQ.M. (42.58 LBF/SQ.FT.) EFFECT. VELOC. RATIO = .294

X/C	ETA = .250 CP		ETA = .387 CP		ETA = .524 CP		ETA = .800 CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			-1.405		-1.503		-1.542	
.010			-.274	.105				
.015					-1.291	.685		
.025			-1.335	.816	-1.272	.905	-1.420	.875
.050			-1.304	.953	-1.275	.947	-1.486	.864
.100			-1.294	.928	-1.283	.865	-1.466	.714
.150			-1.316	.860	-1.294	.771	-1.515	.591
.250	-1.348		-1.314	.664	-1.325	.594	-1.504	.376
.400	-1.355		-1.373	.487	-1.353	.415	-1.554	.166
.550	-1.377	.522	-1.376	.425	-1.361	.341	-1.589	-.018
.700	-1.342	.479	-1.401	.386	-1.345	.365	-1.596	-.160
.850	-1.394	.223	-1.383	.436	-1.390	.507	-1.528	-.382
.950	-1.251	.192	-1.353	.566	-1.408	.584	-1.372	-.663

TABLE 13 (C)

PRESSURE PROFILE ON WING

REAR NOZZLES DEFLECTED 90 DEGREES WITH FLAPS

ALPHA = -.00 DEG Q = 910 N/SQ.M. (19.01 LBF/SQ.FT.) EFFECT. VELOC. RATIO = .196

X/C	ETA = .250 CP		ETA = .387 CP		ETA = .524 CP		ETA = .800 CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			.901		.514		.285	
.010			-1.428	.772				
.015					-1.646	.519		
.025			-1.378	.353	-1.549	.384	-1.295	.221
.050			-1.070	.233	-1.113	.233	-.890	.105
.100			-.871	.017	-.857	.122	-.760	-.026
.150			-.862	-.123	-.897	-.014	-.764	-.177
.250	-.627		-.838	-.134	-.804	-.143	-.701	-.263
.400	-.780		-.829	-.158	-.818	-.188	-.640	-.312
.550	-.750	-.027	-.767	-.070	-.732	-.093	-.533	-.293
.700	-.752	.046	-.756	.030	-.727	.049	-.412	-.269
.850	-.850	-.142	-.806	.176	-.767	.302	-.317	-.230
.950	-1.038	-.179	-.863	.327	-.886	.415	-.238	-.185

ALPHA = 4.00 DEG Q = 904 N/SQ.M. (18.87 LBF/SQ.FT.) EFFECT. VELOC. RATIO = .195

X/C	ETA = .250 CP		ETA = .387 CP		ETA = .524 CP		ETA = .800 CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			-.098		-1.446		-2.178	
.010			-1.396	.930				
.015					-3.017	.896		
.025			-3.052	.758	-3.008	.767	-2.698	.636
.050			-1.713	.597	-1.917	.604	-1.432	.475
.100			-1.249	.282	-1.416	.338	-1.179	.202
.150			-1.198	.172	-1.239	.216	-1.092	.066
.250	-.768		-1.045	.016	-1.096	.049	-.988	-.118
.400	-.895		-.988	-.057	-.976	-.069	-.840	-.232
.550	-.849	.024	-.892	-.016	-.877	-.043	-.641	-.262
.700	-.822	.020	-.847	.050	-.826	.042	-.504	-.256
.850	-.902	-.209	-.886	.182	-.845	.302	-.391	-.251
.950	-1.035	-.236	-.944	.338	-.957	.410	-.298	-.236

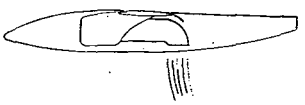
ALPHA = 7.35 DEG Q = 911 N/SQ.M. (19.03 LBF/SQ.FT.) EFFECT. VELOC. RATIO = .196

X/C	ETA = .250 CP		ETA = .387 CP		ETA = .524 CP		ETA = .800 CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			-1.817		-4.195		-4.102	
.010			-1.454	.661				
.015					-5.224	.867		
.025			-3.336	.929	-3.685	.894	-3.104	.846
.050			-2.355	.796	-2.523	.772	-2.144	.675
.100			-1.674	.549	-1.886	.533	-1.632	.417
.150			-1.513	.362	-1.623	.395	-1.399	.222
.250	-.890		-1.247	.175	-1.335	.165	-1.179	.001
.400	-.975		-1.132	.041	-1.147	.030	-.964	-.143
.550	-.905	.092	-.977	.046	-.966	.024	-.746	-.218
.700	-.875	.059	-.921	.075	-.909	.085	-.589	-.238
.850	-.935	-.234	-.938	.163	-.920	.321	-.436	-.264
.950	-1.048	-.219	-.992	.343	-.995	.419	-.348	-.261

TABLE 13 (C) - CONTINUED

PRESSURE PROFILE ON WING

REAR NOZZLES DEFLECTED 90 DEGREES WITH FLAPS



ALPHA = 10.02 DEG Q = 904 N/SQ.M. (18.87 LBF/SQ.FT.) EFFECT. VELOC. RATIO = .195

X/C	ETA = .250 CP		ETA = .387 CP		ETA = .524 CP		ETA = .800 CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			-1.183		-1.645		-2.844	
.010			-1.514	.592				
.015					-2.032	.892		
.025			-2.381	.936	-1.964	.910	-2.381	.875
.050			-2.389	.871	-2.004	.807	-2.401	.719
.100			-2.374	.674	-2.128	.577	-2.424	.502
.150			-2.265	.509	-2.121	.437	-2.417	.332
.250	-1.188		-2.208	.317	-2.063	.260	-2.384	.100
.400	-1.232		-1.811	.181	-1.858	.098	-1.651	-.092
.550	-1.162	.188	-1.279	.165	-1.459	.084	-.883	-.177
.700	-1.021	.127	-1.127	.156	-1.266	.161	-.718	-.218
.850	-1.004	-.168	-1.038	.209	-1.117	.360	-.551	-.276
.950	-1.074	-.147	-1.079	.347	-1.142	.420	-.430	-.303

ALPHA = 11.97 DEG Q = 902 N/SQ.M. (18.85 LBF/SQ.FT.) EFFECT. VELOC. RATIO = .195

X/C	ETA = .250 CP		ETA = .387 CP		ETA = .524 CP		ETA = .800 CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			-1.272		-1.678		-2.913	
.010			-1.546	.458				
.015					-1.822	.846		
.025			-1.960	.926	-1.869	.909	-2.366	.856
.050			-2.077	.900	-1.862	.823	-2.365	.736
.100			-2.167	.753	-1.857	.663	-2.349	.528
.150			-2.180	.602	-1.919	.502	-2.372	.347
.250	-1.516		-2.195	.352	-1.919	.312	-2.436	.134
.400	-1.565		-2.054	.176	-1.986	.154	-2.106	-.055
.550	-1.332	.183	-1.564	.166	-1.824	.114	-1.561	-.148
.700	-1.155	.137	-1.541	.174	-1.558	.170	-.881	-.222
.850	-1.084	-.184	-1.306	.231	-1.352	.373	-.613	-.291
.950	-1.115	-.135	-1.153	.382	-1.212	.448	-.453	-.319

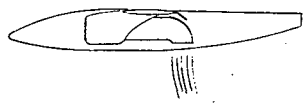
ALPHA = 14.00 DEG Q = 906 N/SQ.M. (18.91 LBF/SQ.FT.) EFFECT. VELOC. RATIO = .195

X/C	ETA = .250 CP		ETA = .387 CP		ETA = .524 CP		ETA = .800 CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			-1.228		-1.762		-2.881	
.010			-1.550	.442				
.015					-1.729	.820		
.025			-1.693	.895	-1.767	.903	-2.343	.861
.050			-1.708	.955	-1.720	.853	-2.284	.785
.100			-1.722	.778	-1.790	.689	-2.312	.569
.150			-1.759	.634	-1.780	.562	-2.291	.397
.250	-1.663		-1.873	.409	-1.854	.382	-2.310	.196
.400	-1.767		-1.922	.229	-1.951	.188	-2.384	-.006
.550	-1.708	.212	-1.911	.201	-1.928	.146	-1.963	-.136
.700	-1.361	.155	-1.767	.193	-1.789	.192	-1.355	-.215
.850	-1.495	-.113	-1.585	.244	-1.642	.385	-.777	-.295
.950	-1.080	-.084	-1.415	.424	-1.381	.456	-.489	-.335

TABLE 13 (C) - CONCLUDED

PRESSURE PROFILE ON WING

REAR NOZZLES DEFLECTED 90 DEGREES WITH FLAPS



ALPHA = 16.00 DEG Q = 908 N/SQ.M. (18.96 LBF/SQ.FT.) EFFECT. VELOC. RATIO = .196

X/C	ETA = .250 CP		ETA = .387 CP		ETA = .524 CP		ETA = .800 CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			-1.326		-1.833		-2.597	
.010			-1.542	.287				
.015					-1.715	.749		
.025			-1.562	.874	-1.706	.890	-2.234	.860
.050			-1.524	.914	-1.701	.877	-2.184	.806
.100			-1.597	.796	-1.752	.733	-2.191	.628
.150			-1.628	.674	-1.729	.610	-2.234	.461
.250	-1.470		-1.632	.465	-1.793	.407	-2.309	.234
.400	-1.672		-1.743	.250	-1.834	.232	-2.277	.019
.550	-1.766	.255	-1.852	.222	-1.860	.170	-2.125	-.117
.700	-1.725	.184	-1.774	.202	-1.843	.203	-1.780	-.209
.850	-1.607	-.073	-1.620	.244	-1.719	.376	-1.188	-.318
.950	-1.230	-.022	-1.530	.437	-1.547	.465	-.664	-.378

ALPHA = 17.98 DEG Q = 907 N/SQ.M. (18.95 LBF/SQ.FT.) EFFECT. VELOC. RATIO = .195

X/C	ETA = .250 CP		ETA = .387 CP		ETA = .524 CP		ETA = .800 CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			-1.444		-1.859		-2.328	
.010			-1.562	.121				
.015					-1.635	.709		
.025			-1.453	.831	-1.656	.883	-2.050	.840
.050			-1.460	.909	-1.599	.899	-2.106	.819
.100			-1.555	.826	-1.639	.779	-2.088	.659
.150			-1.554	.713	-1.642	.663	-2.099	.470
.250	-1.458		-1.547	.498	-1.745	.447	-2.212	.270
.400	-1.619		-1.705	.304	-1.794	.255	-2.166	.041
.550	-1.660	.270	-1.681	.256	-1.823	.189	-2.155	-.121
.700	-1.737	.219	-1.674	.222	-1.816	.196	-1.982	-.230
.850	-1.685	-.080	-1.586	.248	-1.683	.368	-1.495	-.387
.950	-1.438	-.028	-1.582	.433	-1.640	.472	-1.158	-.523

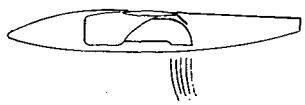
ALPHA = 20.00 DEG Q = 911 N/SQ.M. (19.03 LBF/SQ.FT.) EFFECT. VELOC. RATIO = .196

X/C	ETA = .250 CP		ETA = .387 CP		ETA = .524 CP		ETA = .800 CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			-1.553		-1.863		-2.122	
.010			-1.570	-.030				
.015					-1.565	.609		
.025			-1.494	.792	-1.583	.847	-2.027	.844
.050			-1.508	.906	-1.515	.901	-1.973	.842
.100			-1.486	.864	-1.589	.799	-1.994	.678
.150			-1.519	.749	-1.600	.693	-2.015	.531
.250	-1.529		-1.577	.564	-1.621	.509	-1.991	.330
.400	-1.603		-1.644	.335	-1.729	.293	-2.119	.060
.550	-1.682	.309	-1.678	.286	-1.774	.210	-2.148	-.107
.700	-1.698	.244	-1.645	.243	-1.751	.222	-2.063	-.243
.850	-1.705	-.045	-1.607	.271	-1.654	.376	-1.783	-.427
.950	-1.555	.013	-1.643	.447	-1.721	.471	-1.400	-.680

TABLE 13 (D)

PRESSURE PROFILE ON WING

REAR NOZZLES DEFLECTED 90 DEGREES WITH FLAPS



ALPHA = 0.00 DEG Q = 231 N/SQ.M. (4.83 LBF/SQ.FT.) EFFECT. VELOC. RATIO = .099

X/C	ETA = .250 CP		ETA = .387 CP		ETA = .524 CP		ETA = .800 CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			.641		.635		.478	
.010			-5.648	.144				
.015					-.972	.029		
.025			-1.108	-.250	-1.047	-.264	-1.060	-.461
.050			-.979	-.298	-.979	-.393	-.917	-.420
.100			-.863	-.523	-.849	-.461	-.788	-.509
.150			-.904	-.597	-.965	-.557	-.808	-.638
.250	-.802		-.904	-.645	-.972	-.655	-.829	-.694
.400	-.899		-.958	-.618	-.952	-.668	-.833	-.714
.550	-.882	-.585	-.909	-.519	-.912	-.519	-.763	-.664
.700	-.892	-.569	-.905	-.364	-.902	-.338	-.681	-.589
.850	-.971	-.638	-.938	-.137	-.919	-.028	-.579	-.575
.950	-1.087	-.608	-.988	.032	-1.047	.108	-.516	-.503

ALPHA = 3.97 DEG Q = 231 N/SQ.M. (4.83 LBF/SQ.FT.) EFFECT. VELOC. RATIO = .099

X/C	ETA = .250 CP		ETA = .387 CP		ETA = .524 CP		ETA = .800 CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			.380		-.390		-.711	
.010			-4.672	.639				
.015					-2.449	.476		
.025			-1.836	.353	-2.272	.305	-1.904	.244
.050			-1.822	.073	-1.884	.101	-1.577	-.022
.100			-1.147	-.097	-1.284	-.104	-1.174	-.165
.150			-.990	-.302	-1.236	-.254	-1.099	-.336
.250	-.847		-1.072	-.437	-1.120	-.371	-1.086	-.474
.400	-.989		-1.049	-.474	-1.075	-.441	-.943	-.573
.550	-.956	-.461	-.960	-.391	-.963	-.378	-.814	-.599
.700	-.923	-.467	-.946	-.276	-.950	-.299	-.692	-.560
.850	-.976	-.666	-.993	-.097	-.966	.022	-.599	-.497
.950	-1.062	-.527	-1.032	.058	-1.089	.141	-.520	-.467

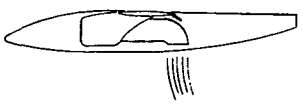
ALPHA = 8.04 DEG Q = 227 N/SQ.M. (4.74 LBF/SQ.FT.) EFFECT. VELOC. RATIO = .098

X/C	ETA = .250 CP		ETA = .387 CP		ETA = .524 CP		ETA = .800 CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			-.927		-2.134		-2.557	
.010			-4.173	.647				
.015					-4.097	.661		
.025			-3.487	.634	-3.924	.578	-3.015	.495
.050			-1.981	.453	-2.287	.377	-1.940	.294
.100			-1.614	.148	-1.718	.169	-1.565	.030
.150			-1.489	-.039	-1.524	.002	-1.406	-.122
.250	-.990		-1.239	-.219	-1.323	-.199	-1.274	-.317
.400	-1.056		-1.167	-.354	-1.228	-.364	-1.066	-.471
.550	-1.006	-.394	-1.066	-.317	-1.090	-.337	-.892	-.522
.700	-.982	-.414	-1.016	-.253	-1.026	-.233	-.767	-.518
.850	-1.026	-.649	-1.033	-.128	-1.023	.036	-.623	-.522
.950	-1.083	-.606	-1.060	.077	-1.113	.154	-.528	-.488

TABLE 13 (D) - CONTINUED

PRESSURE PROFILE ON WING

REAR NOZZLES DEFLECTED 90 DEGREES WITH FLAPS



ALPHA = 10.03 DEG Q = 228 N/SQ.M. (4.77 LBF/SQ.FT.) EFFECT. VELOC. RATIO = .098

	ETA = .250		ETA = .387		ETA = .524		ETA = .800	
X/C	CP		CP		CP		CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			-1.686		-3.639		-4.288	
.010			-3.825	.343				
.015					-5.143	.653		
.025			-3.115	.667	-4.177	.619	-3.653	.563
.050			-2.990	.494	-2.549	.425	-2.266	.412
.100			-1.762	.294	-1.859	.253	-1.776	.129
.150			-1.624	.087	-1.700	.101	-1.562	-.085
.250	-.996		-1.362	-.146	-1.465	-.146	-1.348	-.246
.400	-1.085		-1.219	-.293	-1.279	-.303	-1.075	-.423
.550	-1.035	-.386	-1.069	-.259	-1.112	-.286	-.922	-.490
.700	-.978	-.400	-1.012	-.209	-1.039	-.199	-.764	-.514
.850	-1.015	-.607	-1.019	-.169	-1.032	.055	-.631	-.510
.950	-1.089	-.634	-1.045	.078	-1.102	.172	-.560	-.510

ALPHA = 11.99 DEG Q = 224 N/SQ.M. (4.68 LBF/SQ.FT.) EFFECT. VELOC. RATIO = .097

	ETA = .250		ETA = .387		ETA = .524		ETA = .800	
X/C	CP		CP		CP		CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			-1.262		-1.747		-3.323	
.010			-4.040	.363				
.015					-2.500	.651		
.025			-2.711	.707	-2.240	.637	-2.859	.602
.050			-2.598	.588	-2.317	.482	-2.725	.440
.100			-2.521	.398	-2.169	.264	-2.113	.187
.150			-2.570	.173	-2.310	.138	-2.535	-.024
.250	-1.269		-2.120	-.068	-2.106	-.085	-2.148	-.232
.400	-1.227		-1.275	-.266	-1.701	-.201	-1.418	-.436
.550	-1.142	-.361	-1.224	-.204	-1.231	-.245	-1.009	-.487
.700	-1.074	-.395	-1.108	-.146	-1.166	-.153	-.855	-.521
.850	-1.063	-.579	-1.104	-.102	-1.145	.068	-.743	-.562
.950	-1.077	-.661	-1.108	.116	-1.227	.164	-.665	-.576

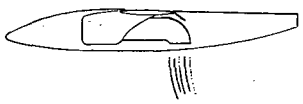
ALPHA = 13.98 DEG Q = 220 N/SQ.M. (4.60 LBF/SQ.FT.) EFFECT. VELOC. RATIO = .096

	ETA = .250		ETA = .387		ETA = .524		ETA = .800	
X/C	CP		CP		CP		CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			-1.414		-1.729		-2.903	
.010			-4.879	.304				
.015					-1.951	.626		
.025			-2.280		-2.116	.662	-2.495	.648
.050			-2.288	.598	-2.116	.541	-2.474	.519
.100			-2.259	.447	-2.044	.347	-2.474	.261
.150			-2.316	.254	-2.073	.197	-2.488	.061
.250	-1.450		-2.345	-.011	-1.994	-.021	-2.381	-.160
.400	-1.527		-2.030	-.191	-2.075	-.163	-1.864	-.365
.550	-1.229	-.271	-1.475	-.188	-1.756	-.209	-1.468	-.465
.700	-1.114	-.299	-1.316	-.146	-1.368	-.101	-1.069	-.490
.850	-1.086	-.531	-1.191	-.094	-1.305	.090	-.785	-.514
.950	-1.086	-.569	-1.201	.083	-1.253	.187	-.667	-.535

TABLE 13 (D) - CONCLUDED

PRESSURE PROFILE ON WING

REAR NOZZLES DEFLECTED 90 DEGREES WITH FLAPS



ALPHA = 16.05 DEG Q = 227 N/SQ.M. (4.74 LBF/SQ.FT.) EFFECT. VELOC. RATIO = .098

X/C	ETA = .250 CP		ETA = .387 CP		ETA = .524 CP		ETA = .800 CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			-1.404		-1.758		-2.819	
.010			-4.949	.240				
.015					-1.827	.649		
.025			-1.855	.677	-1.876	.697	-2.362	.656
.050			-1.966	.663	-1.793	.593	-2.251	.538
.100			-1.737	.496	-1.821	.413	-2.299	.316
.150			-2.077	.330	-1.897	.267	-2.292	.136
.250	-1.758		-2.015	.045	-1.966	.055	-2.313	-.076
.400	-1.582		-1.996	-.140	-1.992	-.113	-2.298	-.288
.550	-1.448	-.274	-1.929	-.130	-1.922	-.150	-1.760	-.406
.700	-1.391	-.271	-1.690	-.090	-1.643	-.090	-1.552	-.463
.850	-1.219	-.483	-1.502	-.049	-1.582	.129	-.920	-.520
.950	-1.098	-.503	-1.256	.163	-1.307	.233	-.732	-.560

ALPHA = 18.01 DEG Q = 227 N/SQ.M. (4.74 LBF/SQ.FT.) EFFECT. VELOC. RATIO = .098

X/C	ETA = .250 CP		ETA = .387 CP		ETA = .524 CP		ETA = .800 CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			-1.095		-1.582		-2.735	
.010			-4.764	.301				
.015					-1.665	.649		
.025			-1.512	.697	-1.554	.732	-2.145	.670
.050			-1.575	.690	-1.776	.670	-2.131	.579
.100			-1.582	.558	-1.686	.496	-2.242	.364
.150			-1.672	.392	-1.735	.357	-2.096	.190
.250	-1.484		-1.707	.132	-1.651	.119	-2.068	-.016
.400	-1.629		-1.733	-.080	-1.787	-.036	-2.218	-.231
.550	-1.683	-.157	-1.700	-.080	-1.794	-.097	-2.063	-.359
.700	-1.578	-.181	-1.669	-.049	-1.723	-.006	-1.673	-.440
.850	-1.296	-.346	-1.538	-.006	-1.585	.176	-1.097	-.528
.950	-1.033	-.379	-1.353	.217	-1.420	.294	-.885	-.585

ALPHA = 20.01 DEG Q = 225 N/SQ.M. (4.70 LBF/SQ.FT.) EFFECT. VELOC. RATIO = .097

X/C	ETA = .250 CP		ETA = .387 CP		ETA = .524 CP		ETA = .800 CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			-1.221		-1.591		-2.480	
.010			-4.586	.235				
.015					-1.514	.662		
.025			-1.437	.711	-1.458	.753	-1.864	.704
.050			-1.416	.753	-1.458	.704	-1.850	.641
.100			-1.416	.585	-1.374	.508	-1.794	.431
.150			-1.444	.452	-1.444	.403	-1.920	.249
.250	-1.409		-1.479	.189	.536	.196	-1.899	.026
.400	-1.496		-1.520	-.031	-1.581	.010	-1.954	-.191
.550	-1.561	-.123	-1.578	-.031	-1.598	-.041	-2.069	-.333
.700	-1.591	-.163	-1.581	-.014	-1.645		-1.734	-.431
.850	-1.510	-.364	-1.527	.016	-1.506	.189	-1.364	-.550
.950	-1.354	-.394	-1.479	.298	-1.442	.325	-1.266	-.652

TABLE 14 (A)

PRESSURE PROFILE ON WING

FORWARD NOZZLES DEFLECTED 90 DEGREES WITH LIFT JET

ALPHA = 0.00 DEG Q = 2038 N/SC.M. (42.57 LBF/SQ.FT.) EFFECT. VELOC. RATIO = .283

	ETA = .250		ETA = .387		ETA = .524		ETA = .800	
X/C	CP		CP		CP		CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			.588		.871		.960	
.010			.322	-.215				
.015					-.549	-.317		
.025			-.777	-.654	-.561	-.707	-.580	-.235
.050			-.571	-.653	-.681	-.541	-.501	-.234
.100			-.521	-.873	-.622	-.692	-.468	-.325
.150			-.556	-.957	-.520	-.862	-.502	-.466
.250	-.405		-.543	-1.204	-.547	-.886	-.535	-.532
.400	-.530		-.555	-1.350	-.543	-.976	-.470	-.549
.550	-.468	-1.617	-.459	-1.351	-.456	-.948	-.373	-.474
.700	-.431	-1.799	-.399	-1.231	-.340	-.784	-.245	-.376
.850	-.484	-1.908	-.373	-.961	-.265	-.565	-.157	-.248
.950	-.672	-1.718	-.377	-.600	-.237	-.350	-.105	-.130

ALPHA = 3.96 DEG Q = 2033 N/SQ.M. (42.46 LBF/SQ.FT.) EFFECT. VELOC. RATIO = .283

	ETA = .250		ETA = .387		ETA = .524		ETA = .800	
X/C	CP		CP		CP		CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			.977		.771		-.199	
.010			.384	.603				
.015					-1.780	.480		
.025			-1.418	-.033	-1.718	.092	-1.465	.326
.050			-1.163	-.061	-1.239	-.050	-1.133	.190
.100			-.873	-.470	-.914	-.243	-.859	-.085
.150			-.842	-.781	-.828	-.539	-.762	-.201
.250	-.528		-.742	-1.027	-.770	-.725	-.722	-.348
.400	-.603		-.670	-1.335	-.669	-.912	-.590	-.424
.550	-.519	-1.615	-.525	-1.361	-.521	-.898	-.450	-.427
.700	-.450	-1.640	-.433	-1.209	-.405	-.785	-.305	-.347
.850	-.461	-1.806	-.380	-.956	-.309	-.570	-.205	-.270
.950	-.596	-1.435	-.370	-.568	-.248	-.379	-.125	-.153

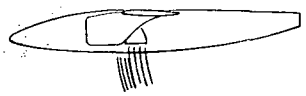
ALPHA = 8.02 DEG Q = 2040 N/SQ.M. (42.61 LBF/SQ.FT.) EFFECT. VELOC. RATIO = .283

	ETA = .250		ETA = .387		ETA = .524		ETA = .800	
X/C	CP		CP		CP		CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			.094		-.983		-2.442	
.010			.380	.994				
.015					-3.346	.909		
.025			-3.013	.620	-3.249	.671	-2.687	.752
.050			-1.712	.359	-1.861	.477	-1.784	.557
.100			-1.288	-.041	-1.501	.042	-1.362	.254
.150			-1.171	-.359	-1.273	-.184	-1.123	.040
.250	-.641		-.969	-.875	-1.042	-.526	-.955	-.159
.400	-.678		-.801	-1.240	-.832	-.731	-.763	-.293
.550	-.558	-1.633	-.619	-1.314	-.617	-.782	-.533	-.349
.700	-.484	-1.697	-.505	-1.129	-.454	-.726	-.375	-.310
.850	-.466	-1.641	-.424	-.908	-.330	-.589	-.249	-.259
.950	-.553	-1.257	-.387	-.569	-.270	-.387	-.162	-.173

TABLE 14 (A) - CONTINUED

PRESSURE PROFILE ON WING

FORWARD NOZZLES DEFLECTED 90 DEGREES WITH LIFT JET



ALPHA = 9.82 DEG Q = 2032 N/SQ.M. (42.44 LBF/SQ.FT.) EFFECT. VELOC. RATIO = .282

X/C	ETA = .250 CP		ETA = .387 CP		ETA = .524 CP		ETA = .800 CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			-.398		-2.260		-3.721	
.010			.329	.971				
.015					-4.592	.965		
.025			-3.199	.774	-3.461	.816	-2.828	.882
.050			-2.138	.535	-2.350	.565	-2.095	.663
.100			-1.419	.104	-1.631	.236	-1.503	.386
.150			-1.285	-.246	-1.372	-.067	-1.288	.141
.250	-.686		-1.034	-.790	-1.102	-.386	-1.053	-.074
.400	-.721		-.833	-1.236	-.881	-.704	-.805	-.251
.550	-.588	-1.632	-.635	-1.280	-.657	-.789	-.574	-.307
.700	-.496	-1.677	-.515	-1.151	-.488	-.717	-.403	-.281
.850	-.467	-1.509	-.416	-.895	-.353	-.594	-.262	-.249
.950	-.546	-1.119	-.379	-.573	-.279	-.396	-.170	-.179

ALPHA = 12.01 DEG Q = 2040 N/SQ.M. (42.61 LBF/SQ.FT.) EFFECT. VELOC. RATIO = .283

X/C	ETA = .250 CP		ETA = .387 CP		ETA = .524 CP		ETA = .800 CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			-.634		-2.027		-1.493	
.010			.268	.964				
.015					-4.388	.967		
.025			-3.269	.882	-2.174	.884	-1.853	.862
.050			-1.918	.658	-2.295	.605	-1.887	.659
.100			-1.941	.230	-1.984	.284	-1.851	.388
.150			-1.851	-.134	-1.562	.017	-1.723	.186
.250	-.773		-1.098	-.693	-1.239	-.356	-1.335	-.042
.400	-.759		-.868	-1.194	-.947	-.670	-1.043	-.238
.550	-.627	-1.542	-.672	-1.315	-.712	-.774	-.769	-.310
.700	-.536	-1.636	-.547	-1.150	-.582	-.731	-.497	-.297
.850	-.478	-1.481	-.432	-.923	-.400	-.668	-.573	-.307
.950	-.511	-1.039	-.428	-.599	-.427	-.494	-.322	-.265

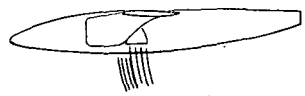
ALPHA = 14.06 DEG Q = 2033 N/SQ.M. (42.47 LBF/SQ.FT.) EFFECT. VELOC. RATIO = .282

X/C	ETA = .250 CP		ETA = .387 CP		ETA = .524 CP		ETA = .800 CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			.102		-.470		-1.419	
.010			.239	.969				
.015					-1.404	.946		
.025			-1.206	.853	-1.125	.801	-1.335	.855
.050			-1.265	.618	-1.223	.569	-1.352	.650
.100			-1.215	.250	-1.124	.264	-1.402	.387
.150			-1.213	-.134	-1.262	.008	-1.352	.192
.250	-.869		-1.284	-.712	-1.316	-.377	-1.415	-.049
.400	-.941		-1.181	-1.160	-1.145	-.688	-1.288	-.248
.550	-.807	-1.635	-1.056	-1.319	-1.043	-.809	-1.114	-.338
.700	-.694	-1.637	-.900	-1.167	-.892	-.798	-.975	-.363
.850	-.671	-1.549	-.711	-.994	-.701	-.788	-.799	-.374
.950	-.607	-1.122	-.564	-.695	-.663	-.657	-.640	-.450

TABLE 14 (A) - CONCLUDED

PRESSURE PROFILE ON WING

FORWARD NOZZLES DEFLECTED 90 DEGREES WITH LIFT JET



ALPHA = 16.00 DEG Q = 2039 N/SQ.M. (42.59 LBF/SQ.FT.) EFFECT. VELOC. RATIO = .282

X/C	ETA = .250 CP		ETA = .387 CP		ETA = .524 CP		ETA = .800 CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			.197		-.355		-1.411	
.010			.223	.960				
.015					-.861	.958		
.025			-.844	.879	-.837	.834	-1.155	.864
.050			-.864	.678	-.867	.608	-1.234	.687
.100			-.807	.287	-.871	.305	-1.181	.441
.150			-.796	-.073	-.865	.053	-1.199	.225
.250	-.883		-.911	-.629	-.906	-.328	-1.197	-.021
.400	-.893		-.948	-1.159	-1.065	-.712	-1.286	-.249
.550	-.937	-1.586	-.984	-1.295	-1.017	-.841	-1.243	-.370
.700	-.999	-1.640	-.993	-1.230	-1.010	-.884	-1.145	-.415
.850	-.917	-1.505	-.906	-1.100	-.931	-.879	-1.007	-.461
.950	-.721	-1.095	-.743	-.880	-.861	-.802	-.882	-.491

ALPHA = 18.00 DEG Q = 2037 N/SQ.M. (42.54 LBF/SQ.FT.) EFFECT. VELOC. RATIO = .282

X/C	ETA = .250 CP		ETA = .387 CP		ETA = .524 CP		ETA = .800 CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			.018		-.417		-1.391	
.010			.213	.931				
.015					-.851	.969		
.025			-.749	.945	-.820	.878	-1.132	.886
.050			-.773	.712	-.842	.680	-1.164	.733
.100			-.770	.374	-.811	.397	-1.139	.466
.150			-.777	.005	-.832	.128	-1.116	.286
.250	-.793		-.814	-.564	-.854	-.271	-1.141	.022
.400	-.885		-.898	-1.132	-.942	-.665	-1.215	-.246
.550	-.937	-1.499	-.963	-1.276	-.973	-.835	-1.220	-.375
.700	-1.017	-1.531	-.998	-1.239	-.976	-.905	-1.202	-.436
.850	-.957	-1.367	-.916	-1.148	-.970	-.913	-1.095	-.543
.950	-.878	-1.125	-.836	-.937	-.912	-.886	-.997	-.629

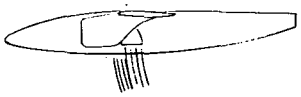
ALPHA = 19.98 DEG Q = 2038 N/SQ.M. (42.56 LBF/SQ.FT.) EFFECT. VELOC. RATIO = .281

X/C	ETA = .250 CP		ETA = .387 CP		ETA = .524 CP		ETA = .800 CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			-.150		-.560		-1.459	
.010			.201	.862				
.015					-.800	.978		
.025			-.754	.970	-.781	.926	-1.081	.913
.050			-.792	.799	-.784	.734	-1.098	.779
.100			-.733	.429	-.737	.451	-1.041	.529
.150			-.755	.102	-.805	.178	-1.058	.333
.250	-.794		-.763	-.494	-.840	-.220	-1.087	.060
.400	-.924		-.907	-1.117	-.904	-.620	-1.089	-.197
.550	-.937	-1.537	-.949	-1.284	-.965	-.832	-1.158	-.359
.700	-.974	-1.485	-.962	-1.227	-.975	-.903	-1.176	-.452
.850	-1.008	-1.424	-.922	-1.133	-.977	-.913	-1.114	-.578
.950	-.954	-1.124	-.857	-.963	-.918	-.874	-1.079	-.701

TABLE 14 (B)

PRESSURE PROFILE ON WING

FORWARD NOZZLES DEFLECTED 90 DEGREES WITH LIFT JET



ALPHA = .01 DEG Q = 911 N/SQ.M. (19.02 LBF/SQ.FT.) EFFECT. VELOC. RATIO = .188

X/C	ETA = .250 CP		ETA = .387 CP		ETA = .524 CP		ETA = .800 CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			-.663		-.080		.950	
.010			-.718	-1.634				
.015					.138	-1.497		
.025			-.208	-1.948	-.120	-1.568	-.407	-.891
.050			-.213	-1.623	-.172	-1.353	-.243	-.696
.100			-.284	-1.570	-.279	-1.322	-.298	-.632
.150			-.443	-1.831	-.391	-1.350	-.393	-.616
.250	-.346		-.424	-1.912	-.396	-1.452	-.391	-.773
.400	-.436		-.461	-2.012	-.461	-1.476	-.421	-.720
.550	-.395	-3.608	-.404	-2.182	-.388	-1.265	-.343	-.612
.700	-.362	-3.374	-.379	-1.831	-.335	-1.064	-.284	-.505
.850	-.412	-2.699	-.374	-1.352	-.289	-.722	-.166	-.326
.950	-.648	-2.000	-.425	-.795	-.270	-.453	-.130	-.176

ALPHA = 3.93 DEG Q = 909 N/SQ.M. (19.00 LBF/SQ.FT.) EFFECT. VELOC. RATIO = .188

X/C	ETA = .250 CP		ETA = .387 CP		ETA = .524 CP		ETA = .800 CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			.675		.911		.789	
.010			-.650	-.377				
.015					-.872	-.758		
.025			-.836	-.704	-.671	-.607	-1.075	-.060
.050			-.631	-.908	-.692	-.716	-.740	-.030
.100			-.576	-1.120	-.626	-.837	-.659	-.316
.150			-.631	-1.409	-.628	-.986	-.631	-.479
.250	-.444		-.621	-1.720	-.652	-1.214	-.623	-.592
.400	-.515		-.588	-2.017	-.592	-1.335	-.531	-.602
.550	-.442	-3.319	-.471	-1.960	-.474	-1.223	-.419	-.557
.700	-.394	-2.992	-.388	-1.813	-.374	-1.042	-.286	-.464
.850	-.421	-2.625	-.390	-1.238	-.299	-.721	-.198	-.326
.950	-.609	-1.743	-.415	-.727	-.286	-.442	-.146	-.180

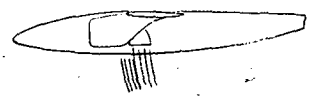
ALPHA = 8.06 DEG Q = 906 N/SQ.M. (18.92 LBF/SQ.FT.) EFFECT. VELOC. RATIO = .187

X/C	ETA = .250 CP		ETA = .387 CP		ETA = .524 CP		ETA = .800 CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			.965		.786		-.685	
.010			-.544	.563				
.015					-1.720	.360		
.025			-1.753	.057	-1.891	.096	-1.851	.429
.050			-1.306	-.289	-1.266	-.160	-1.461	.219
.100			-.943	-.689	-1.004	-.440	-1.118	.052
.150			-.864	-.965	-.925	-.751	-.906	-.153
.250	-.525		-.804	-1.448	-.828	-1.064	-.837	-.376
.400	-.590		-.661	-2.023	-.708	-1.211	-.669	-.467
.550	-.482	-3.170	-.535	-2.135	-.563	-1.197	-.501	-.503
.700	-.398	-2.905	-.443	-1.800	-.428	-1.020	-.353	-.420
.850	-.406	-2.475	-.402	-1.191	-.338	-.713	-.238	-.332
.950	-.546	-1.510	-.419	-.713	-.301	-.460	-.172	-.212

TABLE 14 (B)- CONTINUED

PRESSURE PROFILE ON WING

FORWARD NOZZLES DEFLECTED 90 DEGREES WITH LIFT JET



ALPHA = 9.99 DEG Q = 908 N/SQ.M. (18.95 LBF/SQ.FT.) EFFECT. VELOC. RATIO = .188

	ETA = .250		ETA = .387		ETA = .524		ETA = .800	
X/C	CP		CP		CP		CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			.790		.321		-1.885	
.010			-.458	.867				
.015					-2.819	.596		
.025			-2.048	.269	-2.284	.308	-2.588	.705
.050			-1.501	-.045	-1.626	.094	-1.712	.424
.100			-1.111	-.508	-1.283	-.293	-1.278	.144
.150			-1.033	-.904	-1.100	-.616	-1.114	-.062
.250	-.583		-.854	-1.356	-.942	-.910	-.935	-.269
.400	-.617		-.732	-1.906	-.777	-1.137	-.732	-.424
.550	-.504	-2.963	-.570	-2.062	-.584	-1.123	-.528	-.441
.700	-.408	-2.874	-.463	-1.714	-.440	-.993	-.370	-.390
.850	-.403	-2.530	-.420	-1.170	-.340	-.719	-.253	-.310
.950	-.524	-1.403	-.428	-.689	-.302	-.470	-.176	-.223

ALPHA = 12.01 DEG Q = 906 N/SQ.M. (18.93 LBF/SQ.FT.) EFFECT. VELOC. RATIO = .187

	ETA = .250		ETA = .387		ETA = .524		ETA = .800	
X/C	CP		CP		CP		CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			.453		-.633		-3.030	
.010			-.419	.973				
.015					-3.468	.838		
.025			-2.806	.565	-3.129	.617	-2.980	.874
.050			-1.733	.193	-1.966	.231	-2.057	.599
.100			-1.257	-.338	-1.403	-.112	-1.436	.234
.150			-1.189	-.744	-1.214	-.442	-1.268	.057
.250	-.614		-.936	-1.276	-1.042	-.789	-1.021	-.174
.400	-.647		-.780	-1.877	-.825	-1.125	-.771	-.368
.550	-.524	-3.611	-.585	-2.106	-.628	-1.123	-.567	-.394
.700	-.427	-3.250	-.475	-1.689	-.466	-.967	-.397	-.358
.850	-.387	-2.113	-.417	-1.183	-.350	-.711	-.265	-.291
.950	-.493	-1.317	-.426	-.664	-.307	-.454	-.186	-.220

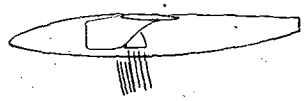
ALPHA = 14.01 DEG Q = 906 N/SQ.M. (18.91 LBF/SQ.FT.) EFFECT. VELOC. RATIO = .187

	ETA = .250		ETA = .387		ETA = .524		ETA = .800	
X/C	CP		CP		CP		CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			-.362		-2.243		-5.103	
.010			-.447	.993				
.015					-4.692	.953		
.025			-3.645	.739	-3.998	.703	-3.356	.946
.050			-2.083	.402	-2.349	.426	-2.413	.694
.100			-1.475	.170	-1.649	.045	-1.729	.414
.150			-1.289	-.604	-1.406	-.287	-1.418	.160
.250	-.654		-1.021	-1.212	-1.155	-.693	-1.138	-.094
.400	-.659		-.814	-1.900	-.891	-1.039	-.835	-.289
.550	-.523	-3.328	-.612	-2.050	-.646	-1.069	-.601	-.354
.700	-.413	-3.003	-.483	-1.705	-.489	-.940	-.423	-.339
.850	-.372	-2.110	-.428	-1.152	-.362	-.714	-.289	-.285
.950	-.429	-1.142	-.423	-.659	-.309	-.466	-.208	-.215

TABLE 14 (B) - CONCLUDED

PRESSURE PROFILE ON WING

FORWARD NOZZLES DEFLECTED 90 DEGREES WITH LIFT JET



ALPHA = 16.01 DEG Q = 908 N/SC.M. (18.96 LBF/SQ.FT.) EFFECT. VELOC. RATIO = .187

X/C	ETA = .250 CP UPPER	ETA = .250 CP LOWER	ETA = .387 CP UPPER	ETA = .387 CP LOWER	ETA = .524 CP UPPER	ETA = .524 CP LOWER	ETA = .800 CP UPPER	ETA = .800 CP LOWER
.000			-.987		-2.518		-1.572	
.010			-.501	.946				
.015					-5.036	.983		
.025			-3.523	.859	-3.894	.833	-1.518	.870
.050			-2.362	.481	-2.336	.514	-1.792	.655
.100			-1.603	-.041	-1.591	.099	-1.624	.368
.150			-1.365	-.479	-1.450	-.227	-1.530	.146
.250	-.652		-1.043	-1.135	-1.176	-.652	-1.490	-.095
.400	-.654		-.822	-1.831	-.943	-.997	-1.096	-.302
.550	-.507	-3.324	-.602	-2.026	-.700	-1.052	-.787	-.376
.700	-.402	-3.037	-.470	-1.658	-.599	-.953	-.632	-.366
.850	-.356	-1.942	-.410	-1.108	-.414	-.755	-.477	-.358
.950	-.412	-1.104	-.409	-.645	-.342	-.516	-.394	-.313

ALPHA = 18.01 DEG Q = 907 N/SQ.M. (18.95 LBF/SQ.FT.) EFFECT. VELOC. RATIO = .187

X/C	ETA = .250 CP UPPER	ETA = .250 CP LOWER	ETA = .387 CP UPPER	ETA = .387 CP LOWER	ETA = .524 CP UPPER	ETA = .524 CP LOWER	ETA = .800 CP UPPER	ETA = .800 CP LOWER
.000			-.068		-1.184		-1.224	
.010			-.498	.976				
.015					-2.920	.961		
.025			-1.839	.820	-1.620	.804	-1.193	.879
.050			-1.870	.495	-1.646	.478	-1.085	.664
.100			-1.645	-.017	-1.481	.110	-1.150	.384
.150			-1.620	-.460	-1.499	-.219	-1.045	.184
.250	-.733		-1.235	-1.156	-1.256	-.623	-1.297	-.074
.400	-.761		-.921	-1.876	-.989	-.975	-1.208	-.309
.550	-.555	-3.302	-.794	-2.011	-.947	-1.053	-1.087	-.396
.700	-.468	-2.989	-.600	-1.701	-.709	-.975	-.903	-.394
.850	-.368	-1.912	-.544	-1.137	-.593	-.794	-.770	-.376
.950	-.430	-.995	-.453	-.689	-.580	-.586	-.605	-.373

ALPHA = 20.04 DEG Q = 906 N/SC.M. (18.93 LBF/SQ.FT.) EFFECT. VELOC. RATIO = .187

X/C	ETA = .250 CP UPPER	ETA = .250 CP LOWER	ETA = .387 CP UPPER	ETA = .387 CP LOWER	ETA = .524 CP UPPER	ETA = .524 CP LOWER	ETA = .800 CP UPPER	ETA = .800 CP LOWER
.000			.298		-.345		-1.127	
.010			-.461	1.011				
.015					-1.151	.933		
.025			-1.061	.803	-.979	.736	-1.061	.891
.050			-1.198	.531	-.998	.465	-.972	.697
.100			-1.183	.006	-1.066	.121	-.986	.436
.150			-1.106	-.446	-1.181	-.230	-1.051	.222
.250	-.906		-1.252	-1.123	-1.073	-.610	-1.078	-.049
.400	-.934		-1.097	-1.805	-1.065	-1.010	-1.155	-.283
.550	-.742	-3.374	-.984	-1.968	-.961	-1.070	-1.128	-.385
.700	-.821	-2.955	-.877	-1.703	-.882	-1.025	-1.053	-.432
.850	-.441	-1.910	-.690	-1.177	-.749	-.865	-.941	-.513
.950	-.477	-1.072	-.568	-.702	-.595	-.699	-.817	-.572

TABLE 14 (C)

PRESSURE PROFILE ON WING

FORWARD NOZZLES DEFLECTED 90 DEGREES WITH LIFT JET

ALPHA = -.01 DEG Q = 231 N/SQ.M. (4.83 LBF/SQ.FT.) EFFECT. VELOC. RATIO = .095

X/C	ETA = .250 CP		ETA = .387 CP		ETA = .524 CP		ETA = .800 CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			-4.729		-4.715		-.794	
.010			-4.095	-5.185				
.015					.978	-4.435		
.025			.617	-4.729	.767	-4.763	.678	-2.479
.050			.378	-2.969	.487	-2.663	.378	-1.926
.100			.126	-2.581	.181	-2.342	.181	-1.354
.150			-.044	-2.554	.003	-2.281	.024	-1.354
.250	-.215		-.174	-2.553	-.140	-2.169	-.160	-1.273
.400	-.255		-.295	-2.563	-.255	-2.020	-.262	-1.112
.550	-.236	-5.313	-.265	-3.181	-.262	-1.773	-.236	-.897
.700	-.246	-7.517	-.285	-2.989	-.275	-1.363	-.189	-.632
.850	-.302	-4.133	-.361	-1.842	-.289	-.867	-.170	-.421
.950	-.560	-2.315	-.536	-1.422	-.368	-.642	-.186	-.252

ALPHA = 3.93 DEG Q = 231 N/SQ.M. (4.83 LBF/SQ.FT.) EFFECT. VELOC. RATIO = .095

X/C	ETA = .250 CP		ETA = .387 CP		ETA = .524 CP		ETA = .800 CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			-2.193		-1.293		.616	
.010			-3.972	-2.984				
.015					.582	-2.725		
.025			.262	-2.766	.412	-2.520	.235	-1.259
.050			.044	-2.213	.207	-2.050	-.024	-1.170
.100			-.059	-2.070	-.059	-1.715	-.134	-.979
.150			-.188	-2.125	-.127	-1.859	-.134	-1.013
.250	-.256		-.290	-2.362	-.243	-1.932	-.270	-.984
.400	-.290		-.333	-2.868	-.333	-1.866	-.303	-.878
.550	-.240	-7.664	-.286	-3.453	-.280	-1.681	-.250	-.769
.700	-.220	-8.543	-.277	-3.208	-.258	-1.304	-.174	-.610
.850	-.257	-4.051	-.333	-1.711	-.243	-.835	-.115	-.346
.950	-.429	-1.628	-.445	-1.298	-.273	-.501	-.111	-.187

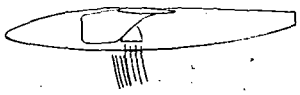
ALPHA = 8.04 DEG Q = 226 N/SQ.M. (4.72 LBF/SQ.FT.) EFFECT. VELOC. RATIO = .094

X/C	ETA = .250 CP		ETA = .387 CP		ETA = .524 CP		ETA = .800 CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			-.322		.320			
.010			-3.843	-1.598				
.015					.125	-1.883		
.025			-.175	-1.681	-.356	-1.674	-.342	-.524
.050			-.266	-1.430	-.196	-1.598	-.384	-.586
.100			-.287	-1.688	-.287	-1.444	-.405	-.593
.150			-.356	-1.835	-.370	-1.570	-.412	-.782
.250	-.273		-.335	-2.185	-.405	-1.763	-.489	-.840
.400	-.299		-.374	-2.777	-.404	-1.881	-.407	-.793
.550	-.232	-7.898	-.286	-3.449	-.323	-1.675	-.316	-.701
.700	-.171	-7.695	-.255	-2.686	-.282	-1.357	-.238	-.546
.850	-.178	-2.787	-.299	-1.797	-.232	-.860	-.113	-.343
.950	-.414	-1.442	-.360	-.979	-.255	-.475	-.086	-.144

TABLE 14 (C) - CONTINUED

PRESSURE PROFILE ON WING

FORWARD NOZZLES DEFLECTED 90 DEGREES WITH LIFT JET



ALPHA = 10.04 DEG Q = 227 N/SQ.M. (4.74 LBF/SQ.FT.) EFFECT. VELOC. RATIO = .094

X/C	ETA = .250 CP		ETA = .387 CP		ETA = .524 CP		ETA = .800 CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			.399		.642		.920	
.010			-3.514	-.762				
.015					-.011	-1.227		
.025			-.546	-1.248	-.512	-1.054	-.637	-.137
.050			-.359	-1.366	-.414	-1.082	-.553	-.338
.100			-.366	-1.353	-.387	-1.248	-.505	-.456
.150			-.512	-1.714	-.414	-1.415	-.470	-.581
.250	-.275		-.428	-2.116	-.408	-1.615	-.540	-.708
.400	-.311		-.392	-2.746	-.426	-1.746	-.405	-.712
.550	-.240	-8.146	-.304	-3.551	-.308	-1.604	-.291	-.634
.700	-.170	-8.183	-.244	-2.945	-.244	-1.268	-.203	-.479
.850	-.190	-2.753	-.267	-1.638	-.203	-.779	-.079	-.345
.950	-.324	-1.419	-.365	-.813	-.196	-.409	-.052	-.089

ALPHA = 12.00 DEG Q = 225 N/SQ.M. (4.70 LBF/SQ.FT.) EFFECT. VELOC. RATIO = .094

X/C	ETA = .250 CP		ETA = .387 CP		ETA = .524 CP		ETA = .800 CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			.893				.718	
.010			-3.354	-.101				
.015					-.710	-.367		
.025			-.878	-.773	-.745	-.871	-1.059	.032
.050			-.731	-.912	-.703	-.843	-.850	-.129
.100			-.542	-1.241	-.556	-.940	-.661	-.213
.150			-.556	-1.528	-.556	-1.248	-.640	-.465
.250	-.311		-.479	-1.920	-.549	-1.442	-.556	-.590
.400	-.343		-.455	-2.785	-.475	-1.638	-.465	-.651
.550	-.254	-8.775	-.346	-3.358	-.370	-1.571	-.312	-.556
.700	-.170	-6.614	-.261	-2.819	-.271	-1.275	-.170	-.455
.850	-.173	-2.809	-.258	-1.611	-.190	-.770	-.088	-.329
.950	-.360	-1.147	-.339	-.936	-.193	-.444	-.034	-.102

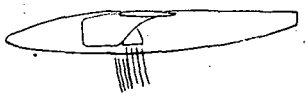
ALPHA = 14.00 DEG Q = 227 N/SQ.M. (4.74 LBF/SQ.FT.) EFFECT. VELOC. RATIO = .094

X/C	ETA = .250 CP		ETA = .387 CP		ETA = .524 CP		ETA = .800 CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			1.029				-.528	
.010			-3.168	.035				
.015					-1.076	-.277		
.025			-1.090	-.507	-1.153	-.673	-1.813	.348
.050			-.819	-.479	-.917	-.479	-1.243	.140
.100			-.757	-.972	-.799	-.854	-.875	-.152
.150			-.673	-1.382	-.722	-1.076	-.750	-.312
.250	-.382		-.597	-1.901	-.632	-1.419	-.673	-.513
.400	-.409		-.510	-2.804	-.577	-1.608	-.543	-.644
.550	-.304	-8.739	-.375	-3.548	-.426	-1.497	-.368	-.577
.700	-.223	-9.032	-.318	-2.665	-.294	-1.325	-.247	-.463
.850	-.213	-2.817	-.308	-1.588	-.227	-.783	-.170	-.341
.950	-.382	-1.099	-.382	-.884	-.247	-.490	-.112	-.237

TABLE 14 (c) -CONCLUDED

PRESSURE PROFILE ON WING

FORWARD NOZZLES DEFLECTED 90 DEGREES WITH LIFT JET



ALPHA = 16.08 DEG Q = 226 N/SQ.M. (4.72 LBF/SQ.FT.) EFFECT. VELOC. RATIO = .094

	ETA = .250		ETA = .387		ETA = .524		ETA = .800	
X/C	CP		CP		CP		CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			.977		.657		-1.066	
.010			-3.046	.629				
.015					-1.902	.287		
.025			-1.554	-.013	-1.826	-.229	-2.000	.496
.050			-1.303	-.417	-1.324	-.403	-1.826	.343
.100			-.898	-.787	-1.003	-.619	-.982	-.020
.150			-.815	-1.087	-.870	-.954	-.912	-.243
.250	-.424		-.682	-1.708	-.759	-1.303	-.815	-.424
.400	-.451		-.573	-2.787	-.627	-1.529	-.607	-.590
.550	-.329	-9.625	-.417	-3.426	-.458	-1.550	-.431	-.552
.700	-.242	-8.333	-.316	-3.054	-.350	-1.215	-.282	-.438
.850	-.218	-1.938	-.306	-1.435	-.262	-.799	-.184	-.346
.950	-.363	-1.228	-.357	-.769	-.258	-.478	-.133	-.235

ALPHA = 18.01 DEG Q = 226 N/SQ.M. (4.72 LBF/SQ.FT.) EFFECT. VELOC. RATIO = .094

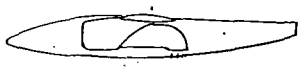
	ETA = .250		ETA = .387		ETA = .524		ETA = .800	
X/C	CP		CP		CP		CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			.873		.106		-2.216	
.010			-2.899	.734				
.015					-2.669	.462		
.025			-2.042	.239	-2.167	.022	-2.899	.650
.050			-1.630	-.124	-1.784	-.180	-1.811	.406
.100			-1.044	-.612	-1.191	-.508	-1.261	.036
.150			-.940	-.856	-1.044	-.835	-1.107	-.236
.250	-.487		-.773	-1.516	-.870	-1.199	-.947	-.438
.400	-.489		-.627	-2.689	-.702	-1.516	-.709	-.587
.550	-.364	-8.743	-.472	-3.450	-.523	-1.523	-.506	-.567
.700	-.273	-7.949	-.357	-2.838	-.391	-1.331	-.350	-.462
.850	-.229	-2.277	-.344	-1.543	-.289	-.840	-.242	-.360
.950	-.374	-.837	-.418	-.935	-.296	-.492	-.188	-.296

ALPHA = 20.00 DEG Q = 228 N/SQ.M. (4.75 LBF/SQ.FT.) EFFECT. VELOC. RATIO = .094

	ETA = .250		ETA = .387		ETA = .524		ETA = .800	
X/C	CP		CP		CP		CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			.363		-.053		-3.404	
.010			-2.705	.986				
.015					-3.051	.778		
.025			-2.552	.488	-2.656	.418	-3.570	.841
.050			-1.694	.211	-1.929	.134	-1.922	.557
.100			-1.188	-.399	-1.320	-.302	-1.493	.238
.150			-1.064	-.780	-1.126	-.627	-1.223	-.046
.250	-.468		-.814	-1.450	-.911	-1.068	-.980	-.269
.400	-.497		-.651	-2.638	-.722	-1.390	-.722	-.436
.550	-.356	-9.335	-.463	-3.152	-.524	-1.376	-.490	-.457
.700	-.248	-8.999	-.359	-2.920	-.376	-1.188	-.336	-.416
.850	-.218	-2.514	-.336	-1.595	-.275	-.792	-.225	-.309
.950	-.322	-1.088	-.386	-.910	-.245	-.457	-.181	-.235

TABLE 15 (A)

PRESSURE PROFILE ON WING



REAR NOZZLES DEFLECTED 90 DEGREES WITH LIFT JET

ALPHA = .00 DEG Q = 2844 N/SQ.M. (59.40 LBF/SQ.FT.) EFFECT. VELOC. RATIO = ∞

	ETA = .250		ETA = .387		ETA = .524		ETA = .800	
X/C	CP		CP		CP		CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			.742		.916		.967	
.010			.710	-.206				
.015					-.235	-.237		
.025			-.335	-.330	-.397	-.407	-.294	-.179
.050			-.328	-.176	-.450	-.263	-.235	-.158
.100			-.299	-.393	-.326	-.294	-.291	-.250
.150			-.345	-.412	-.355	-.339	-.274	-.263
.250	-.273		-.374	-.393	-.364	-.289	-.345	-.350
.400	-.342		-.381	-.326	-.358	-.314	-.326	-.310
.550	-.277	-.200	-.283	-.221	-.270	-.240	-.231	-.252
.700	-.208	-.134	-.183	-.184	-.169	-.156	-.137	-.151
.850	-.145	-.283	-.117	-.212	-.060	-.087	-.030	-.042
.950	-.103	-.090	-.060	-.099	.002	-.003	.030	.049

ALPHA = 3.91 DEG Q = 2845 N/SQ.M. (59.43 LBF/SQ.FT.) EFFECT. VELOC. RATIO = ∞

	ETA = .250		ETA = .387		ETA = .524		ETA = .800	
X/C	CP		CP		CP		CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			.988		.723		.136	
.010			.722	.601				
.015					-1.387	.595		
.025			-1.161	.318	-1.358	.338	-1.159	.373
.050			-.801	.199	-.911	.229	-.803	.198
.100			-.644	-.008	-.683	.097	-.620	.100
.150			-.592	-.120	-.617	.017	-.584	-.060
.250	-.371		-.557	-.157	-.543	-.129	-.531	-.157
.400	-.415		-.484	-.210	-.472	-.174	-.430	-.194
.550	-.318	-.089	-.334	-.139	-.338	-.159	-.299	-.163
.700	-.224	-.085	-.233	-.150	-.212	-.126	-.173	-.117
.850	-.146	-.344	-.132	-.228	-.087	-.083	-.060	-.056
.950	-.090	-.155	-.061	-.134	-.005	-.014	.029	.043

ALPHA = 8.05 DEG Q = 2842 N/SQ.M. (59.35 LBF/SQ.FT.) EFFECT. VELOC. RATIO = ∞

	ETA = .250		ETA = .387		ETA = .524		ETA = .800	
X/C	CP		CP		CP		CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			.204		-.868		-1.975	
.010			.683	1.003				
.015					-2.797	.951		
.025			-2.568	.835	-2.593	.843	-2.463	.816
.050			-1.521	.618	-1.619	.612	-1.562	.595
.100			-1.019	.379	-1.155	.426	-1.066	.372
.150			-.960	.225	-.981	.256	-.904	.239
.250	-.493		-.730	.054	-.788	.105	-.755	.075
.400	-.485		-.600	-.027	-.619	-.016	-.568	-.054
.550	-.360	-.026	-.409	-.040	-.423	-.050	-.390	-.078
.700	-.246	-.068	-.263	-.074	-.259	-.045	-.229	-.051
.850	-.154	-.377	-.152	-.208	-.117	-.038	-.102	-.015
.950	-.093	-.258	-.074	-.135	-.034	.003	-.013	.021

TABLE 15 (A) - CONTINUED

PRESSURE PROFILE ON WING

REAR NOZZLES DEFLECTED 90 DEGREES WITH LIFT JET

ALPHA = 9.99 DEG Q = 2843 N/SQ.M. (59.38 LBF/SQ.FT.) EFFECT. VELOC. RATIO =∞

X/C	ETA = .250 CP		ETA = .387 CP		ETA = .524 CP		ETA = .800 CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			-.610		-2.357		-3.775	
.010			.564	.938				
.015					-4.341	.978		
.025			-2.605	.961	-2.781	.918	-2.588	.924
.050			-1.792	.736	-2.002	.732	-1.799	.732
.100			-1.244	.559	-1.473	.550	-1.286	.474
.150			-1.045	.366	-1.160	.372	-1.042	.340
.250	-.526		-.810	.192	-.892	.194	-.874	.140
.400	-.520		-.643	.017	-.677	.073	-.610	.012
.550	-.390	.004	-.430	.018	-.448	-.003	-.415	-.040
.700	-.268	-.067	-.278	-.045	-.274	-.018	-.245	-.023
.850	-.170	-.383	-.153	-.186	-.122	-.031	-.108	-.011
.950	-.110	-.274	-.081	-.132	-.033	-.004	-.016	.016

ALPHA = 12.00 DEG Q = 2839 N/SQ.M. (59.29 LBF/SQ.FT.) EFFECT. VELOC. RATIO =∞

X/C	ETA = .250 CP		ETA = .387 CP		ETA = .524 CP		ETA = .800 CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			.385		-.146		-.854	
.010			.550	.991				
.015					-1.033	.948		
.025			-1.121	.895	-.872	.834	-1.106	.819
.050			-.971	.739	-.872	.692	-.997	.652
.100			-.982	.543	-.957	.500	-1.064	.438
.150			-.814	.352	-.915	.346	-1.063	.283
.250	-.630		-.822	.148	-.957	.163	-1.065	.102
.400	-.659		-.885	.005	-.895	.018	-.951	-.028
.550	-.515	-.017	-.750	-.032	-.749	-.052	-.733	-.089
.700	-.572	-.078	-.592	-.098	-.584	-.092	-.660	-.095
.850	-.314	-.417	-.360	-.321	-.375	-.155	-.461	-.127
.950	-.176	-.341	-.295	-.250	-.367	-.222	-.246	-.196

ALPHA = 14.02 DEG Q = 2842 N/SQ.M. (59.35 LBF/SQ.FT.) EFFECT. VELOC. RATIO =∞

X/C	ETA = .250 CP		ETA = .387 CP		ETA = .524 CP		ETA = .800 CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			.417		.033		-.855	
.010			.545	.984				
.015					-.583	.962		
.025			-.576	.920	-.584	.858	-.926	.840
.050			-.532	.768	-.564	.704	-.901	.664
.100			-.516	.557	-.553	.514	-.857	.460
.150			-.566	.389	-.574	.365	-.892	.306
.250	-.524		-.570	.184	-.675	.178	-.902	.131
.400	-.665		-.650	-.005	-.682	.015	-.910	-.032
.550	-.698	-.032	-.677	-.040	-.654	-.073	-.903	-.110
.700	-.701	-.101	-.741	-.125	-.689	-.130	-.816	-.147
.850	-.659	-.513	-.661	-.369	-.637	-.218	-.688	-.207
.950	-.516	-.449	-.575	-.430	-.564	-.350	-.506	-.266

TABLE 15 (A) - CONCLUDED
PRESSURE PROFILE ON WING

REAR NOZZLES DEFLECTED 90 DEGREES WITH LIFT JET

ALPHA = 15.99 DEG Q = 2843 N/SQ.M. (59.38 LBF/SQ.FT.) EFFECT. VELOC. RATIO =∞

X/C	ETA = .250 CP		ETA = .387 CP		ETA = .524 CP		ETA = .800 CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			.233		-.097		-.888	
.010			.517	.945				
.015					-.528	.972		
.025			-.515	.951	-.532	.893	-.869	.881
.050			-.529	.835	-.521	.752	-.894	.714
.100			-.528	.644	-.529	.576	-.849	.514
.150			-.539	.486	-.549	.431	-.791	.347
.250	-.569		-.563	.234	-.565	.223	-.855	.158
.400	-.610		-.619	.027	-.626	.049	-.883	-.021
.550	-.669	-.012	-.656	-.023	-.659	-.058	-.895	-.117
.700	-.695	-.124	-.682	-.143	-.643	-.137	-.850	-.171
.850	-.729	-.577	-.661	-.403	-.644	-.269	-.767	-.255
.950	-.649	-.484	-.603	-.461	-.587	-.397	-.664	-.328

ALPHA = 18.02 DEG Q = 2843 N/SQ.M. (59.37 LBF/SQ.FT.) EFFECT. VELOC. RATIO =∞

X/C	ETA = .250 CP		ETA = .387 CP		ETA = .524 CP		ETA = .800 CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			.083		-.263		-.992	
.010			.484	.867				
.015					-.529	.978		
.025			-.547	.987	-.516	.934	-.771	.897
.050			-.555	.874	-.539	.800	-.776	.764
.100			-.539	.711	-.517	.632	-.794	.553
.150			-.565	.533	-.539	.477	-.781	.386
.250	-.563		-.563	.280	-.562	.286	-.816	.188
.400	-.651		-.625	.078	-.635	.089	-.818	.010
.550	-.675	-.001	-.678	.006	-.673	-.022	-.827	-.102
.700	-.729	-.148	-.674	-.127	-.686	-.125	-.850	-.178
.850	-.721	-.626	-.686	-.416	-.668	-.262	-.844	-.305
.950	-.685	-.538	-.663	-.482	-.619	-.405	-.790	-.447

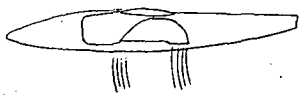
ALPHA = 20.00 DEG Q = 2845 N/SQ.M. (59.43 LBF/SQ.FT.) EFFECT. VELOC. RATIO =∞

X/C	ETA = .250 CP		ETA = .387 CP		ETA = .524 CP		ETA = .800 CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			-.177		-.395		-1.040	
.010			.425	.731				
.015					-.541	.975		
.025			-.590	.986	-.536	.954	-.731	.916
.050			-.598	.929	-.535	.869	-.744	.802
.100			-.577	.760	-.545	.693	-.732	.599
.150			-.586	.600	-.557	.560	-.780	.440
.250	-.627		-.620	.343	-.576	.357	-.773	.255
.400	-.675		-.678	.121	-.634	.125	-.826	.037
.550	-.723	.015	-.703	.041	-.685	.008	-.843	-.090
.700	-.753	-.149	-.719	-.111	-.703	-.098	-.874	-.182
.850	-.764	-.644	-.732	-.415	-.672	-.257	-.878	-.325
.950	-.722	-.565	-.712	-.499	-.605	-.402	-.817	-.476

TABLE 15 (B)

PRESSURE PROFILE ON WING

REAR NOZZLES DEFLECTED 90 DEGREES WITH LIFT JET



ALPHA = -.02 DEG Q = 2033 N/SQ.M. (42.46 LBF/SQ.FT.) EFFECT. VELOC. RATIO = .278

X/C	ETA = .250 CP		ETA = .387 CP		ETA = .524 CP		ETA = .800 CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			.813		.984		.892	
.010			.564	.191				
.015					-.429	.150		
.025			-.490	-.376	-.590	-.205	-.487	-.072
.050			-.469	-.297	-.590	-.160	-.523	-.160
.100			-.453	-.394	-.437	-.287	-.410	-.202
.150			-.487	-.415	-.482	-.312	-.404	-.298
.250	-.383		-.470	-.517	-.439	-.371	-.452	-.318
.400	-.463		-.498	-.506	-.481	-.467	-.415	-.353
.550	-.416	-.395	-.407	-.470	-.400	-.425	-.311	-.306
.700	-.360	-.485	-.326	-.519	-.278	-.386	-.185	-.215
.850	-.365	-1.030	-.295	-.681	-.198	-.326	-.102	-.128
.950	-.399	-.775	-.290	-.487	-.155	-.215	-.032	-.028

ALPHA = 3.96 DEG Q = 2035 N/SQ.M. (42.51 LBF/SQ.FT.) EFFECT. VELOC. RATIO = .278

X/C	ETA = .250 CP		ETA = .387 CP		ETA = .524 CP		ETA = .800 CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			.994		.342		-.147	
.010			.636	.828				
.015					-1.800	.682		
.025			-1.554	.407	-1.495	.485	-1.364	.490
.050			-1.133	.329	-1.253	.288	-1.186	.327
.100			-.819	.039	-.816	.143	-.837	.122
.150			-.737	-.124	-.861	-.095	-.717	-.018
.250	-.481		-.701	-.275	-.695	-.144	-.666	-.147
.400	-.560		-.607	-.402	-.623	-.344	-.536	-.222
.550	-.472	-.417	-.471	-.375	-.482	-.321	-.399	-.226
.700	-.409	-.515	-.388	-.476	-.362	-.336	-.247	-.174
.850	-.398	-1.128	-.339	-.697	-.241	-.320	-.141	-.125
.950	-.439	-.878	-.314	-.501	-.185	-.236	-.054	-.045

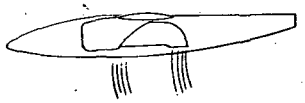
ALPHA = 8.02 DEG Q = 2039 N/SQ.M. (42.59 LBF/SQ.FT.) EFFECT. VELOC. RATIO = .278

X/C	ETA = .250 CP		ETA = .387 CP		ETA = .524 CP		ETA = .800 CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			-.390		-1.653		-3.059	
.010			.582	1.003				
.015					-3.795	1.003		
.025			-2.973	.871	-3.198	.899	-2.498	.883
.050			-1.665	.643	-2.009	.659	-1.697	.673
.100			-1.269	.426	-1.343	.440	-1.262	.419
.150			-1.107	.185	-1.172	.257	-1.058	.210
.250	-.594		-.879	-.011	-.936	.059	-.946	.030
.400	-.646		-.743	-.212	-.781	-.164	-.675	-.101
.550	-.538	-.293	-.572	-.294	-.568	-.235	-.475	-.147
.700	-.462	-.556	-.452	-.410	-.417	-.272	-.314	-.117
.850	-.432	-1.140	-.375	-.681	-.283	-.280	-.181	-.099
.950	-.456	-.951	-.335	-.515	-.212	-.233	-.083	-.057

TABLE 15 (B) - *CONTINUED*

PRESSURE PROFILE ON WING

REAR NOZZLES DEFLECTED 90 DEGREES WITH LIFT JET



ALPHA = 9.98 DEG Q = 2034 N/SQ.M. (42.48 LBF/SQ.FT.) EFFECT. VELOC. RATIO = .278

X/C	ETA = .250 CP		ETA = .387 CP		ETA = .524 CP		ETA = .800 CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			-.983		-3.230		-4.879	
.010			.503	.884				
.015					-4.943	.983		
.025			-3.300		-3.595	.981	-3.246	.961
.050			-2.093	.825	-2.345	.808	-2.239	.801
.100			-1.449	.551	-1.638	.547	-1.523	.533
.150			-1.259	.382	-1.360	.386	-1.186	.348
.250	-.658		-.998	.073	-1.069	.134	-.995	.163
.400	-.678		-.813	-.172	-.841	-.079	-.751	-.032
.550	-.571	-.304	-.607	-.231	-.620	-.192	-.520	-.113
.700	-.481	-.502	-.471	-.393	-.441	-.247	-.344	-.104
.850	-.442	-1.168	-.393	-.674	-.301	-.276	-.197	-.104
.950	-.479	-.986	-.352	-.532	-.222	-.237	-.107	-.066

ALPHA = 11.96 DEG Q = 2038 N/SQ.M. (42.56 LBF/SQ.FT.) EFFECT. VELOC. RATIO = .278

X/C	ETA = .250 CP		ETA = .387 CP		ETA = .524 CP		ETA = .800 CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			-.105		-.618		-1.566	
.010			.490	.974				
.015					-1.295	1.008		
.025			-1.173	.963	-1.299	.920	-1.517	.904
.050			-1.238	.811	-1.098	.743	-1.569	.732
.100			-1.364	.549	-1.274	.534	-1.511	.498
.150			-1.470	.375	-1.183	.354	-1.442	.310
.250	-.838		-1.213	.081	-1.261	.131	-1.406	.118
.400	-.875		-1.089	-.155	-1.133	-.087	-1.152	-.057
.550	-.634	-.277	-.791	-.244	-1.010	-.209	-.926	-.136
.700	-.644	-.504	-.631	-.418	-.732	-.303	-.613	-.154
.850	-.474	-1.185	-.629	-.736	-.601	-.375	-.489	-.186
.950	-.500	-.986	-.503	-.605	-.497	-.410	-.341	-.181

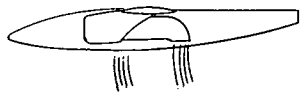
ALPHA = 14.00 DEG Q = 2039 N/SQ.M. (42.59 LBF/SQ.FT.) EFFECT. VELOC. RATIO = .278

X/C	ETA = .250 CP		ETA = .387 CP		ETA = .524 CP		ETA = .800 CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			.018		-.464		-1.677	
.010			.488	.949				
.015					-.997	1.010		
.025			-.968	.980	-.980	.943	-1.367	.925
.050			-.858	.839	-.943	.782	-1.366	.771
.100			-.820	.602	-.963	.564	-1.363	.531
.150			-.967	.421	-1.044	.402	-1.350	.357
.250	-.810		-.950	.132	-1.037	.152	-1.417	.149
.400	-.932		-1.047	-.163	-1.117	-.071	-1.370	-.037
.550	-.871	-.301	-1.043	-.259	-1.086	-.214	-1.178	-.136
.700	-.859	-.554	-.955	-.440	-1.039	-.334	-.919	-.177
.850	-.777	-1.280	-.823	-.802	-.912	-.466	-.641	-.240
.950	-.663	-1.075	-.732	-.740	-.708	-.554	-.494	-.305

TABLE 15 (B)-- CONCLUDED

PRESSURE PROFILE ON WING

REAR NOZZLES DEFLECTED 90 DEGREES WITH LIFT JET



ALPHA = 15.98 DEG Q = 2036 N/SQ.M. (42.53 LBF/SQ.FT.) EFFECT. VELOC. RATIO = .278

	ETA = .250		ETA = .387		ETA = .524		ETA = .800	
X/C	CP		CP		CP		CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			.005		-.535		-1.597	
.010			.437	.925				
.015					-.855	1.009		
.025			-.793		-.885	.976	-1.247	.946
.050			-.756	.876	-.823	.813	-1.222	.797
.100			-.785	.670	-.860	.608	-1.264	.566
.150			-.798	.496	-.830	.437	-1.182	.395
.250	-.794		-.834	.165	-.953	.197	-1.247	.187
.400	-.844		-.883	-.105	-.992	-.040	-1.367	-.013
.550	-.901	-.282	-.933	-.233	-1.031	-.202	-1.256	-.149
.700	-.902	-.519	-.936	-.452	-1.022	-.350	-1.095	-.215
.850	-.879	-1.295	-.915	-.862	-.939	-.534	-.871	-.286
.950	-.807	-1.133	-.790	-.841	-.849	-.662	-.738	-.400

ALPHA = 17.96 DEG Q = 2041 N/SQ.M. (42.62 LBF/SQ.FT.) EFFECT. VELOC. RATIO = .278

	ETA = .250		ETA = .387		ETA = .524		ETA = .800	
X/C	CP		CP		CP		CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			-.132		-.644		-1.567	
.010			.400	.843				
.015					-.811	1.003		
.025			-.738		-.826	.993	-1.141	.950
.050			-.752	.933	-.819	.867	-1.182	.847
.100			-.741	.741	-.813	.670	-1.109	.612
.150			-.755	.553	-.836	.507	-1.161	.443
.250	-.769		-.791	.222	-.844	.246	-1.137	.233
.400	-.805		-.849	-.053	-.926	-.010	-1.176	.009
.550	-.898	-.306	-.877	-.222	-.965	-.197	-1.242	-.142
.700	-.936	-.576	-.936	-.462	-.961	-.343	-1.192	-.235
.850	-.973	-1.361	-.900	-.877	-.932	-.534	-1.036	-.336
.950	-.913	-1.184	-.819	-.861	-.876	-.693	-.932	-.518

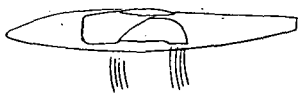
ALPHA = 20.00 DEG Q = 2038 N/SQ.M. (42.57 LBF/SQ.FT.) EFFECT. VELOC. RATIO = .278

	ETA = .250		ETA = .387		ETA = .524		ETA = .800	
X/C	CP		CP		CP		CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			-.295		-.757		-1.454	
.010			.397	.751				
.015					-.798	.993		
.025			-.746		-.770	1.008	-1.063	.970
.050			-.721	.969	-.782	.920	-1.048	.872
.100			-.756	.800	-.792	.740	-1.076	.664
.150			-.748	.603	-.795	.560	-1.061	.494
.250	-.760		-.801	.329	-.823	.320	-1.094	.270
.400	-.854		-.831	-.017	-.909	.034	-1.109	.038
.550	-.875	-.273	-.874	-.194	-.957	-.160	-1.124	-.128
.700	-.959	-.612	-.940	-.455	-.963	-.341	-1.154	-.243
.850	-.948	-1.380	-.912	-.888	-.937	-.545	-1.118	-.405
.950	-.914	-1.192	-.808	-.880	-.894	-.699	-1.029	-.584

TABLE 15 (C)

PRESSURE PROFILE ON WING

REAR NOZZLES DEFLECTED 90 DEGREES WITH LIFT JET



ALPHA = -.04 DEG Q = 908 N/SC.M. (18.97 LBF/SQ.FT.) EFFECT. VELOC. RATIO = .186

	ETA = .250		ETA = .387		ETA = .524		ETA = .800	
X/C	CP		CP		CP		CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			.618		.845		.999	
.010			-.152	-.300				
.015					-.381	-.449		
.025			-.404	-.548	-.294	-.485	-.296	-.133
.050			-.404	-.449	-.428	-.419	-.345	-.284
.100			-.372	-.568	-.430	-.511	-.317	-.296
.150			-.490	-.650	-.584	-.525	-.376	-.348
.250	-.366		-.508	-.694	-.451	-.543	-.463	-.455
.400	-.458		-.481	-.643	-.466	-.563	-.405	-.463
.550	-.432	-.587	-.416	-.606	-.399	-.506	-.294	-.385
.700	-.395	-.660	-.374	-.640	-.326	-.499	-.218	-.285
.850	-.399	-1.196	-.333	-.797	-.233	-.420	-.111	-.191
.950	-.489	-.893	-.343	-.587	-.204	-.279	-.063	-.073

ALPHA = 3.97 DEG Q = 912 N/SQ.M. (19.05 LBF/SQ.FT.) EFFECT. VELOC. RATIO = .187

	ETA = .250		ETA = .387		ETA = .524		ETA = .800	
X/C	CP		CP		CP		CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			1.006		.852		.347	
.010			-.087	.666				
.015					-1.382	.429		
.025			-1.230	.193	-1.273	.179	-1.241	.300
.050			-.919	-.037	-1.014	.060	-.919	.163
.100			-.722	-.156	-.809	-.002	-.741	-.030
.150			-.721	-.279	-.766	-.239	-.672	-.147
.250	-.462		-.667	-.500	-.667	-.358	-.641	-.288
.400	-.561		-.620	-.576	-.618	-.461	-.516	-.339
.550	-.486	-.522	-.484	-.545	-.502	-.472	-.414	-.325
.700	-.444	-.737	-.430	-.628	-.372	-.470	-.261	-.263
.850	-.449	-1.314	-.392	-.838	-.288	-.438	-.167	-.194
.950	-.526	-1.045	-.385	-.615	-.242	-.341	-.093	-.089

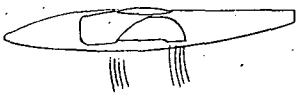
ALPHA = 7.98 DEG Q = 903 N/SC.M. (18.86 LBF/SQ.FT.) EFFECT. VELOC. RATIO = .186

	ETA = .250		ETA = .387		ETA = .524		ETA = .800	
X/C	CP		CP		CP		CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			.496		-.479		-1.733	
.010			-.038					
.015					-2.807	.899		
.025			-2.329	.744	-2.629	.758	-2.727	.786
.050			-1.508	.459	-1.571	.524	-1.672	.548
.100			-1.088	.229	-1.257	.266	-1.255	.269
.150			-.995	.009	-1.056	.060	-1.011	.109
.250	-.592		-.845	-.258	-.899	-.166	-.857	-.043
.400	-.645		-.762	-.444	-.779	-.321	-.678	-.178
.550	-.556	-.556	-.583	-.473	-.584	-.404	-.484	-.251
.700	-.497	-.729	-.488	-.594	-.440	-.428	-.334	-.213
.850	-.493	-1.382	-.431	-.848	-.333	-.416	-.214	-.171
.950	-.557	-1.126	-.413	-.637	-.280	-.346	-.130	-.111

TABLE 15 (C) - CONTINUED

PRESSURE PROFILE ON WING

REAR NOZZLES DEFLECTED 90 DEGREES WITH LIFT JET



ALPHA = 9.98 DEG Q = 909 N/SQ.M. (18.99 LBF/SQ.FT.) EFFECT. VELOC. RATIO = .186

	ETA = .250		ETA = .387		ETA = .524		ETA = .800	
X/C	CP		CP		CP		CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			-.348		-1.851		-3.656	
.010			-.071	1.003				
.015					-4.875	1.010		
.025			-2.974	.891	-3.377	.884	-2.816	.920
.050			-1.960	.633	-2.017	.634	-1.932	.679
.100			-1.326	.354	-1.459	.433	-1.364	.425
.150			-1.206	.173	-1.274	.165	-1.168	.229
.250	-.636		-.955	-.139	-1.034	-.053	-.962	.006
.400	-.689		-.799	-.357	-.831	-.276	-.731	-.170
.550	-.586	-.331	-.630	-.428	-.638	-.360	-.536	-.213
.700	-.517	-.775	-.511	-.567	-.480	-.398	-.358	-.203
.850	-.501	-1.375	-.439	-.843	-.360	-.395	-.239	-.176
.950	-.579	-1.151	-.425	-.654	-.299	-.331	-.147	-.119

ALPHA = 11.99 DEG Q = 908 N/SQ.M. (18.96 LBF/SQ.FT.) EFFECT. VELOC. RATIO = .186

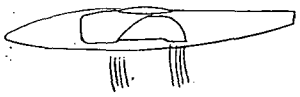
	ETA = .250		ETA = .387		ETA = .524		ETA = .800	
X/C	CP		CP		CP		CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			-1.524		-1.748		-5.298	
.010			-.161	.858				
.015					-5.302	1.007		
.025			-3.388	1.007	-3.707	.986	-3.548	.978
.050			-2.283	.809	-2.409	.823	-2.288	.787
.100			-1.505	.544	-1.693	.535	-1.623	.558
.150			-1.308	.291	-1.434	.315	-1.323	.334
.250	-.693		-1.046	-.041	-1.169	.047	-1.061	.095
.400	-.740		-.847	-.290	-.891	-.184	-.808	-.085
.550	-.612	-.518	-.663	-.372	-.654	-.309	-.560	-.164
.700	-.534	-.761	-.525	-.544	-.519	-.344	-.380	-.173
.850	-.509	-1.386	-.451	-.829	-.361	-.384	-.242	-.169
.950	-.567	-1.175	-.424	-.655	-.297	-.314	-.179	-.128

ALPHA = 13.97 DEG Q = 910 N/SQ.M. (19.01 LBF/SQ.FT.) EFFECT. VELOC. RATIO = .186

	ETA = .250		ETA = .387		ETA = .524		ETA = .800	
X/C	CP		CP		CP		CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			-.180		-.680		-1.758	
.010			-.114	.971				
.015					-1.566			
.025			-1.601	.980	-1.381	.935	-1.608	.936
.050			-1.468	.813	-1.419	.756	-1.649	.749
.100			-1.481	.540	-1.421	.516	-1.535	.495
.150			-1.609	.324	-1.495	.308	-1.511	.308
.250	-.891		-1.471	-.002	-1.346	.056	-1.485	.081
.400	-.853		-1.159	-.272	-1.216	-.195	-1.209	-.100
.550	-.704	-.545	-.818	-.363	-1.021	-.326	-1.039	-.189
.700	-.609	-.771	-.674	-.551	-.845	-.403	-.657	-.209
.850	-.523	-1.376	-.531	-.857	-.559	-.476	-.444	-.219
.950	-.610	-1.149	-.528	-.726	-.521	-.484	-.351	-.231

TABLE 15 (C) - CONCLUDED
PRESSURE PROFILE ON WING

REAR NOZZLES DEFLECTED 90 DEGREES WITH LIFT JET



ALPHA = 15.98 DEG Q = 907 N/SQ.M. (18.94 LBF/SQ.FT.) EFFECT. VELOC. RATIO = .186

X/C	ETA = .250 CP		ETA = .387 CP		ETA = .524 CP		ETA = .800 CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			-.011		-.536		-1.606	
.010			-.068	.962				
.015					-1.107	1.030		
.025			-.996	.992	-1.050	.966	-1.373	.959
.050			-1.076	.835	-1.053	.764	-1.436	.787
.100			-1.137	.595	-1.048	.552	-1.357	.543
.150			-1.189	.345	-1.088	.359	-1.385	.359
.250	-.899		-1.105	.027	-1.083	.089	-1.429	.126
.400	-.968		-1.141	-.240	-1.157	-.177	-1.423	-.071
.550	-.834	-.542	-1.102	-.371	-1.115	-.329	-1.321	-.184
.700	-.763	-.752	-.926	-.570	-1.022	-.435	-1.036	-.226
.850	-.677	-1.432	-.823	-.926	-.830	-.532	-.770	-.253
.950	-.667	-1.174	-.691	-.777	-.757	-.608	-.592	-.324

ALPHA = 17.98 DEG Q = 907 N/SQ.M. (18.95 LBF/SQ.FT.) EFFECT. VELOC. RATIO = .186

X/C	ETA = .250 CP		ETA = .387 CP		ETA = .524 CP		ETA = .800 CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			-.069		-.566		-1.611	
.010			-.028	.929				
.015					-1.000	1.032		
.025			-.913	1.020	-.926	.983	-1.349	.985
.050			-.872	.875	-.929	.823	-1.285	.823
.100			-.833	.624	-.913	.592	-1.301	.585
.150			-1.011	.417	-.943	.379	-1.370	.379
.250	-.825		-.972	.083	-.978	.118	-1.341	.168
.400	-.951		-1.008	-.215	-1.091	-.150	-1.357	-.053
.550	-1.001	-.507	-1.007	-.372	-1.069	-.342	-1.348	-.180
.700	-.905	-.778	-1.045	-.591	-1.069	-.473	-1.217	-.251
.850	-.862	-1.463	-.952	-.934	-.969	-.636	-1.045	-.350
.950	-.756	-1.228	-.789	-.894	-.889	-.739	-.773	-.397

ALPHA = 19.97 DEG Q = 908 N/SQ.M. (18.95 LBF/SQ.FT.) EFFECT. VELOC. RATIO = .185

X/C	ETA = .250 CP		ETA = .387 CP		ETA = .524 CP		ETA = .800 CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			-.205		-.625		-1.593	
.010			-.028	.868				
.015					-.925	1.033		
.025			-.821	1.033	-.894	1.016	-1.244	.991
.050			-.814	.927	-.876	.861	-1.198	.861
.100			-.835	.703	-.870	.642	-1.231	.636
.150			-.831	.505	-.885	.453	-1.220	.432
.250	-.833		-.892	.131	-.929	.181	-1.244	.196
.400	-.896		-.964	-.197	-1.010	-.113	-1.346	-.025
.550	-1.019	-.503	-1.019	-.380	-1.031	-.333	-1.301	-.183
.700	-.958	-.817	-1.016	-.610	-1.024	-.480	-1.286	-.287
.850	-.901	-1.543	-.940	-.995	-1.003	-.665	-1.126	-.393
.950	-.822	-1.275	-.853	-.921	-.918	-.774	-.992	-.539

TABLE 15 (D)

PRESSURE PROFILE ON WING

REAR NOZZLES DEFLECTED 90 DEGREES WITH LIFT JET

ALPHA = -.02 DEG Q = 231 N/SQ.M. (4.83 LBF/SQ.FT.) EFFECT. VELOC. RATIO = .094

X/C	ETA = .250 CP		ETA = .387 CP		ETA = .524 CP		ETA = .800 CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			-1.141		-.092		.645	
.010			-6.016	-1.469				
.015					.481	-1.564		
.025			.106	-1.666	.345	-1.619	.038	-.780
.050			-.057	-1.141	.079	-1.128	-.064	-.766
.100			-.112	-1.128	-.146	-1.019	-.153	-.705
.150			-.235	-1.128	-.214	-.896	-.146	-.732
.250	-.303		-.337	-.997	-.255	-.914	-.337	-.852
.400	-.362		-.356	-.766	-.366	-.762	-.323	-.670
.550	-.343	-.584	-.339	-.733	-.306	-.686	-.277	-.584
.700	-.343	-.703	-.319	-.726	-.283	-.647	-.217	-.455
.850	-.376	-1.053	-.326	-.762	-.237	-.495	-.154	-.273
.950	-.495	-.868	-.366	-.514	-.224	-.333	-.118	-.151

ALPHA = 3.99 DEG Q = 231 N/SQ.M. (4.83 LBF/SQ.FT.) EFFECT. VELOC. RATIO = .094

X/C	ETA = .250 CP		ETA = .387 CP		ETA = .524 CP		ETA = .800 CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			.662		.914		.921	
.010			-5.256	.007				
.015					-.122	-.484		
.025			-.552	-.668	-.593	-.368	-.463	-.402
.050			-.327	-.525	-.354	-.375	-.532	-.225
.100			-.368	-.532	-.375	-.532	-.457	-.293
.150			-.409	-.627	-.395	-.559	-.429	-.375
.250	-.354		-.443	-.770	-.457	-.604	-.484	-.452
.400	-.439		-.485	-.783	-.459	-.684	-.426	-.482
.550	-.396	-.545	-.393	-.627	-.366	-.604	-.333	-.442
.700	-.350	-.598	-.350	-.614	-.320	-.525	-.251	-.350
.850	-.406	-1.084	-.327	-.746	-.254	-.459	-.158	-.294
.950	-.459	-.780	-.343	-.565	-.198	-.317	-.099	-.122

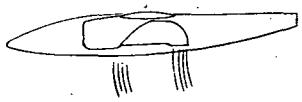
ALPHA = 7.99 DEG Q = 224 N/SQ.M. (4.68 LBF/SQ.FT.) EFFECT. VELOC. RATIO = .092

X/C	ETA = .250 CP		ETA = .387 CP		ETA = .524 CP		ETA = .800 CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			.937		.783		-.040	
.010			-4.682	.684				
.015					-1.468	.501		
.025			-1.510	.347	-1.531	.305	-1.285	.347
.050			-.947	.072	-.954	.030	-1.201	.164
.100			-.666	-.033	-.750	.023	-.856	-.005
.150			-.666	-.321	-.743	-.406	-.708	-.160
.250	-.406		-.603	-.513	-.666	-.367	-.673	-.223
.400	-.438		-.540	-.554	-.571	-.493	-.503	-.288
.550	-.370	-.459	-.418	-.486	-.411	-.455	-.377	-.309
.700	-.349	-.609	-.329	-.520	-.315	-.431	-.234	-.274
.850	-.332	-1.065	-.298	-.595	-.223	-.360	-.155	-.206
.950	-.380	-.765	-.322	-.547	-.199	-.322	-.083	-.141

TABLE 15 (D) - CONTINUED

PRESSURE PROFILE ON WING

REAR NOZZLES DEFLECTED 90 DEGREES WITH LIFT JET



ALPHA = 10.01 DEG Q = 226 N/SQ.M. (4.73 LBF/SQ.FT.) EFFECT. VELOC. RATIO = .093

	ETA = .250		ETA = .387		ETA = .524		ETA = .800	
X/C	CP		CP		CP		CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			.898		.425		-.793	
.010			-3.955	.905				
.015					-2.262	.703		
.025			-1.615	.425	-1.768	.613	-1.747	.620
.050			-1.232	.167	-1.441	.237	-1.468	.306
.100			-.807	.105	-.835	.139	-.877	.098
.150			-.738	-.230	-.856	-.111	-.821	-.042
.250	-.459		-.675	-.449	-.772	-.273	-.745	-.195
.400	-.523		-.590	-.550	-.651	-.486	-.587	-.337
.550	-.435	-.536	-.452	-.489	-.486	-.432	-.435	-.297
.700	-.395	-.573	-.388	-.536	-.371	-.422	-.297	-.246
.850	-.398	-1.117	-.351	-.658	-.314	-.361	-.206	-.175
.950	-.449	-.867	-.364	-.577	-.246	-.283	-.131	-.158

ALPHA = 11.98 DEG Q = 226 N/SQ.M. (4.73 LBF/SQ.FT.) EFFECT. VELOC. RATIO = .092

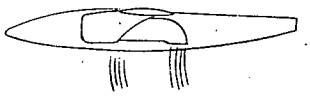
	ETA = .250		ETA = .387		ETA = .524		ETA = .800	
X/C	CP		CP		CP		CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			.053		-.267		-1.624	
.010			-3.559					
.015					-2.299	.958		
.025			-3.016	.805	-2.313	.777	-2.320	.742
.050			-1.527	.561	-1.596	.436	-1.568	.596
.100			-.991	.227	-1.053	.318	-1.130	.332
.150			-.886	-.086	-.998	.067	-.984	.102
.250	-.496		-.733	-.268	-.796	-.177	-.810	-.113
.400	-.558		-.636	-.396	-.676	-.410	-.606	-.214
.550	-.461	-.494	-.515	-.504	-.515	-.403	-.437	-.224
.700	-.410	-.602	-.417	-.491	-.383	-.413	-.292	-.214
.850	-.390	-1.156	-.366	-.599	-.295	-.322	-.191	-.167
.950	-.488	-.923	-.376	-.515	-.251	-.278	-.140	-.099

ALPHA = 13.99 DEG Q = 227 N/SQ.M. (4.74 LBF/SQ.FT.) EFFECT. VELOC. RATIO = .093

	ETA = .250		ETA = .387		ETA = .524		ETA = .800	
X/C	CP		CP		CP		CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			-.157		-1.709		-3.450	
.010			-3.318					
.015					-4.539			
.025			-3.360	.919	-3.499	.892	-3.311	.933
.050			-1.674	.566	-1.813	.635	-1.869	.663
.100			-1.127	.365	-1.494	.378	-1.286	.462
.150			-1.009	.191	-1.071	.240	-1.127	.247
.250	-.516		-.794	-.173	-.884	-.075	-.898	.043
.400	-.546		-.643	-.294	-.704	-.176	-.647	-.153
.550	-.445	-.546	-.492	-.401	-.516	-.388	-.448	-.186
.700	-.368	-.613	-.395	-.452	-.381	-.348	-.300	-.200
.850	-.364	-1.087	-.341	-.580	-.267	-.307	-.183	-.132
.950	-.408	-.916	-.354	-.516	-.223	-.223	-.109	-.092

TABLE 15 (D) - CONCLUDED
PRESSURE PROFILE ON WING

REAR NOZZLES DEFLECTED 90 DEGREES WITH LIFT JET



ALPHA = 15.97 DEG Q = 227 N/SQ.M. (4.74 LBF/SQ.FT.) EFFECT. VELOC. RATIO = .092

	ETA = .250		ETA = .387		ETA = .524		ETA = .800	
X/C	CP		CP		CP		CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			-.768		-1.956		-3.172	
.010			-3.207	.948				
.015					-3.082			
.025			-3.464		-2.908	.997	-3.304	
.050			-1.755	.781	-2.074	.788	-2.033	.809
.100			-1.595	.573	-1.456	.441	-1.477	.496
.150			-1.345	.184	-1.289	.295	-1.317	.309
.250	-.532		-.942	-.083	-1.004	.079	-.970	.085
.400	-.588		-.673	-.279	-.791	-.215	-.720	-.103
.550	-.474	-.541	-.504	-.322	-.555	-.279	-.487	-.181
.700	-.390	-.568	-.407	-.440	-.417	-.305	-.322	-.164
.850	-.366	-1.090	-.346	-.592	-.326	-.305	-.268	-.137
.950	-.407	-.908	-.353	-.481	-.231	-.265	-.127	-.103

ALPHA = 17.97 DEG Q = 226 N/SQ.M. (4.71 LBF/SQ.FT.) EFFECT. VELOC. RATIO = .092

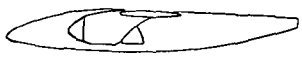
	ETA = .250		ETA = .387		ETA = .524		ETA = .800	
X/C	CP		CP		CP		CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			-.050		-1.531		-1.741	
.010			-3.109	.913				
.015					-2.502			
.025			-2.146		-2.167	.983	-2.118	.969
.050			-1.531	.843	-1.377	.836	-1.391	.795
.100			-1.363	.550	-1.489	.571	-1.580	.550
.150			-1.524	.306	-1.447	.362	-1.405	.362
.250	-.637		-1.266	.055	-1.049	.122	-1.308	.105
.400	-.690		-.863	-.230	-.829	-.155	-.819	-.084
.550	-.480	-.413	-.744	-.284	-.717	-.237	-.663	-.176
.700	-.447	-.477	-.514	-.409	-.531	-.338	-.372	-.176
.850	-.379	-1.110	-.355	-.579	-.365	-.342	-.328	-.179
.950	-.399	-.897	-.345	-.433	-.314	-.270	-.206	-.142

ALPHA = 19.99 DEG Q = 228 N/SQ.M. (4.77 LBF/SQ.FT.) EFFECT. VELOC. RATIO = .092

	ETA = .250		ETA = .387		ETA = .524		ETA = .800	
X/C	CP		CP		CP		CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			-.410		-.376		-1.404	
.010			-2.888	.997				
.015					-.921			
.025			-.921		-1.156	.990	-1.225	.997
.050			-.887	.859	-.935	.797	-1.225	.804
.100			-.887	.638	-.790	.542	-1.266	.542
.150			-1.073	.342	-1.128	.362	-1.204	.349
.250	-.755		-.838	.093	-.997	.133	-1.211	.133
.400	-.726		-1.001	-.175	-.971	-.111	-1.145	-.054
.550	-.713	-.422	-.690	-.302	-.817	-.275	-1.024	-.158
.700	-.456	-.496	-.613	-.379	-.780	-.345	-.850	-.221
.850	-.412	-1.084	-.616	-.563	-.629	-.402	-.579	-.258
.950	-.502	-.820	-.519	-.482	-.452	-.412	-.502	-.245

TABLE 16

PRESSURE PROFILE ON WING



FORWARD NOZZLES DEFLECTED 90 DEGREES WITH INLETS CLOSED

ALPHA = -.03 DEG Q = 2839 N/SC.M. (59.29 LBF/SQ.FT.) EFFECT. VELOC. RATIO = ∞

	ETA = .250		ETA = .387		ETA = .524		ETA = .800	
X/C	CP		CP		CP		CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			.456		.809		.951	
.010			.644	-.563				
.015					-.023	-.610		
.025			-.188	-.649	-.195	-.354	-.205	-.449
.050			-.293	-.507	-.281	-.394	-.195	-.332
.100			-.272	-.510	-.251	-.425	-.210	-.325
.150			-.368	-.511	-.294	-.444	-.281	-.364
.250	-.260		-.357	-.448	-.336	-.433	-.331	-.407
.400	-.331		-.350	-.365	-.337	-.424	-.317	-.370
.550	-.263	-.332	-.266	-.311	-.256	-.343	-.230	-.280
.700	-.201	-.371	-.170	-.324	-.173	-.202	-.140	-.160
.850	-.162	-.303	-.125	-.298	-.054	-.113	-.023	-.066
.950	-.154	-.212	-.094	-.182	.007	-.003	.029	.038

ALPHA = 3.99 DEG Q = 2836 N/SQ.M. (59.23 LBF/SQ.FT.) EFFECT. VELOC. RATIO = ∞

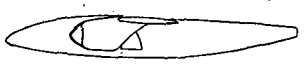
	ETA = .250		ETA = .387		ETA = .524		ETA = .800	
X/C	CP		CP		CP		CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			.958		.928		.523	
.010			.587	.405				
.015					-.948	.367		
.025			-.985	.016	-.975	.126	-.842	.227
.050			-.637	-.013	-.732	.067	-.730	.167
.100			-.576	-.033	-.579	-.101	-.532	-.040
.150			-.526	-.339	-.545	-.191	-.497	-.060
.250	-.321		-.460	-.341	-.432	-.287	-.491	-.209
.400	-.357		-.427	-.361	-.478	-.327	-.417	-.258
.550	-.256	-.354	-.282	-.326	-.280	-.270	-.271	-.216
.700	-.163	-.369	-.180	-.270	-.177	-.199	-.153	-.147
.850	-.092	-.266	-.086	-.197	-.056	-.067	-.047	-.053
.950	-.069	-.167	-.037	-.052	.027	.016	.035	.040

ALPHA = 8.03 DEG Q = 2848 N/SQ.M. (59.49 LBF/SQ.FT.) EFFECT. VELOC. RATIO = ∞

	ETA = .250		ETA = .387		ETA = .524		ETA = .800	
X/C	CP		CP		CP		CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			.746		-.277		-1.205	
.010			.543	.953				
.015					-2.157	.855		
.025			-1.762	.590	-1.873	.665	-1.884	.694
.050			-1.106	.385	-1.221	.442	-1.447	.516
.100			-.818	.144	-.968	.221	-.904	.330
.150			-.772	-.002	-.872	.065	-.836	.134
.250	-.405		-.653	-.192	-.675	-.079	-.673	.025
.400	-.414		-.502	-.278	-.541	-.173	-.530	-.133
.550	-.279	-.337	-.333	-.237	-.355	-.148	-.348	-.117
.700	-.159	-.280	-.191	-.184	-.203	-.114	-.201	-.096
.850	-.059	-.161	-.067	-.070	-.058	-.037	-.080	-.043
.950	-.001	-.053	.020	.021	.036	.023	.015	.016

TABLE 16 - CONCLUDED

PRESSURE PROFILE ON WING



FORWARD NOZZLES DEFLECTED 90 DEGREES WITH INLETS CLOSED

ALPHA = 10.02 DEG Q = 2847 N/SQ.M. (59.46 LBF/SQ.FT.) EFFECT. VELOC. RATIO = ∞

	ETA = .250		ETA = .387		ETA = .524		ETA = .800	
X/C	CP		CP		CP		CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			.127		-1.395		-2.923	
.010			.470	.993				
.015					-2.888	.953		
.025			-2.532	.838	-3.121	.825	-2.461	.876
.050			-1.490	.568	-1.712	.576	-1.612	.619
.100			-1.043	.298	-1.190	.378	-1.163	.378
.150			-.896	.172	-1.019	.202	-.977	.251
.250	-.455		-.731	-.063	-.796	.037	-.792	.063
.400	-.440		-.554	-.216	-.598	-.099	-.572	-.054
.550	-.307	-.279	-.356	-.197	-.396	-.096	-.390	-.078
.700	-.185	-.182	-.204	-.116	-.224	-.055	-.223	-.055
.850	-.079	-.039	-.072	-.013	-.070	-.018	-.093	-.017
.950	-.006	.012	.026	.041	.021	.026	-.008	.016

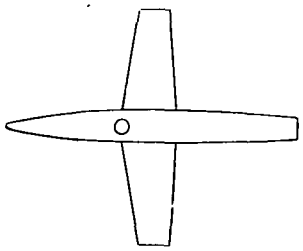
ALPHA = 12.00 DEG Q = 2845 N/SQ.M. (59.41 LBF/SQ.FT.) EFFECT. VELOC. RATIO = ∞

	ETA = .250		ETA = .387		ETA = .524		ETA = .800	
X/C	CP		CP		CP		CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			-.644		-2.452		-4.104	
.010			.335	.947				
.015					-4.577	.973		
.025			-2.773	.901	-2.831	.935	-2.853	.931
.050			-1.798	.696	-2.159	.746	-1.903	.730
.100			-1.211	.440	-1.409	.486	-1.404	.512
.150			-1.045	.286	-1.195	.318	-1.130	.353
.250	-.480		-.807	.021	-.897	.142	-.877	.132
.400	-.446		-.583	-.161	-.644	-.035	-.640	.017
.550	-.305	-.251	-.370	-.152	-.414	-.049	-.415	-.041
.700	-.176	-.140	-.208	-.097	-.229	-.026	-.238	-.040
.850	-.064	-.033	-.062	-.010	-.071	-.005	-.104	-.022
.950	.019	.014	.030	.049	.013	.023	-.019	.009

TABLE 17 (A)

PRESSURE PROFILE ON FUSELAGE

LIFT JET ALONE



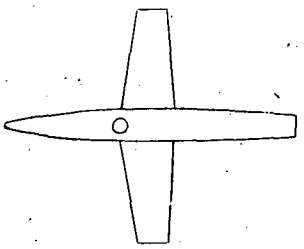
	ALPHA = .00 DEG EFF. VEL. RATIO =∞			ALPHA = 4.00 DEG EFF. VEL. RATIO =∞			ALPHA = 7.96 DEG EFF. VEL. RATIO =∞		
	CP			CP			CP		
Y/YMAX =	0.0	0.5	0.8	0.0	0.5	0.8	0.0	0.5	0.8
X/L									
.130	-.068			-.013			.044		
.219	-.063			-.011			.034		
.286	-.098			-.049			-.006		
.331	-.094			-.049			-.001		
.353	-.076	-.055	-.109	-.031	-.006	-.024	.037	.046	.015
.375	-.116			-.084			-.056		
.397		-.065	-.103		-.007	-.025		.053	.031
.420	-.121	-.087	-.073	-.016	-.042	-.046	.030	.017	.015
.442	-.070	-.092	-.095	-.012	-.036	-.046	.039	.011	-.004
.464	-.077	-.121	-.124	-.042	-.072	-.068	-.004	-.025	-.026
.487	-.113	-.099	-.113	-.072	-.062	-.066	-.021	-.026	-.026
.509	-.104	-.109	-.111	-.064	-.070	-.066	-.017	-.028	-.025
.531	-.115	-.104	-.095	-.081	-.072	-.067	-.042	-.037	-.034
.554	-.115			-.084			-.053		
.576	-.101			-.079			-.056		
.598	-.092	-.077	-.086	-.074	-.055	-.044	-.052	-.038	-.014
.620	-.069			-.056			-.038		
.643	-.056	-.054	-.059	-.047	-.042	-.041	-.032	-.014	-.011
.665	-.034	-.036	-.062	-.029	-.028	-.034	-.012	-.008	-.009
.687	-.021			-.013			-.015		
.710	-.017			-.008			-.008		
.732	-.001			-.002			-.003		

	ALPHA = 9.97 DEG EFF. VEL. RATIO =∞			ALPHA =12.01 DEG EFF. VEL. RATIO =∞			ALPHA =14.03 DEG EFF. VEL. RATIO =∞		
	CP			CP			CP		
Y/YMAX =	0.0	0.5	0.8	0.0	0.5	0.8	0.0	0.5	0.8
X/L									
.130	.071			.102			.138		
.219	.059			.092			.120		
.286	.032			.064			.100		
.331	.034			.056			.083		
.353	.062	.067	.030	.081	.096	.048	.111	.117	.052
.375	-.037			-.018			.001		
.397		.078	.052		.098	.058		.113	.072
.420	.045	.044	.036	.056	.056	.067	.063	.069	.075
.442	.057	.035	.024	.060	.051	.027	.070	.062	.014
.464	.025	-.005	-.016	.022	.004	-.013	.012	.011	-.020
.487	-.007	.010	-.012	-.001	-.001	-.021	.003	-.002	-.035
.509	-.012	-.012	-.005	-.009	-.009	-.012	-.012	-.016	-.020
.531	-.024	-.018	-.003	-.026	-.028	-.022	-.029	-.031	-.035
.554	-.035			-.040			-.048		
.576	-.042			-.044			-.058		
.598	-.040	-.029	-.008	-.050	-.040	-.024	-.058	-.052	-.031
.620	-.033			-.044			-.058		
.643	-.029	-.014	-.007	-.041	-.029	-.022	-.051	-.040	-.040
.665	-.011	-.004	-.007	-.023	-.016	-.018	-.031	-.029	-.038
.687	.000			-.018			-.029		
.710	-.001			-.014			-.025		
.732	.002			-.014			-.018		

TABLE 17 (A) - CONCLUDED

PRESSURE PROFILE ON FUSELAGE

LIFT JET ALONE

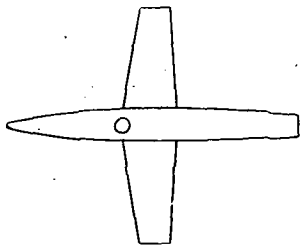


	ALPHA =15.98 DEG EFF. VEL. RATIO =∞			ALPHA =18.04 DEG EFF. VEL. RATIO =∞			ALPHA =20.05 DEG EFF. VEL. RATIO =∞		
	CP			CP			CP		
Y/YMAX =	0.0	0.5	0.8	0.0	0.5	0.8	0.0	0.5	0.8
X/L									
.130	.184			.213			.253		
.219	.155			.201			.233		
.286	.114			.151			.184		
.331	.109			.143			.166		
.353	.140	.135	.056	.168	.165	.050	.205	.195	.044
.375	.019			.046			.070		
.397		.127	.070		.155	.083		.182	.103
.420	.062	.072	.069	.085	.093	.079	.105	.132	.130
.442	.069	.075	.014	.072	.087	.009	.089	.122	.024
.464	.010	.019	-.029	.015	.032	-.047	.033	.059	.000
.487	.006	.019	-.057	.019	.012	-.063	.037	.041	-.041
.509	-.028	-.031	-.052	-.021	-.015	-.064	-.004	-.004	-.069
.531	-.038	-.050	-.075	-.034	-.040	-.085	-.017	-.028	-.078
.554	-.063			-.063			-.039		
.576	-.067			-.076			-.056		
.598	-.075	-.071	-.064	-.089	-.098	-.104	-.073	-.087	-.109
.620	-.073			-.082			-.081		
.643	-.072	-.060	-.100	-.082	-.088	-.122	-.081	-.083	-.129
.665	-.052	-.060	-.069	-.068	-.079	-.104	-.065	-.077	-.131
.687	-.042			-.059			-.060		
.710	-.045			-.053			-.055		
.732	-.034			-.047			-.050		

TABLE 17 (B)

PRESSURE PROFILE ON FUSELAGE

LIFT JET ALONE



ALPHA = .02 DEG
EFF. VEL. RATIO = .318

	CP		
Y/YMAX =	0.0	0.5	0.8
X/L			
.130	-.046		
.219	-.017		
.286	.019		
.331	.145		
.353	.357	.140	-.064
.375	.475		
.397		-4.278	-1.312
.420	-.939	-2.659	-2.371
.442	-1.583	-.903	-1.298
.464	-1.719	-.602	-.997
.487	-.568	-.695	-.533
.509	-.022	-.403	-.511
.531	-.429	-.359	-.449
.554	-.401		
.576	-.331		
.598	-.345	-.314	-.269
.620	-.354		
.643	-.258	-.219	-.199
.665	-.229	-.263	-.157
.687	-.218		
.710	-.214		
.732	-.202		

ALPHA = 4.04 DEG
EFF. VEL. RATIO = .319

	CP		
	0.0	0.5	0.8
	-.004		
	.028		
	.058		
	.187		
	.382	.162	-.088
	.478		
		-4.444	-1.338
	-1.285	-2.490	-2.216
	-1.551	-.846	-1.152
	-.782	-.496	-.909
	-.407	-.246	-.584
	-.084	-.379	-.451
	-.302	-.286	-.317
	-.298		
	-.311		
	-.297	-.209	-.188
	-.292		
	-.218	-.204	-.155
	-.238	-.171	-.137
	-.223		
	-.153		
	-.195		

ALPHA = 8.01 DEG
EFF. VEL. RATIO = .319

	CP		
	0.0	0.5	0.8
	.054		
	.086		
	.110		
	.220		
	.423	.191	-.114
	.477		
		-4.632	-1.309
	-1.140	-2.495	-1.999
	-1.347	-.767	-1.047
	-.848	-.459	-.771
	-.376	-.776	-.463
	-.551	-.245	-.346
	-.275	-.215	-.309
	-.215		
	-.207		
	-.232	-.162	-.135
	-.196		
	-.135	-.158	-.111
	-.152	-.114	-.082
	-.165		
	-.138		
	-.124		

ALPHA =10.00 DEG
EFF. VEL. RATIO = .319

	CP		
Y/YMAX =	0.0	0.5	0.8
X/L			
.130	.083		
.219	.105		
.286	.127		
.331	.256		
.353	.430	.198	-.112
.375	.477		
.397		-4.555	-1.308
.420	-1.312	-2.709	-1.525
.442	-1.434	-.788	-1.038
.464	-1.310	-.400	-.759
.487	-.311	-.217	-.484
.509	-.305	-.259	-.388
.531	-.243	-.190	-.283
.554	-.210		
.576	-.201		
.598	-.168	-.114	-.107
.620	-.155		
.643	-.171	-.069	-.043
.665	-.145	-.097	-.062
.687	-.156		
.710	-.130		
.732	-.118		

ALPHA =12.00 DEG
EFF. VEL. RATIO = .319

	CP		
	0.0	0.5	0.8
	.128		
	.133		
	.164		
	.262		
	.447	.199	-.209
	.481		
		-4.760	-1.313
	-1.399	-2.615	-3.478
	-1.286	-.770	-.989
	-.882	-.432	-.740
	-.267	-.463	-.464
	-.731	-.242	-.368
	-.189	-.177	-.244
	-.173		
	-.211		
	-.168	-.105	-.085
	-.098		
	-.110	-.089	-.054
	-.106	-.081	-.043
	-.118		
	-.140		
	-.072		

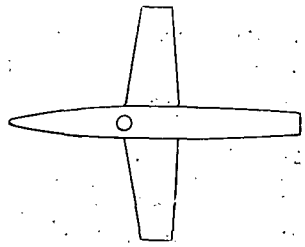
ALPHA =13.96 DEG
EFF. VEL. RATIO = .319

	CP		
	0.0	0.5	0.8
	.159		
	.170		
	.198		
	.294		
	.469	.218	-.166
	.477		
		-4.709	-1.272
	-1.272	-2.744	-2.573
	-1.352	-.838	-1.048
	-.814	-.415	-.790
	-.333	-.519	-.469
	-.268	-.245	-.307
	-.189	-.161	-.281
	-.175		
	-.095		
	-.126	-.075	-.061
	-.106		
	-.063	-.078	-.027
	-.097	-.023	-.022
	-.060		
	-.103		
	-.057		

TABLE 17 (B) - CONCLUDED

PRESSURE PROFILE ON FUSELAGE

LIFT JET ALONE

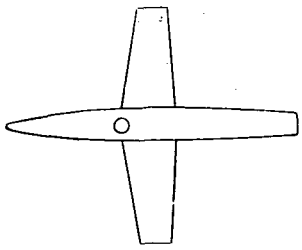


	ALPHA =16.06 DEG EFF. VEL. RATIO = .319			ALPHA =18.03 DEG EFF. VEL. RATIO = .319			ALPHA =20.00 DEG EFF. VEL. RATIO = .318		
	CP			CP			CP		
Y/YMAX =	0.0	0.5	0.8	0.0	0.5	0.8	0.0	0.5	0.8
X/L									
.130	.194			.225			.272		
.219	.202			.241			.260		
.286	.227			.263			.273		
.331	.314			.340			.365		
.353	.474	.221	-.238	.493	.238	-.285	.513	.265	-.314
.375	.485			.484			.490		
.397		-4.728	-1.246		-4.686	-1.280		-4.509	-1.214
.420	-1.151	-2.658	-2.880	-1.344	-2.667	-2.835	-1.265	-2.611	-2.777
.442	-1.295	-.829	-1.075	-1.305	-.931	-1.038	-1.188	-.948	-1.002
.464	-1.300	-.463	-.762	-1.330	-.435	-.768	-1.410	-.421	-.736
.487	-.306	-.299	-.481	-.316	-.465	-.565	-.319	-.370	-.489
.509	-.409	-.293	-.476	-.339	-.333	-.417	-.312	-.201	-.368
.531	-.174	-.154	-.295	-.178	-.157	-.250	-.152	-.159	-.215
.554	-.142			-.169			-.109		
.576	-.138			-.107			-.101		
.598	-.126	-.053	-.083	-.097	-.092	-.073	-.087	-.074	-.083
.620	-.093			-.097			-.113		
.643	-.079	-.081	-.043	-.155	-.073	-.080	-.083	-.035	-.059
.665	-.060	-.045	-.070	-.094	-.062	-.041	-.072	-.041	-.052
.687	-.115			-.085			-.070		
.710	-.077			-.044			-.077		
.732	-.037			-.058			-.067		

TABLE 17 (C)

PRESSURE PROFILE ON FUSELAGE

LIFT JET ALONE



ALPHA = -.02 DEG
EFF. VEL. RATIO = .200

	CP		
Y/YMAX =	0.0	0.5	0.8
X/L			
.130	-.056		
.219	-.036		
.286	-.037		
.331	.053		
.353	.262	.004	-.190
.375	.379		
.397		-4.492	-1.473
.420	-1.560	-1.947	-3.707
.442	-1.462	-.749	-1.132
.464	-.337	-.499	-.920
.487	-.532	-.851	-.485
.509	-.470	-.360	-.482
.531	-.509	-.405	-.474
.554	-.382		
.576	-.438		
.598	-.421	-.434	-.281
.620	-.475		
.643	-.399	-.309	-.223
.665	-.347	-.324	-.226
.687	-.293		
.710	-.251		
.732	-.268		

ALPHA = 4.00 DEG
EFF. VEL. RATIO = .200

	CP		
	0.0	0.5	0.8
	.002		
	.002		
	.005		
	.076		
	.279	.027	-.193
	.366		
		-4.100	-1.358
	-1.398	-1.757	-1.526
	-1.342	-.664	-1.206
	-1.113	-.473	-.823
	-.550	-.380	-.526
	-.486	-.527	-.457
	-.440	-.419	-.499
	-.518		
	-.508		
	-.488	-.333	-.290
	-.468		
	-.373	-.435	-.274
	-.423	-.451	-.244
	-.350		
	-.326		
	-.315		

ALPHA = 8.04 DEG
EFF. VEL. RATIO = .200

	CP		
	0.0	0.5	0.8
	.043		
	.052		
	.058		
	.114		
	.299	.064	-.233
	.365		
		-4.004	-1.336
	-.637	-1.637	-.427
	-1.285	-.568	-1.078
	-1.262	-.453	-.743
	-.575	-.870	-.487
	-.759	-.534	-.418
	-.440	-.440	-.367
	-.599		
	-.604		
	-.508	-.428	-.339
	-.470		
	-.533	-.432	-.255
	-.421	-.362	-.255
	-.420		
	-.425		
	-.375		

ALPHA = 9.96 DEG
EFF. VEL. RATIO = .200

	CP		
Y/YMAX =	0.0	0.5	0.8
X/L			
.130	.078		
.219	.086		
.286	.074		
.331	.127		
.353	.317	.041	-.241
.375	.360		
.397		-3.826	-1.215
.420	-1.205	-1.621	-2.418
.442	-1.220	-.598	-1.215
.464	-.426	-.521	-.667
.487	-.587	-.128	-.434
.509	-.406	-.441	-.465
.531	-.557	-.381	-.373
.554	-.688		
.576	-.633		
.598	-.459	-.528	-.275
.620	-.522		
.643	-.489	-.316	-.285
.665	-.432	-.429	-.224
.687	-.517		
.710	-.482		
.732	-.402		

ALPHA =11.98 DEG
EFF. VEL. RATIO = .199

	CP		
	0.0	0.5	0.8
	.109		
	.123		
	.089		
	.148		
	.332	.068	-.273
	.362		
		-3.616	-1.257
	-1.098	-1.549	-.601
	-1.198	-.561	-1.044
	-1.017	-.466	-.827
	-.595	-.415	-.443
	-.381	-.508	-.473
	-.455	-.428	-.377
	-.572		
	-.591		
	-.642	-.525	-.290
	-.595		
	-.629	-.396	-.215
	-.498	-.388	-.217
	-.483		
	-.374		
	-.427		

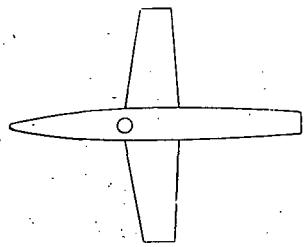
ALPHA =14.02 DEG
EFF. VEL. RATIO = .199

	CP		
	0.0	0.5	0.8
	.154		
	.139		
	.119		
	.167		
	.341	.098	-.274
	.364		
		-3.765	-1.140
	-1.989	-1.578	-.823
	-1.135	-.654	-.985
	-.114	-.453	-.627
	-.503	-1.190	-.422
	-.667	-.549	-.476
	-.601	-.457	-.352
	-.725		
	-.610		
	-.599	-.504	-.300
	-.615		
	-.525	-.405	-.258
	-.506	-.374	-.241
	-.536		
	-.532		
	-.438		

TABLE 17 (C) - CONCLUDED

PRESSURE PROFILE ON FUSELAGE

LIFT JET ALONE

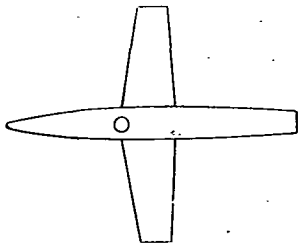


	ALPHA =16.00 DEG EFF. VEL. RATIO = .199			ALPHA =17.98 DEG EFF. VEL. RATIO = .199			ALPHA =19.98 DEG EFF. VEL. RATIO = .198		
		CP			CP			CP	
Y/YMAX =	0.0	0.5	0.8	0.0	0.5	0.8	0.0	0.5	0.8
X/L									
.130	.183			.223			.267		
.219	.175			.200			.240		
.286	.139			.175			.209		
.331	.178			.208			.227		
.353	.352	.091	-.317	.364	.110	-.332	.382	.107	-.346
.375	.366			.365			.373		
.397		-3.552	-1.223		-3.722	-1.182		-3.766	-1.063
.420	-.771	-1.568	-1.617	-1.129	-1.414	-2.723	-1.700	-1.410	-2.653
.442	-1.140	-.546	-.977	-1.168	-.387	-.781	-1.120	-.439	-.885
.464	-.814	-.477	-.645	-.958	-.469	-.773	-.856	-.423	-.566
.487	-.610	-.893	-.438	-.602	-.328	-.472	-.790	-1.577	-.482
.509	-.752	-.778	-.387	-.496	-.389	-.449	-.820	-.451	-.418
.531	-.571	-.492	-.433	-.851	-.510	-.351	-.603	-.471	-.443
.554	-.718			-.667			-.677		
.576	-.672			-.715			-.767		
.598	-.531	-.558	-.297	-.711	-.522	-.316	-.714	-.532	-.285
.620	-.605			-.604			-.643		
.643	-.566	-.511	-.260	-.715	-.524	-.286	-.730	-.507	-.251
.665	-.578	-.491	-.243	-.593	-.469	-.241	-.596	-.428	-.224
.687	-.572			-.519			-.597		
.710	-.532			-.522			-.548		
.732	-.490			-.465			-.585		

TABLE 17 (D)

PRESSUPE PROFILE ON FUSELAGE

LIFT JET ALONE



ALPHA = .01 DEG
EFF. VEL. RATIO = .099

	CP		
Y/YMAX =	0.0	0.5	0.8
X/L			
.130	-.081		
.219	-.111		
.286	-.186		
.331	-.163		
.353	-.037	-.251	-.346
.375	.039		
.397		-4.457	-1.360
.420	.044	-2.188	-2.828
.442	-1.632	-.581	-1.371
.464	-1.953	-.442	-1.600
.487	-.431	-2.556	-.591
.509	-1.050	-.495	-.730
.531	-.426	-.410	-.453
.554	-.497		
.576	-.469		
.598	-.482	-.321	-.344
.620	-.408		
.643	-.368	-.324	-.272
.665	-.303	-.329	-.303
.687	-.300		
.710	-.274		
.732	-.212		

ALPHA = .02 DEG
EFF. VEL. RATIO = .100

	CP		
Y/YMAX =	0.0	0.5	0.8
X/L			
.130	-.074		
.219	-.110		
.286	-.169		
.331	-.150		
.353	-.037	-.204	-.403
.375	.019		
.397		-5.061	-1.415
.420	-2.795	-1.796	.218
.442	-1.329	-.567	-1.904
.464	.164	-.491	-1.104
.487	-.432	-.357	-.830
.509	-.357	-.459	-.819
.531	-.443	-.406	-.636
.554	-.506		
.576	-.467		
.598	-.443	-.388	-.359
.620	-.458		
.643	-.381	-.322	-.293
.665	-.340	-.319	-.282
.687	-.305		
.710	-.279		
.732	-.260		

ALPHA = 3.96 DEG
EFF. VEL. RATIO = .100

	CP		
Y/YMAX =	0.0	0.5	0.8
X/L			
.130	-.046		
.219	-.078		
.286	-.148		
.331	-.151		
.353	-.027	-.253	-.407
.375	.010		
.397		-4.069	-1.391
.420	-.416	-1.897	-3.535
.442	-.998	-.594	-1.714
.464	.597	-.330	-1.165
.487	-.383	-.443	-.626
.509	-.432	-.335	-.583
.531	-.416	-.367	-.507
.554	-.418		
.576	-.367		
.598	-.433	-.326	-.289
.620	-.349		
.643	-.311	-.293	-.264
.665	-.292	-.217	-.245
.687	-.282		
.710	-.221		
.732	-.200		

ALPHA = 7.99 DEG
EFF. VEL. RATIO = .099

	CP		
Y/YMAX =	0.0	0.5	0.8
X/L			
.130	.008		
.219	-.039		
.286	-.110		
.331	-.120		
.353	.008	-.154	-.346
.375	.036		
.397		-4.185	-1.154
.420	-.483	-1.566	-.169
.442	-.072	-.467	-1.062
.464	-1.771	-.272	-.754
.487	-.445	-.251	-.624
.509	-.619	-.316	-.537
.531	-.397	-.370	-.391
.554	-.424		
.576	-.382		
.598	-.375	-.353	-.276
.620	-.383		
.643	-.316	-.380	-.280
.665	-.336	-.305	-.277
.687	-.315		
.710	-.297		
.732	-.263		

ALPHA = 9.95 DEG
EFF. VEL. RATIO = .099

	CP		
Y/YMAX =	0.0	0.5	0.8
X/L			
.130	.034		
.219	-.021		
.286	-.091		
.331	-.098		
.353	.023	-.207	-.340
.375	.022		
.397		-3.803	-1.018
.420	.158	-1.398	-4.952
.442	-.737	-.514	-.861
.464	-.119	-.314	-1.040
.487	-.341	-.043	-.617
.509	-.596	-.346	-.579
.531	-.319	-.309	-.444
.554	-.321		
.576	-.343		
.598	-.387	-.451	-.286
.620	-.355		
.643	-.341	-.427	-.202
.665	-.343	-.331	-.206
.687	-.271		
.710	-.290		
.732	-.269		

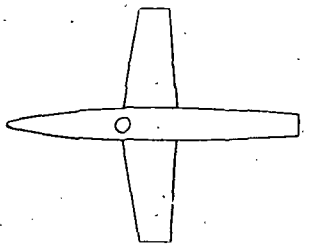
ALPHA =12.06 DEG
EFF. VEL. RATIO = .099

	CP		
Y/YMAX =	0.0	0.5	0.8
X/L			
.130	.074		
.219	.025		
.286	-.069		
.331	-.073		
.353	.026	-.195	-.255
.375	.017		
.397		-3.414	-1.085
.420	.088	-1.150	-2.360
.442	-.621	-.426	-1.020
.464	.277	-.259	-.907
.487	-.313	-.248	-.583
.509	-1.269	-.313	-.388
.531	-.329	-.291	-.253
.554	-.334		
.576	-.310		
.598	-.324	-.312	-.208
.620	-.328		
.643	-.283	-.379	-.250
.665	-.244	-.350	-.199
.687	-.274		
.710	-.250		
.732	-.257		

TABLE 17 (D) - CONCLUDED

PRESSURE PROFILE ON FUSELAGE

LIFT JET ALONE

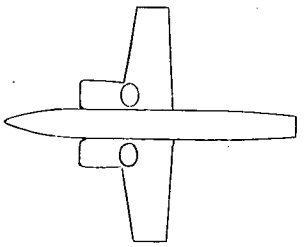


	ALPHA =14.00 DEG EFF. VEL. RATIO = .099			ALPHA =15.98 DEG EFF. VEL. RATIO = .099			ALPHA =17.98 DEG EFF. VEL. RATIO = .099		
	CP			CP			CP		
Y/YMAX =	0.0	0.5	0.8	0.0	0.5	0.8	0.0	0.5	0.8
X/L									
.130	.106			.114			.169		
.219	.044			.067			.107		
.286	-.043			-.023			.012		
.331	-.056			-.052			-.023		
.353	.031	-.136	-.272	.061	-.149	-.311	.073	-.123	-.279
.375	.016			.040			.033		
.397		-3.495	-1.057		-3.659	-.822		-3.124	-.964
.420	.114	-1.117	.281	-2.060	-1.279	-3.611	-.133	-1.146	.077
.442	-1.332	-.329	-1.257	-1.086	-.370	-.833	.211	-.347	-.567
.464	.604	-.232	-.566	-.660	-.197	-.671	-.369	-.176	-.744
.487	-.334	-.038	-.421	-.197	-.192	-.445	-.272	-.111	-.412
.509	-.232	-.270	-.356	.045	-.321	-.375	-.363	-.261	-.438
.531	-.232	-.318	-.512	-.256	-.251	-.396	-.197	-.358	-.390
.554	-.266			-.248			-.215		
.576	-.290			-.271			-.199		
.598	-.323	-.348	-.247	-.341	-.311	-.248	-.256	-.283	-.264
.620	-.263			-.289			-.228		
.643	-.273	-.377	-.181	-.282	-.295	-.228	-.311	-.319	-.243
.665	-.328	-.343	-.204	-.233	-.323	-.200	-.199	-.305	-.190
.687	-.255			-.202			-.199		
.710	-.268			-.244			-.267		
.732	-.248			-.273			-.239		

TABLE 18 (A)

PRESSURE PROFILE ON FUSELAGE

FORWARD NOZZLES DEFLECTED 45 DEGREES



ALPHA = 0.00 DEG
EFF. VEL. RATIO =∞

	CP		
Y/YMAX =	0.0	0.5	0.8
X/L			
.130	-.019		
.219	-.033		
.286	-.116		
.331	-.161		
.353	-.112	-.172	-.217
.375	-.116		
.397		-.116	-.118
.420	-.129	-.080	-.069
.442	-.081	-.055	-.075
.464	-.067	-.115	-.101
.487	-.116	-.109	-.127
.509	-.129	-.132	-.145
.531	-.137	-.137	-.161
.554	-.151		
.576	-.140		
.598	-.127	-.127	-.128
.620	-.093		
.643	-.065	-.057	-.065
.665	-.031	-.036	-.037
.687	.006		
.710	.015		
.732	.020		

ALPHA = 4.04 DEG
EFF. VEL. RATIO =∞

	CP		
	0.0	0.5	0.8
	.024		
	.038		
	-.054		
	-.104		
	-.067	-.107	-.119
	-.084		
		-.086	-.086
	-.092	-.056	-.052
	-.029	-.025	-.043
	-.063	-.094	-.084
	-.105	-.105	-.106
	-.101	-.115	-.126
	-.129	-.124	-.148
	-.135		
	-.126		
	-.111	-.099	-.107
	-.081		
	-.060	-.055	-.055
	-.033	-.029	-.028
	-.005		
	.008		
	.021		

ALPHA = 7.98 DEG
EFF. VEL. RATIO =∞

	CP		
	0.0	0.5	0.8
	.075		
	.102		
	.024		
	-.034		
	-.024	-.042	-.080
	-.042		
		-.045	-.042
	-.050	-.020	-.018
	-.008	.005	-.028
	-.016	-.073	-.067
	-.084	-.084	-.079
	-.084	-.093	-.101
	-.115	-.107	-.115
	-.125		
	-.116		
	-.105	-.090	-.083
	-.082		
	-.065	-.051	-.050
	-.037	-.034	-.045
	-.017		
	-.007		
	.007		

ALPHA = 9.98 DEG
EFF. VEL. RATIO =∞

	CP		
Y/YMAX =	0.0	0.5	0.8
X/L			
.130	.114		
.219	.128		
.286	.059		
.331	-.008		
.353	.021	-.007	-.041
.375	-.007		
.397		-.021	-.026
.420	-.025	-.000	.004
.442	.012	.012	-.007
.464	-.021	-.059	-.055
.487	-.064	-.065	-.068
.509	-.065	-.089	-.089
.531	-.099	-.098	-.096
.554	-.112		
.576	-.109		
.598	-.103	-.088	-.074
.620	-.079		
.643	-.066	-.052	-.047
.665	-.043	-.038	-.038
.687	-.023		
.710	-.012		
.732	.003		

ALPHA =12.01 DEG
EFF. VEL. RATIO =∞

	CP		
	0.0	0.5	0.8
	.159		
	.173		
	.093		
	.034		
	.038	.019	.004
	.018		
		-.005	-.005
	-.008	.013	.017
	.024	.021	-.001
	-.001	-.043	-.044
	-.053	-.057	-.056
	-.055	-.077	-.082
	-.097	-.094	-.095
	-.103		
	-.103		
	-.101	-.081	-.063
	-.078		
	-.065	-.051	-.046
	-.049	-.037	-.033
	-.034		
	-.023		
	-.013		

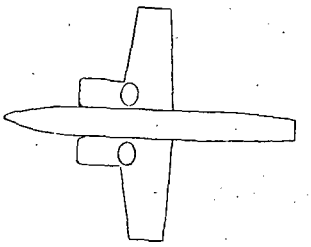
ALPHA =13.98 DEG
EFF. VEL. RATIO =∞

	CP		
	0.0	0.5	0.8
	.178		
	.201		
	.135		
	.057		
	.064	.030	.009
	.023		
		-.000	-.001
	-.003	.010	.017
	.014	.021	-.022
	-.018	-.064	-.067
	-.069	-.080	-.098
	-.093	-.097	-.118
	-.110	-.117	-.134
	-.123		
	-.131		
	-.124	-.119	-.108
	-.109		
	-.100	-.083	-.072
	-.070	-.062	-.053
	-.050		
	-.046		
	-.034		

TABLE 18 (A) - CONCLUDED

PRESSURE PROFILE ON FUSELAGE

FORWARD NOZZLES DEFLECTED 45 DEGREES

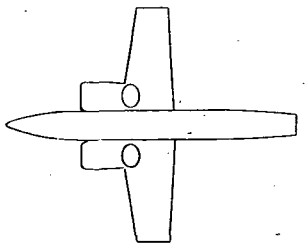


	ALPHA =16.02 DEG EFF. VEL. RATIO =∞			ALPHA =17.99 DEG EFF. VEL. RATIO =∞			ALPHA =20.00 DEG EFF. VEL. RATIO =∞		
	CP			CP			CP		
Y/YMAX =	0.0	0.5	0.8	0.0	0.5	0.8	0.0	0.5	0.8
X/L									
.130	.199			.259			.292		
.219	.235			.278			.310		
.286	.158			.201			.235		
.331	.081			.118			.161		
.353	.089	.059	.046	.122	.085	.069	.153	.116	.113
.375	.030			.074			.097		
.397		.010	.012		.026	.026		.052	.037
.420	.002	.016	.010	.022	.032	.026	.029	.039	.048
.442	.019	.026	-.016	.013	.033	-.031	.026	.058	-.021
.464	-.024	-.057	-.071	-.016	-.051	-.081	-.023	-.021	-.078
.487	-.069	-.071	-.115	-.059	-.052	-.131	-.052	-.059	-.118
.509	-.109	-.108	-.129	-.099	-.107	-.156	-.127	-.101	-.166
.531	-.119	-.129	-.151	-.112	-.126	-.171	-.105	-.124	-.179
.554	-.130			-.135			-.123		
.576	-.131			-.134			-.131		
.598	-.133	-.132	-.121	-.135	-.146	-.149	-.132	-.148	-.165
.620	-.121			-.122			-.119		
.643	-.111	-.101	-.104	-.112	-.112	-.128	-.112	-.122	-.141
.665	-.089	-.085	-.087	-.092	-.089	-.100	-.093	-.107	-.138
.687	-.062			-.072			-.071		
.710	-.053			-.063			-.064		
.732	-.046			-.052			-.050		

TABLE 18 (B)

PRESSURE PROFILE ON FUSELAGE

FORWARD NOZZLES DEFLECTED 45 DEGREES



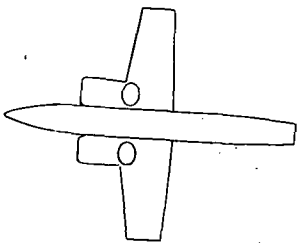
	ALPHA = -.01 DEG EFF. VEL. RATIO = .296			ALPHA = 4.00 DEG EFF. VEL. RATIO = .295			ALPHA = 8.04 DEG EFF. VEL. RATIO = .295		
	CP			CP			CP		
Y/YMAX =	0.0	0.5	0.8	0.0	0.5	0.8	0.0	0.5	0.8
X/L									
.130	-.031			.019			.089		
.219	.000			.045			.121		
.286	-.056			.032			.094		
.331	-.053			.014			.091		
.353	.015	-.034	-.064	.092	.034	.009	.141	.105	.091
.375	.018			.070			.146		
.397		.053	.073		.104	.128		.172	.197
.420	.071	.113	.089	.132	.167	.156	.207	.219	.225
.442	.144	.122	.155	.213	.169	.209	.265	.223	.274
.464	.135	.091	-.017	.215	.126	.088	.270	.186	.146
.487	.005	.031	-.322	.059	.070	-.229	.128	.165	-.173
.509	-.178	-.104	-.640	-.127	-.035	-.579	-.286	.024	-.447
.531	-.143	-.251	-.880	-.072	-.178	-.842	-.003	-.100	-.744
.554	-.173			-.108			-.047		
.576	-.122			-.078			-.001		
.598	-.017	-.275	-.507	.013	-.216	-.489	.046	-.141	-.411
.620	.118			.116			.139		
.643	.174	.030	.008	.170	-.045	.025	.159	-.141	.017
.665	.171	.042	-.122	.163	.029	-.081	.135	-.065	-.054
.687	.124			.130			.110		
.710	.092			.064			.059		
.732	.047			.041			.023		

	ALPHA = 9.99 DEG EFF. VEL. RATIO = .294			ALPHA =12.01 DEG EFF. VEL. RATIO = .294			ALPHA =14.01 DEG EFF. VEL. RATIO = .294		
	CP			CP			CP		
Y/YMAX =	0.0	0.5	0.8	0.0	0.5	0.8	0.0	0.5	0.8
X/L									
.130	.121			.157			.187		
.219	.156			.207			.242		
.286	.141			.203			.237		
.331	.131			.170			.208		
.353	.184	.152	.122	.219	.196	.177	.258	.231	.212
.375	.174			.216			.243		
.397		.196	.222		.250	.254		.263	.278
.420	.213	.242	.257	.250	.276	.269	.281	.305	.325
.442	.290	.246	.308	.336	.289	.345	.352	.312	.367
.464	.332	.213	.185	.342	.259	.232	.367	.281	.248
.487	.151	.162	-.126	.199	.216	-.040	.208	.232	-.029
.509	.033	.059	-.416	.115	.102	-.343	.132	.122	-.343
.531	.039	-.069	-.705	.064	-.016	-.608	.091	-.002	-.582
.554	.006			.044			.053		
.576	.009			.056			.062		
.598	.069	-.118	-.381	.105	-.079	-.320	.096	-.068	-.332
.620	.133			.158			.148		
.643	.166	-.182	.007	.176	-.106	.024	.157	-.162	-.004
.665	.132	-.059	-.060	.153	-.085	-.038	.161	-.138	-.052
.687	.107			.098			.107		
.710	.057			.068			.053		
.732	.010			.023			.014		

TABLE 18 (B) - CONCLUDED

PRESSURE PROFILE ON FUSELAGE

FORWARD NOZZLES DEFLECTED 45 DEGREES

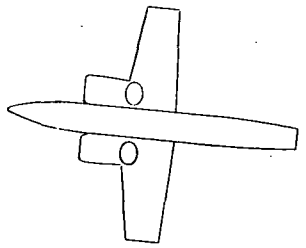


	ALPHA =15.99 DEG EFF. VEL. RATIO = .297			ALPHA =18.04 DEG EFF. VEL. RATIO = .297			ALPHA =20.01 DEG EFF. VEL. RATIO = .297		
	CP			CP			CP		
Y/YMAX =	0.0	0.5	0.8	0.0	0.5	0.8	0.0	0.5	0.8
X/L									
.130	.224			.280			.311		
.219	.271			.310			.349		
.286	.277			.324			.351		
.331	.253			.287			.330		
.353	.284	.250	.240	.305	.291	.277	.336	.308	.300
.375	.262			.303			.319		
.397		.279	.303		.299	.318		.320	.339
.420	.293	.317	.328	.305	.334	.361	.339	.350	.339
.442	.365	.324	.397	.383	.334	.414	.395	.351	.437
.464	.410	.288	.279	.409	.303	.296	.415	.320	.346
.487	.226	.257	.011	.232	.217	.020	.255	.236	.068
.509	.047	.120	-.327	.036	.113	-.296	.182	.160	-.268
.531	.100	.002	-.584	.093	.007	-.596	.107	.035	-.553
.554	.052			.050			.055		
.576	.068			.052			.038		
.598	.097	-.062	-.281	.090	-.055	-.336	.077	-.067	-.300
.620	.134			.120			.100		
.643	.158	-.201	.005	.113	-.169	-.017	.117	-.113	-.099
.665	.142	-.156	-.055	.142	-.178	-.036	.118	-.211	-.090
.687	.105			.096			.091		
.710	.022			.029			.042		
.732	-.010			.004			-.019		

TABLE 18 (C)

PRESSURE PROFILE ON FUSELAGE

FORWARD NOZZLES DEFLECTED 45 DEGREES



ALPHA = -.01 DEG
EFF. VEL. RATIO = .198

	CP		
Y/YMAX =	0.0	0.5	0.8
X/L			
.130	-.038		
.219	-.012		
.286	-.024		
.331	-.010		
.353	.003	-.004	-.044
.375	.032		
.397		.068	.073
.420	.068	.113	.130
.442	.162	.138	.166
.464	.244	.060	.036
.487	-.025	-.037	-.307
.509	-.131	-.201	-.934
.531	-.205	-.364	-1.273
.554	-.191		
.576	-.078		
.598	.092	-.148	-.056
.620	.184		
.643	.205	.064	-.218
.665	.189	.076	-.309
.687	.179		
.710	.155		
.732	.150		

ALPHA = 4.03 DEG
EFF. VEL. RATIO = .197

	CP		
Y/YMAX =	0.0	0.5	0.8
X/L			
.130	.045		
.219	.070		
.286	.086		
.331	.054		
.353	.116	.083	.047
.375	.122		
.397		.130	.134
.420	.155	.179	.192
.442	.237	.187	.241
.464	.233	.134	.142
.487	.044	-.001	-.219
.509	-.100	-.137	-.773
.531	-.141	-.313	-1.257
.554	-.147		
.576	-.052		
.598	.108	-.036	-.047
.620	.175		
.643	.185	.112	-.266
.665	.163	.054	-.239
.687	.142		
.710	.109		
.732	.137		

ALPHA = 8.03 DEG
EFF. VEL. RATIO = .197

	CP		
Y/YMAX =	0.0	0.5	0.8
X/L			
.130	.098		
.219	.131		
.286	.144		
.331	.140		
.353	.197	.142	.124
.375	.192		
.397		.188	.212
.420	.175	.237	.249
.442	.282	.241	.315
.464	.319	.184	.212
.487	.114	.077	-.087
.509	-.095	-.062	-.656
.531	-.087	-.238	-1.057
.554	-.104		
.576	-.007		
.598	.097	.021	-.039
.620	.175		
.643	.160	.020	-.222
.665	.148	.017	-.265
.687	.109		
.710	.111		
.732	.117		

ALPHA = 9.98 DEG
EFF. VEL. RATIO = .197

	CP		
Y/YMAX =	0.0	0.5	0.8
X/L			
.130	.123		
.219	.165		
.286	.202		
.331	.177		
.353	.224	.178	.151
.375	.214		
.397		.226	.226
.420	.226	.259	.271
.442	.308	.275	.345
.464	.336	.226	.250
.487	.132	.128	-.093
.509	.034	-.048	-.731
.531	-.040	-.220	-1.087
.554	-.073		
.576	.007		
.598	.119	-.057	-.077
.620	.158		
.643	.154	.055	-.230
.665	.139	.023	-.266
.687	.109		
.710	.112		
.732	.107		

ALPHA =12.03 DEG
EFF. VEL. RATIO = .197

	CP		
Y/YMAX =	0.0	0.5	0.8
X/L			
.130	.148		
.219	.199		
.286	.233		
.331	.213		
.353	.247	.205	.217
.375	.237		
.397		.257	.257
.420	.253	.311	.274
.442	.347	.327	.372
.464	.352	.249	.315
.487	.176	.233	-.037
.509	.078	.008	-.646
.531	-.033	-.159	-.981
.554	-.048		
.576	.004		
.598	.100	-.061	-.070
.620	.145		
.643	.144	.020	-.211
.665	.125	-.022	-.282
.687	.106		
.710	.102		
.732	.088		

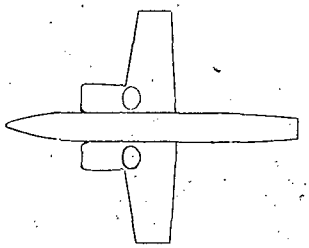
ALPHA =14.02 DEG
EFF. VEL. RATIO = .198

	CP		
Y/YMAX =	0.0	0.5	0.8
X/L			
.130	.211		
.219	.264		
.286	.255		
.331	.250		
.353	.280	.256	.237
.375	.293		
.397		.289	.281
.420	.281	.334	.359
.442	.355	.342	.424
.464	.481	.297	.334
.487	.208	.232	.016
.509	.110	.033	-.619
.531	.012	-.118	-.892
.554	-.021		
.576	.058		
.598	.109	-.087	-.068
.620	.166		
.643	.157	-.043	-.240
.665	.130	.015	-.235
.687	.116		
.710	.082		
.732	.070		

TABLE 18 (C) - CONCLUDED

PRESSURE PROFILE ON FUSELAGE

FORWARD NOZZLES DEFLECTED 45 DEGREES

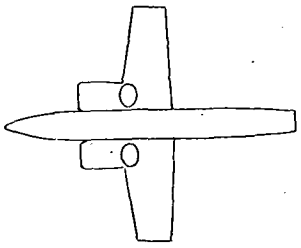


	ALPHA =16.01 DEG EFF. VEL. RATIO = .197			ALPHA =18.00 DEG EFF. VEL. RATIO = .197			ALPHA =19.39 DEG EFF. VEL. RATIO = .197		
		CP			CP			CP	
Y/YMAX =	0.0	0.5	0.8	0.0	0.5	0.8	0.0	0.5	0.8
X/L									
.130	.256			.277			.316		
.219	.301			.351			.395		
.286	.326			.391			.413		
.331	.303			.347			.359		
.353	.323	.295	.296	.378	.344	.328	.387	.365	.362
.375	.331			.349			.386		
.397		.331	.331		.361	.341		.374	.358
.420	.339	.359	.371	.328	.402	.386	.370	.415	.411
.442	.396	.384	.437	.414	.427	.464	.403	.440	.489
.464	.400	.347	.396	.423	.374	.394	.448	.399	.440
.487	.236	.298	.048	.271	.259	.079	.288	.280	.104
.509	-.005	.073	-.492	.132	.120	-.515	.211	.129	-.484
.531	.040	-.083	-1.016	.066	-.081	-.920	.076	-.047	-.937
.554	-.016			.003			.022		
.576	.028			.062			.043		
.598	.092	-.097	-.088	.112	-.034	-.107	.110	-.036	-.141
.620	.158			.175			.152		
.643	.123	-.035	-.164	.160	-.060	-.155	.173	-.066	-.176
.665	.135	.008	-.231	.134	-.029	-.257	.134	-.056	-.244
.687	.110			.105			.091		
.710	.085			.064			.089		
.732	.075			.047			.046		

TABLE 18 (D)

PRESSURE PROFILE ON FUSELAGE

FORWARD NOZZLES DEFLECTED 45 DEGREES



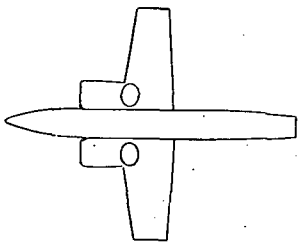
	ALPHA = .02 DEG EFF. VEL. RATIO = .098			ALPHA = 4.03 DEG EFF. VEL. RATIO = .098			ALPHA = 8.03 DEG EFF. VEL. RATIO = .098		
		CP			CP			CP	
Y/YMAX =	0.0	0.5	0.8	0.0	0.5	0.8	0.0	0.5	0.8
X/L									
.130	.088			.179			.213		
.219	.146			.209			.287		
.286	.186			.277			.345		
.331	.193			.287			.342		
.353	.207	.197	.183	.317	.283	.260	.386	.372	.342
.375	.449			.306			.363		
.397		.449	.466		.306	.306		.363	.330
.420	.466	.549	.482	.306	.355	.339	.396	.413	.479
.442	.549	.615	.648	.405	.438	.471	.429	.462	.495
.464	.664	.549	.565	.536	.372	.405	.479	.429	.479
.487	.416	.300	.218	.257	.257	.026	.297	.413	.032
.509	.383	.119	-.841	.142	-.138	-1.208	.131	-.068	-1.161
.531	-.096	-.675	-2.826	-.303	-.846	-2.245	-.250	-.830	-2.140
.554	-.269			-.304			-.243		
.576	.027			.152			.121		
.598	.306	.095	-.235	.382	.280	-.233	.318	.240	-.240
.620	.339			.274			.325		
.643	.387	.238	.125	.267	.152	.064	.260	.148	.012
.665	.316	.217	.193	.317	.219	.138	.264	.138	.169
.687	.326			.314			.267		
.710	.326			.297			.264		
.732	.312			.297			.284		

	ALPHA = 9.87 DEG EFF. VEL. RATIO = .099			ALPHA =11.97 DEG EFF. VEL. RATIO = .098			ALPHA =13.99 DEG EFF. VEL. RATIO = .098		
		CP			CP			CP	
Y/YMAX =	0.0	0.5	0.8	0.0	0.5	0.8	0.0	0.5	0.8
X/L									
.130	.264			.296			.292		
.219	.304			.327			.350		
.286	.385			.405			.421		
.331	.378			.391			.401		
.353	.472	.371	.358	.391	.388	.361	.431	.394	.373
.375	.388			.454			.426		
.397		.421	.355		.421	.338		.459	.377
.420	.372	.404	.388	.404	.454	.421	.393	.459	.393
.442	.437	.519	.519	.437	.520	.570	.459	.525	.558
.464	.584	.453	.519	.471	.471	.520	.476	.459	.509
.487	.306	.388	.094	.355	.355	.156	.294	.393	.112
.509	-.069	-.069	-1.082	-.026	-.009	-1.018	.046	-.020	-1.258
.531	-.200	-.723	-2.587	-.241	-.770	-2.160	-.267	-.664	-1.837
.554	-.317			-.254			-.298		
.576	.089			.100			.062		
.598	.308	.133	-.229	.310	.201	-.376	.292	.163	-.379
.620	.318			.358			.285		
.643	.281	.163	-.015	.310	.093	-.023	.252	.038	-.054
.665	.244	.053	.046	.259	.079	.001	.163	.014	.004
.687	.254			.201			.180		
.710	.240			.222			.207		
.732	.240			.235			.201		

TABLE 18 (D) – CONCLUDED

PRESSURE PROFILE ON FUSELAGE

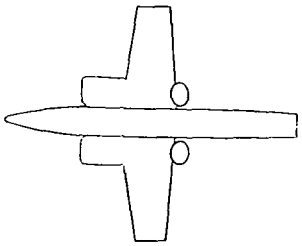
FORWARD NOZZLES DEFLECTED 45 DEGREES



	ALPHA =16.00 DEG EFF. VEL. RATIO = .098			ALPHA =18.04 DEG EFF. VEL. RATIO = .099			ALPHA =20.02 DEG EFF. VEL. RATIO = .099		
	CP			CP			CP		
Y/YMAX =	0.0	0.5	0.8	0.0	0.5	0.8	0.0	0.5	0.8
X/L									
.130	.329			.399			.446		
.219	.420			.483			.535		
.286	.464			.530			.601		
.331	.470			.524			.561		
.353	.484	.467	.426	.571	.527	.483	.584	.551	.502
.375	.448			.478			.581		
.397		.448	.399		.494	.445		.597	.516
.420	.366	.497	.464	.396	.543	.560	.516	.629	.693
.442	.497	.563	.596	.511	.609	.609	.581	.645	.693
.464	.645	.546	.596	.576	.576	.625	.709	.629	.693
.487	.333	.514	.202	.413	.314	.216	.484	.452	.307
.509	.071	-.044	-.913	.183	.036	-.848	.163	.083	-.817
.531	-.224	-.667	-1.848	-.242	-.651	-2.387	-.174	-.480	-1.749
.554	-.280			-.205			-.184		
.576	.103			.050			.139		
.598	.255	.252	-.220	.329	.137	-.380	.340	.264	-.151
.620	.292			.295			.311		
.643	.252	.019	-.176	.258	.057	-.128	.291	.053	-.161
.665	.215	.006	.023	.205	-.024	-.041	.182	-.006	-.072
.687	.157			.188			.152		
.710	.171			.191			.166		
.732	.184			.208			.182		

TABLE 19 (A)
PRESSURE PROFILE ON FUSELAGE

REAR NOZZLES DEFLECTED 45 DEGREES



ALPHA = .01 DEG
EFF. VEL. RATIO =∞

	CP		
Y/YMAX =	0.0	0.5	0.8
X/L			
.130	-.037		
.219	-.054		
.286	-.160		
.331	-.205		
.353	-.170	-.185	-.301
.375	-.191		
.397		-.191	-.223
.420	-.238	-.142	-.150
.442	-.173	-.092	-.152
.464	-.160	-.122	-.135
.487	-.093	-.101	-.083
.509	-.078	-.066	-.040
.531	-.062	-.045	-.015
.554	-.052		
.576	-.037		
.598	-.027	-.003	.011
.620	-.010		
.643	-.006	-.008	-.016
.665	.005	-.008	-.035
.687	-.020		
.710	-.009		
.732	.001		

ALPHA = 3.98 DEG
EFF. VEL. RATIO =∞

	CP		
Y/YMAX =	0.0	0.5	0.8
X/L			
.130	.018		
.219	.013		
.286	-.082		
.331	-.134		
.353	-.119	-.127	-.219
.375	-.134		
.397		-.147	-.180
.420	-.183	-.105	-.100
.442	-.125	-.094	-.122
.464	-.125	-.101	-.117
.487	-.068	-.075	-.055
.509	-.054	-.046	-.025
.531	-.045	-.033	-.004
.554	-.047		
.576	-.031		
.598	-.025	-.005	.023
.620	-.008		
.643	-.007	-.005	-.002
.665	.006	-.005	-.018
.687	.009		
.710	.012		
.732	.022		

ALPHA = 8.02 DEG
EFF. VEL. RATIO =∞

	CP		
Y/YMAX =	0.0	0.5	0.8
X/L			
.130	.081		
.219	.088		
.286	.012		
.331	-.073		
.353	-.044	-.051	-.130
.375	-.067		
.397		-.086	-.118
.420	-.117	-.050	-.057
.442	-.088	-.040	-.091
.464	-.097	-.062	-.089
.487	-.036	-.033	-.041
.509	-.025	-.020	.003
.531	-.024	-.007	.027
.554	-.026		
.576	-.020		
.598	-.021	.003	.034
.620	-.007		
.643	-.005	.000	.006
.665	.007	.002	-.010
.687	.010		
.710	.015		
.732	.023		

ALPHA =10.06 DEG
EFF. VEL. RATIO =∞

	CP		
Y/YMAX =	0.0	0.5	0.8
X/L			
.130	.113		
.219	.133		
.286	.040		
.331	-.033		
.353	-.013	-.008	-.083
.375	-.046		
.397		-.067	-.102
.420	-.086	-.046	-.034
.442	-.069	-.055	-.082
.464	-.069	-.046	-.072
.487	-.027	-.034	-.043
.509	-.023	-.012	.009
.531	-.022	-.005	.027
.554	-.012		
.576	-.011		
.598	-.016	.014	.042
.620	-.006		
.643	-.009	.000	.006
.665	-.002	-.000	-.006
.687	.007		
.710	.010		
.732	.016		

ALPHA =12.02 DEG
EFF. VEL. RATIO =∞

	CP		
Y/YMAX =	0.0	0.5	0.8
X/L			
.130	.144		
.219	.170		
.286	.089		
.331	.016		
.353	.023	.027	-.034
.375	-.015		
.397		-.035	-.068
.420	-.060	-.015	-.005
.442	-.052	-.005	-.062
.464	-.047	-.027	-.055
.487	-.006	-.008	-.025
.509	-.008	.000	.020
.531	-.005	.007	.029
.554	-.013		
.576	-.013		
.598	-.018	-.001	.031
.620	-.016		
.643	-.023	-.010	-.004
.665	-.013	-.011	-.019
.687	-.006		
.710	-.002		
.732	.004		

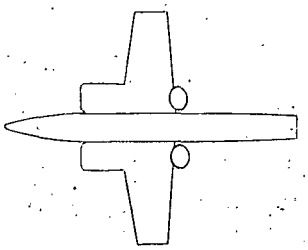
ALPHA =14.01 DEG
EFF. VEL. RATIO =∞

	CP		
Y/YMAX =	0.0	0.5	0.8
X/L			
.130	.183		
.219	.190		
.286	.124		
.331	.042		
.353	.054	.056	.000
.375	.011		
.397		-.019	-.053
.420	-.056	-.007	.002
.442	-.039	.006	-.052
.464	-.058	-.016	-.063
.487	-.002	.011	-.032
.509	-.008	.005	-.002
.531	-.005	.005	.027
.554	-.017		
.576	-.023		
.598	-.037	-.016	.012
.620	-.033		
.643	-.037	-.031	-.021
.665	-.039	-.029	-.043
.687	-.035		
.710	-.029		
.732	-.027		

TABLE 19 (A) - CONCLUDED

PRESSURE PROFILE ON FUSELAGE

REAR NOZZLES DEFLECTED 45 DEGREES

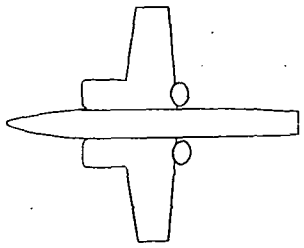


	ALPHA =15.99 DEG EFF. VEL. RATIO =∞			ALPHA =18.00 DEG EFF. VEL. RATIO =∞			ALPHA =20.03 DEG EFF. VEL. RATIO =∞		
	CP			CP			CP		
Y/YMAX =	0.0	0.5	0.8	0.0	0.5	0.8	0.0	0.5	0.8
X/L									
.130	.220			.260			.300		
.219	.225			.259			.322		
.286	.164			.197			.248		
.331	.068			.125			.162		
.353	.083	.077	.036	.105	.115	.049	.154	.149	.084
.375	.037			.074			.103		
.397		-.002	-.041		.034	-.025		.054	-.000
.420	-.042	.003	.006	-.031	.038	.025	.002	.055	.044
.442	-.039	.005	-.040	-.013	.027	-.035	-.007	.055	-.029
.464	-.048	-.015	-.079	-.030	.006	-.055	-.043	.035	-.059
.487	.005	.012	-.049	.027	.033	-.041	.050	.035	-.017
.509	-.037	.005	-.010	-.001	.018	-.012	.014	.034	.008
.531	-.003	-.005	.005	.003	.004	.012	.019	.010	.016
.554	-.027			-.020			-.008		
.576	-.032			-.036			-.028		
.598	-.045	-.031	-.002	-.051	-.039	-.013	-.049	-.039	-.010
.620	-.045			-.057			-.051		
.643	-.059	-.059	-.046	-.067	-.058	-.063	-.067	-.064	-.078
.665	-.052	-.048	-.057	-.065	-.065	-.075	-.062	-.068	-.094
.687	-.053			-.064			-.073		
.710	-.057			-.064			-.075		
.732	-.046			-.061			-.071		

TABLE 19 (B)

PRESSURE PROFILE ON FUSELAGE

REAR NOZZLES DEFLECTED 45 DEGREES



ALPHA = .00 DEG
EFF. VEL. RATIO = .291

	CP		
Y/YMAX =	0.0	0.5	0.8
X/L			
.130	-.045		
.219	-.042		
.286	-.104		
.331	-.154		
.353	-.112	-.148	-.235
.375	-.119		
.397		-.125	-.174
.420	-.167	-.075	-.079
.442	-.107	-.039	-.064
.464	-.070	-.033	-.053
.487	.016	.031	.018
.509	.040	.062	.071
.531	.087	.097	.113
.554	.129		
.576	.171		
.598	.202	.230	.260
.620	.227		
.643	.225	.235	.309
.665	.175	.164	-.158
.687	.093		
.710	-.011		
.732	-.085		

ALPHA = 4.01 DEG
EFF. VEL. RATIO = .292

	CP		
	0.0	0.5	0.8
	.004		
	-.003		
	-.027		
	-.062		
	-.035	-.082	-.139
	-.069		
		-.081	-.113
	-.120	-.047	-.039
	-.078	-.021	-.063
	-.072	-.005	-.036
	.039	.061	.017
	.065	.079	.079
	.110	.119	.121
	.143		
	.184		
	.222	.250	.275
	.250		
	.235	.265	.352
	.179	.168	-.105
	.101		
	-.011		
	-.077		

ALPHA = 7.99 DEG
EFF. VEL. RATIO = .292

	CP		
	0.0	0.5	0.8
	.072		
	.107		
	.044		
	-.006		
	.030	-.009	-.054
	.002		
		-.029	-.071
	-.074	.009	.006
	-.029	.026	-.027
	-.016	.046	-.011
	.090	.082	.042
	.093	.119	.108
	.143	.155	.148
	.178		
	.211		
	.252	.286	.316
	.273		
	.250	.272	.359
	.209	.203	-.097
	.118		
	.017		
	-.095		

ALPHA =10.05 DEG
EFF. VEL. RATIO = .292

	CP		
Y/YMAX =	0.0	0.5	0.8
X/L			
.130	.095		
.219	.135		
.286	.102		
.331	.035		
.353	.077	.024	-.015
.375	.019		
.397		.010	-.041
.420	-.034	.030	.027
.442	-.005	.049	-.006
.464	.005	.072	.010
.487	.105	.114	.054
.509	.113	.133	.113
.531	.164	.164	.171
.554	.187		
.576	.239		
.598	.258	.297	.323
.620	.289		
.643	.257	.293	.376
.665	.211	.195	-.082
.687	.125		
.710	.015		
.732	-.088		

ALPHA =11.99 DEG
EFF. VEL. RATIO = .292

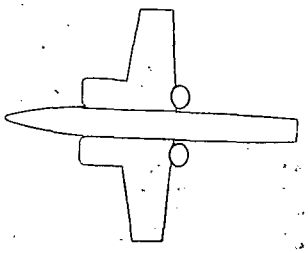
	CP		
	0.0	0.5	0.8
	.138		
	.170		
	.137		
	.078		
	.102	.064	.040
	.064		
		.035	-.022
	-.024	.066	.075
	.006	.084	.009
	.020	.079	.022
	.119	.134	.066
	.128	.148	.112
	.176	.177	.165
	.220		
	.248		
	.276	.306	.332
	.290		
	.279	.295	.375
	.214	.200	-.094
	.119		
	.001		
	-.110		

ALPHA =13.99 DEG
EFF. VEL. RATIO = .292

	CP		
	0.0	0.5	0.8
	.170		
	.210		
	.180		
	.113		
	.134	.090	.071
	.085		
		.053	.001
	-.006	.082	.078
	.003	.087	.009
	.020	.096	.016
	.140	.144	.071
	.144	.149	.118
	.184	.184	.169
	.212		
	.249		
	.274	.308	.344
	.290		
	.251	.280	.359
	.179	.175	-.081
	.075		
	-.047		
	-.158		

TABLE 19 (B) - CONCLUDED
PRESSURE PROFILE ON FUSELAGE

REAR NOZZLES DEFLECTED 45 DEGREES

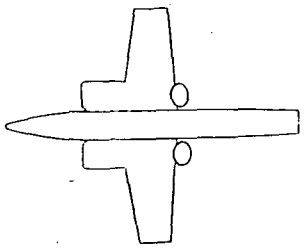


	ALPHA =15.99 DEG EFF. VEL. RATIO = .291			ALPHA =18.03 DEG EFF. VEL. RATIO = .291			ALPHA =19.99 DEG EFF. VEL. RATIO = .291		
	CP			CP			CP		
Y/YMAX =	0.0	0.5	0.8	0.0	0.5	0.8	0.0	0.5	0.8
X/L									
.130	.200			.255			.273		
.219	.244			.282			.317		
.286	.221			.242			.308		
.331	.141			.185			.216		
.353	.152	.127	.097	.190	.164	.125	.232	.190	.161
.375	.116			.154			.174		
.397		.074	.026		.112	.032		.141	.068
.420	.023	.099	.110	.035	.118	.120	.068	.154	.152
.442	.015	.112	.019	.043	.131	.006	.056	.158	.045
.464	-.001	.112	.019	.006	.136	.021	.067	.156	.030
.487	.149	.154	.052	.173	.178	.054	.194	.196	.065
.509	.147	.167	.118	.178	.178	.120	.182	.204	.121
.531	.202	.189	.165	.215	.204	.165	.233	.227	.172
.554	.223			.229			.252		
.576	.248			.261			.282		
.598	.284	.316	.347	.285	.322	.359	.307	.336	.364
.620	.286			.289			.309		
.643	.245	.281	.353	.246	.276	.338	.258	.288	.355
.665	.170	.172	-.104	.170	.161	-.061	.170	.170	-.083
.687	.053			.045			.053		
.710	-.076			-.087			-.098		
.732	-.185			-.207			-.211		

TABLE 19 (C)

PRESSURE PROFILE ON FUSELAGE

REAR NOZZLES DEFLECTED 45 DEGREES



ALPHA = .01 DEG
EFF. VEL. RATIO = .194

	CP		
Y/YMAX =	0.0	0.5	0.8
X/L			
.130	-.083		
.219	-.071		
.286	-.113		
.331	-.162		
.353	-.132	-.153	-.243
.375	-.147		
.397		-.156	-.188
.420	-.188	-.107	-.102
.442	-.143	-.066	-.135
.464	-.135	-.090	-.123
.487	-.062	-.045	-.070
.509	-.053	-.025	-.021
.531	-.017	-.012	-.008
.554	.003		
.576	.006		
.598	-.019	-.005	-.077
.620	-.072		
.643	-.186	-.209	-.566
.665	-.297	-.421	-1.059
.687	-.471		
.710	-.517		
.732	-.459		

ALPHA = 4.02 DEG
EFF. VEL. RATIO = .194

	CP		
Y/YMAX =	0.0	0.5	0.8
X/L			
.130	-.035		
.219	-.019		
.286	-.044		
.331	-.086		
.353	-.061	-.078	-.152
.375	-.091		
.397		-.124	-.144
.420	-.152	-.066	-.058
.442	-.116	-.058	-.116
.464	-.103	-.066	-.103
.487	-.026	-.026	-.050
.509	-.050	.003	-.009
.531	.007	.015	.024
.554	.014		
.576	.019		
.598	.004	.016	-.045
.620	-.035		
.643	-.154	-.187	-.521
.665	-.291	-.330	-.960
.687	-.437		
.710	-.517		
.732	-.462		

ALPHA = 8.00 DEG
EFF. VEL. RATIO = .194

	CP		
Y/YMAX =	0.0	0.5	0.8
X/L			
.130	.023		
.219	.059		
.286	.023		
.331	-.005		
.353	.003	-.034	-.057
.375	-.025		
.397		-.058	-.095
.420	-.103	-.034	-.025
.442	-.066	-.009	-.075
.464	-.091	-.009	-.066
.487	.024	.016	-.042
.509	-.042	.049	.003
.531	.065	.061	.044
.554	.060		
.576	.057		
.598	.051	.081	-.007
.620	.021		
.643	-.090	-.059	-.269
.665	-.214	-.279	-.687
.687	-.397		
.710	-.493		
.732	-.484		

ALPHA =10.01 DEG
EFF. VEL. RATIO = .194

	CP		
Y/YMAX =	0.0	0.5	0.8
X/L			
.130	.043		
.219	.085		
.286	.062		
.331	.010		
.353	.038	.006	-.023
.375	-.005		
.397		-.041	-.086
.420	-.082	-.017	-.017
.442	-.066	.004	-.082
.464	-.082	.008	-.058
.487	.032	.045	-.033
.509	.045	.053	.024
.531	.069	.069	.057
.554	.078		
.576	.088		
.598	.090	.101	.029
.620	.056		
.643	-.041	-.014	-.177
.665	-.158	-.160	-.651
.687	-.363		
.710	-.495		
.732	-.501		

ALPHA =11.97 DEG
EFF. VEL. RATIO = .194

	CP		
Y/YMAX =	0.0	0.5	0.8
X/L			
.130	.070		
.219	.118		
.286	.113		
.331	.051		
.353	.066	.059	.007
.375	.014		
.397		-.015	-.076
.420	-.068	-.003	.006
.442	-.068	.067	-.072
.464	-.060	.018	-.064
.487	.067	.067	-.048
.509	.047	.067	.022
.531	.092	.075	.055
.554	.093		
.576	.109		
.598	.128	.133	.069
.620	.082		
.643	.001	.065	-.021
.665	-.148	-.120	-.663
.687	-.338		
.710	-.456		
.732	-.515		

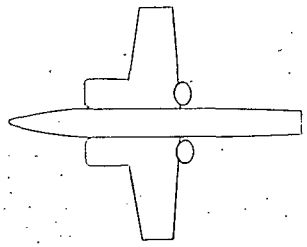
ALPHA =14.00 DEG
EFF. VEL. RATIO = .194

	CP		
Y/YMAX =	0.0	0.5	0.8
X/L			
.130	.100		
.219	.158		
.286	.140		
.331	.075		
.353	.081	.101	.028
.375	.040		
.397		.011	-.050
.420	-.046	.019	.031
.442	-.050	.036	-.074
.464	-.058	.036	-.058
.487	.076	.089	-.022
.509	.076	.072	.007
.531	.113	.105	.056
.554	.108		
.576	.137		
.598	.156	.166	.083
.620	.160		
.643	.061	.129	.097
.665	-.038	-.049	-.648
.687	-.224		
.710	-.420		
.732	-.512		

TABLE 19 (C) – CONCLUDED

PRESSURE PROFILE ON FUSELAGE

REAR NOZZLES DEFLECTED 45 DEGREES

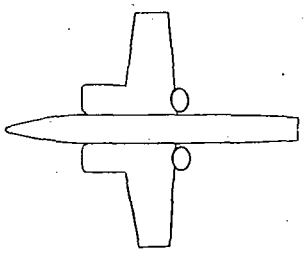


	ALPHA =16.01 DEG EFF. VEL. RATIO = .194			ALPHA =17.97 DEG EFF. VEL. RATIO = .194			ALPHA =20.01 DEG EFF. VEL. RATIO = .194		
	CP			CP			CP		
Y/YMAX =	0.0	0.5	0.8	0.0	0.5	0.8	0.0	0.5	0.8
X/L									
.130	.146			.182			.220		
.219	.188			.227			.255		
.286	.190			.219			.244		
.331	.126			.150			.188		
.353	.112	.106	.075	.174	.170	.103	.184	.192	.130
.375	.088			.124			.142		
.397		.047	-.023		.129	-.022		.089	.012
.420	-.019	.051	.063	-.022	.080	.080	.020	.110	.097
.442	-.031	.071	-.051	-.010	.092	-.030	.008	.101	-.037
.464	-.068	.051	-.064	-.043	.076	-.022	-.037	.097	-.037
.487	.092	.117	-.039	.120	.141	-.026	.126	.142	-.045
.509	.071	.092	.010	.104	.120	.018	.114	.118	.020
.531	.133	.112	.051	.145	.133	.059	.171	.138	.057
.554	.154			.161			.181		
.576	.175			.195			.218		
.598	.196	.215	.097	.221	.233	.123	.240	.241	.149
.620	.218			.220			.229		
.643	.143	.175	.215	.163	.227	.280	.178	.248	.271
.665	.028	.049	-.738	.048	.076	-.794	.062	.074	-.737
.687	-.188			-.138			-.126		
.710	-.368			-.362			-.369		
.732	-.502			-.495			-.533		

TABLE 19 (D)

PRESSURE PROFILE ON FUSELAGE

REAR NOZZLES DEFLECTED 45 DEGREES



ALPHA = 0.00 DEG
EFF. VEL. RATIO = .097

	CP		
Y/YMAX =	0.0	0.5	0.8
X/L			
.130	-.400		
.219	-.387		
.286	-.333		
.331	-.384		
.353	-.367	-.333	-.434
.375	-.412		
.397		-.428	-.477
.420	-.494	-.412	-.412
.442	-.461	-.412	-.477
.464	-.461	-.461	-.461
.487	-.444	-.412	-.461
.509	-.428	-.477	-.461
.531	-.494	-.477	-.494
.554	-.528		
.576	-.606		
.598	-.707	-.751	-.896
.620	-.879		
.643	-1.014	-1.219	-1.638
.665	-1.044	-1.105	-.754
.687	-.855		
.710	-.596		
.732	-.410		

ALPHA = 4.06 DEG
EFF. VEL. RATIO = .097

	CP		
Y/YMAX =	0.0	0.5	0.8
X/L			
.130	-.340		
.219	-.313		
.286	-.269		
.331	-.323		
.353	-.306	-.293	-.343
.375	-.329		
.397		-.378	-.411
.420	-.411	-.378	-.183
.442	-.411	-.346	-.443
.464	-.443	-.459	-.476
.487	-.394	-.378	-.411
.509	-.394	-.411	-.427
.531	-.443	-.459	-.394
.554	-.506		
.576	-.566		
.598	-.683	-.700	-.856
.620	-.850		
.643	-1.000	-1.200	-1.647
.665	-1.033	-1.103	-.863
.687	-.920		
.710	-.636		
.732	-.446		

ALPHA = 7.99 DEG
EFF. VEL. RATIO = .097

	CP		
Y/YMAX =	0.0	0.5	0.8
X/L			
.130	-.283		
.219	-.246		
.286	-.199		
.331	-.242		
.353	-.242	-.226	-.273
.375	-.307		
.397		-.356	-.388
.420	-.356	-.323	-.356
.442	-.372	-.323	-.404
.464	-.421	-.404	-.470
.487	-.356	-.356	-.421
.509	-.421	-.388	-.404
.531	-.404	-.437	-.437
.554	-.486		
.576	-.546		
.598	-.663	-.707	-.860
.620	-.830		
.643	-1.021	-1.232	-1.713
.665	-1.064	-1.188	-.998
.687	-.944		
.710	-.687		
.732	-.493		

ALPHA =10.04 DEG
EFF. VEL. RATIO = .097

	CP		
Y/YMAX =	0.0	0.5	0.8
X/L			
.130	-.281		
.219	-.240		
.286	-.197		
.331	-.247		
.353	-.254	-.247	-.277
.375	-.306		
.397		-.388	-.404
.420	-.388	-.372	-.339
.442	-.454	-.372	-.437
.464	-.470	-.421	-.470
.487	-.404	-.404	-.470
.509	-.437	-.404	-.503
.531	-.503	-.470	-.503
.554	-.489		
.576	-.563		
.598	-.674	-.715	-.893
.620	-.846		
.643	-1.034	-1.243	-1.805
.665	-1.098	-1.257	-1.078
.687	-.997		
.710	-.745		
.732	-.556		

ALPHA =12.02 DEG
EFF. VEL. RATIO = .097

	CP		
Y/YMAX =	0.0	0.5	0.8
X/L			
.130	-.264		
.219	-.217		
.286	-.167		
.331	-.217		
.353	-.231	-.231	-.241
.375	-.307		
.397		-.339	-.405
.420	-.388	-.356	-.339
.442	-.388	-.339	-.454
.464	-.388	-.405	-.454
.487	-.372	-.323	-.470
.509	-.405	-.437	-.454
.531	-.437	-.454	-.486
.554	-.482		
.576	-.556		
.598	-.683	-.707	-.904
.620	-.841		
.643	-1.045	-1.277	-1.837
.665	-1.143	-1.334	-1.153
.687	-1.019		
.710	-.800		
.732	-.579		

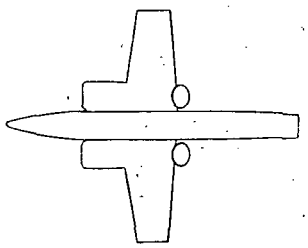
ALPHA =14.04 DEG
EFF. VEL. RATIO = .097

	CP		
Y/YMAX =	0.0	0.5	0.8
X/L			
.130	-.262		
.219	-.198		
.286	-.151		
.331	-.215		
.353	-.225	-.228	-.238
.375	-.315		
.397		-.315	-.396
.420	-.380	-.315	-.331
.442	-.429	-.331	-.478
.464	-.445	-.396	-.478
.487	-.413	-.364	-.478
.509	-.429	-.429	-.494
.531	-.462	-.478	-.511
.554	-.503		
.576	-.570		
.598	-.687	-.738	-.939
.620	-.872		
.643	-1.080	-1.342	-2.009
.665	-1.204	-1.435	-1.305
.687	-1.137		
.710	-.868		
.732	-.637		

TABLE 19 (D) - CONCLUDED

PRESSURE PROFILE ON FUSELAGE

REAR NOZZLES DEFLECTED 45 DEGREES

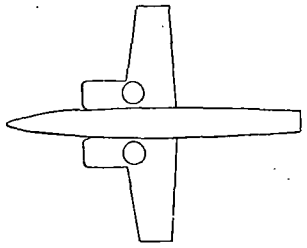


	ALPHA =16.00 DEG EFF. VEL. RATIO = .096			ALPHA =18.01 DEG EFF. VEL. RATIO = .096			ALPHA =20.00 DEG EFF. VEL. RATIO = .097		
	CP			CP			CP		
Y/YMAX =	0.0	0.5	0.8	0.0	0.5	0.8	0.0	0.5	0.8
X/L									
.130	-.247			-.215			-.202		
.219	-.193			-.157			-.152		
.286	-.159			-.137			-.098		
.331	-.210			-.184			-.165		
.353	-.227	-.240	-.254	-.201	-.215	-.215	-.182	-.182	-.219
.375	-.292			-.260			-.275		
.397		-.325	-.375		-.326	-.441		-.357	-.407
.420	-.375	-.375	-.309	-.392	-.293	-.343	-.407	-.357	-.341
.442	-.457	-.309	-.457	-.409	-.343	-.458	-.439	-.341	-.489
.464	-.457	-.424	-.474	-.441	-.376	-.507	-.472	-.390	-.521
.487	-.408	-.424	-.523	-.425	-.425	-.573	-.390	-.374	-.571
.509	-.457	-.441	-.507	-.441	-.458	-.557	-.505	-.472	-.571
.531	-.457	-.507	-.523	-.458	-.474	-.540	-.456	-.489	-.571
.554	-.501			-.508			-.522		
.576	-.583			-.593			-.600		
.598	-.719	-.766	-.977	-.714	-.765	-.988	-.721	-.785	-1.032
.620	-.916			-.907			-.944		
.643	-1.140	-1.418	-2.186	-1.157	-1.413	-2.224	-1.177	-1.443	-2.333
.665	-1.262	-1.530	-1.517	-1.285	-1.559	-1.559	-1.332	-1.618	-1.625
.687	-1.187			-1.221			-1.227		
.710	-.984			-.998			-1.035		
.732	-.702			-.741			-.779		

TABLE 20 (A)

PRESSURE PROFILE ON FUSELAGE

FORWARD NOZZLES DEFLECTED 90 DEGREES



ALPHA = .02 DEG
EFF. VEL. RATIO =∞

	CP		
Y/YMAX =	0.0	0.5	0.8
X/L			
.130	.013		
.219	.015		
.286	-.055		
.331	-.070		
.353	-.003	-.057	-.118
.375	-.017		
.397		.025	.025
.420	.025	.033	.046
.442	.072	.011	-.017
.464	.022	-.071	-.166
.487	-.143	-.127	-.293
.509	-.219	-.232	-.363
.531	-.274	-.310	-.432
.554	-.286		
.576	-.267		
.598	-.223	-.259	-.330
.620	-.153		
.643	-.109	-.206	-.239
.665	-.074	-.179	-.147
.687	-.061		
.710	-.059		
.732	-.067		

ALPHA = 4.04 DEG
EFF. VEL. RATIO =∞

	CP		
	0.0	0.5	0.8
	.052		
	.070		
	.012		
	-.016		
	.050	.000	-.044
	.033		
		.045	.084
	.076	.066	.086
	.113	.043	.050
	.036	-.057	-.124
	-.119	-.117	-.233
	-.212	-.204	-.333
	-.241	-.241	-.375
	-.263		
	-.245		
	-.193	-.217	-.274
	-.132		
	-.088	-.172	-.193
	-.048	-.132	-.141
	-.043		
	-.039		
	-.028		

ALPHA = 8.03 DEG
EFF. VEL. RATIO =∞

	CP		
	0.0	0.5	0.8
	.110		
	.135		
	.075		
	.058		
	.094	.056	.049
	.078		
		.101	.116
	.117	.117	.120
	.150	.076	.091
	.075	-.016	-.052
	-.091	-.080	-.215
	-.156	-.178	-.290
	-.202	-.264	-.351
	-.246		
	-.226		
	-.182	-.173	-.203
	-.106		
	-.057	-.103	-.132
	-.014	-.106	-.111
	-.013		
	-.006		
	-.017		

ALPHA =10.04 DEG
EFF. VEL. RATIO =∞

	CP		
Y/YMAX =	0.0	0.5	0.8
X/L			
.130	.154		
.219	.189		
.286	.132		
.331	.089		
.353	.115	.098	.070
.375	.111		
.397		.118	.152
.420	.131	.127	.125
.442	.178	.097	.107
.464	.107	-.003	-.035
.487	-.080	-.063	-.177
.509	-.158	-.156	-.254
.531	-.186	-.231	-.341
.554	-.233		
.576	-.216		
.598	-.180	-.168	-.193
.620	-.108		
.643	-.046	-.087	-.121
.665	-.003	-.063	-.084
.687	.015		
.710	.004		
.732	-.002		

ALPHA =12.02 DEG
EFF. VEL. RATIO =∞

	CP		
	0.0	0.5	0.8
	.172		
	.204		
	.165		
	.114		
	.143	.122	.102
	.126		
		.127	.149
	.148	.136	.122
	.187	.085	.125
	.117	-.002	-.046
	-.082	-.064	-.167
	-.170	-.180	-.257
	-.220	-.221	-.349
	-.243		
	-.228		
	-.195	-.196	-.202
	-.124		
	-.072	-.089	-.135
	-.020	-.063	-.116
	-.002		
	.001		
	-.000		

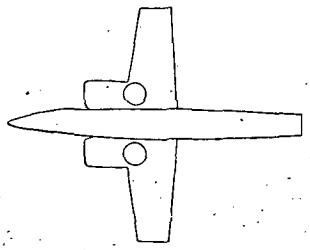
ALPHA =14.03 DEG
EFF. VEL. RATIO =∞

	CP		
	0.0	0.5	0.8
	.212		
	.243		
	.182		
	.137		
	.163	.133	.132
	.151		
		.151	.175
	.168	.157	.142
	.201	.111	.145
	.148	-.008	-.017
	-.064	-.060	-.156
	-.142	-.182	-.262
	-.211	-.232	-.327
	-.263		
	-.253		
	-.220	-.234	-.217
	-.171		
	-.106	-.113	-.161
	-.077	-.087	-.148
	-.028		
	-.005		
	-.012		

TABLE 20 (A) - CONCLUDED

PRESSURE PROFILE ON FUSELAGE

FORWARD NOZZLES DEFLECTED 90 DEGREES

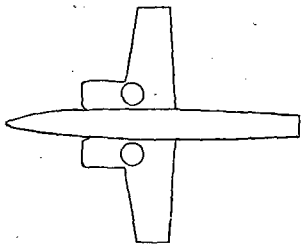


	ALPHA =16.02 DEG EFF. VEL. RATIO =∞			ALPHA =18.02 DEG EFF. VEL. RATIO =∞			ALPHA =20.04 DEG EFF. VEL. RATIO =∞		
	CP			CP			CP		
Y/YMAX =	0.0	0.5	0.8	0.0	0.5	0.8	0.0	0.5	0.8
X/L									
.130	.249			.290			.322		
.219	.274			.318			.343		
.286	.229			.251			.296		
.331	.169			.199			.228		
.353	.191	.160	.161	.210	.181	.184	.245	.219	.227
.375	.150			.172			.213		
.397		.152	.192		.164	.183		.188	.217
.420	.180	.151	.143	.198	.158	.155	.213	.190	.184
.442	.214	.103	.146	.209	.117	.144	.230	.151	.178
.464	.138	-.003	-.001	.136	.004	.001	.168	.027	.027
.487	-.070	-.065	-.164	-.081	-.058	-.147	-.040	-.064	-.100
.509	-.160	-.172	-.275	-.142	-.178	-.287	-.126	-.151	-.219
.531	-.211	-.275	-.328	-.182	-.262	-.356	-.176	-.204	-.355
.554	-.255			-.254			-.230		
.576	-.259			-.248			-.236		
.598	-.235	-.250	-.256	-.232	-.245	-.242	-.220	-.244	-.267
.620	-.179			-.178			-.185		
.643	-.146	-.121	-.155	-.140	-.119	-.130	-.147	-.132	-.122
.665	-.088	-.089	-.143	-.088	-.081	-.118	-.100	-.097	-.108
.687	-.048			-.067			-.067		
.710	-.022			-.043			-.048		
.732	-.023			-.026			-.037		

TABLE 20 (B)

PRESSURE PROFILE ON FUSELAGE

FORWARD NOZZLES DEFLECTED 90 DEGREES



ALPHA = .01 DEG
EFF. VEL. RATIO = .302

	CP		
Y/YMAX =	0.0	0.5	0.8
X/L			
.130	.102		
.219	.219		
.286	.309		
.331	.377		
.353	.409	.402	.397
.375	.309		
.397		.350	.449
.420	.450	.403	.295
.442	.408	-.158	-.164
.464	-.254	-.715	-1.854
.487	-1.106	-1.179	-1.938
.509	-1.240	-1.616	-1.163
.531	-1.260	-1.568	-1.331
.554	-1.151		
.576	-.924		
.598	-.566	-.606	-.912
.620	-.389		
.643	-.260	-.193	-.859
.665	-.183	-.193	-.872
.687	-.104		
.710	-.077		
.732	-.054		

ALPHA = 4.01 DEG
EFF. VEL. RATIO = .302

	CP		
	0.0	0.5	0.8
	.167		
	.299		
	.374		
	.443		
	.477	.463	.467
	.269		
		.374	.313
	.430	.284	.366
	.317	-.169	-.134
	-.366	-.880	-1.824
	-1.291	-1.167	-1.890
	-1.287	-1.603	-1.748
	-1.587	-1.611	-1.475
	-1.222		
	-1.043		
	-.741	-.853	-.933
	-.507		
	-.428	-.380	-.716
	-.253	-.313	-.968
	-.216		
	-.118		
	-.105		

ALPHA = 8.03 DEG
EFF. VEL. RATIO = .302

	CP		
	0.0	0.5	0.8
	.210		
	.351		
	.438		
	.493		
	.535	.532	.522
	.413		
		.463	.647
	.472	.344	.203
	.468	-.160	-.087
	-.074	-.693	-1.751
	-1.077	-.996	-1.692
	-1.219	-1.409	-1.608
	-1.650	-1.221	-1.438
	-1.265		
	-1.124		
	-.871	-.975	-1.082
	-.632		
	-.436	-.490	-.601
	-.386	-.316	-.541
	-.309		
	-.182		
	-.142		

ALPHA =10.02 DEG
EFF. VEL. RATIO = .302

	CP		
Y/YMAX =	0.0	0.5	0.8
X/L			
.130	.244		
.219	.392		
.286	.463		
.331	.513		
.353	.563	.554	.548
.375	.431		
.397		.514	.576
.420	.633	.420	.503
.442	.595	.079	.040
.464	-.032	-.970	-1.877
.487	-1.048	-1.026	-1.671
.509	-1.274	-1.533	-1.757
.531	-1.274	-1.560	-1.053
.554	-1.280		
.576	-1.115		
.598	-.843	-1.050	-1.122
.620	-.592		
.643	-.432	-.561	-.557
.665	-.385	-.353	-.474
.687	-.289		
.710	-.224		
.732	-.118		

ALPHA =12.02 DEG
EFF. VEL. RATIO = .301

	CP		
	0.0	0.5	0.8
	.281		
	.418		
	.514		
	.543		
	.578	.580	.578
	.538		
		.587	.699
	.594	.558	.457
	.541	.072	.028
	-.029	-.872	-1.535
	-.809	-.934	-1.776
	-1.210	-1.422	-1.804
	-1.402	-1.742	-1.681
	-1.281		
	-1.090		
	-.826	-1.033	-1.224
	-.601		
	-.464	-.502	-.574
	-.398	-.382	-.516
	-.265		
	-.203		
	-.122		

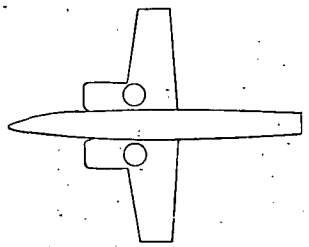
ALPHA =14.01 DEG
EFF. VEL. RATIO = .301

	CP		
	0.0	0.5	0.8
	.330		
	.458		
	.530		
	.568		
	.605	.603	.597
	.595		
		.540	.700
	.629	.292	.573
	.423	.044	-.047
	-.129	-.806	-1.596
	-1.005	-1.047	-1.866
	-1.313	-1.532	-1.503
	-1.452	-1.481	-1.745
	-1.257		
	-1.151		
	-.921	-1.088	-1.204
	-.688		
	-.528	-.618	-.598
	-.447	-.446	-.467
	-.386		
	-.293		
	-.203		

TABLE 20 (B) - CONCLUDED

PRESSURE PROFILE ON FUSELAGE

FORWARD NOZZLES DEFLECTED 90 DEGREES

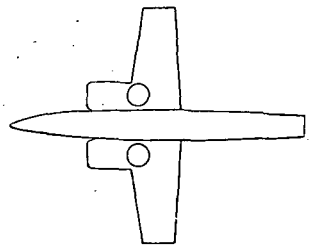


	ALPHA =16.04 DEG EFF. VEL. RATIO = .302			ALPHA =18.03 DEG EFF. VEL. RATIO = .302			ALPHA =20.02 DEG EFF. VEL. RATIO = .302		
	CP			CP			CP		
Y/YMAX =	0.0	0.5	0.8	0.0	0.5	0.8	0.0	0.5	0.8
X/L									
.130	.359			.401			.426		
.219	.501			.526			.553		
.286	.576			.604			.629		
.331	.607			.622			.644		
.353	.631	.625	.620	.653	.644	.638	.658	.661	.643
.375	.760			.568			.679		
.397		.527	.756		.601	.801		.668	.537
.420	.865	.616	.545	.669	.421	.454	.623	.449	.462
.442	.701	.132	.199	.709	-.023	.197	.477	-.115	-.099
.464	-.098	-.620	-1.801	.065	-.707	-1.570	-.225	-1.088	-1.611
.487	-.887	-.685	-1.783	-.916	-.752	-1.625	-.935	-.778	-1.748
.509	-.833	-1.222	-1.239	-.985	-1.284	-1.746	-1.178	-1.271	-1.452
.531	-1.135	-1.639	-1.501	-1.257	-1.220	-1.286	-1.220	-1.428	-1.698
.554	-1.228			-1.232			-1.225		
.576	-1.104			-1.103			-1.116		
.598	-.933	-1.078	-1.318	-.986	-1.122	-1.230	-.980	-1.164	-1.158
.620	-.753			-.734			-.790		
.643	-.652	-.817	-.699	-.676	-.824	-.749	-.730	-.868	-.939
.665	-.530	-.589	-.528	-.555	-.606	-.606	-.630	-.692	-.738
.687	-.419			-.461			-.540		
.710	-.352			-.434			-.466		
.732	-.306			-.324			-.354		

TABLE 20-(C)

PRESSURE PROFILE ON FUSELAGE

FORWARD NOZZLES DEFLECTED 90 DEGREES



ALPHA = .00 DEG
EFF. VEL. RATIO = .203

	CP		
Y/YMAX =	0.0	0.5	0.8
X/L			
.130	.045		
.219	.127		
.286	.172		
.331	.165		
.353	.160	.143	.169
.375	.007		
.397		-.242	-.209
.420	-.445	-.943	-.987
.442	-.890	-1.407	-4.285
.464	-4.529	-3.425	-4.171
.487	-2.504	-3.172	-2.944
.509	-2.006	-2.349	-4.790
.531	-.951	-2.752	-3.229
.554	-.949		
.576	-.479		
.598	-.366	-.347	-.876
.620	-.140		
.643	-.190	-.227	-.929
.665	-.098	-.129	-.740
.687	.013		
.710	.030		
.732	.060		

ALPHA = 4.00 DEG
EFF. VEL. RATIO = .202

	CP		
Y/YMAX =	0.0	0.5	0.8
X/L			
.130	.098		
.219	.187		
.286	.234		
.331	.215		
.353	.209	.211	.241
.375	-.389		
.397		-.306	-.117
.420	-.306	-.829	-1.187
.442	-.998	-1.422	-4.081
.464	-5.023	-3.735	-4.846
.487	-2.228	-2.249	-2.385
.509	-2.035	-2.829	-4.410
.531	-2.138	-2.920	-2.825
.554	-.865		
.576	-.519		
.598	-.295	-.519	-1.305
.620	-.166		
.643	-.146	-.316	-1.030
.665	-.072	-.115	-.848
.687	-.102		
.710	-.013		
.732	.070		

ALPHA = 7.98 DEG
EFF. VEL. RATIO = .201

	CP		
Y/YMAX =	0.0	0.5	0.8
X/L			
.130	.158		
.219	.253		
.286	.313		
.331	.295		
.353	.276	.274	.296
.375	.389		
.397		.233	.040
.420	.159	-.335	-.750
.442	-.211	-1.400	-3.421
.464	-3.766	-2.927	-3.684
.487	-2.199	-2.186	-1.944
.509	-2.009	-2.602	-4.268
.531	-1.302	-.960	-2.462
.554	-.963		
.576	-.466		
.598	-.248	-.299	-1.211
.620	-.212		
.643	-.141	-.347	-1.306
.665	-.085	-.264	-.979
.687	-.052		
.710	.009		
.732	-.007		

ALPHA =10.02 DEG
EFF. VEL. RATIO = .202

	CP		
Y/YMAX =	0.0	0.5	0.8
X/L			
.130	.209		
.219	.301		
.286	.372		
.331	.324		
.353	.319	.316	.348
.375	.146		
.397		-.310	-.014
.420	.093	-.795	-.627
.442	-.684	-1.461	-3.077
.464	-3.509	-3.023	-5.210
.487	-2.588	-2.226	-2.658
.509	-2.954	-2.501	-4.655
.531	-.845	-3.011	-3.488
.554	-1.004		
.576	-.475		
.598	-.310	-.512	-1.295
.620	-.237		
.643	-.214	-.368	-1.234
.665	-.189	-.214	-.899
.687	-.072		
.710	.022		
.732	.021		

ALPHA =12.00 DEG
EFF. VEL. RATIO = .202

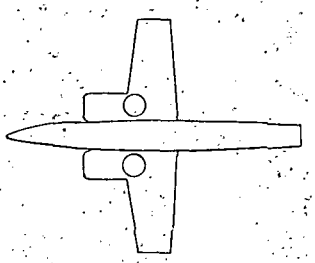
	CP		
Y/YMAX =	0.0	0.5	0.8
X/L			
.130	.266		
.219	.349		
.286	.430		
.331	.386		
.353	.370	.376	.386
.375	.008		
.397		-.325	-.308
.420	-.321	-.755	-.571
.442	-.251	-1.227	-3.496
.464	-1.978	-2.921	-5.087
.487	-2.519	-2.753	-2.441
.509	-1.863	-2.876	-3.980
.531	-1.559	-2.757	-3.249
.554	-.995		
.576	-.539		
.598	-.368	-.386	-1.460
.620	-.350		
.643	-.213	-.276	-1.353
.665	-.196	-.210	-.984
.687	-.092		
.710	-.015		
.732	.041		

ALPHA =13.98 DEG
EFF. VEL. RATIO = .202

	CP		
Y/YMAX =	0.0	0.5	0.8
X/L			
.130	.282		
.219	.394		
.286	.461		
.331	.438		
.353	.419	.421	.423
.375	.306		
.397		.326	.228
.420	.191	-.465	-.215
.442	.015	-1.506	-2.973
.464	-2.604	-2.932	-4.264
.487	-2.281	-2.121	-2.801
.509	-2.481	-2.453	-4.133
.531	-1.654	-2.436	-3.182
.554	-1.156		
.576	-.496		
.598	-.414	-.590	-1.484
.620	-.297		
.643	-.234	-.381	-1.380
.665	-.208	-.222	-1.041
.687	-.103		
.710	-.027		
.732	.011		

TABLE 20 (C) - CONCLUDED
PRESSURE PROFILE ON FUSELAGE

FORWARD NOZZLES DEFLECTED 90 DEGREES

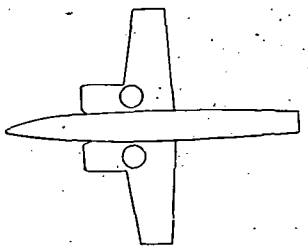


	ALPHA =16.01 DEG EFF. VEL. RATIO = .202			ALPHA =17.99 DEG EFF. VEL. RATIO = .202			ALPHA =20.02 DEG EFF. VEL. RATIO = .202		
	CP			CP			CP		
Y/YMAX =	0.0	0.5	0.8	0.0	0.5	0.8	0.0	0.5	0.8
X/L									
.130	.334			.369			.402		
.219	.432			.468			.512		
.286	.496			.530			.601		
.331	.452			.479			.557		
.353	.455	.443	.447	.478	.489	.458	.527	.515	.494
.375	.080			.995			.720		
.397		.064	-.263		.731	.292		.401	.581
.420	.366	-.304	-.034	.511	.312	.357	.348	.045	.557
.442	-.202	-1.063	-3.125	.410	-.115	-2.103	.467	-.532	-2.082
.464	-1.055	-2.888	-4.444	-1.717	-2.099	-5.050	-1.677	-2.406	-6.125
.487	-2.345	-1.541	-2.631	-1.859	-1.895	-1.761	-1.788	-1.739	-2.107
.509	-2.194	-2.664	-3.721	-1.879	-1.973	-3.355	-2.835	-2.140	-3.527
.531	-2.059	-2.100	-2.986	-1.351	-2.700	-2.883	-1.407	-1.244	-2.823
.554	-1.238			-1.318			-1.564		
.576	-.884			-.820			-1.007		
.598	-.534	-.564	-1.623	-.563	-.612	-1.403	-.626	-.538	-1.231
.620	-.367			-.530			-.449		
.643	-.267	-.336	-1.380	-.351	-.451	-1.593	-.447	-.570	-1.743
.665	-.230	-.244	-1.055	-.274	-.367	-1.366	-.281	-.370	-1.479
.687	-.165			-.194			-.267		
.710	-.109			-.112			-.148		
.732	-.005			-.033			-.088		

TABLE 20 (D)

PRESSURE PROFILE ON FUSELAGE

FORWARD NOZZLES DEFLECTED 90 DEGREES



ALPHA = -.01 DEG
EFF. VEL. RATIO = .100

	CP		
Y/YMAX =	0.0	0.5	0.8
X/L			
.130	.064		
.219	.074		
.286	.044		
.331	-.134		
.353	-.263	-.253	-.178
.375	-.203		
.397		.644	-2.561
.420	-.070	-1.698	-1.681
.442	-1.714	-1.930	-3.060
.464	-4.288	-2.445	-1.548
.487	-1.233	1.972	-7.410
.509	-4.919	-2.794	-3.940
.531	.378	.046	-2.644
.554	-.611		
.576	-.989		
.598	-.601	-1.290	-1.992
.620	-.485		
.643	-.256	-.345	-1.010
.665	.115	-.110	-.536
.687	.265		
.710	.265		
.732	.313		

ALPHA = 4.02 DEG
EFF. VEL. RATIO = .100

	CP		
	0.0	0.5	0.8
	.149		
	.139		
	.119		
	-.072		
	-.211	-.187	-.099
	-.397		
		.033	-1.754
	-.910	-2.350	-1.175
	-2.002	-2.532	-2.780
	-6.636	-3.194	-2.896
	-.811	-2.135	-6.636
	-3.061	-1.820	-4.865
	-.579	-3.210	-3.392
	-.731		
	-.833		
	-.374	-1.095	-2.104
	-.418		
	-.092	-.041	-.755
	.119	.044	-.269
	.169		
	.343		
	.356		

ALPHA = 8.00 DEG
EFF. VEL. RATIO = .100

	CP		
	0.0	0.5	0.8
	.220		
	.220		
	.192		
	-.030		
	-.157	-.150	-.013
	-.353		
		-.186	-.987
	-.169	-3.441	-2.022
	-2.122	-1.588	-3.291
	-2.773	-3.775	-1.922
	-.553	-.937	-7.297
	-3.241	-2.556	-4.710
	-1.254	-.753	-3.191
	-.555		
	-.390		
	-.740	-.744	-1.823
	-.363		
	-.236	-.205	-.449
	.031	.148	.007
	.182		
	.179		
	.384		

ALPHA =10.01 DEG
EFF. VEL. RATIO = .100

	CP		
Y/YMAX =	0.0	0.5	0.8
X/L			
.130	.273		
.219	.284		
.286	.263		
.331	.047		
.353	-.108	-.070	.009
.375	.231		
.397		-.572	-.204
.420	-.371	-1.859	-1.792
.442	-2.110	-2.378	-2.930
.464	-2.294	-2.194	-4.585
.487	-1.140	-1.960	-10.238
.509	-.471	-1.341	-5.856
.531	-2.746	-2.110	-1.943
.554	-.489		
.576	-.537		
.598	-.510	-.468	-1.423
.620	-.386		
.643	.043	-.187	-.207
.665	.194	.088	-.128
.687	.249		
.710	.328		
.732	.359		

ALPHA =11.99 DEG
EFF. VEL. RATIO = .100

	CP		
	0.0	0.5	0.8
	.335		
	.420		
	.304		
	.100		
	-.064	-.040	.086
	.801		
		-1.840	-1.093
	-1.342	-3.650	-2.122
	-4.264	-3.517	-2.952
	-2.255	-3.434	-5.310
	-2.986	-3.251	-12.152
	-4.613	-3.617	-8.067
	-2.654	-.927	-2.952
	-.446		
	-.374		
	-.402	-.432	-.845
	-.170		
	.045	.062	-.385
	.205	.164	.079
	.168		
	.369		
	.492		

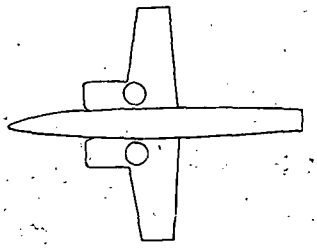
ALPHA =14.02 DEG
EFF. VEL. RATIO = .100

	CP		
	0.0	0.5	0.8
	.392		
	.402		
	.368		
	.147		
	-.007	.021	.130
	.823		
		-.503	-1.995
	-.752	-3.852	-2.343
	-2.807	-2.923	-4.730
	-5.907	-3.321	-4.432
	-3.338	-1.647	-9.040
	-2.923	-1.730	-5.426
	-2.128	-4.100	-2.227
	-.279		
	-.350		
	-.306	-.544	-.738
	-.170		
	-.017	-.109	-.275
	.106	.082	.150
	.296		
	.392		
	.385		

TABLE 20 (D) - *CONCLUDED*

PRESSURE PROFILE ON FUSELAGE

FORWARD NOZZLES DEFLECTED 90 DEGREES

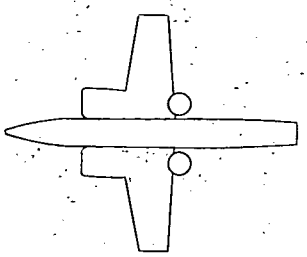


	ALPHA =15.98 DEG EFF. VEL. RATIO = .100			ALPHA =18.00 DEG EFF. VEL. RATIO = .101			ALPHA =19.99 DEG EFF. VEL. RATIO = .101		
		CP			CP			CP	
Y/YMAX =	0.0	0.5	0.8	0.0	0.5	0.8	0.0	0.5	0.8
X/L									
.130	.437			.465			.518		
.219	.471			.485			.518		
.286	.403			.472			.491		
.331	.181			.239			.259		
.353	.028	.069	.191	.087	.117	.198	.091	.104	.259
.375	-.756			-1.396			.862		
.397		-1.038	-.623		-1.939	-1.001		-1.381	.289
.420	-1.238	-2.068	-1.537	-.540	-2.713	-1.067	.027	-1.774	-1.119
.442	-2.981	-3.645	-3.430	-2.203	-3.026	-2.647	-1.823	-1.594	-2.789
.464	-2.666	-4.492	-5.107	-5.232	-3.553	-4.524	-1.774	-2.691	-4.493
.487	-2.583	-.939	-14.373	-.655	-2.285	-16.853	-1.512	-1.316	-19.656
.509	-5.157	-1.437	-8.511	-2.071	-1.594	-6.795	-3.019	-2.953	-4.198
.531	-.042	-1.337	-1.570	-.721	1.090	-2.483	-.333	1.305	-.857
.554	-.347			-.255			-.279		
.576	-.457			-.126			-.222		
.598	-.218	-.361	-1.002	-.262	-.515	-.938	-.101	-.558	-.716
.620	-.109			-.045			-.064		
.643	.202	.048	-.266	.019	-.197	-.096	.205	.212	.071
.665	.120	.280	.260	.174	.137	.026	.205	.299	.202
.687	.277			.333			.293		
.710	.464			.512			.468		
.732	.491			.482			.501		

TABLE 21 (A)

PRESSURE PROFILE ON FUSELAGE

REAR NOZZLES DEFLECTED 90 DEGREES



ALPHA = -.00 DEG
EFF. VEL. RATIO =∞

	CP		
Y/YMAX =	0.0	0.5	0.8
X/L			
.130	-.024		
.219	-.033		
.286	-.127		
.331	-.188		
.353	-.142	-.201	-.276
.375	-.150		
.397		-.139	-.170
.420	-.163	-.089	-.097
.442	-.114	-.030	-.089
.464	-.088	-.042	-.046
.487	.001	-.001	.017
.509	.037	.050	.078
.531	.061	.087	.131
.554	.091		
.576	.104		
.598	.088	.098	.140
.620	.040		
.643	-.019	-.099	-.281
.665	-.059	-.141	-.320
.687	-.094		
.710	-.108		
.732	-.109		

ALPHA = 3.99 DEG
EFF. VEL. RATIO =∞

	CP		
Y/YMAX =	0.0	0.5	0.8
X/L			
.130	.029		
.219	.033		
.286	-.062		
.331	-.116		
.353	-.079	-.130	-.181
.375	-.093		
.397		-.102	-.131
.420	-.131	-.056	-.057
.442	-.078	.008	-.063
.464	-.059	-.021	-.035
.487	.017	.013	.020
.509	.050	.066	.096
.531	.076	.098	.147
.554	.090		
.576	.096		
.598	.074	.096	.146
.620	.020		
.643	-.048	-.113	-.293
.665	-.094	-.153	-.345
.687	-.118		
.710	-.130		
.732	-.124		

ALPHA = 8.05 DEG
EFF. VEL. RATIO =∞

	CP		
Y/YMAX =	0.0	0.5	0.8
X/L			
.130	.089		
.219	.104		
.286	.030		
.331	-.032		
.353	-.026	-.053	-.110
.375	-.035		
.397		-.052	-.075
.420	-.075	-.006	-.003
.442	-.036	-.001	-.028
.464	-.020	.015	-.009
.487	.050	.062	.058
.509	.080	.095	.117
.531	.099	.123	.173
.554	.103		
.576	.103		
.598	.070	.100	.169
.620	.015		
.643	-.061	-.128	-.314
.665	-.114	-.186	-.381
.687	-.144		
.710	-.144		
.732	-.131		

ALPHA =10.00 DEG
EFF. VEL. RATIO =∞

	CP		
Y/YMAX =	0.0	0.5	0.8
X/L			
.130	.120		
.219	.136		
.286	.081		
.331	-.001		
.353	.011	-.026	-.057
.375	-.006		
.397		-.023	-.055
.420	-.057	.013	.018
.442	-.016	.031	-.011
.464	-.012	.035	.001
.487	.065	.086	.062
.509	.097	.101	.129
.531	.105	.133	.185
.554	.113		
.576	.106		
.598	.071	.105	.176
.620	.013		
.643	-.070	-.138	-.316
.665	-.127	-.180	-.399
.687	-.156		
.710	-.157		
.732	-.144		

ALPHA =11.99 DEG
EFF. VEL. RATIO =∞

	CP		
Y/YMAX =	0.0	0.5	0.8
X/L			
.130	.155		
.219	.167		
.286	.108		
.331	.027		
.353	.058	.021	-.027
.375	.018		
.397		-.006	-.034
.420	-.032	.034	.034
.442	-.002	.044	-.006
.464	-.011	.043	.013
.487	.081	.086	.062
.509	.104	.111	.133
.531	.112	.134	.185
.554	.111		
.576	.105		
.598	.068	.104	.185
.620	.006		
.643	-.076	-.142	-.315
.665	-.138	-.207	-.412
.687	-.167		
.710	-.185		
.732	-.154		

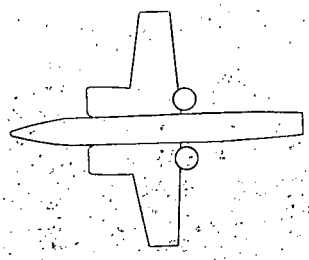
ALPHA =14.04 DEG
EFF. VEL. RATIO =∞

	CP		
Y/YMAX =	0.0	0.5	0.8
X/L			
.130	.192		
.219	.218		
.286	.146		
.331	.066		
.353	.079	.036	.018
.375	.044		
.397		.016	-.016
.420	-.018	.057	.062
.442	.010	.058	.005
.464	.013	.058	.001
.487	.099	.107	.058
.509	.115	.117	.130
.531	.111	.132	.183
.554	.117		
.576	.105		
.598	.066	.112	.194
.620	-.001		
.643	-.103	-.148	-.308
.665	-.165	-.239	-.437
.687	-.208		
.710	-.216		
.732	-.197		

TABLE 21 (A) - CONCLUDED

PRESSURE PROFILE ON FUSELAGE

REAR NOZZLES DEFLECTED 90 DEGREES

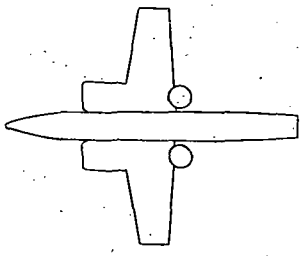


	ALPHA =16.01 DEG EFF. VEL. RATIO =∞			ALPHA =18.02 DEG EFF. VEL. RATIO =∞			ALPHA =20.02 DEG EFF. VEL. RATIO =∞		
		CP			CP			CP	
Y/YMAX =	0.0	0.5	0.8	0.0	0.5	0.8	0.0	0.5	0.8
X/L									
.130	.231			.278			.308		
.219	.249			.281			.327		
.286	.179			.209			.260		
.331	.099			.131			.170		
.353	.118	.090	.049	.147	.114	.088	.175	.152	.117
.375	.080			.114			.138		
.397		.043	-.008		.074	.020		.115	.038
.420	.004	.069	.068	.025	.098	.101	.037	.123	.131
.442	.016	.084	.012	.028	.104	.023	.053	.121	.040
.464	.016	.090	.025	.023	.088	.026	.016	.114	.046
.487	.105	.120	.067	.132	.114	.085	.146	.143	.100
.509	.125	.127	.139	.139	.143	.139	.153	.156	.151
.531	.128	.148	.193	.135	.207	.195	.151	.169	.202
.554	.117			.133			.140		
.576	.104			.112			.119		
.598	.059	.105	.195	.062	.114	.204	.064	.119	.214
.620	-.012			-.011			-.017		
.643	-.111	-.157	-.312	-.114	-.158	-.315	-.121	-.162	-.301
.665	-.192	-.270	-.464	-.200	-.270	-.470	-.209	-.276	-.464
.687	-.229			-.266			-.268		
.710	-.247			-.279			-.295		
.732	-.220			-.258			-.263		

TABLE 21 (B)

PRESSURE PROFILE ON FUSELAGE

REAR NOZZLES DEFLECTED 90 DEGREES



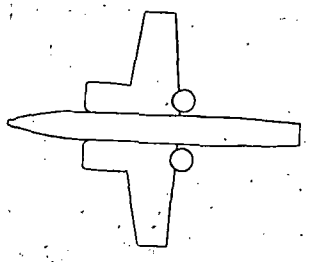
	ALPHA = .00 DEG EFF. VEL. RATIO = .294			ALPHA = 4.02 DEG EFF. VEL. RATIO = .294			ALPHA = 8.04 DEG EFF. VEL. RATIO = .294		
	CP			CP			CP		
Y/YMAX =	0.0	0.5	0.8	0.0	0.5	0.8	0.0	0.5	0.8
X/L									
.130	-.006			.055			.111		
.219	.029			.095			.161		
.286	.010			.062			.158		
.331	-.022			.052			.115		
.353	.039	-.032	-.085	.091	.038	-.015	.158	.121	.075
.375	.013			.082			.139		
.397		.019	-.031		.074	.012		.116	.065
.420	-.032	.085	.079	.016	.122	.126	.063	.161	.156
.442	.033	.156	.063	.067	.155	.073	.094	.191	.103
.464	.059	.160	.103	.071	.181	.100	.107	.220	.139
.487	.205	.209	.158	.232	.245	.164	.267	.275	.163
.509	.235	.240	.211	.252	.261	.219	.289	.291	.240
.531	.205	.237	.251	.225	.261	.263	.260	.286	.262
.554	.106			.115			.144		
.576	-.137			-.122			-.087		
.598	-.568	-.664	-1.872	-.553	-.643	-1.851	-.524	-.651	-1.853
.620	-1.040			-1.043			-1.006		
.643	-1.291	-1.749	-2.552	-1.295	-1.796	-2.667	-1.291	-1.954	-2.863
.665	-1.166	-1.688	-1.787	-1.174	-1.753	-1.776	-1.202	-1.821	-1.556
.687	-.901			-.877			-.885		
.710	-.606			-.597			-.547		
.732	-.526			-.514			-.431		

	ALPHA =10.02 DEG EFF. VEL. RATIO = .294			ALPHA =12.02 DEG EFF. VEL. RATIO = .294			ALPHA =14.01 DEG EFF. VEL. RATIO = .294		
	CP			CP			CP		
Y/YMAX =	0.0	0.5	0.8	0.0	0.5	0.8	0.0	0.5	0.8
X/L									
.130	.157			.192			.226		
.219	.202			.230			.277		
.286	.196			.250			.289		
.331	.152			.209			.232		
.353	.202	.155	.118	.248	.202	.160	.286	.243	.204
.375	.182			.223			.255		
.397		.158	.090		.210	.130		.237	.153
.420	.087	.195	.207	.131	.235	.228	.157	.259	.253
.442	.116	.226	.125	.148	.268	.142	.180	.288	.160
.464	.116	.247	.145	.148	.268	.166	.168	.308	.186
.487	.306	.320	.191	.321	.326	.188	.355	.332	.217
.509	.304	.304	.238	.336	.330	.252	.372	.372	.273
.531	.273	.291	.268	.323	.325	.290	.350	.354	.304
.554	.179			.210			.269		
.576	-.059			.005			.057		
.598	-.470	-.635	-2.014	-.349	-.624	-1.985	-.294	-.528	-1.887
.620	-.947			-.892			-.771		
.643	-1.329	-2.035	-2.999	-1.320	-2.029	-3.141	-1.193	-1.880	-3.050
.665	-1.285	-1.876	-1.542	-1.373	-1.945	-1.711	-1.358	-1.918	-2.135
.687	-.967			-1.124			-1.235		
.710	-.614			-.764			-.977		
.732	-.389			-.463			-.667		

TABLE 21 (B) - CONCLUDED

PRESSURE PROFILE ON FUSELAGE

REAR NOZZLES DEFLECTED 90 DEGREES

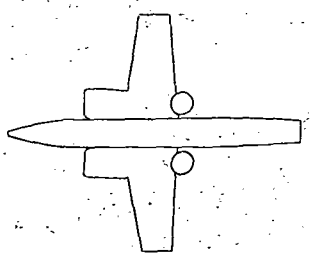


	ALPHA =16.01 DEG EFF. VEL. RATIO = .294			ALPHA =18.01 DEG EFF. VEL. RATIO = .293			ALPHA =20.00 DEG EFF. VEL. RATIO = .293		
	CP			CP			CP		
Y/YMAX =	0.0	0.5	0.8	0.0	0.5	0.8	0.0	0.5	0.8
X/L									
.130	.254			.296			.348		
.219	.307			.371			.394		
.286	.327			.359			.408		
.331	.292			.328			.378		
.353	.312	.283	.244	.346	.316	.265	.388	.351	.313
.375	.270			.314			.360		
.397		.269	.175		.299	.193		.324	.216
.420	.166	.301	.311	.182	.328	.336	.212	.351	.357
.442	.185	.312	.179	.213	.336	.197	.210	.360	.221
.464	.212	.320	.208	.181	.354	.208	.205	.373	.210
.487	.358	.385	.230	.387	.350	.223	.415	.430	.225
.509	.396	.373	.269	.410	.394	.285	.422	.413	.273
.531	.382	.382	.298	.365	.372	.308	.393	.408	.327
.554	.282			.248			.282		
.576	.119			.010			.044		
.598	-.239	-.533	-1.857	-.221	-.503	-1.729	-.382	-.483	-1.718
.620	-.688			-.910			-.838		
.643	-1.262	-1.799	-2.598	-1.228	-1.588	-2.444	-1.166	-1.614	-2.310
.665	-1.237	-1.870	-2.253	-1.215	-1.652	-2.150	-1.185	-1.615	-2.012
.687	-1.247			-1.129			-1.085		
.710	-1.117			-.999			-1.024		
.732	-.899			-.880			-.933		

TABLE 21 (C)

PRESSURE PROFILE ON FUSELAGE

REAR NOZZLES DEFLECTED 90 DEGREES



ALPHA = .01 DEG
EFF. VEL. RATIO = .196

	CP		
Y/YMAX =	0.0	0.5	0.8
X/L			
.130	-.017		
.219	.008		
.286	-.016		
.331	-.039		
.353	.002	-.055	-.117
.375	-.017		
.397		-.045	-.090
.420	-.086	-.004	-.017
.442	-.057	.004	-.070
.464	-.070	.008	-.041
.487	.024	.041	-.025
.509	-.029	-.021	-.017
.531	-.188	-.110	-.086
.554	-.507		
.576	-1.124		
.598	-1.981	-2.808	-4.511
.620	-2.454		
.643	-1.986	-1.940	-2.174
.665	-.942	-.794	-1.384
.687	-.615		
.710	-.610		
.732	-.524		

ALPHA = 4.00 DEG
EFF. VEL. RATIO = .196

	CP		
	0.0	0.5	0.8
	.022		
	.042		
	.040		
	.008		
	.041	-.004	-.047
	.024		
		-.013	-.066
	-.054	.032	.044
	-.049	.028	-.093
	-.086	.012	-.070
	.032	.024	-.058
	-.029	-.017	-.037
	-.180	-.131	-.103
	-.499		
	-1.156		
	-2.019	-2.911	-4.611
	-2.464		
	-2.012	-2.194	-2.218
	-1.005	-.861	-1.299
	-.617		
	-.588		
	-.523		

ALPHA = 8.02 DEG
EFF. VEL. RATIO = .196

	CP		
	0.0	0.5	0.8
	.079		
	.119		
	.125		
	.072		
	.099	.055	.022
	.069		
		.032	-.038
	-.046	.057	.053
	-.042	.044	-.062
	-.079	.036	-.075
	.061	.081	-.070
	-.009	-.009	-.066
	-.169	-.112	-.124
	-.492		
	-1.120		
	-1.985	-2.870	-4.572
	-2.509		
	-2.063	-2.367	-2.283
	-1.065	-.834	-1.298
	-.638		
	-.556		
	-.476		

ALPHA =10.00 DEG
EFF. VEL. RATIO = .195

	CP		
Y/YMAX =	0.0	0.5	0.8
X/L			
.130	.102		
.219	.154		
.286	.156		
.331	.113		
.353	.132	.088	.051
.375	.095		
.397		.062	-.025
.420	-.033	.078	.099
.442	-.004	.070	-.058
.464	-.049	.058	-.082
.487	.058	.037	-.086
.509	.016	.000	-.066
.531	-.156	-.115	-.119
.554	-.500		
.576	-1.123		
.598	-2.028	-2.914	-4.640
.620	-2.546		
.643	-2.064	-2.461	-2.304
.665	-1.077	-.791	-1.312
.687	-.658		
.710	-.568		
.732	-.488		

ALPHA =12.02 DEG
EFF. VEL. RATIO = .196

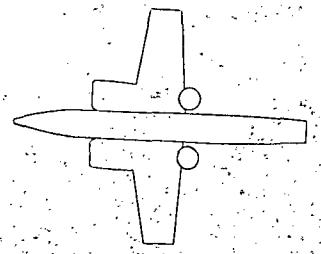
	CP		
	0.0	0.5	0.8
	.128		
	.182		
	.192		
	.142		
	.160	.118	.081
	.115		
		.099	-.004
	.013	.103	.115
	-.004	.095	-.045
	-.045	.074	-.069
	.091	.095	-.074
	.029	.017	-.061
	-.131	-.115	-.131
	-.480		
	-1.132		
	-2.018	-2.888	-4.667
	-2.603		
	-2.131	-2.502	-2.322
	-1.140	-.857	-1.309
	-.604		
	-.541		
	-.478		

ALPHA =14.00 DEG
EFF. VEL. RATIO = .196

	CP		
	0.0	0.5	0.8
	.175		
	.232		
	.237		
	.177		
	.192	.149	.107
	.169		
		.104	.014
	.046	.149	.173
	.018	.124	-.031
	-.044	.112	-.056
	.116	.116	-.064
	.046	.059	-.056
	-.101	-.081	-.126
	-.461		
	-1.070		
	-2.005	-2.830	-4.772
	-2.590		
	-2.202	-2.927	-2.331
	-1.249	-.822	-1.242
	-.647		
	-.546		
	-.508		

TABLE 21 (C) - CONCLUDED
PRESSURE PROFILE ON FUSELAGE

REAR NOZZLES DEFLECTED 90 DEGREES

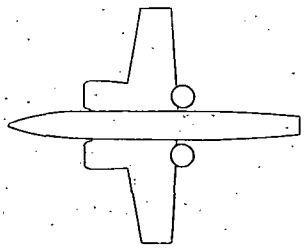


	ALPHA =15.98 DEG EFF. VEL. RATIO = .197			ALPHA =18.03 DEG EFF. VEL. RATIO = .197			ALPHA =19.99 DEG EFF. VEL. RATIO = .196		
	CP			CP			CP		
Y/YMAX =	0.0	0.5	0.8	0.0	0.5	0.8	0.0	0.5	0.8
X/L									
.130	.211			.258			.274		
.219	.276			.299			.343		
.286	.273			.327			.359		
.331	.231			.259			.294		
.353	.219	.195	.158	.269	.234	.179	.309	.281	.216
.375	.200			.232			.250		
.397		.143	.037		.183	.069		.185	.058
.420	.037	.159	.176	.057	.183	.187	.050	.205	.176
.442	.028	.151	-.025	.040	.175	-.008	.054	.205	.009
.464	-.045	.135	-.066	.012	.142	-.045	-.028	.176	-.040
.487	.127	.139	-.070	.158	.179	-.078	.189	.213	-.065
.509	.069	.069	-.061	.093	.101	-.074	.111	.107	-.065
.531	-.078	-.053	-.123	-.053	-.025	-.102	-.032	-.016	-.122
.554	-.413			-.372			-.350		
.576	-1.033			-.945			-.908		
.598	-1.978	-2.780	-4.865	-1.865	-2.763	-4.888	-1.804	-2.654	-4.959
.620	-2.577			-2.581			-2.493		
.643	-2.422	-3.244	-2.318	-2.450	-3.446	-2.423	-2.448	-3.644	-2.406
.665	-1.576	-1.004	-1.211	-1.627	-1.222	-1.185	-1.776	-1.526	-1.340
.687	-.833			-.887			-1.049		
.710	-.493			-.533			-.593		
.732	-.477			-.496			-.547		

TABLE 21 (D)

PRESSURE PROFILE ON FUSELAGE

REAR NOZZLES DEFLECTED 90 DEGREES



ALPHA = .00 DEG
EFF. VEL. RATIO = .098

	CP		
Y/YMAX =	0.0	0.5	0.8
X/L			
.130	-.280		
.219	-.280		
.286	-.277		
.331	-.334		
.353	-.348	-.378	-.402
.375	-.420		
.397		-.503	-.569
.420	-.552	-.503	-.569
.442	-.536	-.486	-.980
.464	-.503	-.618	-.601
.487	-.717	-.684	-.634
.509	-.865	-.782	-.700
.531	-1.194	-1.062	-.881
.554	-1.685		
.576	-2.358		
.598	-2.530	-2.604	-3.284
.620	-1.723		
.643	-1.371	-1.560	-1.993
.665	-1.341	-1.476	-1.300
.687	-.918		
.710	-.766		
.732	-.611		

ALPHA = 3.99 DEG
EFF. VEL. RATIO = .098

	CP		
	0.0	0.5	0.8
	-.196		
	-.192		
	-.172		
	-.243		
	-.260	-.284	-.290
	-.322		
		-.388	-.420
	-.420	-.404	-.338
	-.437	-.437	-.519
	-.453	-.519	-.569
	-.618	-.585	-.585
	-.799	-.766	-.651
	-1.144	-1.046	-.881
	-1.706		
	-2.446		
	-2.814	-3.125	-3.655
	-2.294		
	-.986	-1.817	-1.983
	-.722	-1.094	-1.199
	-.658		
	-.601		
	-.554		

ALPHA = 7.98 DEG
EFF. VEL. RATIO = .097

	CP		
	0.0	0.5	0.8
	-.171		
	-.147		
	-.130		
	-.219		
	-.229	-.253	-.284
	-.302		
		-.386	-.435
	-.352	-.419	-.352
	-.452	-.435	-.502
	-.502	-.519	-.585
	-.635	-.485	-.618
	-.818	-.801	-.718
	-1.200	-1.084	-.901
	-1.775		
	-2.513		
	-2.888	-3.182	-3.684
	-2.383		
	-.956	-1.826	-2.052
	-.689	-1.062	-1.208
	-.662		
	-.580		
	-.529		

ALPHA =10.04 DEG
EFF. VEL. RATIO = .098

	CP		
Y/YMAX =	0.0	0.5	0.8
X/L			
.130	-.132		
.219	-.115		
.286	-.095		
.331	-.177		
.353	-.204	-.224	-.248
.375	-.258		
.397		-.340	-.423
.420	-.456	-.357	-.340
.442	-.439	-.423	-.505
.464	-.604	-.538	-.637
.487	-.604	-.555	-.621
.509	-.836	-.786	-.720
.531	-1.182	-1.100	-.935
.554	-1.797		
.576	-2.563		
.598	-2.950	-3.316	-3.838
.620	-2.516		
.643	-1.095	-1.949	-2.129
.665	-.702	-.997	-1.210
.687	-.681		
.710	-.627		
.732	-.532		

ALPHA =11.97 DEG
EFF. VEL. RATIO = .098

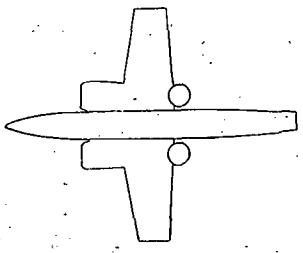
	CP		
	0.0	0.5	0.8
	-.149		
	-.095		
	-.081		
	-.175		
	-.192	-.223	-.253
	-.306		
		-.387	-.469
	-.420	-.387	-.355
	-.469	-.420	-.584
	-.535	-.535	-.535
	-.649	-.535	-.682
	-.862	-.797	-.780
	-1.222	-1.091	-.944
	-1.836		
	-2.656		
	-3.069	-3.442	-4.017
	-2.609		
	-1.012	-1.889	-2.192
	-.689	-1.116	-1.254
	-.703		
	-.646		
	-.568		

ALPHA =14.01 DEG
EFF. VEL. RATIO = .098

	CP		
	0.0	0.5	0.8
	-.130		
	-.103		
	-.079		
	-.154		
	-.198	-.215	-.225
	-.273		
		-.355	-.404
	-.371	-.355	-.371
	-.437	-.437	-.552
	-.503	-.519	-.618
	-.651	-.618	-.684
	-.832	-.816	-.750
	-1.227	-1.112	-.980
	-1.883		
	-2.667		
	-3.062	-3.461	-4.015
	-2.660		
	-1.177	-1.660	-2.208
	-.728	-1.093	-1.251
	-.718		
	-.647		
	-.525		

TABLE 21 (D) - CONCLUDED
PRESSURE PROFILE ON FUSELAGE

REAR NOZZLES DEFLECTED 90 DEGREES

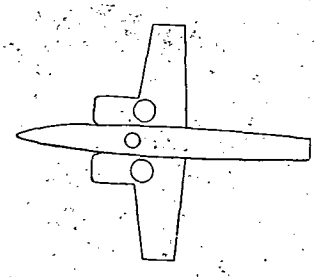


	ALPHA =16.03 DEG EFF. VEL. RATIO = .097			ALPHA =18.00 DEG EFF. VEL. RATIO = .098			ALPHA =20.00 DEG EFF. VEL. RATIO = .098		
	CP			CP			CP		
Y/YMAX =	0.0	0.5	0.8	0.0	0.5	0.8	0.0	0.5	0.8
X/L									
.130	-.126			-.090			-.062		
.219	-.106			-.036			-.038		
.286	-.058			-.012			-.011		
.331	-.167			-.103			-.116		
.353	-.195	-.201	-.225	-.117	-.158	-.185	-.130	-.174	-.184
.375	-.251			-.278			-.218		
.397		-.351	-.434		-.344	-.377		-.349	-.448
.420	-.500	-.400	-.417	-.377	-.344	-.145	-.399	-.399	-.349
.442	-.550	-.467	-.583	-.459	-.443	-.575	-.465	-.432	-.597
.464	-.616	-.600	-.666	-.509	-.542	-.657	-.597	-.547	-.696
.487	-.683	-.749	-.716	-.674	-.657	-.690	-.679	-.696	-.696
.509	-.915	-.849	-.749	-.839	-.806	-.773	-.893	-.827	-.827
.531	-1.264	-1.147	-.965	-1.235	-1.037	-1.037	-1.256	-1.190	-.976
.554	-1.847			-1.859			-1.903		
.576	-2.652			-2.616			-2.695		
.598	-3.088	-3.490	-4.056	-3.114	-3.412	-3.928	-3.189	-3.541	-4.042
.620	-2.597			-2.612			-2.708		
.643	-1.104	-1.748	-2.253	-1.195	-1.853	-2.175	-1.087	-1.811	-2.241
.665	-.729	-1.026	-1.281	-.730	-1.012	-1.232	-.810	-1.158	-1.229
.687	-.722			-.727			-.870		
.710	-.620			-.669			-.766		
.732	-.562			-.622			-.732		

TABLE 22 (A)

PRESSURE PROFILE ON FUSELAGE

FORWARD NOZZLES DEFLECTED 90 DEGREES WITH LIFT JET



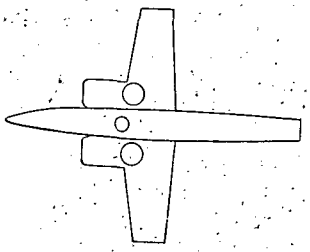
	ALPHA = 0.00 DEG EFF. VEL. RATIO = .283			ALPHA = 3.96 DEG EFF. VEL. RATIO = .283			ALPHA = 8.02 DEG EFF. VEL. RATIO = .283		
		CP			CP			CP	
Y/YMAX =	0.0	0.5	0.8	0.0	0.5	0.8	0.0	0.5	0.8
X/L									
.130	.086			.138			.207		
.219	.211			.275			.336		
.286	.308			.356			.413		
.331	.385			.438			.491		
.353	.507	.354	.261	.551	.408	.325	.584	.453	.392
.375	.422			.374			.346		
.397		-1.950	-.257		-.740	.249		-2.957	.506
.420	-.346	-1.069	-.313	-.437	-.193	-1.009	-.608	-1.408	-1.862
.442	-.877	-.558	-1.364	-.298	-.526	.140	-.890	-.618	-1.165
.464	-1.098	-.884	-2.649	-2.848	-1.940	-2.385	-1.986	-1.397	-1.750
.487	-1.368	-1.532	-1.784	-1.340	-1.161	-2.003	-1.561	-1.893	-1.998
.509	-1.534	-1.784	-2.362	-1.534	-1.664	-1.580	-1.692	-1.823	-1.555
.531	-1.671	-1.631	-2.045	-1.437	-1.050	-1.747	-1.891	-.839	-1.752
.554	-1.330			-1.404			-1.454		
.576	-1.209			-1.358			-1.388		
.598	-.980	-.981	-1.165	-1.057	-1.077	-1.314	-1.212	-1.173	-1.310
.620	-.796			-.897			-.984		
.643	-.657	-.776	-1.475	-.729	-.890	-1.635	-.893	-1.001	-1.643
.665	-.495	-.625	-1.460	-.629	-.798	-1.797	-.782	-.854	-1.774
.687	-.389			-.465			-.662		
.710	-.248			-.340			-.499		
.732	-.205			-.308			-.429		

	ALPHA = 9.82 DEG EFF. VEL. RATIO = .282			ALPHA =12.01 DEG EFF. VEL. RATIO = .283			ALPHA =14.06 DEG EFF. VEL. RATIO = .282		
		CP			CP			CP	
Y/YMAX =	0.0	0.5	0.8	0.0	0.5	0.8	0.0	0.5	0.8
X/L									
.130	.223			.268			.297		
.219	.358			.390			.426		
.286	.451			.474			.503		
.331	.503			.533			.544		
.353	.594	.469	.382	.608	.499	.415	.629	.495	.425
.375	.266			.376			.451		
.397		-.096	-.301		-1.024	.442		-2.201	-.509
.420	-.892	-1.352	.272	-.470	-.439	-2.437	-.387	-1.384	-.632
.442	-.667	-.482	-1.749	-.550	-.650	-.681	-.383	-.537	-.703
.464	-1.044	-.546	-1.743	.321	-.804	-2.054	-2.170	-1.389	-1.920
.487	-1.500	-1.626	-1.745	-1.600	-1.496	-1.928	-1.554	-1.354	-2.196
.509	-1.668	-1.775	-1.555	-1.662	-2.116	-1.830	-1.570	-1.733	-1.799
.531	-1.815	-2.342	-1.505	-1.903	-1.604	-1.888	-1.995	-1.647	-1.510
.554	-1.493			-1.539			-1.512		
.576	-1.428			-1.380			-1.417		
.598	-1.153	-1.191	-1.241	-1.247	-1.184	-1.307	-1.198	-1.264	-1.293
.620	-.997			-1.059			-1.049		
.643	-.863	-.927	-1.716	-.895	-.899	-1.618	-.960	-.916	-1.563
.665	-.732	-.797	-1.855	-.834	-.785	-1.540	-.818	-.842	-1.535
.687	-.658			-.702			-.677		
.710	-.562			-.533			-.553		
.732	-.438			-.432			-.448		

TABLE 22 (A) - CONCLUDED

PRESSURE PROFILE ON FUSELAGE

FORWARD NOZZLES DEFLECTED 90 DEGREES WITH LIFT JET

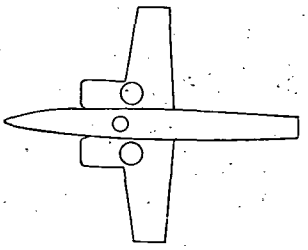


	ALPHA =16.00 DEG EFF. VEL. RATIO = .282			ALPHA =18.00 DEG EFF. VEL. RATIO = .282			ALPHA =19.98 DEG EFF. VEL. RATIO = .281		
		CP			CP			CP	
Y/YMAX =	0.0	0.5	0.8	0.0	0.5	0.8	0.0	0.5	0.8
X/L									
.130	.232			.374			.418		
.219	.455			.491			.527		
.266	.534			.571			.601		
.331	.568			.592			.612		
.353	.649	.525	.442	.657	.553	.464	.673	.581	.467
.375	.469			.372			.193		
.397		-.897	-.187		-.490	.492		-1.291	.029
.420	.272	-1.285	-.041	-.505	-1.458	-.835	-.345	.209	.120
.442	-.507	-.392	-.009	.222	-.107	-.534	-.296	-.353	-1.103
.464	.024	-1.665	-1.723	-.497	-.538	-1.947	.443	-1.360	-1.935
.487	-1.400	-1.546	-2.073	-1.184	-1.277	-1.940	-1.543	-1.491	-2.181
.509	-1.528	-1.690	-1.366	-1.735	-1.849	-1.843	-1.561	-1.566	-1.867
.531	-.789	-1.406	-1.594	-1.323	-1.540	-1.390	-1.889	-1.661	-1.473
.554	-1.491			-1.468			-1.409		
.576	-1.346			-1.346			-1.358		
.598	-1.213	-1.294	-1.327	-1.301	-1.276	-1.290	-1.326	-1.309	-1.344
.620	-1.061			-1.053			-1.049		
.643	-.950	-.909	-1.166	-.952	-.936	-1.075	-.907	-.959	-1.032
.665	-.835	-.364	-1.325	-.794	-.855	-1.093	-.866	-.883	-1.063
.687	-.752			-.787			-.754		
.710	-.651			-.683			-.665		
.732	-.491			-.538			-.586		

TABLE 22 (B)

PRESSURE PROFILE ON FUSELAGE

FORWARD NOZZLES DEFLECTED 90 DEGREES WITH LIFT JET



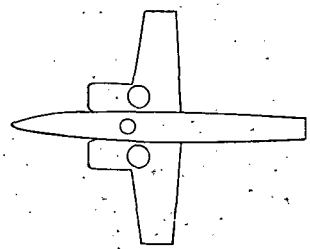
	ALPHA = .01 DEG EFF. VEL. RATIO = .188			ALPHA = 3.93 DEG EFF. VEL. RATIO = .188			ALPHA = 8.06 DEG EFF. VEL. RATIO = .187		
	CP			CP			CP		
Y/YMAX =	0.0	0.5	0.8	0.0	0.5	0.8	0.0	0.5	0.8
X/L									
.130	.063			.093			.176		
.219	.133			.219			.256		
.286	.193			.254			.304		
.331	.231			.287			.323		
.353	.340	.185	.071	.355	.214	.129	.399	.241	.152
.375	.244			.201			.759		
.397		-5.016	-1.528		.217	-.077		-5.309	-.891
.420	-1.487	-.168	-2.459	-1.169	-5.970	-.085	-.760	-3.872	-3.026
.442	-1.745	-1.798	-4.162	-2.253	-2.097	-3.995	-1.729	-2.538	-5.099
.464	-5.465	-1.124	-3.697	-2.895	-2.621	-3.843	-2.480	-2.107	-7.833
.487	-1.839	-2.868	-2.537	-2.408	-1.590	-4.334	-2.657	-2.197	-3.580
.509	-1.487	-2.639	-1.549	-1.594	-3.009	-2.661	-2.066	-2.579	-2.312
.531	-.907	-1.659	-3.137	-1.500	-2.338	-2.449	-1.191	-2.648	-2.714
.554	-1.641			-1.551			-1.445		
.576	-1.545			-1.395			-1.352		
.598	-1.016	-1.421	-2.300	-1.083	-1.304	-2.526	-1.163	-1.365	-2.718
.620	-.522			-.596			-.801		
.643	-.314	-.301	-1.336	-.282	-.386	-1.399	-.416	-.389	-1.559
.665	-.180	-.104	-.726	-.166	-.133	-.902	-.239	-.187	-1.021
.687	-.079			-.103			-.194		
.710	.010			-.041			-.051		
.732	.054			.018			-.013		

	ALPHA = 9.99 DEG EFF. VEL. RATIO = .188			ALPHA =12.01 DEG EFF. VEL. RATIO = .187			ALPHA =14.01 DEG EFF. VEL. RATIO = .187		
	CP			CP			CP		
Y/YMAX =	0.0	0.5	0.8	0.0	0.5	0.8	0.0	0.5	0.8
X/L									
.130	.201			.221			.250		
.219	.293			.326			.350		
.286	.335			.368			.382		
.331	.331			.358			.368		
.353	.398	.242	.147	.425	.248	.140	.421	.241	.128
.375	-.571			.046			.101		
.397		-.748	-1.477		-6.027	-.984		-4.462	-1.604
.420	-.891	-5.874	-1.830	-.356	.518	-2.268	-.335	-2.372	.166
.442	-1.825	-3.018	-7.169	-3.011	-2.818	-6.117	-2.663	-2.347	-4.598
.464	-5.497	-4.415	-3.846	-7.361	-3.294	-3.360	-5.009	-5.505	-4.618
.487	-2.776	-4.022	-3.649	-2.375	-1.997	-4.775	-2.610	-3.029	-3.493
.509	-1.928	-3.166	-2.473	-2.001	-2.900	-3.310	-1.797	-3.325	-4.208
.531	-1.694	-1.985	-2.420	-1.558	-2.178	-1.423	-1.288	-1.916	-1.205
.554	-1.452			-1.286			-1.446		
.576	-1.344			-1.262			-1.264		
.598	-1.092	-1.345	-2.762	-1.032	-1.228	-2.887	-.905	-1.054	-2.795
.620	-.760			-.702			-.691		
.643	-.434	-.512	-1.457	-.467	-.392	-1.426	-.549	-.366	-1.627
.665	-.189	-.237	-1.165	-.331	-.231	-1.062	-.223	-.258	-1.059
.687	-.144			-.186			-.133		
.710	-.095			-.048			-.040		
.732	.017			.020			-.008		

TABLE 22 (B) - CONCLUDED

PRESSURE PROFILE ON FUSELAGE

FORWARD NOZZLES DEFLECTED 90 DEGREES WITH LIFT JET

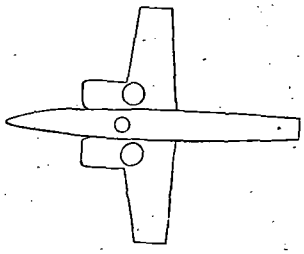


	ALPHA =16.01 DEG EFF. VEL. RATIO = .187			ALPHA =18.01 DEG EFF. VEL. RATIO = .187			ALPHA =20.04 DEG EFF. VEL. RATIO = .187		
	CP			CP			CP		
Y/YMAX =	0.0	0.5	0.8	0.0	0.5	0.8	0.0	0.5	0.8
X/L									
.130	.303			.355			.379		
.219	.382			.417			.483		
.286	.429			.463			.497		
.331	.393			.425			.446		
.353	.441	.264	.146	.465	.304	.178	.477	.312	.182
.375	.215			.369			.185		
.397		-4.398	-1.108		-2.706	-.516		-3.647	-1.456
.420	-1.477	-6.102	.624	-.455	-2.378	-2.997	-1.727	.624	-.406
.442	-1.879	-2.403	-7.335	-1.615	-1.697	-5.576	-.430	-2.519	-5.477
.464	-3.615	-4.980	-3.140	-5.043	-3.981	-3.054	-3.130	-2.777	-4.000
.487	-2.829	-2.448	-4.197	-2.812	-3.300	-3.124	-3.557	-1.854	-3.939
.509	-2.382	-3.550	-1.940	-1.988	-3.181	-4.268	-2.191	-3.155	-5.055
.531	-.666	-3.726	-2.440	-1.082	-.320	-2.689	-1.612	-.955	-1.239
.554	-1.242			-1.224			-1.273		
.576	-1.254			-1.237			-1.137		
.598	-1.077	-1.130	-2.869	-1.034	-1.099	-2.505	-.987	-1.188	-2.731
.620	-.696			-.686			-.776		
.643	-.615	-.354	-1.600	-.529	-.423	-1.676	-.428	-.508	-1.680
.665	-.306	-.263	-1.035	-.282	-.316	-1.165	-.338	-.308	-1.271
.687	-.150			-.136			-.279		
.710	-.072			-.072			-.114		
.732	.027			.018			-.026		

TABLE 22 (C)

PRESSURE PROFILE ON FUSELAGE

FORWARD NOZZLES DEFLECTED 90 DEGREES WITH LIFT JET



ALPHA = -0.00 DEG
EFF. VEL. RATIO = .093

	CP		
Y/YMAX =	0.0	0.5	0.8
X/L			
.130	-.124		
.219	-.127		
.286	-.107		
.331	-.178		
.353	-.141	-.366	-.400
.375	.202		
.397		-9.574	-3.794
.420	-1.780	-10.916	.152
.442	-5.349	-2.631	-7.527
.464	-3.630	-2.320	-4.498
.487	-1.551	-1.551	-5.513
.509	-2.156	-4.252	-1.927
.531	-2.648	-2.320	-3.761
.554	-2.140		
.576	-1.545		
.598	-.766	-.883	-1.418
.620	-.335		
.643	.024	.118	-.114
.665	.209	.139	.041
.687	.270		
.710	.310		
.732	.323		

ALPHA = -.01 DEG
EFF. VEL. RATIO = .095

	CP		
	0.0	0.5	0.8
	-.084		
	-.123		
	-.117		
	-.179		
	-.150	-.325	-.434
	-.701		
		-9.215	-6.334
	-4.676	-1.860	-7.573
	-7.927	-1.876	-6.350
	-3.421	-2.149	-4.628
	-1.715	-2.117	-4.403
	-1.473	-3.437	-3.871
	-3.582	-2.745	-6.946
	-2.495		
	-1.781		
	-.810	-.764	-1.137
	-.328		
	.059	.056	-.196
	.274	.181	.132
	.241		
	.260		
	.227		

ALPHA = 3.93 DEG
EFF. VEL. RATIO = .095

	CP		
	0.0	0.5	0.8
	.001		
	.001		
	-.019		
	-.124		
	-.091	-.253	-.316
	-2.234		
		-11.521	-5.984
	-5.437	-1.832	-3.168
	-3.699	-4.987	-4.987
	-4.488	-3.924	-6.789
	-3.055	-4.053	-4.327
	-3.136	-4.327	-5.824
	-3.120	-7.417	-7.578
	-2.434		
	-1.707		
	-.838	-.828	-1.323
	-.408		
	.064	.054	-.088
	.140	.068	.078
	.299		
	.236		
	.259		

ALPHA = 8.04 DEG
EFF. VEL. RATIO = .094

	CP		
Y/YMAX =	0.0	0.5	0.8
X/L			
.130	.089		
.219	.100		
.286	.073		
.331	-.073		
.353	-.032	-.269	-.296
.375	-.326		
.397		-6.745	-3.684
.420	-3.091	.300	.234
.442	-2.449	-1.676	-2.548
.464	-4.984	.431	-2.137
.487	-2.268	.579	-4.441
.509	-1.445	.036	-4.408
.531	-1.857	-1.988	-3.157
.554	-1.978		
.576	-1.508		
.598	-.772	-.927	-1.393
.620	-.197		
.643	.086	.066	-.076
.665	.309	.191	.070
.687	.259		
.710	.313		
.732	.343		

ALPHA =10.04 DEG
EFF. VEL. RATIO = .094

	CP		
	0.0	0.5	0.8
	.134		
	.144		
	.150		
	-.008		
	.009	-.170	-.213
	.936		
		-6.150	-2.541
	.215	-3.049	.034
	-4.034	-1.016	-.966
	-.031	-.540	-4.444
	.789	-.934	-4.969
	-1.048	-2.246	-1.458
	-2.426	-2.787	-4.116
	-2.341		
	-1.698		
	-.768	-.657	-1.085
	-.230		
	-.001	.188	-.098
	.137	.252	.043
	.326		
	.390		
	.393		

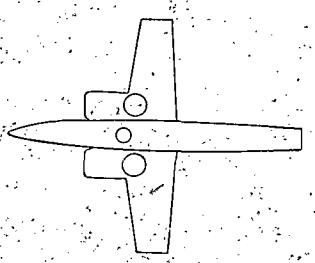
ALPHA =12.00 DEG
EFF. VEL. RATIO = .094

	CP		
	0.0	0.5	0.8
	.159		
	.193		
	.149		
	-.010		
	-.000	-.227	-.193
	.280		
		-5.072	-2.016
	-3.668	-1.587	-2.099
	-1.075	-2.083	-2.215
	-1.009	-2.330	-2.958
	-.117	1.386	-5.485
	-2.314	-2.248	-3.635
	-1.785	.395	-3.305
	-2.037		
	-1.522		
	-.634	-.959	-1.346
	-.166		
	-.071	.211	-.237
	.183	.092	.156
	.353		
	.234		
	.424		

TABLE 22 (C) - CONCLUDED

PRESSURE PROFILE ON FUSELAGE

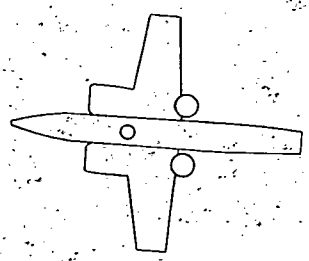
FORWARD NOZZLES DEFLECTED 90 DEGREES WITH LIFT JET



	ALPHA =14.00 DEG EFF. VEL. RATIO = .094			ALPHA =16.08 DEG EFF. VEL. RATIO = .094			ALPHA =18.01 DEG EFF. VEL. RATIO = .094		
	CP			CP			CP		
Y/YMAX =	0.0	0.5	0.8	0.0	0.5	0.8	0.0	0.5	0.8
X/L									
.130	.147			.157			.191		
.219	.174			.191			.177		
.286	.124			.157			.136		
.331	-.028			-.018			-.036		
.353	-.035	-.159	-.112	-.002	-.164	-.171	-.009	-.191	-.232
.375	-.820			.246			1.716		
.397		-6.577	-4.281		-3.063	-1.960		-4.440	-2.564
.420	-3.116	-8.627	-4.756	-1.137	-5.351	-4.330	-1.099	.630	-2.284
.442	-2.280	-.131	-3.526	-8.165	-2.273	-4.725	-6.415	-1.955	-2.235
.464	-6.249	-3.936	-4.756	-1.170	-1.186	-3.277	.136	.070	-2.514
.487	-.852	-2.870	-5.429	-1.137	-.627	-3.984	-1.033	-1.099	-4.736
.509	-2.214	-3.772	-4.149	-.758	-1.762	-4.017	-1.593	-3.173	-1.576
.531	-3.657	-4.067	-1.952	-3.754	-4.001	-3.705	-1.329	-5.312	-.638
.554	-1.944			-1.866			-2.157		
.576	-1.556			-1.356			-1.292		
.598	-.651	-.896	-1.533	-.836	-.684	-1.315	-.752	-.941	-1.015
.620	-.270			-.268			-.428		
.643	.019	-.018	-.199	.009	-.001	-.079	-.025	.073	-.073
.665	.195	.168	.178	.002	.273	.151	.089	.238	.113
.687	.245			.232			.235		
.710	.349			.300			.265		
.732	.397			.347			.285		

TABLE 23 (A)
PRESSURE PROFILE ON FUSELAGE

REAR NOZZLES DEFLECTED 90 DEGREES WITH LIFT JET



ALPHA = .00 DEG
EFF. VEL. RATIO =∞

	CP		
Y/YMAX =	0.0	0.5	0.8
X/L			
.130	-.019		
.219	-.014		
.286	-.116		
.331	-.178		
.353	-.126	-.178	-.273
.375	-.194		
.397		-.096	-.168
.420	-.170	-.114	-.105
.442	-.104	-.063	-.083
.464	-.092	-.038	-.055
.487	.005	.014	.007
.509	.048	.053	.091
.531	.075	.092	.143
.554	.092		
.576	.102		
.598	.084	.099	.136
.620	.041		
.643	-.021	-.089	-.270
.665	-.062	-.134	-.300
.687	-.089		
.710	-.108		
.732	-.105		

ALPHA = 3.91 DEG
EFF. VEL. RATIO =∞

	CP		
	0.0	0.5	0.8
	.031		
	.037		
	-.053		
	-.116		
	-.074	-.106	-.181
	-.158		
		-.066	-.117
	-.117	-.079	-.087
	-.072	-.031	-.056
	-.070	-.022	-.039
	.020	.012	.025
	.051	.064	.098
	.081	.101	.153
	.092		
	.097		
	.072	.096	.153
	.020		
	-.045	-.108	-.301
	-.092	-.159	-.335
	-.118		
	-.118		
	-.116		

ALPHA = 8.05 DEG
EFF. VEL. RATIO =∞

	CP		
	0.0	0.5	0.8
	.108		
	.114		
	.038		
	-.033		
	-.017	-.026	-.098
	-.105		
		.002	-.067
	-.066	-.029	-.028
	-.024	.009	-.028
	-.027	.018	-.002
	.050	.048	.054
	.078	.086	.117
	.102	.120	.166
	.106		
	.105		
	.071	.100	.170
	.018		
	-.058	-.126	-.309
	-.119	-.178	-.381
	-.139		
	-.143		
	-.125		

ALPHA = 9.99 DEG
EFF. VEL. RATIO =∞

	CP		
Y/YMAX =	0.0	0.5	0.8
X/L			
.130	.124		
.219	.146		
.286	.078		
.331	-.001		
.353	.019	.008	-.055
.375	-.103		
.397		.022	-.050
.420	-.052	-.015	-.011
.442	-.010	.011	-.023
.464	-.010	.034	-.003
.487	.073	.065	.069
.509	.099	.107	.137
.531	.111	.134	.184
.554	.106		
.576	.105		
.598	.072	.107	.181
.620	.021		
.643	-.064	-.128	-.308
.665	-.120	-.190	-.408
.687	-.153		
.710	-.155		
.732	-.131		

ALPHA =12.00 DEG
EFF. VEL. RATIO =∞

	CP		
	0.0	0.5	0.8
	.161		
	.179		
	.108		
	.032		
	.050	.034	-.017
	-.085		
		.071	-.023
	-.013	.007	.007
	.003	.047	-.010
	-.005	.045	-.001
	.083	.077	.056
	.101	.115	.140
	.113	.138	.189
	.110		
	.104		
	.069	.111	.181
	.003		
	-.079	-.134	-.318
	-.144	-.210	-.420
	-.167		
	-.169		
	-.146		

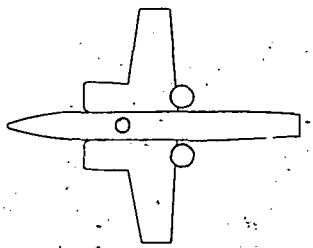
ALPHA =14.02 DEG
EFF. VEL. RATIO =∞

	CP		
	0.0	0.5	0.8
	.195		
	.203		
	.145		
	.055		
	.075	.065	.020
	-.084		
		.088	-.005
	-.004	.020	.012
	.024	.055	.002
	-.002	.055	.008
	.093	.096	.087
	.113	.128	.140
	.123	.139	.194
	.115		
	.108		
	.064	.109	.189
	-.001		
	-.094	-.153	-.314
	-.159	-.235	-.426
	-.201		
	-.209		
	-.184		

TABLE 23 (A) - CONCLUDED

PRESSURE PROFILE ON FUSELAGE

REAR NOZZLES DEFLECTED 90 DEGREES WITH LIFT JET

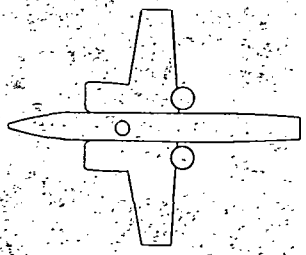


	ALPHA =15.99 DEG EFF. VEL. RATIO =∞			ALPHA =18.02 DEG EFF. VEL. RATIO =∞			ALPHA =20.00 DEG EFF. VEL. RATIO =∞		
	CP			CP			CP		
Y/YMAX =	0.0	0.5	0.8	0.0	0.5	0.8	0.0	0.5	0.8
X/L									
.130	.230			.266			.310		
.219	.251			.278			.316		
.286	.181			.215			.264		
.331	.103			.145			.177		
.353	.110	.085	.052	.146	.121	.080	.175	.150	.126
.375	-.074			-.065			-.041		
.397		.120	.017		.137	.026		.177	.052
.420	.011	.044	.048	.027	.071	.071	.049	.104	.099
.442	.041	.082	.018	.043	.096	.013	.043	.122	.039
.464	.011	.073	.023	.018	.091	.022	.040	.112	.041
.487	.111	.115	.073	.125	.125	.073	.146	.138	.094
.509	.119	.130	.130	.133	.138	.145	.158	.153	.141
.531	.133	.151	.196	.132	.156	.196	.155	.166	.201
.554	.122			.127			.136		
.576	.105			.115			.118		
.598	.063	.112	.189	.065	.111	.199	.064	.113	.205
.620	-.009			-.010			-.013		
.643	-.112	-.164	-.313	-.119	-.170	-.313	-.122	-.164	-.305
.665	-.191	-.260	-.453	-.208	-.265	-.455	-.208	-.269	-.471
.687	-.236			-.257			-.264		
.710	-.242			-.262			-.278		
.732	-.230			-.241			-.263		

TABLE 23 (B)

PRESSURE PROFILE ON FUSELAGE

REAR NOZZLES DEFLECTED 90 DEGREES WITH LIFT JET



ALPHA = -.02 DEG
EFF. VEL. RATIO = .278

	CP		
Y/YMAX =	0.0	0.5	0.8
X/L			
.130	.034		
.219	.080		
.286	.073		
.331	.085		
.353	.288	.027	-.241
.375	.370		
.397		-4.337	-1.489
.420	-1.375	-1.586	-.424
.442	-1.150	-.832	-.686
.464	-.570	-.289	-.444
.487	-.426	-.623	-.307
.509	-.190	-.076	-.173
.531	-.126	-.192	-.096
.554	-.357		
.576	-.426		
.598	-.926	-1.215	-2.919
.620	-1.570		
.643	-1.641	-2.130	-2.042
.665	-1.273	-1.473	-1.309
.687	-.860		
.710	-.678		
.732	-.533		

ALPHA = 3.96 DEG
EFF. VEL. RATIO = .278

	CP		
	0.0	0.5	0.8
	.094		
	.131		
	.136		
	.158		
	.322	.073	-.193
	.377		
		-4.505	-1.452
	-1.658	-1.529	-2.117
	-1.432	-.514	-.728
	-.346	-.322	-.392
	-.291	.065	-.218
	-.180	-.343	-.191
	-.195	-.193	-.151
	-.384		
	-.531		
	-1.131	-1.350	-2.963
	-1.678		
	-1.695	-2.197	-1.930
	-1.260	-1.377	-1.285
	-.799		
	-.650		
	-.475		

ALPHA = 8.02 DEG
EFF. VEL. RATIO = .278

	CP		
	0.0	0.5	0.8
	.153		
	.206		
	.204		
	.206		
	.359	.112	-.153
	.391		
		-4.109	-1.276
	-1.388	-1.866	-3.877
	-1.326	-.489	-.722
	-.955	-.310	-.425
	-.368	-.222	-.390
	-.219	-.386	-.115
	-.140	-.147	-.011
	-.240		
	-.628		
	-1.154	-1.361	-3.005
	-1.692		
	-1.762	-2.232	-2.036
	-1.314	-1.479	-1.347
	-.860		
	-.645		
	-.580		

ALPHA = 9.98 DEG
EFF. VEL. RATIO = .278

	CP		
Y/YMAX =	0.0	0.5	0.8
X/L			
.130	.184		
.219	.240		
.286	.231		
.331	.223		
.353	.383	.123	-.124
.375	.398		
.397		-4.383	-1.284
.420	-1.875	-1.578	-.681
.442	-1.161	-.472	-.715
.464	-.715	-.319	-.410
.487	-.260	-.686	-.238
.509	-.138	-.170	-.189
.531	-.233	-.196	-.130
.554	-.441		
.576	-.714		
.598	-1.283	-1.452	-3.082
.620	-1.730		
.643	-1.817	-2.288	-2.156
.665	-1.416	-1.451	-1.383
.687	-.929		
.710	-.676		
.732	-.525		

ALPHA =11.96 DEG
EFF. VEL. RATIO = .278

	CP		
	0.0	0.5	0.8
	.235		
	.268		
	.279		
	.261		
	.398	.152	-.093
	.399		
		-3.798	-1.156
	-1.051	-1.509	.081
	-1.138	-.521	-.706
	-.337	-.282	-.468
	-.499	.194	-.412
	-.251	-.547	-.293
	-.244	-.311	-.079
	-.437		
	-.627		
	-1.225	-1.375	-3.104
	-1.803		
	-1.827	-2.331	-2.135
	-1.359	-1.451	-1.387
	-.979		
	-.733		
	-.632		

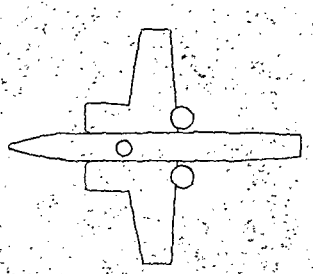
ALPHA =14.00 DEG
EFF. VEL. RATIO = .278

	CP		
	0.0	0.5	0.8
	.254		
	.307		
	.307		
	.281		
	.420	.199	-.097
	.393		
		-4.137	-1.234
	-1.894	-1.363	-2.016
	-.781	-.504	-.515
	-1.401	-.265	-.415
	-.416	-.431	-.183
	-.232	-.453	-.167
	-.265	-.187	-.090
	-.337		
	-.720		
	-1.206	-1.365	-3.158
	-1.781		
	-1.818	-2.348	-2.143
	-1.425	-1.563	-1.432
	-.952		
	-.759		
	-.630		

TABLE 23.(B). - CONCLUDED

PRESSURE PROFILE ON FUSELAGE

REAR NOZZLES DEFLECTED 90 DEGREES WITH LIFT JET

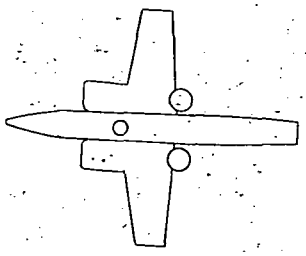


	ALPHA =15.98 DEG EFF. VEL. RATIO = .278			ALPHA =17.96 DEG EFF. VEL. RATIO = .278			ALPHA =20.00 DEG EFF. VEL. RATIO = .278		
	CP			CP			CP		
Y/YMAX =	0.0	0.5	0.8	0.0	0.5	0.8	0.0	0.5	0.8
X/L									
.130	.299			.327			.377		
.219	.359			.386			.443		
.286	.349			.373			.421		
.331	.308			.345			.376		
.353	.430	.219	-.056	.452	.215	-.037	.460	.225	-.024
.375	.399			.399			.414		
.397		-4.029	-1.076		-3.895	-1.106		-4.182	-1.168
.420	-1.151	-1.257	-2.225	-1.751	-1.212	-2.141	-1.449	-1.268	.179
.442	-.706	-.366	-.567	-1.192	-.239	-.514	-.843	-.378	-.401
.464	-1.305	-.313	-.370	-.769	-.235	-.328	-.237	-.157	-.259
.487	-.395	-.253	-.342	-.270	-.093	-.204	-.387	-.098	-.142
.509	-.138	-.477	-.083	-.091	-.142	-.113	-.089	-.051	-.075
.531	-.295	-.194	-.059	-.204	-.166	-.073	-.186	-.100	.033
.554	-.502			-.438			-.342		
.576	-.800			-.569			-.594		
.598	-1.291	-1.447	-3.204	-1.270	-1.382	-3.121	-1.182	-1.321	-3.167
.620	-1.783			-1.757			-1.751		
.643	-1.924	-2.419	-2.298	-1.944	-2.528	-2.327	-1.978	-2.517	-2.539
.665	-1.492	-1.644	-1.509	-1.590	-1.758	-1.644	-1.654	-1.914	-1.636
.687	-1.059			-1.136			-1.075		
.710	-.822			-.855			-.886		
.732	-.730			-.744			-.744		

TABLE 23 (C)

PRESSURE PROFILE ON FUSELAGE

REAR NOZZLES DEFLECTED 90 DEGREES WITH LIFT JET



ALPHA = -.04 DEG
EFF. VEL. RATIO = .186

	CP		
Y/YMAX =	0.0	0.5	0.8
X/L			
.130	.004		
.219	.014		
.286	-.028		
.331	-.055		
.353	.086	-.161	-.454
.375	.162		
.397		-5.095	-1.386
.420	-.567	-1.803	-4.428
.442	-1.742	-.788	-.845
.464	-.169	-.378	-.735
.487	-.305	-.264	-.456
.509	-.354	-.333	-.423
.531	-.497	-.432	-.337
.554	-.799		
.576	-1.409		
.598	-2.231	-2.981	-4.222
.620	-2.339		
.643	-1.595	-1.506	-2.007
.665	-1.034	-1.219	-1.152
.687	-.909		
.710	-.720		
.732	-.614		

ALPHA = 3.97 DEG
EFF. VEL. RATIO = .187

	CP		
	0.0	0.5	0.8
	.050		
	.069		
	.016		
	-.032		
	.119	-.150	-.365
	.172		
		-5.059	-1.561
	-.199	-1.842	-1.972
	-2.307	-.660	-.957
	-2.405	-.391	-.688
	-.386	-.696	-.476
	-.350	-.272	-.333
	-.586	-.391	-.415
	-.861		
	-1.473		
	-2.268	-3.041	-4.272
	-2.445		
	-1.554	-1.615	-2.088
	-1.037	-1.107	-1.304
	-.936		
	-.719		
	-.551		

ALPHA = 7.98 DEG
EFF. VEL. RATIO = .186

	CP		
	0.0	0.5	0.8
	.108		
	.138		
	.077		
	.034		
	.139	-.078	-.341
	.175		
		-5.059	-1.707
	-.212	-1.670	.413
	-1.732	-.534	-.958
	-.702	-.357	-.641
	-.464	-.373	-.492
	-.414	-.352	-.381
	-.492	-.422	-.348
	-.859		
	-1.497		
	-2.257	-3.066	-4.374
	-2.465		
	-1.584	-1.571	-2.193
	-1.059	-1.293	-1.291
	-.850		
	-.661		
	-.568		

ALPHA = 9.98 DEG
EFF. VEL. RATIO = .186

	CP		
Y/YMAX =	0.0	0.5	0.8
X/L			
.130	.140		
.219	.159		
.286	.106		
.331	.057		
.353	.164	-.086	-.342
.375	.199		
.397		-5.062	-1.573
.420	-1.516	-1.867	-3.209
.442	-1.082	-.436	-.894
.464	-.628	-.407	-.587
.487	-.391	-1.111	-.411
.509	-.374	-.362	-.341
.531	-.558	-.509	-.370
.554	-.868		
.576	-1.446		
.598	-2.290	-3.095	-4.485
.620	-2.452		
.643	-1.605	-1.607	-2.266
.665	-1.097	-1.249	-1.430
.687	-.853		
.710	-.721		
.732	-.588		

ALPHA =11.99 DEG
EFF. VEL. RATIO = .186

	CP		
	0.0	0.5	0.8
	.196		
	.198		
	.141		
	.085		
	.184	-.069	-.300
	.191		
		-4.708	-1.447
	-2.680	-1.587	-.583
	-1.832	-.431	-.841
	-.124	-.259	-.567
	-.411	-.104	-.329
	-.370	-.370	-.366
	-.485	-.423	-.354
	-.824		
	-1.475		
	-2.278	-3.113	-4.518
	-2.532		
	-1.602	-1.672	-2.263
	-1.059	-1.200	-1.433
	-.827		
	-.771		
	-.503		

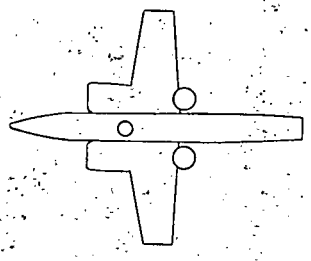
ALPHA =13.97 DEG
EFF. VEL. RATIO = .186

	CP		
	0.0	0.5	0.8
	.218		
	.248		
	.184		
	.102		
	.208	-.040	-.232
	.199		
		-4.928	-1.472
	-1.059	-1.688	-1.108
	-1.513	-.638	-.785
	-.900	-.266	-.507
	-.344	-.119	-.377
	-.369	-.303	-.275
	-.536	-.381	-.315
	-.837		
	-1.461		
	-2.293	-3.105	-4.579
	-2.480		
	-1.559	-1.720	-2.185
	-1.042	-1.098	-1.332
	-.791		
	-.668		
	-.525		

TABLE 23 (c) - CONCLUDED

PRESSURE PROFILE ON FUSELAGE

REAR NOZZLES DEFLECTED 90 DEGREES WITH LIFT JET

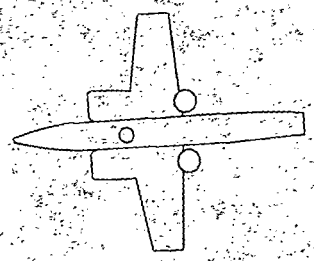


	ALPHA =15.98 DEG EFF. VEL. RATIO = .186			ALPHA =17.98 DEG EFF. VEL. RATIO = .186			ALPHA =19.97 DEG EFF. VEL. RATIO = .185		
	CP			CP			CP		
Y/YMAX =	0.0	0.5	0.8	0.0	0.5	0.8	0.0	0.5	0.8
X/L									
.130	.272			.310			.347		
.219	.283			.319			.358		
.286	.233			.265			.324		
.331	.152			.170			.198		
.353	.237	-.009	-.246	.255	-.017	-.217	.283	.027	-.181
.375	.215			.225			.216		
.397		-5.105	-1.405		-4.970	-1.403		-4.496	-1.394
.420	-1.217	-1.656	-1.258	-1.145	-1.657	-1.190	-1.140	-1.603	-1.058
.442	-1.635	-.359	-.798	-1.465	-.349	-.780	-1.902	-.378	-.726
.464	.215	-.306	-.507	-.349	-.210	-.522	-.316	-.263	-.550
.487	-.265	.227	-.388	-.288	-1.588	-.403	-.292	-1.587	-.357
.509	-.331	-.257	-.318	-.399	-.251	-.333	-.378	-.280	-.353
.531	-.470	-.417	-.351	-.485	-.399	-.321	-.431	-.398	-.312
.554	-.795			-.810			-.787		
.576	-1.451			-1.484			-1.523		
.598	-2.305	-3.132	-4.536	-2.319	-3.159	-4.646	-2.443	-3.244	-4.814
.620	-2.476			-2.559			-2.560		
.643	-1.570	-1.596	-2.266	-1.594	-1.592	-2.234	-1.514	-1.735	-2.321
.665	-.970	-1.187	-1.328	-1.033	-1.161	-1.432	-1.067	-1.204	-1.443
.687	-.805			-.887			-.836		
.710	-.694			-.729			-.699		
.732	-.605			-.602			-.627		

TABLE 23 (D)

PRESSURE PROFILE ON FUSELAGE

REAR NOZZLES DEFLECTED 90 DEGREES WITH LIFT JET



ALPHA = -.02 DEG
EFF. VEL. RATIO = .094

Y/YMAX =	CP 0.0	0.5	0.8
X/L			
.130	-.164		
.219	-.263		
.286	-.329		
.331	-.415	-.567	-.752
.353	-.346		
.375	-.384		
.397		-5.503	-1.930
.420	-3.733	-2.171	.211
.442	-4.666	-1.013	-1.914
.464	.066	-.513	-1.045
.487	-.642	-.545	-.948
.509	-.755	-.497	-.803
.531	-.980	-.916	-.932
.554	-1.261		
.576	-1.660		
.598	-1.664	-1.763	-2.510
.620	-1.244		
.643	-.917	-1.003	-1.145
.665	-.706	-.861	-.947
.687	-.663		
.710	-.481		
.732	-.421		

ALPHA = 3.99 DEG
EFF. VEL. RATIO = .094

	CP 0.0	0.5	0.8
	-.099		
	-.122		
	-.221		
	-.323		
	-.271	-.446	-.581
	-.363		
		-3.679	-2.616
	-.411	-2.150	-2.456
	-3.067	-.604	-1.120
	-2.777	-.491	-.814
	-.475	-.685	-.653
	-.653	-.621	-.717
	-.878	-.814	-.733
	-1.189		
	-1.641		
	-1.826	-2.064	-2.414
	-1.189		
	-.858	-1.017	-1.243
	-.670	-.835	-.865
	-.591		
	-.495		
	-.349		

ALPHA = 7.99 DEG
EFF. VEL. RATIO = .092

	CP 0.0	0.5	0.8
	-.046		
	-.063		
	-.145		
	-.295	-.442	-.823
	-.240		
	-.261		
		-5.109	-1.340
	-1.340	-1.174	-3.017
	.420	-.476	-.776
	-2.386	-.294	-.526
	-.360	.105	-.593
	-.427	-.460	-.526
	-.676	-.676	-.593
	-1.170		
	-1.579		
	-1.682	-2.104	-2.561
	-1.225		
	-.772	-.976	-1.290
	-.659	-.697	-.939
	-.543		
	-.427		
	-.448		

ALPHA =10.01 DEG
EFF. VEL. RATIO = .093

Y/YMAX =	CP 0.0	0.5	0.8
X/L			
.130	.000		
.219	-.027		
.286	-.128		
.331	-.256	-.273	-.617
.353	-.202		
.375	-.262		
.397		-4.108	-2.218
.420	-3.204	-1.331	.099
.442	-.443	-.706	-.739
.464	-.098	-.131	-.558
.487	-.394	-.772	-.443
.509	-.427	-.574	-.492
.531	-.706	-.591	-.706
.554	-1.119		
.576	-1.622		
.598	-1.720	-2.128	-2.559
.620	-1.180		
.643	-.775	-.903	-1.261
.665	-.542	-.687	-.913
.687	-.505		
.710	-.333		
.732	-.273		

ALPHA =11.98 DEG
EFF. VEL. RATIO = .092

	CP 0.0	0.5	0.8
	.022		
	-.049		
	-.072		
	-.238		
	-.187	-.450	-.592
	-.277		
		-4.336	-1.427
	-2.265	-1.247	.052
	-3.005	-.162	-.836
	-1.132	-.129	-.572
	-.408	-.540	-.441
	-.490	-.392	-.638
	-.720	-.655	-.704
	-1.168		
	-1.674		
	-1.856	-2.136	-2.511
	-1.013		
	-.652	-.821	-1.300
	-.608	-.807	-.922
	-.490		
	-.409		
	-.254		

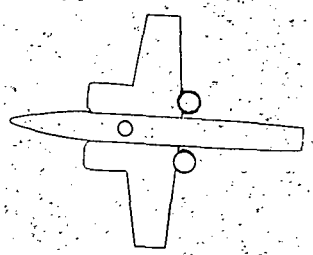
ALPHA =13.99 DEG
EFF. VEL. RATIO = .093

	CP 0.0	0.5	0.8
	.093		
	.043		
	-.018		
	-.163	-.371	-.590
	-.139		
	-.193		
		-3.353	-1.143
	-2.338	-.635	.282
	.495	-.193	-.455
	-.553	-.209	-.684
	-.225	.069	-.389
	-.389	-.471	-.536
	-.700	-.586	-.569
	-1.123		
	-1.655		
	-1.765	-2.091	-2.542
	-1.184		
	-.747	-.885	-1.231
	-.512	-.740	-.888
	-.418		
	-.280		
	-.186		

TABLE 23 (D) - CONCLUDED

PRESSURE PROFILE ON FUSELAGE

REAR NOZZLES DEFLECTED 90 DEGREES WITH LIFT JET

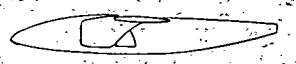


	ALPHA =15.97 DEG EFF. VEL. RATIO = .092			ALPHA =17.97 DEG EFF. VEL. RATIO = .092			ALPHA =19.99 DEG EFF. VEL. RATIO = .092		
		CP			CP			CP	
Y/YMAX =	0.0	0.5	0.8	0.0	0.5	0.8	0.0	0.5	0.8
X/L									
.130	.149			.200			.260		
.219	.116			.153			.200		
.286	.072			.092			.140		
.331	-.117			-.091			-.031		
.353	-.087	-.279	-.407	-.105	-.355	-.331	-.044	-.338	-.362
.375	-.234			-.153			-.141		
.397		-4.171	-1.153		-4.852	-1.439		-3.562	-1.445
.420	-.381	-1.153	.455	-2.808	-1.159	.243	-.434	-.597	-3.090
.442	-2.826	-.447	-.266	.358	.012	-.894	-.451	-.272	-.858
.464	.225	-.299	-.578	-.367	-.120	-.383	-1.461	-.109	-.597
.487	-.234	-.168	-.332	-.268	-.565	-.449	-.386	.135	-.288
.509	-.512	-.414	-.545	-.449	-.285	-.416	-.304	-.239	-.369
.531	-.726	-.594	-.726	-.631	-.614	-.548	-.614	-.548	-.532
.554	-1.167			-1.055			-1.033		
.576	-1.679			-1.519			-1.395		
.598	-1.786	-2.036	-2.531	-1.600	-1.857	-2.372	-1.592	-1.836	-2.381
.620	-1.258			-1.140			-.936		
.643	-.665	-.834	-1.359	-.629	-.788	-.930	-.642	-.559	-.970
.665	-.689	-.722	-.803	-.490	-.581	-.913	-.472	-.639	-.873
.687	-.433			-.459			-.298		
.710	-.352			-.290			-.495		
.732	-.180			-.233			-.238		

TABLE 24 (A)

PRESSURE PROFILE ON NACELLE

FORWARD NOZZLES DEFLECTED 90 DEGREES



ALPHA = .02 DEG
EFF. VEL. RATIO =∞

X/C	CP UPPER	CP LOWER
-.620	.690	.690
-.684	.895	.912
-.716	-.264	-.296
-.684	-.957	-.998
-.593	-.445	
-.501	-.336	
-.380	-.233	-.156
-.268	-.184	-.031
-.148	-.173	.093
-.029	-.241	
.060	-.236	
.250	-.312	
.400	-.403	
.550	-.323	
.700	-.278	
.850	-.260	
.950	-.299	

ALPHA = 4.04 DEG
EFF. VEL. RATIO =∞

X/C	CP UPPER	CP LOWER
	.793	.640
	.990	.797
	-1.041	.386
	-1.126	-.499
	-.583	
	-.443	
	-.308	-.085
	-.229	.023
	-.231	.128
	-.323	
	-.318	
	-.405	
	-.455	
	-.366	
	-.290	
	-.254	
	-.250	

ALPHA = 8.03 DEG
EFF. VEL. RATIO =∞

X/C	CP UPPER	CP LOWER
	.861	.600
	1.013	.653
	-2.032	.763
	-1.594	-.151
	-.763	
	-.555	
	-.389	-.030
	-.294	.045
	-.309	.146
	-.412	
	-.429	
	-.490	
	-.492	
	-.382	
	-.280	
	-.201	
	-.170	

ALPHA = 10.04 DEG
EFF. VEL. RATIO =∞

X/C	CP UPPER	CP LOWER
-.620	.910	.537
-.684	1.008	.553
-.716	-2.676	.865
-.684	-1.907	.026
-.593	-.890	
-.501	-.607	
-.380	-.424	-.002
-.268	-.335	.069
-.148	-.353	.166
-.029	-.473	
.060	-.488	
.250	-.535	
.400	-.518	
.550	-.392	
.700	-.277	
.850	-.194	
.950	-.159	

ALPHA = 12.02 DEG
EFF. VEL. RATIO =∞

X/C	CP UPPER	CP LOWER
	.920	.495
	1.004	.449
	-1.537	.933
	-1.405	.107
	-1.536	
	-.816	
	-.417	.034
	-.313	.084
	-.301	.166
	-.371	
	-.409	
	-.624	
	-.586	
	-.539	
	-.421	
	-.254	
	-.239	

ALPHA = 14.03 DEG
EFF. VEL. RATIO =∞

X/C	CP UPPER	CP LOWER
	.922	.496
	.991	.374
	-1.425	.973
	-1.273	.246
	-1.422	
	-1.107	
	-.500	.033
	-.329	.090
	-.273	.163
	-.314	
	-.429	
	-.536	
	-.614	
	-.668	
	-.661	
	-.512	
	-.416	

ALPHA = 16.02 DEG
EFF. VEL. RATIO =∞

X/C	CP UPPER	CP LOWER
-.620	.933	.506
-.684	.978	.297
-.716	-1.865	.988
-.684	-1.108	.337
-.593	-1.294	
-.501	-.927	
-.380	-.550	.052
-.268	-.388	.102
-.148	-.328	.170
-.029	-.316	
.060	-.471	
.250	-.535	
.400	-.602	
.550	-.660	
.700	-.721	
.850	-.658	
.950	-.485	

ALPHA = 18.02 DEG
EFF. VEL. RATIO =∞

X/C	CP UPPER	CP LOWER
	.957	.489
	.963	.208
	-1.956	.983
	-1.081	.437
	-1.227	
	-.942	
	-.561	.080
	-.412	.120
	-.352	.180
	-.370	
	-.526	
	-.556	
	-.639	
	-.699	
	-.716	
	-.673	
	-.565	

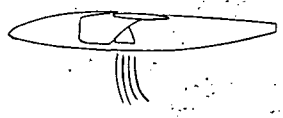
ALPHA = 20.04 DEG
EFF. VEL. RATIO =∞

X/C	CP UPPER	CP LOWER
	.971	.478
	.947	.081
	-2.007	.969
	-1.107	.525
	-1.171	
	-1.031	
	-.600	.106
	-.447	.130
	-.458	.167
	-.436	
	-.574	
	-.577	
	-.641	
	-.710	
	-.750	
	-.685	
	-.595	

TABLE 24.(B)

PRESSURE PROFILE ON NACELLE

FORWARD NOZZLES DEFLECTED 90 DEGREES



ALPHA = .01 DEG
EFF. VEL. RATIO = .302

X/C	CP UPPER	CP LOWER
-.620	-2.463	-2.952
-.684	-1.196	-2.344
-.716	.899	.992
-.684	-.399	.246
-.593	-.343	
-.501	-.305	
-.380	-.232	.189
-.268	-.191	.264
-.148	-.197	.364
-.029	-.298	
.060	-.276	
.250	-.400	
.400	-.492	
.550	-.431	
.700	-.398	
.850	-.431	
.950	-.629	

ALPHA = 4.01 DEG
EFF. VEL. RATIO = .302

X/C	CP UPPER	CP LOWER
-.620	-2.142	-3.260
-.684	-.752	-3.174
-.716	.570	.788
-.684	-.857	.516
-.593	-.522	
-.501	-.427	
-.380	-.322	.226
-.268	-.267	.278
-.148	-.297	.348
-.029	-.426	
.060	-.401	
.250	-.491	
.400	-.580	
.550	-.485	
.700	-.441	
.850	-.437	
.950	-.576	

ALPHA = 8.03 DEG
EFF. VEL. RATIO = .302

X/C	CP UPPER	CP LOWER
-.620	-1.869	-3.501
-.684	-.225	-3.985
-.716	.032	.242
-.684	-1.342	.735
-.593	-.743	
-.501	-.553	
-.380	-.442	.256
-.268	-.370	.295
-.148	-.392	.362
-.029	-.528	
.060	-.539	
.250	-.608	
.400	-.660	
.550	-.553	
.700	-.464	
.850	-.450	
.950	-.528	

ALPHA = 10.02 DEG
EFF. VEL. RATIO = .302

X/C	CP UPPER	CP LOWER
-.620	-1.762	-3.642
-.684	-.101	-4.426
-.716	-.447	-.073
-.684	-1.628	.816
-.593	-.841	
-.501	-.626	
-.380	-.491	.302
-.268	-.421	.308
-.148	-.471	.363
-.029	-.605	
.060	-.616	
.250	-.683	
.400	-.708	
.550	-.592	
.700	-.492	
.850	-.456	
.950	-.526	

ALPHA = 12.02 DEG
EFF. VEL. RATIO = .301

X/C	CP UPPER	CP LOWER
-.620	-1.680	-3.773
-.684	.058	-4.846
-.716	-.593	-.413
-.684	-1.641	.864
-.593	-.902	
-.501	-.715	
-.380	-.554	.314
-.268	-.472	.312
-.148	-.501	.354
-.029	-.671	
.060	-.661	
.250	-.714	
.400	-.737	
.550	-.601	
.700	-.526	
.850	-.474	
.950	-.509	

ALPHA = 14.01 DEG
EFF. VEL. RATIO = .301

X/C	CP UPPER	CP LOWER
-.620	-1.565	-3.833
-.684	.202	-5.236
-.716	-.937	-.985
-.684	-1.624	.911
-.593	-.868	
-.501	-.639	
-.380	-.489	.344
-.268	-.440	.338
-.148	-.418	.383
-.029	-.548	
.060	-.536	
.250	-.915	
.400	-.972	
.550	-.892	
.700	-.789	
.850	-.672	
.950	-.608	

ALPHA = 16.04 DEG
EFF. VEL. RATIO = .302

X/C	CP UPPER	CP LOWER
-.620	-1.480	-3.932
-.684	.327	-5.585
-.716	-1.100	-1.265
-.684	-1.719	.917
-.593	-.868	
-.501	-.637	
-.380	-.523	.361
-.268	-.433	.352
-.148	-.316	.392
-.029	-.440	
.060	-.550	
.250	-.780	
.400	-.934	
.550	-.958	
.700	-.936	
.850	-.815	
.950	-.742	

ALPHA = 18.03 DEG
EFF. VEL. RATIO = .302

X/C	CP UPPER	CP LOWER
-.620	-1.421	-4.070
-.684	.425	-5.971
-.716	-1.451	-1.976
-.684	-1.825	.953
-.593	-.861	
-.501	-.633	
-.380	-.521	.372
-.268	-.455	.363
-.148	-.408	.405
-.029	-.435	
.060	-.650	
.250	-.767	
.400	-.884	
.550	-.912	
.700	-.974	
.850	-.974	
.950	-.867	

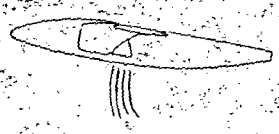
ALPHA = 20.02 DEG
EFF. VEL. RATIO = .302

X/C	CP UPPER	CP LOWER
-.620	-1.327	-4.162
-.684	.495	-6.192
-.716	-1.765	-2.554
-.684	-1.915	.941
-.593	-.866	
-.501	-.632	
-.380	-.558	.394
-.268	-.514	.375
-.148	-.421	.394
-.029	-.464	
.060	-.698	
.250	-.707	
.400	-.867	
.550	-.905	
.700	-.971	
.850	-.967	
.950	-.906	

TABLE 24 (C)

PRESSURE PROFILE ON NACELLE

FORWARD NOZZLES DEFLECTED 90 DEGREES



ALPHA = .00 DEG
EFF. VEL. RATIO = .203

X/C	CP UPPER	CP LOWER
-.020	-6.695	-7.494
-.034	-4.999	-6.467
-.716	.930	-.552
-.034	.009	-.390
-.593	-.282	
-.501	-.232	
-.380	-.233	
-.268	-.196	.020
-.148	-.217	.080
-.029	-.301	.176
.060	-.243	
.250	-.354	
.400	-.465	
.550	-.413	
.700	-.385	
.850	-.445	
.950	-.697	

ALPHA = 4.00 DEG
EFF. VEL. RATIO = .202

CP UPPER	CP LOWER
-6.574	-8.012
-4.413	-7.648
-.585	-.160
-.422	-.621
-.418	
-.362	
-.308	
-.262	.035
-.276	.058
-.362	.149
-.390	
-.453	
-.523	
-.449	
-.402	
-.428	
-.594	

ALPHA = 7.98 DEG
EFF. VEL. RATIO = .201

CP UPPER	CP LOWER
-6.192	-8.454
-3.435	-8.829
.944	-1.061
-.716	.890
-.586	
-.493	
-.361	
-.314	.092
-.347	.082
-.479	.152
-.481	
-.532	
-.576	
-.479	
-.408	
-.408	
-.547	

ALPHA = 10.02 DEG
EFF. VEL. RATIO = .202

X/C	CP UPPER	CP LOWER
-.020	-5.895	-9.013
-.084	-3.071	-9.454
-.716	.850	-1.454
-.084	.943	.879
-.593	-.678	
-.501	-.532	
-.380	-.416	
-.268	-.362	.121
-.148	-.380	.090
-.029	-.546	.144
.060	-.596	
.250	-.588	
.400	-.607	
.550	-.495	
.700	-.405	
.850	-.406	
.950	-.507	

ALPHA = 12.02 DEG
EFF. VEL. RATIO = .202

CP UPPER	CP LOWER
-5.731	-8.837
-2.707	-10.274
.684	-2.230
-1.130	.926
-.730	
-.629	
-.480	
-.403	.144
-.436	.100
-.562	.121
-.557	
-.593	
-.623	
-.502	
-.399	
-.376	
-.481	

ALPHA = 13.98 DEG
EFF. VEL. RATIO = .202

CP UPPER	CP LOWER
-5.588	-9.012
-2.298	-10.988
.478	-2.915
-1.358	.956
-.833	
-.667	
-.521	
-.460	.149
-.479	.093
-.632	.112
-.607	
-.643	
-.654	
-.514	
-.403	
-.352	
-.424	

ALPHA = 16.01 DEG
EFF. VEL. RATIO = .202

X/C	CP UPPER	CP LOWER
-.020	-5.403	-9.147
-.084	-2.091	-11.344
-.716	.338	-3.633
-.084	-1.502	.948
-.593	-.878	
-.501	-.716	
-.380	-.578	
-.268	-.503	.183
-.148	-.513	.121
-.029	-.646	.095
.060	-.644	
.250	-.646	
.400	-.653	
.550	-.510	
.700	-.404	
.850	-.352	
.930	-.410	

ALPHA = 17.99 DEG
EFF. VEL. RATIO = .202

CP UPPER	CP LOWER
-5.188	-9.204
-1.852	-11.908
.133	-4.250
-1.644	.950
-.887	
-.699	
-.585	
-.494	.212
-.486	.119
-.605	.091
-.635	
-.763	
-.759	
-.586	
-.515	
-.414	
-.436	

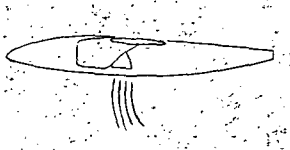
ALPHA = 20.02 DEG
EFF. VEL. RATIO = .202

CP UPPER	CP LOWER
-5.123	-9.466
-1.661	-12.531
.032	-5.080
-1.745	.930
-.852	
-.662	
-.500	
-.469	.216
-.420	.132
-.480	.095
-.571	
-.882	
-.874	
-.918	
-.877	
-.678	
-.583	

TABLE 24 (D)

PRESSURE PROFILE ON NACELLE

FORWARD NOZZLES DEFLECTED 90 DEGREES



	ALPHA = -.01 DEG EFF. VEL. RATIO = .100		ALPHA = 4.02 DEG EFF. VEL. RATIO = .100		ALPHA = 8.00 DEG EFF. VEL. RATIO = .100	
	CP		CP		CP	
X/C	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
-.620	-31.365	-32.025	-30.591	-32.804	-30.219	-33.699
-.684	-27.407	-29.336	-25.778	-31.093	-24.301	-33.300
-.716	-4.313	-4.901	-3.118	-6.723	-1.890	-8.529
-.684	.943	.964	.805	1.066	.712	1.126
-.593	.269		.169		.099	
-.501	.064		.027		-.043	
-.380	.042	-.085	-.001	-.057	-.036	-.015
-.268	.035	-.085	.013	-.079	-.022	-.036
-.148	.007	.007	-.036	.013	-.036	-.015
-.029	-.050		-.093		-.129	
.060	-.043		-.079		-.122	
.250	-.181		-.209		-.224	
.400	-.297		-.282		-.298	
.550	-.280		-.235		-.229	
.700	-.263		-.225		-.188	
.850	-.369		-.276		-.243	
.950	-.679		-.544		-.480	

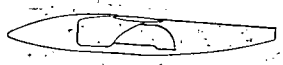
	ALPHA = 10.01 DEG EFF. VEL. RATIO = .100		ALPHA = 11.99 DEG EFF. VEL. RATIO = .100		ALPHA = 14.02 DEG EFF. VEL. RATIO = .100	
	CP		CP		CP	
X/C	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
-.620	-29.712	-34.190	-29.166	-34.145	-28.628	-34.441
-.684	-23.740	-34.476	-22.796	-35.131	-21.562	-35.864
-.716	-1.404	-9.483	-1.078	-10.398	-.391	-11.635
-.684	.589	1.161	.554	1.178	.416	1.230
-.593	-.025		-.023		-.101	
-.501	-.097		-.106		-.143	
-.380	-.054	.039	-.085	.093	-.115	.098
-.268	-.032	-.025	-.014	-.014	-.065	.041
-.148	-.089	.011	-.049	.022	-.087	.027
-.029	-.154		-.149		-.179	
.060	-.161		-.170		-.193	
.250	-.242		-.222		-.237	
.400	-.286		-.259		-.272	
.550	-.214		-.190		-.190	
.700	-.166		-.143		-.122	
.850	-.183		-.160		-.146	
.950	-.451		-.340		-.337	

	ALPHA = 15.98 DEG EFF. VEL. RATIO = .100		ALPHA = 18.00 DEG EFF. VEL. RATIO = .101		ALPHA = 19.99 DEG EFF. VEL. RATIO = .101	
	CP		CP		CP	
X/C	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
-.620	-28.359	-34.771	-27.848	-34.886	-27.390	-35.028
-.684	-21.082	-37.474	-19.881	-38.240	-19.073	-39.127
-.716	-.080	-12.854	.150	-13.898	.477	-15.240
-.684	.346	1.197	.179	1.163	.092	1.127
-.593	-.151		-.236		-.293	
-.501	-.179		-.229		-.272	
-.380	-.130	.140	-.159	.179	-.195	.197
-.268	-.073	.041	-.110	.059	-.125	.064
-.148	-.087	.041	-.138	.045	-.146	.036
-.029	-.222		-.264		-.279	
.060	-.222		-.250		-.244	
.250	-.252		-.266		-.273	
.400	-.276		-.292		-.276	
.550	-.143		-.174		-.161	
.700	-.092		-.089		-.064	
.850	-.068		-.052		-.050	
.950	-.286		-.258		-.205	

TABLE 25.(A)

PRESSURE PROFILE ON NACELLE

REAR NOZZLES DEFLECTED 90 DEGREES



	ALPHA = -.00 DEG EFF. VEL. RATIO = ∞		ALPHA = 3.99 DEG EFF. VEL. RATIO = ∞		ALPHA = 8.05 DEG EFF. VEL. RATIO = ∞	
	CP		CP		CP	
X/C	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
-.620	.570	.579	.666	.506	.753	.427
-.684	.822	.355	.948	.699	.998	.456
-.716	.981	-.141	-.671	.309	-1.636	.760
-.684	-.820	-1.155	-1.037	-.620	-1.518	-.170
-.593	-.366		-.555		-.744	
-.501	-.309		-.414		-.542	
-.380	-.234	-.319	-.298	-.256	-.387	-.180
-.288	-.133	-.258	-.224	-.207	-.297	-.147
-.196	-.157	-.264	-.226	-.225	-.303	-.175
-.029	-.195		-.283		-.413	
.060	-.197		-.313		-.440	
.250	-.231		-.376		-.485	
.400	-.343		-.430		-.492	
.550	-.275		-.325		-.368	
.700	-.204		-.233		-.258	
.850	-.146		-.150		-.163	
.950	-.107		-.088		-.100	

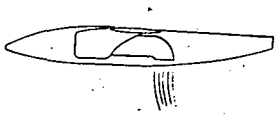
	ALPHA = 10.00 DEG EFF. VEL. RATIO = ∞		ALPHA = 11.99 DEG EFF. VEL. RATIO = ∞		ALPHA = 14.04 DEG EFF. VEL. RATIO = ∞	
	CP		CP		CP	
X/C	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
-.620	.795	.354	.829	.318	.833	.314
-.684	1.002	.520	.998	.251	.997	.177
-.716	-2.210	.373	-1.473	.943	-1.407	.970
-.684	-1.822	-.028	-1.448	.087	-1.293	.237
-.593	-.827		-1.415		-1.480	
-.501	-.596		-.714		-1.016	
-.380	-.422	-.137	-.385	-.108	-.435	-.089
-.288	-.340	-.115	-.304	-.095	-.305	-.084
-.196	-.391	-.134	-.280	-.138	-.248	-.129
-.029	-.448		-.356		-.278	
.060	-.491		-.459		-.400	
.250	-.537		-.655		-.551	
.400	-.515		-.683		-.633	
.550	-.365		-.553		-.654	
.700	-.269		-.471		-.678	
.850	-.169		-.273		-.673	
.950	-.109		-.184		-.550	

	ALPHA = 16.01 DEG EFF. VEL. RATIO = ∞		ALPHA = 18.02 DEG EFF. VEL. RATIO = ∞		ALPHA = 20.02 DEG EFF. VEL. RATIO = ∞	
	CP		CP		CP	
X/C	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
-.620	.869	.314	.879	.291	.896	.272
-.684	.991	.038	.972	-.060	.959	-.195
-.716	-1.339	.982	-1.574	.962	-1.986	.917
-.684	-1.115	.332	-1.155	.480	-1.119	.583
-.593	-1.358		-1.297		-1.208	
-.501	-.953		-.893		-1.015	
-.380	-.488	-.049	-.513	-.017	-.550	.023
-.288	-.389	-.036	-.399	-.041	-.480	-.002
-.196	-.347	-.104	-.335	-.088	-.446	-.050
-.029	-.340		-.404		-.536	
.060	-.522		-.576		-.612	
.250	-.599		-.591		-.586	
.400	-.624		-.642		-.665	
.550	-.676		-.682		-.722	
.700	-.701		-.725		-.753	
.850	-.723		-.753		-.781	
.950	-.670		-.576		-.745	

TABLE 25 (B)

PRESSURE PROFILE ON NACELLE

REAR NOZZLES DEFLECTED 90 DEGREES



ALPHA = .00 DEG
EFF. VEL. RATIO = .294

ALPHA = 4.02 DEG
EFF. VEL. RATIO = .294

ALPHA = 8.04 DEG
EFF. VEL. RATIO = .294

	CP		CP		CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
X/C						
-.020	-3.114	-3.348	-2.881	-3.616	-2.595	-3.983
-.084	-1.973	-2.218	-1.993	-3.026	-.872	-3.953
-.716	.988	.994	.793	.925	.430	.586
-.684	-.228	-.039	-.614	.232	-1.072	.492
-.593	-.272		-.436		-.595	
-.501	-.239		-.375		-.511	
-.380	-.212	-.101	-.293	-.094	-.386	-.031
-.268	-.155	-.128	-.233	-.074	-.330	-.041
-.148	-.171	-.179	-.266	-.126	-.376	-.090
-.029	-.261		-.364		-.497	
.060	-.283		-.403		-.560	
.250	-.383		-.502		-.633	
.400	-.483		-.581		-.659	
.550	-.427		-.491		-.558	
.700	-.384		-.426		-.469	
.850	-.373		-.388		-.426	
.950	-.371		-.401		-.431	

ALPHA = 10.02 DEG
EFF. VEL. RATIO = .294

ALPHA = 12.02 DEG
EFF. VEL. RATIO = .294

ALPHA = 14.01 DEG
EFF. VEL. RATIO = .294

	CP		CP		CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
X/C						
-.020	-2.943	-4.116	-2.298	-4.224	-2.196	-4.334
-.084	-.659	-4.242	-.472	-4.565	-.279	-4.833
-.716	.253	.344	.010	.120	-.279	-.273
-.684	-1.256	.330	-1.530	.710	-1.532	.786
-.593	-.684		-.695		-.723	
-.501	-.361		-.541		-.543	
-.380	-.420	-.001	-.385	.044	-.381	.081
-.268	-.364	-.021	-.298	.024	-.278	.052
-.148	-.413	-.075	-.291	-.039	-.270	-.022
-.029	-.570		-.419		-.347	
.060	-.628		-.602		-.569	
.250	-.681		-.843		-.805	
.400	-.714		-.908		-.879	
.550	-.578		-.872		-.901	
.700	-.491		-.800		-.921	
.850	-.448		-.722		-.884	
.950	-.442		-.624		-.721	

ALPHA = 16.01 DEG
EFF. VEL. RATIO = .294

ALPHA = 18.01 DEG
EFF. VEL. RATIO = .293

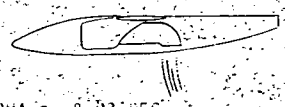
ALPHA = 20.00 DEG
EFF. VEL. RATIO = .293

	CP		CP		CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
X/C						
-.020	-2.113	-4.445	-1.991	-4.631	-1.914	-4.656
-.084	-.159	-5.212	-.028	-5.351	.132	-6.036
-.716	-.557	-.642	-.822	-.979	-1.218	-1.620
-.684	-1.441	.839	-1.526	.883	-1.655	.927
-.593	-.746		-.769		-.809	
-.501	-.556		-.579		-.582	
-.380	-.415	.127	-.442	.142	-.468	.204
-.268	-.316	.074	-.346	.091	-.390	.133
-.146	-.272	-.008	-.298	.010	-.328	.041
-.029	-.383		-.349		-.473	
.060	-.674		-.762		-.795	
.250	-.775		-.797		-.802	
.400	-.834		-.846		-.846	
.550	-.876		-.884		-.899	
.700	-.927		-.946		-.922	
.850	-.961		-.968		-.976	
.950	-.941		-.982		-1.003	

TABLE 25 (C)

PRESSURE PROFILE ON NACELLE

REAR NOZZLES DEFLECTED 90 DEGREES



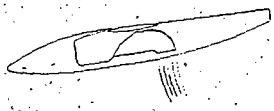
	ALPHA = .01 DEG EFF. VEL. RATIO = .196		ALPHA = 4.00 DEG EFF. VEL. RATIO = .196		ALPHA = 8.02 DEG EFF. VEL. RATIO = .196	
	CP		CP		CP	
X/C	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
-.620	-8.321	-8.037	-7.875	-9.044	-7.472	-9.607
-.684	-6.152	-7.129	-5.388	-8.220	-4.528	-9.209
-.716	.734	.513	.921	.065	.960	-.772
-.684	.160	.312	-.203	.506	-.526	.709
-.593	-.220		-.374		-.532	
-.501	-.259		-.387		-.481	
-.380	-.195	-.150	-.308	-.132	-.402	-.051
-.268	-.157	-.140	-.285	-.130	-.334	-.093
-.148	-.183	-.215	-.306	-.215	-.402	-.185
-.029	-.281		-.402		-.542	
.060	-.389		-.463		-.588	
.250	-.432		-.550		-.685	
.400	-.554		-.640		-.753	
.550	-.493		-.561		-.652	
.700	-.454		-.507		-.581	
.850	-.486		-.515		-.567	
.950	-.544		-.571		-.616	

	ALPHA = 10.00 DEG EFF. VEL. RATIO = .195		ALPHA = 12.02 DEG EFF. VEL. RATIO = .196		ALPHA = 14.00 DEG EFF. VEL. RATIO = .196	
	CP		CP		CP	
X/C	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
-.620	-7.299	-9.645	-7.139	-10.052	-6.941	-10.251
-.684	-4.209	-9.907	-3.793	-10.393	-3.566	-10.835
-.716	.908	-1.298	.842	-1.830	.777	-2.479
-.684	.755	.778	-.893	.828	-1.031	.873
-.593	-.634		-.564		-.685	
-.501	-.561		-.566		-.550	
-.380	-.429	-.018	-.438	-.002	-.426	.035
-.268	-.392	-.063	-.383	-.050	-.335	-.030
-.148	-.447	-.159	-.397	-.159	-.359	-.132
-.029	-.608		-.526		-.464	
.060	-.657		-.770		-.683	
.250	-.750		-1.006		-1.067	
.400	-.798		-.895		-1.069	
.550	-.682		-.785		-1.057	
.700	-.601		-.726		-.963	
.850	-.562		-.641		-.878	
.950	-.635		-.674		-.762	

	ALPHA = 15.98 DEG EFF. VEL. RATIO = .197		ALPHA = 18.03 DEG EFF. VEL. RATIO = .197		ALPHA = 19.99 DEG EFF. VEL. RATIO = .196	
	CP		CP		CP	
X/C	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
-.620	-6.766	-10.384	-6.549	-10.551	-6.472	-10.746
-.684	-3.241	-11.430	-2.959	-11.848	-2.693	-12.639
-.716	.649	-3.159	.481	-3.891	.312	-4.632
-.684	-1.155	.891	-1.232	.909	-1.383	.897
-.593	-.693		-.709		-.756	
-.501	-.567		-.583		-.608	
-.380	-.440	.072	-.449	.111	-.482	.146
-.268	-.345	-.003	-.354	.029	-.384	.057
-.148	-.358	-.115	-.338	-.076	-.358	-.072
-.029	-.457		-.451		-.486	
.060	-.719		-.804		-.882	
.250	-.931		-.864		-.954	
.400	-1.051		-.980		-1.006	
.550	-1.079		-1.045		-1.051	
.700	-1.084		-1.017		-1.126	
.850	-1.067		-1.051		-1.167	
.950	-.992		-1.043		-1.111	

TABLE 25.(D)

PRESSURE PROFILE ON NACELLE



REAR NOZZLES DEFLECTED 90 DEGREES

ALPHA = .00 DEG
EFF. VEL. RATIO = .098

X/C	CP UPPER	CP LOWER
-.620	-36.381	-37.195
-.684	-31.041	-33.312
-.716	-9.894	-8.814
-.684	.666	.701
-.593	.053	
-.501	-.157	
-.380	-.227	-.340
-.268	-.241	-.390
-.148	-.241	-.422
-.029	-.335	
.060	-.347	
.250	-.352	
.500	-.655	
.650	-.622	
.700	-.615	
.850	-.646	
.950	-.733	

ALPHA = 4.99 DEG
EFF. VEL. RATIO = .098

CP UPPER	CP LOWER
-35.659	-36.029
-29.318	-35.470
-9.155	-8.407
.539	.729
-.129	
-.283	
-.319	-.269
-.326	-.375
-.396	-.487
-.407	
-.529	
-.691	
-.689	
-.645	
-.649	
-.696	
-.704	

ALPHA = 7.98 DEG
EFF. VEL. RATIO = .097

CP UPPER	CP LOWER
-35.258	-39.173
-28.253	-37.645
-3.629	-11.295
.414	.740
-.247	
-.389	
-.418	-.219
-.396	-.325
-.446	-.475
-.595	
-.610	
-.739	
-.806	
-.724	
-.697	
-.707	
-.813	

ALPHA = 10.04 DEG
EFF. VEL. RATIO = .098

X/C	CP UPPER	CP LOWER
-.620	-34.630	-39.421
-.684	-27.237	-36.490
-.716	-6.691	-11.940
-.684	.317	.726
-.593	.023	
-.501	-.457	
-.380	-.429	-.258
-.268	-.422	-.316
-.148	-.492	-.471
-.029	-.633	
.060	-.607	
.250	-.829	
.500	-.845	
.650	-.767	
.700	-.728	
.850	-.745	
.950	-.852	

ALPHA = 11.97 DEG
EFF. VEL. RATIO = .098

CP UPPER	CP LOWER
-33.928	-39.572
-25.830	-38.803
-2.345	-13.137
.138	.740
-.463	
-.654	
-.540	-.239
-.512	-.391
-.575	-.512
-.722	
-.797	
-.911	
-.939	
-.848	
-.767	
-.801	
-.892	

ALPHA = 14.01 DEG
EFF. VEL. RATIO = .098

CP UPPER	CP LOWER
-33.645	-40.120
-25.384	-40.809
-1.931	-14.677
-.005	.586
-.525	
-.610	
-.581	-.251
-.581	-.371
-.638	-.518
-.835	
-.919	
-1.045	
-1.032	
-.928	
-.823	
-.820	
-.911	

ALPHA = 15.03 DEG
EFF. VEL. RATIO = .097

X/C	CP UPPER	CP LOWER
-.620	-33.723	-40.303
-.684	-24.520	-41.061
-.716	-1.564	-16.316
-.684	-.052	.744
-.593	-.520	
-.501	-.593	
-.380	-.563	-.187
-.268	-.527	-.307
-.148	-.512	-.393
-.029	-.818	
.060	-.960	
.250	-1.207	
.400	-1.243	
.550	-1.030	
.740	-1.095	
.850	-.924	
.950	-.941	

ALPHA = 18.00 DEG
EFF. VEL. RATIO = .098

CP UPPER	CP LOWER
-33.240	-40.852
-23.006	-42.503
-1.119	-17.077
-.103	.483
-.576	
-.611	
-.562	-.181
-.533	-.329
-.547	-.477
-.703	
-.907	
-1.293	
-1.321	
-1.304	
-1.155	
-1.053	
-1.148	

ALPHA = 20.00 DEG
EFF. VEL. RATIO = .098

CP UPPER	CP LOWER
-32.831	-41.177
-23.359	-43.142
-.550	-18.662
-.203	.367
-.620	
-.640	
-.591	-.154
-.527	-.302
-.548	-.457
-.696	
-.914	
-1.174	
-1.230	
-1.382	
-1.315	
-1.264	
-1.155	

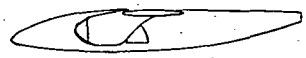


TABLE 26

PRESSURE PROFILE ON NACELLE

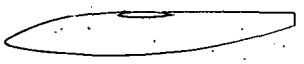
FORWARD NOZZLES DEFLECTED 90 DEGREES WITH INLETS CLOSED

	ALPHA = -.03 DEG EFF. VEL. RATIO =∞		ALPHA = 3.99 DEG EFF. VEL. RATIO =∞		ALPHA = 8.03 DEG EFF. VEL. RATIO =∞	
	CP		CP		CP	
X/C	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
-.620						
-.684						
-.716	-.190	.689	-.126	.574	-.057	.517
-.684	.005	-.106	-.290	.004	-.427	.100
-.593	-.336		-.456		-.542	
-.501	-.301		-.380		-.460	
-.380	-.216	-.350	-.277	-.295	-.331	-.267
-.268	-.154	-.265	-.208	-.257	-.256	-.218
-.148	-.155	-.246	-.206	-.241	-.266	-.207
-.029	-.209		-.291		-.358	
.060	-.179		-.245		-.340	
.250	-.260		-.321		-.405	
.400	-.331		-.357		-.414	
.550	-.263		-.256		-.279	
.700	-.201		-.163		-.159	
.850	-.162		-.092		-.059	
.950	-.154		-.069		-.001	

	ALPHA = 10.02 DEG EFF. VEL. RATIO =∞		ALPHA = 12.00 DEG EFF. VEL. RATIO =∞	
	CP		CP	
X/C	UPPER	LOWER	UPPER	LOWER
-.620				
-.684				
-.716	-.002	.475	.042	.418
-.684	-.415	.160	-.493	.182
-.593	-.603		-.641	
-.501	-.522		-.569	
-.380	-.384	-.240	-.451	-.217
-.268	-.298	-.190	-.369	-.175
-.148	-.308	-.183	-.351	-.172
-.029	-.417		-.453	
.060	-.392		-.440	
.250	-.455		-.480	
.400	-.440		-.446	
.550	-.307		-.305	
.700	-.185		-.176	
.850	-.079		-.064	
.950	-.006		.009	

TABLE 27.(A)
PRESSURE PROFILE ON WING

LIFT JET ALONE



BETA = -11.96 DEG Q = 2850 N/SQ.M. (59.52 LBF/SQ.FT.) EFFECT. VELOC. RATIO = ∞

	ETA = .250		ETA = .387		ETA = .524		ETA = .800	
X/C	CP		CP		CP		CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000	.98		.94		1.00		1.02	
.010	-.88	.65	.53	.26				
.015					-.20	.03		
.025	-.917	.246	-.509	.095	-.329	-.022	-.184	-.234
.050	-.561	.158	-.365	-.043	-.234	-.159	-.219	-.165
.100	-.428	-.011	-.328	-.131	-.294	-.120	-.205	-.145
.150	-.389	-.061	-.355	-.179	-.300	-.210	-.229	-.218
.250	-.370	-.198	-.354	-.230	-.303	-.289	-.311	-.287
.400	-.309	-.216	-.338	-.250	-.293	-.257	-.261	-.273
.550	-.209	-.171	-.214	-.174	-.241	-.189	-.202	-.207
.700	-.118	-.093	-.097	-.094	-.129	-.123	-.120	-.127
.850	-.022	.004	-.022	-.002	.010	.004	.012	-.003
.950	.056	.076	.052	.076	.061	.072	.057	.079

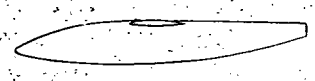
BETA = -7.93 DEG Q = 2844 N/SQ.M. (59.41 LBF/SQ.FT.) EFFECT. VELOC. RATIO = ∞

	ETA = .250		ETA = .387		ETA = .524		ETA = .800	
X/C	CP		CP		CP		CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000	1.01		.91		.97		1.01	
.010	-.59	.41	.56	.18				
.015					-.16	-.10		
.025	-.585	.076	-.445	-.120	-.341	-.169	-.159	-.233
.050	-.473	-.009	-.418	-.121	-.238	-.165	-.188	-.185
.100	-.445	-.103	-.284	-.151	-.241	-.175	-.200	-.228
.150	-.357	-.215	-.332	-.234	-.331	-.279	-.238	-.269
.250	-.316	-.253	-.311	-.266	-.315	-.317	-.294	-.324
.400	-.298	-.284	-.293	-.279	-.307	-.303	-.275	-.295
.550	-.192	-.209	-.210	-.213	-.251	-.206	-.205	-.218
.700	-.098	-.117	-.094	-.107	-.126	-.136	-.126	-.147
.850	-.016	-.016	-.018	-.016	.001	-.017	.000	-.013
.950	.066	.070	.053	.077	.055	.060	.050	.074

BETA = -3.96 DEG Q = 2846 N/SQ.M. (59.43 LBF/SQ.FT.) EFFECT. VELOC. RATIO = ∞

	ETA = .250		ETA = .387		ETA = .524		ETA = .800	
X/C	CP		CP		CP		CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000	1.00		.73		.96		.99	
.010	-.15	.24	.58	.22				
.015					-.09	-.23		
.025	-.495	-.101	-.253	-.204	-.255	-.240	-.270	-.459
.050	-.372	-.109	-.347	-.166	-.245	-.114	-.235	-.188
.100	-.310	-.196	-.229	-.255	-.209	-.241	-.201	-.248
.150	-.297	-.268	-.274	-.261	-.293	-.288	-.245	-.295
.250	-.318	-.332	-.289	-.328	-.287	-.322	-.308	-.303
.400	-.239	-.315	-.306	-.332	-.279	-.324	-.297	-.296
.550	-.191	-.241	-.207	-.243	-.255	-.223	-.188	-.222
.700	-.104	-.120	-.109	-.125	-.116	-.125	-.115	-.144
.850	-.024	-.035	-.019	-.025	.006	-.024	.001	-.025
.950	.062	.056	.052	.069	.050	.064	.061	.076

TABLE 27 (A) Continued
PRESSURE PROFILE ON WING



LIFT JET ALONE

BETA = .04 DEG Q = 2842 N/SQ.M. (59.37 LBF/SQ.FT.) EFFECT. VELOC. RATIO = ∞

	ETA = .250		ETA = .387		ETA = .524		ETA = .800	
X/C	CP		CP		CP		CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000	.97		.68		.89		.98	
.010	-.01	-.14	.59	-.07				
.015					-.16	-.28		
.025	-.361	-.227	-.282	-.274	-.174	-.274	-.269	-.248
.050	-.278	-.212	-.203	-.248	-.288	-.231	-.167	-.168
.100	-.262	-.274	-.266	-.297	-.193	-.235	-.203	-.244
.150	-.317	-.315	-.249	-.287	-.291	-.299	-.208	-.320
.250	-.273	-.321	-.295	-.307	-.270	-.311	-.334	-.339
.400	-.281	-.357	-.304	-.346	-.283	-.334	-.288	-.301
.550	-.183	-.275	-.198	-.242	-.255	-.230	-.197	-.234
.700	-.100	-.120	-.095	-.115	-.122	-.122	-.107	-.137
.850	-.022	-.039	-.016	-.031	-.001	-.029	.006	-.016
.950	.065	.058	.056	.067	.063	.063	.062	.080

BETA = 4.02 DEG Q = 2841 N/SQ.M. (59.34 LBF/SQ.FT.) EFFECT. VELOC. RATIO = ∞

	ETA = .250		ETA = .387		ETA = .524		ETA = .800	
X/C	CP		CP		CP		CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000	.89		.49		.85		.94	
.010	.13	-.30	.57	-.30				
.015					-.03	-.21		
.025	-.174	-.298	-.235	-.372	-.299	-.421	-.090	-.273
.050	-.260	-.376	-.188	-.239	-.123	-.164	-.128	-.273
.100	-.201	-.319	-.237	-.268	-.161	-.350	-.202	-.244
.150	-.234	-.366	-.243	-.285	-.261	-.304	-.227	-.298
.250	-.236	-.392	-.278	-.326	-.223	-.379	-.294	-.361
.400	-.274	-.381	-.284	-.351	-.289	-.346	-.297	-.307
.550	-.175	-.277	-.185	-.250	-.256	-.233	-.172	-.233
.700	-.098	-.115	-.101	-.122	-.112	-.127	-.108	-.146
.850	-.024	-.041	-.016	-.031	.007	-.024	.007	-.019
.950	.064	.059	.061	.070	.067	.065	.063	.076

BETA = 8.04 DEG Q = 2842 N/SQ.M. (59.35 LBF/SQ.FT.) EFFECT. VELOC. RATIO = ∞

	ETA = .250		ETA = .387		ETA = .524		ETA = .800	
X/C	CP		CP		CP		CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000	.66		.33		.84		.86	
.010	.19	-.52	.57	-.14				
.015					.10	-.60		
.025	-.030	-.740	-.061	-.443	-.110	-.283	-.148	-.275
.050	-.087	-.360	-.121	-.317	-.193	-.356	-.091	-.316
.100	-.117	-.429	-.111	-.288	-.084	-.295	-.227	-.185
.150	-.137	-.462	-.217	-.419	-.239	-.359	-.161	-.252
.250	-.240	-.455	-.240	-.402	-.282	-.346	-.255	-.374
.400	-.234	-.358	-.237	-.355	-.274	-.378	-.252	-.292
.550	-.175	-.280	-.168	-.257	-.257	-.240	-.186	-.252
.700	-.092	-.141	-.086	-.109	-.101	-.125	-.115	-.138
.850	-.015	-.034	-.009	-.033	.012	-.042	.017	-.013
.950	.069	.063	.062	.072	.070	.062	.065	.078

TABLE 27 (A) Concluded

PRESSURE PROFILE ON WING

LIFT JET ALONE

BETA = 12.04 DEG Q = 2843 N/SQ.M. (59.38 LBF/SQ.FT.) EFFECT. VELOC. RATIO = ∞

X/C	ETA = .250 CP		ETA = .387 CP		ETA = .524 CP		ETA = .800 CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000	.50		.33		.73		.86	
.010	.39		.55					
.015		-.94		-.39				
.025	.103		-.105		.21	-.57	-.105	-.411
.050	.087	-.821	-.051	-.559	-.053	-.446	-.138	-.309
.100	-.061	-.518	-.083	-.403	-.028	-.395	-.216	-.254
.150	-.105	-.459	-.136	-.380	-.130	.352	-.188	-.390
.250	-.206	-.490	-.223	-.438	-.147	.369	-.238	-.342
.400	-.225	-.464	-.232	-.403	-.137	.403	-.229	-.290
.550	-.162	-.376	-.166	-.386	-.234	.343	-.165	-.236
.700	-.092	-.276	-.082	-.258	-.255	.266	-.106	-.145
.850	-.015	-.139	-.009	-.113	-.089	.119	.021	-.017
.950	.060	-.036	.066	-.026	.013	.031	.074	.076
		.060		.077	.071	.064		

TABLE 27 (B)
PRESSURE PROFILE ON WING

LIFT JET ALONE

BETA = -11.99 DEG Q = 2846 N/SQ.M. (59.43 LBF/SQ.FT.) EFFECT. VELOC. RATIO = .317

X/C	ETA = .250 CP		ETA = .387 CP		ETA = .524 CP		ETA = .800 CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000	.91		.99		1.00		1.00	
.010	-.82	.64	.41	.44				
.015					-.42	-.06		
.025	-1.008	.210	-.648	.006	-.392	-.098	-.268	-.127
.050	-.570	.088	-.382	-.030	-.336	-.093	-.254	-.181
.100	-.436	.063	-.365	-.130	-.327	-.193	-.211	-.157
.150	-.543	-.136	-.405	-.181	-.318	-.246	-.249	-.251
.250	-.406	-.168	-.399	-.251	-.350	-.282	-.316	-.292
.400	-.343	-.266	-.351	-.279	-.312	-.280	-.299	-.288
.550	-.224	-.218	-.239	-.219	-.164	-.221	-.207	-.217
.700	-.136	-.102	-.127	-.104	-.139	-.136	-.130	-.136
.850	-.041	-.032	-.038	-.026	-.018	-.028	-.012	-.020
.950	.047	.054	.035	.058	.045	.048	.042	.055

BETA = -7.99 DEG Q = 2841 N/SQ.M. (59.34 LBF/SQ.FT.) EFFECT. VELOC. RATIO = .317

X/C	ETA = .250 CP		ETA = .387 CP		ETA = .524 CP		ETA = .800 CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000	.99		.90		1.00		1.01	
.010	-.57	.37	.46	.21				
.015					-.26	-.15		
.025	-.639	.033	-.456	.048	-.360	-.089	-.313	-.276
.050	-.421	-.032	-.435	-.101	-.363	-.140	-.245	-.209
.100	-.394	-.162	-.297	-.182	-.259	-.220	-.215	-.154
.150	-.380	-.212	-.379	-.224	-.350	-.240	-.332	-.283
.250	-.368	-.317	-.348	-.315	-.324	-.297	-.335	-.319
.400	-.318	-.325	-.340	-.312	-.318	-.324	-.304	-.312
.550	-.215	-.261	-.228	-.247	-.175	-.226	-.209	-.212
.700	-.122	-.117	-.118	-.123	-.126	-.132	-.119	-.145
.850	-.036	-.044	-.033	-.032	-.012	-.037	-.006	-.036
.950	.049	.049	.038	.059	.048	.049	.040	.054

BETA = -3.97 DEG Q = 2849 N/SQ.M. (59.50 LBF/SQ.FT.) EFFECT. VELOC. RATIO = .317

X/C	ETA = .250 CP		ETA = .387 CP		ETA = .524 CP		ETA = .800 CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000	.99		.82		.95		1.00	
.010	-.21	.13	.48	-.01				
.015					-.29	-.27		
.025	-.485	-.115	-.532	-.116	-.262	-.243	-.192	-.272
.050	-.369	-.075	-.326	-.209	-.295	-.175	-.261	-.263
.100	-.319	-.259	-.284	-.256	-.321	-.271	-.297	-.259
.150	-.376	-.295	-.353	-.345	-.279	-.279	-.248	-.302
.250	-.336	-.366	-.317	-.365	-.309	-.343	-.344	-.325
.400	-.321	-.391	-.335	-.354	-.321	-.318	-.302	-.294
.550	-.209	-.305	-.238	-.283	-.190	-.255	-.204	-.244
.700	-.118	-.148	-.126	-.145	-.116	-.140	-.099	-.131
.850	-.043	-.059	-.045	-.049	-.020	-.048	-.008	-.039
.950	.041	.042	.032	.048	.043	.048	.043	.054

TABLE 27 (B) CONTINUED
PRESSURE PROFILE ON WING

LIFT JET ALONE

BETA = .03 DEG Q = 2839 N/SQ.M. (59.29 LBF/SQ.FT.) EFFECT. VELOC. RATIO = .317

X/C	ETA = .250 CP		ETA = .387 CP		ETA = .524 CP		ETA = .800 CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000	.93		.77		.91		.98	
.010	-.17	-.02	.51	-.16				
.015					-.13	-.25		
.025	-.272	-.475	-.206	-.378	-.298	-.309	-.194	-.290
.050	-.243	-.244	-.238	-.286	-.216	-.301	-.189	-.229
.100	-.264	-.360	-.207	-.298	-.212	-.269	-.189	-.270
.150	-.309	-.403	-.353	-.344	-.329	-.326	-.267	-.359
.250	-.272	-.423	-.287	-.415	-.292	-.387	-.312	-.357
.400	-.292	-.437	-.318	-.357	-.317	-.376	-.280	-.324
.550	-.208	-.313	-.231	-.292	-.208	-.259	-.211	-.237
.700	-.114	-.183	-.114	-.160	-.111	-.157	-.095	-.134
.850	-.036	-.066	-.033	-.059	-.014	-.054	-.010	-.040
.950	.044	.034	.039	.044	.042	.046	.050	.059

BETA = 4.01 DEG Q = 2851 N/SQ.M. (59.55 LBF/SQ.FT.) EFFECT. VELOC. RATIO = .318

X/C	ETA = .250 CP		ETA = .387 CP		ETA = .524 CP		ETA = .800 CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000	.72		.54		.88		.95	
.010	-.00	-.47	.50	-.19				
.015					-.16	-.41		
.025	-.095	-.596	-.146	-.525	-.226	-.490	-.225	-.336
.050	-.280	-.456	-.234	-.330	-.267	-.325	-.192	-.278
.100	-.192	-.397	-.182	-.373	-.209	-.313	-.227	-.271
.150	-.282	-.495	-.289	-.460	-.263	-.400	-.242	-.300
.250	-.264	-.472	-.280	-.449	-.278	-.372	-.340	-.363
.400	-.268	-.469	-.298	-.406	-.299	-.377	-.289	-.318
.550	-.207	-.347	-.213	-.309	-.210	-.276	-.206	-.247
.700	-.113	-.195	-.105	-.170	-.108	-.160	-.104	-.136
.850	-.046	-.078	-.037	-.069	-.022	-.061	-.007	-.041
.950	.040	.035	.036	.047	.048	.042	.051	.058

BETA = 8.00 DEG Q = 2843 N/SQ.M. (59.38 LBF/SQ.FT.) EFFECT. VELOC. RATIO = .318

X/C	ETA = .250 CP		ETA = .387 CP		ETA = .524 CP		ETA = .800 CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000	.60		.40		.84		.91	
.010	.33	-.68	.48	-.32				
.015					.10	-.65		
.025	.006	-.746	-.018	-.703	-.058	-.501	-.090	-.518
.050	.018	-.511	-.042	-.341	-.042	-.296	-.078	-.268
.100	-.096	-.495	-.141	-.424	-.197	-.332	-.150	-.266
.150	-.188	-.550	-.245	-.453	-.250	-.445	-.229	-.345
.250	-.241	-.550	-.223	-.462	-.275	-.471	-.305	-.414
.400	-.266	-.498	-.275	-.425	-.277	-.388	-.292	-.358
.550	-.192	-.354	-.199	-.327	-.220	-.280	-.208	-.262
.700	-.118	-.202	-.107	-.178	-.104	-.167	-.116	-.146
.850	-.046	-.081	-.031	-.068	-.007	-.071	-.005	-.050
.950	.029	.029	.038	.044	.047	.036	.052	.055

TABLE 27 (B) CONCLUDED

PRESSURE PROFILE ON WING

LIFT JET ALONE

BETA = 12.03 DEG Q = 2840 N/SQ.M. (59.31 LBF/SQ.FT.) EFFECT. VELOC. RATIO = .318

	ETA = .250		ETA = .387		ETA = .524		ETA = .800	
X/C	CP		CP		CP		CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000	.16		-.04		.55		.84	
.010	.46	-1.30	.44	-.92				
.015					.11	-.94		
.025	.240	-1.253	.100	-.730	.078	-.653	-.193	-.365
.050	-.025	-.903	-.089	-.591	-.126	-.518	-.187	-.336
.100	-.049	-.623	-.008	-.510	-.208	-.414	-.080	-.371
.150	-.170	-.676	-.169	-.584	-.119	-.468	-.280	-.368
.250	-.177	-.531	-.218	-.555	-.249	-.448	-.251	-.374
.400	-.232	-.512	-.238	-.470	-.253	-.397	-.246	-.384
.550	-.178	-.371	-.180	-.340	-.226	-.323	-.193	-.247
.700	-.105	-.208	-.110	-.202	-.101	-.175	-.114	-.145
.850	-.042	-.087	-.032	-.068	-.013	-.078	.003	-.032
.950	.031	.024	.040	.040	.044	.033	.055	.060

TABLE 27 (c)
PRESSURE PROFILE ON WING

LIFT JET ALONE

BETA = -12.01 DEG Q = 2746 N/SQ.M. (57.36 LBF/SQ.FT.) EFFECT. VELOC. RATIO = .199

X/C	ETA = .250 CP		ETA = .387 CP		ETA = .524 CP		ETA = .800 CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000	.96		.87		.98		1.01	
.010	-.72	.63	1.01	.47				
.015					-.17	.03		
.025	-.884	.156	-.440	-.105	-.440	-.097	-.252	-.096
.050	-.622	.066	-.391	-.123	-.448	-.127	-.213	-.199
.100	-.468	-.066	-.348	-.119	-.326	-.192	-.255	-.206
.150	-.438	-.157	-.389	-.235	-.354	-.280	-.249	-.242
.250	-.398	-.240	-.349	-.270	-.349	-.262	-.327	-.315
.400	-.339	-.278	-.342	-.277	-.331	-.286	-.296	-.288
.550	-.217	-.201	-.225	-.225	.849	-.211	-.208	-.221
.700	-.130	-.091	-.124	-.086	-.130	-.128	-.118	-.138
.850	-.039	-.010	-.039	-.002	-.009	-.025	-.007	-.020
.950	.050	.064	.035	.066	.045	.052	.042	

BETA = -8.02 DEG Q = 2745 N/SQ.M. (57.32 LBF/SQ.FT.) EFFECT. VELOC. RATIO = .199

X/C	ETA = .250 CP		ETA = .387 CP		ETA = .524 CP		ETA = .800 CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000	1.01		.91		.99		1.01	
.010	-.61	.31		.01				
.015					-.21	-.15		
.025	-.613	.040	-.431	-.117	-.338	-.208	-.216	-.206
.050	-.425	-.082	-.474	-.188	-.268	-.233	-.251	-.243
.100	-.339	-.139	-.274	-.183	-.306	-.225	-.224	-.211
.150	-.406	-.258	-.368	-.297	-.297	-.269	-.263	-.298
.250	-.345	-.284	-.359	-.345	-.331	-.348	-.317	-.316
.400	-.321	-.329	-.326	-.328	-.334	-.315	-.285	-.301
.550	-.211	-.241	-.230	-.234	.770	-.239	-.211	-.231
.700	-.120	-.108	-.119	-.130	-.123	-.120	-.114	-.131
.850	-.033	-.028	-.034	-.028	-.012	-.036	-.009	-.027
.950	.058	.062	.041	.060	.051	.050	.044	.058

BETA = -4.02 DEG Q = 2746 N/SQ.M. (57.35 LBF/SQ.FT.) EFFECT. VELOC. RATIO = .199

X/C	ETA = .250 CP		ETA = .387 CP		ETA = .524 CP		ETA = .800 CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000	.99		.73		.95		1.00	
.010	-.20	-.04		.01				
.015					-.05	-.18		
.025	-.639	-.163	-.348	-.255	-.275	-.093	-.262	-.245
.050	-.259	-.140	-.283	-.238	-.236	-.200	-.235	-.202
.100	-.286	-.218	-.225	-.298	-.262	-.265	-.228	-.267
.150	-.368	-.376	-.308	-.306	-.244	-.333	-.204	-.294
.250	-.336	-.348	-.300	-.402	-.307	-.366	-.302	-.337
.400	-.313	-.354	-.307	-.366	-.307	-.337	-.292	-.315
.550	-.199	-.283		-.276	.710	-.250	-.207	-.244
.700	-.125	-.141	-.098	-.131	-.111	-.137	-.098	-.133
.850	-.028	-.032	-.035	-.039	-.009	-.036	-.015	-.036
.950	.057	.055	.044	.054	.052	.052	.043	.057

TABLE 27 (C) CONTINUED
PRESSURE PROFILE ON WING

LIFT JET ALONE

BETA = -.05 DEG Q = 2742 N/SQ.M. (57.27 LBF/SQ.FT.) EFFECT. VELOC. RATIO = .198

	ETA = .250		ETA = .387		ETA = .524		ETA = .800	
X/C	CP		CP		CP		CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000	.86		.63		.86		.98	
.010	.06	-.35		-.36				
.015					-.03	-.42		
.025	-.226	-.635	-.144	-.327	-.236	-.404	-.166	-.312
.050	-.224	-.321	-.282	-.336	-.136	-.351	-.188	-.318
.100	-.202	-.363	-.170	-.284	-.190	-.263	-.214	-.266
.150	-.297	-.383	-.323	-.387	-.286	-.377	-.214	-.338
.250	-.310	-.430	-.286	-.431	-.285	-.434	-.308	-.365
.400	-.282	-.435	-.319	-.391	-.329	-.383	-.277	-.325
.550	-.201	-.318	-.212	-.287	.657	-.264	-.220	-.256
.700	-.112	-.188	-.101	-.167	-.103	-.155	-.094	-.141
.850	-.035	-.053	-.030	-.058	-.009	-.051	-.011	-.043
.950	.052	.050	.041	.055	.048	.048	.047	.051

BETA = 4.03 DEG Q = 2748 N/SQ.M. (57.40 LBF/SQ.FT.) EFFECT. VELOC. RATIO = .200

	ETA = .250		ETA = .387		ETA = .524		ETA = .800	
X/C	CP		CP		CP		CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000	.62		.40		.79		.95	
.010	.12	-.64		-.47				
.015					.08	-.49		
.025	-.163	-.695	-.078	-.587	-.040	-.537	-.213	-.260
.050	-.097	-.563	-.137	-.427	-.174	-.392	-.103	-.306
.100	-.102	-.499	-.116	-.443	-.214	-.375	-.257	-.310
.150	-.237	-.544	-.279	-.560	-.212	-.503	-.229	-.399
.250	-.231	-.497	-.285	-.495	-.290	-.436	-.289	-.401
.400	-.263	-.514	-.277	-.418	-.289	-.400	-.275	-.313
.550	-.202	-.339	-.206	-.321	.604	-.303	-.200	-.255
.700	-.119	-.194	-.101	-.180	-.100	-.170	-.100	-.144
.850	-.042	-.070	-.032	-.062	-.009	-.065	-.011	-.042
.950	.038	.033	.041	.044	.047	.036	.049	.054

BETA = 7.94 DEG Q = 2745 N/SQ.M. (57.34 LBF/SQ.FT.) EFFECT. VELOC. RATIO = .199

	ETA = .250		ETA = .387		ETA = .524		ETA = .800	
X/C	CP		CP		CP		CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000	.38		.10		.27		.92	
.010	.47	-.83		-.86				
.015					.00			
.025	.131	-1.149	-.111	-.902	-.009	-.540	-.027	-.526
.050	.022	-.679	.002	-.698	.041	-.467	-.093	-.421
.100	-.091	-.573	-.069	-.469	-.167	-.414	-.187	-.271
.150	-.142	-.616	-.165	-.589	-.201	-.448	-.198	-.395
.250	-.230	-.645	-.224	-.546	-.236	-.557	-.281	-.425
.400	-.249	-.522	-.263	-.510	-.288	-.426	-.273	-.336
.550	-.182	-.367	-.189	-.360	.545	-.314	-.197	-.265
.700	-.111	-.216	-.103	-.207	-.113	-.192	-.104	-.147
.850	-.045	-.090	-.039	-.090	-.015	-.068	-.003	-.049
.950	.030	.031	.030	.036	.041	.031	.052	.056

27 (C) CONCLUDED
PRESSURE PROFILE ON WING

LIFT JET ALONE

BETA = 11.63 DEG Q = 2738 N/SQ.M. (57.19 LBF/SQ.FT.) EFFECT. VELOC. RATIO = .200

	ETA = .250		ETA = .387		ETA = .524		ETA = .800	
X/C	CP		CP		CP		CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000	.02		-.30		.49		.88	
.010	.64	-1.47		-1.03				
.015					.28	-.93		
.025	.275	-1.645	.294	-1.157	.114	-.987	.074	-.619
.050	.088	-1.031	.073	-.679	-.052	-.653	-.032	-.371
.100	.001	-.706	-.051	-.611	-.131	-.537	-.134	-.387
.150	-.069	-.815	-.142	-.693	-.113	-.581	-.145	-.516
.250	-.178	-.720	-.181	-.580	-.210	-.507	-.254	-.461
.400	-.235		-.217	-.518	-.238	-.446	-.238	-.389
.550	-.170	-.413	-.178	-.390	.482	-.338	-.185	-.292
.700	-.118	-.262	-.103	-.226	-.103	-.205	-.112	-.163
.850	-.059	-.109	-.043	-.093	-.011	-.091	-.004	-.064
.950	.011	.013	.020	.033	.035	.022	.049	.045

TABLE 27 (D)
PRESSURE PROFILE ON WING

LIFT JET ALONE

BETA = -12.02 DEG Q = 693 N/SQ.M. (14.47 LBF/SQ.FT.) EFFECT. VELOC. RATIO = .102

X/C	ETA = .250 CP		ETA = .387 CP		ETA = .524 CP		ETA = .800 CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000	1.01		.85		.92		1.05	
.010	-.17	.45	1.11	.09				
.015					-.09	-.24		
.025	-.465	-.081	-.288	-.165	-.115	-.117	-.142	-.174
.050	-.308	.142	-.194	-.151	-.169	-.169	-.046	-.103
.100	-.263	-.031	-.276	-.133	-.208	-.210	-.110	-.183
.150	-.292	-.183	-.215	-.240	-.208	-.253	-.128	-.228
.250	-.249	-.213	-.269	-.197	-.203	-.241	-.228	-.274
.400	-.230	-.204	-.226	-.235	-.230	-.218	-.186	-.227
.550	-.113	-.138	-.141	-.154	-.806	-.141	-.120	-.149
.700	-.027	-.043	-.037	-.046	-.052	-.071	-.046	-.075
.850	.052	.074	.053	.054	.080	.046	.062	.018
.950	.130	.145	.124	.138	.126	.133	.114	.129

BETA = -8.03 DEG Q = 695 N/SQ.M. (14.51 LBF/SQ.FT.) EFFECT. VELOC. RATIO = .102

X/C	ETA = .250 CP		ETA = .387 CP		ETA = .524 CP		ETA = .800 CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000	.97		.59		.99		1.04	
.010	-.24	.19	1.00	.06				
.015					-.13	-.14		
.025	-.477	-.053	-.196	-.141	-.182	-.182	-.060	-.203
.050	-.209	-.144	-.184	-.200	-.166	-.268	-.046	-.157
.100	-.277	-.155	-.116	-.200	-.212	-.184	-.107	-.198
.150	-.246	-.218	-.159	-.191	-.178	-.243	-.091	-.275
.250	-.230	-.286	-.230	-.302	-.221	-.268	-.209	-.257
.400	-.193	-.297	-.223	-.289	-.212	-.246	-.168	-.249
.550	-.119	-.172	-.119	-.154	-.733	-.172	-.104	-.169
.700	-.030	-.038	-.039	-.052	-.050	-.064	-.043	-.059
.850	.048	.047	.054	.057	.079	.051	.061	.044
.950	.131	.117	.122	.139	.135	.128	.125	.140

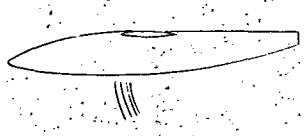
BETA = -4.02 DEG Q = 697 N/SQ.M. (14.55 LBF/SQ.FT.) EFFECT. VELOC. RATIO = .102

X/C	ETA = .250 CP		ETA = .387 CP		ETA = .524 CP		ETA = .800 CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000	1.00		.56		.77		1.05	
.010	.11	-.21	1.00	-.26				
.015					-.01	-.35		
.025	-.260	-.348	-.163	-.321	-.088	-.387	-.106	-.502
.050	-.167	-.242	-.061	-.172	-.117	-.416	-.011	-.212
.100	-.135	-.205	-.054	-.190	-.061	-.273	-.088	-.251
.150	-.192	-.312	-.190	-.285	-.104	-.298	-.165	-.233
.250	-.210	-.343	-.167	-.323	-.187	-.292	-.174	-.299
.400	-.209	-.263	-.183	-.282	-.212	-.290	-.155	-.239
.550	-.099	-.212	-.103	-.183	-.635	-.184	-.090	-.147
.700	-.021	-.065	-.021	-.057	-.031	-.068	-.027	-.070
.850	.067	.050	.073	.041	.083	.046	.086	.038
.950	.135	.140	.138	.150	.147	.127	.127	.140

TABLE 27 (D) CONTINUED

PRESSURE PROFILE ON WING

LIFT JET ALONE



BETA = -.04 DEG Q = 691 N/SQ.M. (14.42 LBF/SQ.FT.) EFFECT. VELOC. RATIO = .101

	ETA = .250		ETA = .387		ETA = .524		ETA = .800	
X/C	CP		CP		CP		CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000	.69		.19		.54		1.01	
.010	.30	-.38	.88	-.77				
.015					.20	-.66		
.025	-.057	-.762	.126	-.778	.005	-.557	.080	-.429
.050	.014	-.605	-.030	-.425	-.073	-.441	-.066	-.292
.100	-.036	-.484	.007	-.315	-.018	-.399	-.082	-.313
.150	-.096	-.457	-.087	-.425	-.091	-.349	-.110	-.288
.250	-.103	-.445	-.135	-.421	-.162	-.272	-.162	-.299
.400	-.153	-.413	-.180	-.374	-.200	-.361	-.163	-.267
.550	-.095	-.239	-.086	-.254	-.610	-.227	-.097	-.195
.700	-.026	-.079	-.021	-.109	-.032	-.071	-.043	-.078
.850	.055	.022	.066	.007	.079	.022	.076	.017
.950	.129	.120	.124	.128	.131	.123	.127	.128

BETA = 3.99 DEG Q = 695 N/SQ.M. (14.52 LBF/SQ.FT.) EFFECT. VELOC. RATIO = .102

	ETA = .250		ETA = .387		ETA = .524		ETA = .800	
X/C	CP		CP		CP		CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000	.22		-.24		.45		.94	
.010	.42	-1.15	.81	-.62				
.015					.32	-.91		
.025	.187	-1.081	.248	-.833	.142	-.720	-.031	-.491
.050	.099	-.729	-.008	-.584	.126	-.500	.049	-.389
.100	-.015	-.539	.037	-.489	-.071	-.498	.015	-.280
.150	-.062	-.579	-.067	-.484	-.192	-.507	-.103	-.321
.250	-.105	-.499	-.105	-.489	-.139	-.457	-.185	-.263
.400	-.141	-.446	-.147	-.374	-.152	-.322	-.164	-.296
.550	-.091	-.287	-.083	-.253	-.557	-.194	-.083	-.200
.700	-.017	-.136	-.030	-.106	-.032	-.105	-.025	-.080
.850	.052	.009	.054	.008	.070	.002	.071	.027
.950	.112	.115	.121	.119	.129	.119	.126	.125

BETA = 7.98 DEG Q = 694 N/SQ.M. (14.50 LBF/SQ.FT.) EFFECT. VELOC. RATIO = .102

	ETA = .250		ETA = .387		ETA = .524		ETA = .800	
X/C	CP		CP		CP		CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000	-.08		-.24		.51		.85	
.010	.76	-1.60	.69	-.71				
.015					.44	-1.22		
.025	.374	-1.375	.331	-.805	.324	-.896	.067	-.646
.050	.221	-.853	.153	-.701	.181	-.508	.065	-.453
.100	.078	-.689	.085	-.585	.069	-.421	-.006	-.308
.150	-.010	-.644	-.026	-.589	-.099	-.521	-.053	-.403
.250	-.097	-.552	-.074	-.468	-.137	-.478	-.128	-.356
.400	-.123	-.381	-.144	-.430	-.125	-.369	-.130	-.302
.550	-.065	-.318	-.090	-.270	-.516	-.203	-.083	-.196
.700	-.015	-.137	-.011	-.100	-.028	-.085	-.010	-.097
.850	.038	-.005	.050	.022	.064	.004	.068	.025
.950	.105	.086	.123	.115	.118	.124	.133	.122

TABLE 27 (D) CONCLUDED

PRESSURE PROFILE ON WING

LIFT JET ALONE

BETA = 11.96 DEG Q = 695 N/SQ.M. (14.52 LBF/SQ.FT.) EFFECT. VELOC. RATIO = .102

X/C	ETA = .250 CP		ETA = .387 CP		ETA = .524 CP		ETA = .800 CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000	-.76		-.82		.04		.79	
.010	.78	-2.18	.63	-1.53				
.015					.47	-1.19		
.025	.512	-1.502	.271	-1.166	.423	-1.062	.224	-.640
.050	.299	-1.017	.210	-.699	.163	-.642	.085	-.436
.100	.079	-.726	.151	-.622	.113	-.450	.015	-.404
.150	.067	-.636	.051	-.645	.033	-.540	-.030	-.441
.250	-.064	-.605	-.078	-.540	-.037	-.444	-.116	-.380
.400	-.093	-.422	-.129	-.433	-.111	-.398	-.114	-.304
.550	-.056	-.326	-.059	-.250	-.472	-.265	-.067	-.192
.700	-.009	-.133	-.012	-.136	-.024	-.144	-.005	-.090
.850	.042	-.012	.051	.001	.057	.012	.068	.024
.950	.090	.081	.110	.115	.120	.109	.128	.130

TABLE 28 (A)

PRESSURE PROFILE ON WING

FORWARD NOZZLES DEFLECTED 90 DEGREES

BETA = -12.13 DEG Q = 2848 N/SQ.M. (59.49 LBF/SQ.FT.) EFFECT. VELOC. RATIO = ∞

	ETA = .250		ETA = .387		ETA = .524		ETA = .800	
X/C	CP		CP		CP		CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000							1.000	
.010			.694	.742				
.015					-.515	.057		
.025			-.987	.301	-.634	-.029	-.376	-.110
.050			-.744	.105	-.575	-.069	-.329	-.150
.100			-.571	-.036	-.458	-.122	-.287	-.189
.150			-.538	-.272	-.482	-.272	-.319	-.234
.250	-.364		-.487	-.357	-.442	-.312	-.364	-.307
.400	-.412		-.443	-.376	-.430	-.356	-.338	-.306
.550	-.331	-.331	-.331	-.312	-.300	-.294	-.257	-.239
.700	-.272	-.383	-.237	-.273	-.216	-.231	-.165	-.172
.850	-.261	-.465	-.175	-.278	-.084	-.123	-.040	-.058
.950	-.329	-.490	-.144	-.170	-.028	-.033	.015	.038

BETA = -8.16 DEG Q = 2846 N/SQ.M. (59.44 LBF/SQ.FT.) EFFECT. VELOC. RATIO = ∞

	ETA = .250		ETA = .387		ETA = .524		ETA = .800	
X/C	CP		CP		CP		CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			.999				.992	
.010			.704	.418				
.015					-.577	.041		
.025			-.876	.073	-.605	-.129	-.351	-.051
.050			-.581	-.086	-.562	-.065	-.250	-.211
.100			-.571	-.186	-.442	-.165	-.326	-.196
.150			-.544	-.316	-.462	-.276	-.360	-.286
.250	-.348		-.471	-.381	-.421	-.336	-.373	-.351
.400	-.420		-.440	-.404	-.427	-.378	-.356	-.308
.550	-.346	-.326	-.341	-.339	-.332	-.320	-.255	-.244
.700	-.282	-.396	-.238	-.290	-.214	-.249	-.157	-.184
.850	-.267	-.459	-.180	-.301	-.094	-.122	-.042	-.056
.950	-.333	-.480	-.155	-.201	-.030	-.033	.018	.032

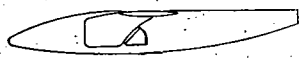
BETA = -4.10 DEG Q = 2843 N/SQ.M. (59.39 LBF/SQ.FT.) EFFECT. VELOC. RATIO = ∞

	ETA = .250		ETA = .387		ETA = .524		ETA = .800	
X/C	CP		CP		CP		CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			.953		.984		.955	
.010			.726	.227				
.015					-.527	-.111		
.025			-.725	-.091	-.579	-.189	-.385	-.151
.050			-.529	-.176	-.448	-.178	-.314	-.172
.100			-.464	-.215	-.460	-.218	-.363	-.218
.150			-.468	-.392	-.452	-.338	-.330	-.294
.250	-.322		-.434	-.456	-.405	-.398	-.381	-.356
.400	-.401		-.439	-.434	-.439	-.397	-.352	-.316
.550	-.330	-.352	-.340	-.360	-.325	-.341	-.272	-.269
.700	-.274	-.397	-.230	-.311	-.199	-.269	-.158	-.193
.850	-.261	-.443	-.182	-.308	-.098	-.147	-.054	-.065
.950	-.308	-.414	-.145	-.192	-.033	-.049	.010	.027

TABLE 28 (A) CONTINUED

PRESSURE PROFILE ON WING

FORWARD NOZZLES DEFLECTED 90 DEGREES



BETA = -.01 DEG Q = 2850 N/SQ.M. (59.52 LBF/SQ.FT.) EFFECT. VELOC. RATIO = ∞

X/C	ETA = .250 CP		ETA = .387 CP		ETA = .524 CP		ETA = .800 CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			.712		.943		.936	
.010			.727	.018				
.015					-.299	-.230		
.025			-.405	-.301	-.464	-.258	-.406	-.250
.050			-.414	-.344	-.323	-.229	-.330	-.159
.100			-.352	-.433	-.405	-.278	-.316	-.241
.150			-.406	-.431	-.364	-.354	-.358	-.301
.250	-.330		-.448	-.514	-.419	-.447	-.387	-.359
.400	-.407		-.431	-.456	-.387	-.418	-.353	-.345
.550	-.332	-.358	-.324	-.358	-.332	-.368	-.266	-.273
.700	-.278	-.377	-.237	-.311	-.203	-.273	-.144	-.201
.850	-.269	-.421	-.188	-.307	-.095	-.157	-.052	-.076
.950	-.296	-.412	-.150	-.206	-.037	-.053	.013	.021

BETA = 3.96 DEG Q = 2848 N/SQ.M. (59.48 LBF/SQ.FT.) EFFECT. VELOC. RATIO = ∞

X/C	ETA = .250 CP		ETA = .387 CP		ETA = .524 CP		ETA = .800 CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			.546		.916		.938	
.010			.700	-.487				
.015					-.206	-.253		
.025			-.388	-.560	-.454	-.380	-.428	-.264
.050			-.245	-.334	-.354	-.222	-.323	-.223
.100			-.311	-.409	-.354	-.241	-.336	-.215
.150			-.415	-.458	-.365	-.398	-.336	-.355
.250	-.303		-.387	-.514	-.417	-.433	-.375	-.363
.400	-.390		-.404	-.436	-.407	-.430	-.359	-.341
.550	-.331	-.353	-.321	-.337	-.318	-.377	-.281	-.281
.700	-.274	-.378	-.239	-.313	-.198	-.285	-.161	-.206
.850	-.268	-.405	-.185	-.333	-.095	-.164	-.047	-.071
.950	-.302	-.396	-.167	-.205	-.033	-.059	.016	.025

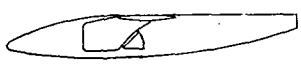
BETA = 7.97 DEG Q = 2844 N/SQ.M. (59.40 LBF/SQ.FT.) EFFECT. VELOC. RATIO = ∞

X/C	ETA = .250 CP		ETA = .387 CP		ETA = .524 CP		ETA = .800 CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			.162		.828		.903	
.010			.656	-.385				
.015					-.277	-.148		
.025			-.181	-.737	-.316	-.336	-.274	-.232
.050			-.276	-.659	-.299	-.322	-.179	-.238
.100			-.226	-.526	-.304	-.328	-.296	-.315
.150			-.400	-.520	-.365	-.445	-.294	-.284
.250	-.277		-.344	-.548	-.338	-.443	-.396	-.419
.400	-.358		-.390	-.419	-.367	-.441	-.338	-.348
.550	-.324	-.395	-.304	-.343	-.295	-.367	-.252	-.289
.700	-.276	-.401	-.244	-.341	-.206	-.286	-.162	-.198
.850	-.255	-.408	-.179	-.356	-.096	-.162	-.043	-.068
.950	-.293	-.364	-.166	-.271	-.028	-.065	.020	.024

TABLE 28 (A) CONCLUDED

PRESSURE PROFILE ON WING

FORWARD NOZZLES DEFLECTED 90 DEGREES



BETA = 11.98 DEG Q = 2838 N/SQ.M. (59.28 LBF/SQ.FT.) EFFECT. VELOC. RATIO = ∞

	ETA = .250		ETA = .387		ETA = .524		ETA = .800	
X/C	CP		CP		CP		CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			-.370		.610		.868	
.010			.574	-1.018				
.015					.146	-.432		
.025			.005	-.992	-.249	-.565	-.206	-.194
.050			-.049	-.495	-.185	-.374	-.287	-.312
.100			-.162	-.593	-.244	-.372	-.240	-.262
.150			-.317	-.608	-.283	-.431	-.297	-.313
.250	-.260		-.317	-.557	-.292	-.472	-.332	-.372
.400	-.350		-.377	-.508	-.344	-.476	-.322	-.369
.550	-.306	-.363	-.280	-.474	-.270	-.406	-.243	-.293
.700	-.263	-.382	-.219	-.396	-.185	-.312	-.161	-.218
.850	-.229	-.347	-.174	-.358	-.083	-.190	-.029	-.068
.950	-.243	-.321	-.194	-.195	-.026	-.075	.025	.023

TABLE 28 (B)
PRESSURE PROFILE ON WING

FORWARD NOZZLES DEFLECTED 90 DEGREES

BETA = -12.01 DEG Q = 2035 N/SQ.M. (42.49 LBF/SQ.FT.) EFFECT. VELOC. RATIO = .303

	ETA = .250		ETA = .387		ETA = .524		ETA = .800	
X/C	CP		CP		CP		CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000							.969	
.010			.399	.579				
.015					-.626	-.118		
.025			-1.071	-.005	-.857	-.238	-.685	-.201
.050			-.823	-.110	-.588	-.377	-.426	-.157
.100			-.679	-.405	-.536	-.486	-.444	-.297
.150			-.606	-.644	-.531	-.568	-.460	-.424
.250	-.425		-.560	-.825	-.527	-.750	-.435	-.452
.400	-.485		-.536	-1.116	-.513	-.823	-.407	-.438
.550	-.397	-1.723	-.417	-1.247	-.398	-.792	-.323	-.386
.700	-.354	-2.242	-.322	-1.210	-.309	-.663	-.211	-.313
.850	-.362	-2.204	-.306	-.928	-.203	-.452	-.117	-.182
.950	-.519	-1.679	-.317	-.564	-.179	-.275	-.067	-.065

BETA = -8.04 DEG Q = 2036 N/SQ.M. (42.53 LBF/SQ.FT.) EFFECT. VELOC. RATIO = .303

	ETA = .250		ETA = .387		ETA = .524		ETA = .800	
X/C	CP		CP		CP		CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			.996		.985		1.007	
.010			.423	.494				
.015					-.677	-.359		
.025			-1.119	-.318	-.829	-.430	-.433	-.106
.050			-.687	-.358	-.538	-.471	-.437	-.231
.100			-.597	-.599	-.520	-.541	-.412	-.335
.150			-.569	-.722	-.529	-.618	-.420	-.406
.250	-.402		-.535	-.963	-.494	-.824	-.452	-.468
.400	-.481		-.537	-1.128	-.500	-.887	-.414	-.474
.550	-.401	-1.509	-.415	-1.191	-.413	-.816	-.324	-.412
.700	-.362	-2.021	-.333	-1.191	-.312	-.684	-.221	-.317
.850	-.390	-2.146	-.317	-.939	-.211	-.459	-.114	-.186
.950	-.601	-1.778	-.318	-.551	-.179	-.280	-.065	-.069

BETA = -4.02 DEG Q = 2031 N/SQ.M. (42.43 LBF/SQ.FT.) EFFECT. VELOC. RATIO = .303

	ETA = .250		ETA = .387		ETA = .524		ETA = .800	
X/C	CP		CP		CP		CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			.827		.966		1.003	
.010			.484	-.091				
.015					-.688	-.195		
.025			-.711	-.395	-.516	-.439	-.527	-.280
.050			-.596	-.519	-.551	-.374	-.433	-.253
.100			-.528	-.745	-.453	-.589	-.415	-.307
.150			-.572	-.902	-.482	-.747	-.413	-.368
.250	-.392		-.507	-1.098	-.522	-.854	-.503	-.489
.400	-.482		-.537	-1.239	-.496	-.952	-.448	-.498
.550	-.422	-1.628	-.427	-1.227	-.410	-.867	-.335	-.439
.700	-.387	-1.728	-.346	-1.151	-.310	-.719	-.214	-.326
.850	-.438	-1.890	-.325	-.927	-.225	-.492	-.120	-.199
.950	-.639	-1.761	-.335	-.554	-.188	-.297	-.074	-.079

TABLE 28 (B) CONTINURD
PRESSURE PROFILE ON WING

FORWARD NOZZLES DEFLECTED 90 DEGREES

BETA = .01 DEG Q = 2034 N/SQ.M. (42.48 LBF/SQ.FT.) EFFECT. VELOC. RATIO = .302

	ETA = .250		ETA = .387		ETA = .524		ETA = .800	
X/C	CP		CP		CP		CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			.567		.928		.968	
.010			.542	-.564				
.015					-.537	-.580		
.025			-.708	-.562	-.607	-.666	-.559	-.199
.050			-.446	-.708	-.435	-.373	-.447	-.245
.100			-.481	-.876	-.473	-.606	-.425	-.297
.150			-.497	-1.085	-.424	-.794	-.439	-.377
.250	-.378		-.521	-1.225	-.505	-.935	-.472	-.506
.400	-.492		-.526	-1.335	-.512	-1.001	-.447	-.511
.550	-.441	-1.596	-.422	-1.288	-.416	-.911	-.345	-.447
.700	-.405	-1.610	-.345	-1.158	-.308	-.724	-.218	-.336
.850	-.433	-1.609	-.326	-.896	-.222	-.517	-.133	-.218
.950	-.624	-1.478	-.334	-.532	-.198	-.310	-.077	-.092

BETA = 4.01 DEG Q = 2038 N/SQ.M. (42.56 LBF/SQ.FT.) EFFECT. VELOC. RATIO = .303

	ETA = .250		ETA = .387		ETA = .524		ETA = .800	
X/C	CP		CP		CP		CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			.703		.889		.902	
.010			.565	-.671				
.015					-.418	-.357		
.025			-.505	-.589	-.578	-.595	-.417	-.265
.050			-.490	-.701	-.545	-.554	-.457	-.179
.100			-.506	-.882	-.536	-.635	-.413	-.345
.150			-.490	-1.156	-.504	-.783	-.483	-.467
.250	-.398		-.522	-1.165	-.500	-.860	-.466	-.507
.400	-.481		-.529	-1.302	-.519	-.938	-.442	-.529
.550	-.442	-1.385	-.447	-1.245	-.413	-.875	-.347	-.442
.700	-.417	-1.438	-.360	-1.099	-.320	-.740	-.219	-.354
.850	-.456	-1.462	-.332	-.858	-.245	-.534	-.140	-.231
.950	-.687	-1.424	-.339	-.549	-.216	-.331	-.084	-.107

BETA = 7.99 DEG Q = 2030 N/SQ.M. (42.40 LBF/SQ.FT.) EFFECT. VELOC. RATIO = .302

	ETA = .250		ETA = .387		ETA = .524		ETA = .800	
X/C	CP		CP		CP		CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			.430		.891		.912	
.010			.606	-.661				
.015					-.319	-.771		
.025			-.389	-1.273	-.389	-.518	-.515	-.226
.050			-.354	-.953	-.484	-.592	-.431	-.289
.100			-.393	-.864	-.420	-.600	-.367	-.351
.150			-.505	-1.059	-.414	-.706	-.431	-.472
.250	-.361		-.514	-1.228	-.538	-.869	-.448	-.541
.400	-.490		-.514	-1.242	-.522	-.979	-.441	-.559
.550	-.452	-1.298	-.437	-1.193	-.418	-.900	-.357	-.481
.700	-.431	-1.299	-.380	-1.051	-.337	-.743	-.241	-.368
.850	-.473	-1.291	-.360	-.902	-.242	-.556	-.137	-.243
.950	-.684	-1.277	-.360	-.565	-.218	-.342	-.091	-.108

TABLE 28 (B) CONCLUDED
PRESSURE PROFILE ON WING

FORWARD NOZZLES DEFLECTED 90 DEGREES

BETA = 12.04 DEG Q = 2035 N/SQ.M. (42.49 LBF/SQ.FT.) EFFECT. VELOC. RATIO = .302

	ETA = .250		ETA = .387		ETA = .524		ETA = .800	
X/C	CP		CP		CP		CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			.190		.785		.874	
.010			.587	-.963				
.015					-.295	-.529		
.025			-.308	-1.383	-.327	-.817	-.412	-.239
.050			-.272	-.858	-.407	-.439	-.470	-.339
.100			-.329	-.967	-.309	-.606	-.427	-.343
.150			-.384	-.998	-.494	-.825	-.387	-.521
.250	-.365		-.443	-1.212	-.467	-.889	-.451	-.534
.400	-.464		-.457	-1.248	-.495	-.995	-.472	-.549
.550	-.429	-1.317	-.442	-1.138	-.436	-.894	-.362	-.500
.700	-.428	-1.280	-.380	-1.059	-.349	-.768	-.258	-.385
.850	-.485	-1.276	-.373	-.932	-.255	-.572	-.136	-.256
.950	-.662	-1.224	-.362	-.571	-.237	-.354	-.091	-.122

TABLE 28 (C)
PRESSURE PROFILE ON WING

FORWARD NOZZLES DEFLECTED 90 DEGREES

BETA = -11.99 DEG Q = 910 N/SQ.M. (19.00 LBF/SQ.FT.) EFFECT. VELOC. RATIO = .202

	ETA = .250		ETA = .387		ETA = .524		ETA = .800	
X/C	CP		CP		CP		CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			.905		.827		1.016	
.010			.065	.096				
.015					-.472	-.704		
.025			-.744	-.441	-.389	-.829	-.268	-.548
.050			-.617	-.482	-.361	-.730	-.285	-.510
.100			-.474	-.742	-.387	-.820	-.304	-.462
.150			-.514	-.981	-.481	-.959	-.354	-.526
.250	-.392		-.520	-1.166	-.468	-1.041	-.399	-.594
.400	-.445		-.468	-1.581	-.458	-1.104	-.380	-.578
.550	-.369	-3.502	-.392	-1.854	-.373	-1.026	-.303	-.495
.700	-.340	-4.036	-.328	-1.757	-.311	-.893	-.228	-.398
.850	-.334	-3.057	-.322	-1.231	-.228	-.595	-.135	-.228
.950	-.479	-1.748	-.361	-.706	-.221	-.364	-.085	-.100

BETA = -8.00 DEG Q = 912 N/SQ.M. (19.05 LBF/SQ.FT.) EFFECT. VELOC. RATIO = .202

	ETA = .250		ETA = .387		ETA = .524		ETA = .800	
X/C	CP		CP		CP		CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			.736		.741		1.021	
.010			.081	-.377				
.015					-.149	-.963		
.025			-.467	-.792	-.315	-.970	-.175	-.474
.050			-.469	-.840	-.427	-.873	-.185	-.559
.100			-.408	-.818	-.346	-.916	-.287	-.562
.150			-.483	-1.055	-.384	-.958	-.325	-.604
.250	-.375		-.491	-1.295	-.420	-1.063	-.419	-.674
.400	-.430		-.496	-1.610	-.464	-1.187	-.382	-.625
.550	-.381	-2.995	-.404	-1.919	-.382	-1.121	-.300	-.526
.700	-.344	-3.592	-.351	-1.824	-.323	-.918	-.237	-.416
.850	-.371	-3.292	-.329	-1.266	-.249	-.627	-.131	-.246
.950	-.573	-2.134	-.387	-.739	-.227	-.378	-.087	-.114

BETA = -3.96 DEG Q = 906 N/SQ.M. (18.93 LBF/SQ.FT.) EFFECT. VELOC. RATIO = .201

	ETA = .250		ETA = .387		ETA = .524		ETA = .800	
X/C	CP		CP		CP		CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			.110		.503		.985	
.010			.032	-1.057				
.015					.100	-1.343		
.025			-.265	-1.086	-.172	-1.171	-.203	-.832
.050			-.344	-.952	-.222	-.971	-.229	-.596
.100			-.370	-1.152	-.307	-.971	-.229	-.563
.150			-.417	-1.316	-.377	-1.107	-.319	-.657
.250	-.349		-.464	-1.434	-.399	-1.245	-.385	-.690
.400	-.425		-.466	-1.729	-.437	-1.286	-.396	-.671
.550	-.376	-3.475	-.380	-2.057	-.369	-1.177	-.319	-.557
.700	-.349	-3.765	-.355	-1.901	-.310	-.954	-.236	-.444
.850	-.375	-3.259	-.349	-1.390	-.246	-.676	-.130	-.262
.950	-.592	-2.395	-.386	-.776	-.231	-.396	-.086	-.118

TABLE 28 (C) CONTINUED
PRESSURE PROFILE ON WING

FORWARD NOZZLES DEFLECTED 90 DEGREES

BETA = -.02 DEG Q = 909 N/SQ.M. (18.98 LBF/SQ.FT.) EFFECT. VELOC. RATIO = .202

X/C	ETA = .250 CP		ETA = .387 CP		ETA = .524 CP		ETA = .800 CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			-.272		.517		.968	
.010			.005	-1.339				
.015					.159	-1.634		
.025			-.073	-1.400	-.106	-1.408	-.012	-.373
.050			-.182	-1.386	-.193	-1.153	-.174	-.585
.100			-.253	-1.365	-.207	-1.202	-.217	-.633
.150			-.396	-1.528	-.293	-1.256	-.318	-.666
.250	-.319		-.406	-1.566	-.416	-1.292	-.382	-.718
.400	-.414		-.447	-1.815	-.424	-1.376	-.378	-.679
.550	-.359	-3.182	-.381	-2.080	-.350	-1.252	-.290	-.559
.700	-.339	-3.807	-.343	-2.026	-.311	-1.003	-.218	-.428
.850	-.397	-3.425	-.342	-1.400	-.233	-.696	-.125	-.252
.950	-.632	-2.400	-.407	-.798	-.220	-.400	-.083	-.112

BETA = 3.99 DEG Q = 914 N/SQ.M. (19.09 LBF/SQ.FT.) EFFECT. VELOC. RATIO = .202

X/C	ETA = .250 CP		ETA = .387 CP		ETA = .524 CP		ETA = .800 CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			-.813		.268		.936	
.010			-.006	-1.291				
.015					.344	-1.727		
.025			.086	-1.984	.015	-1.725	-.106	-.768
.050			-.051	-1.477	-.045	-1.273	-.190	-.615
.100			-.216	-1.406	-.214	-1.139	-.194	-.627
.150			-.358	-1.582	-.249	-1.360	-.285	-.689
.250	-.302		-.378	-1.671	-.361	-1.410	-.420	-.709
.400	-.423		-.447	-1.894	-.417	-1.413	-.386	-.706
.550	-.379	-3.061	-.384	-2.154	-.344	-1.287	-.320	-.580
.700	-.367	-3.667	-.349	-1.953	-.303	-1.041	-.217	-.417
.850	-.433	-3.373	-.357	-1.400	-.239	-.719	-.127	-.270
.950	-.720	-2.289	-.424	-.784	-.229	-.416	-.080	-.117

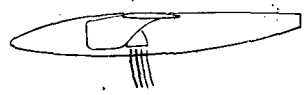
BETA = 7.99 DEG Q = 910 N/SQ.M. (19.00 LBF/SQ.FT.) EFFECT. VELOC. RATIO = .201

X/C	ETA = .250 CP		ETA = .387 CP		ETA = .524 CP		ETA = .800 CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			-1.382		-.280		.801	
.010			-.077	-2.669				
.015					.447	-1.955		
.025			.241	-2.262	.143	-1.874	-.055	-.947
.050			-.020	-1.834	.006	-1.375	-.244	-.625
.100			-.091	-1.721	-.154	-1.290	-.219	-.632
.150			-.268	-1.846	-.245	-1.401	-.266	-.704
.250	-.309		-.377	-1.818	-.318	-1.445	-.337	-.695
.400	-.388		-.415	-1.913	-.383	-1.428	-.364	-.702
.550	-.369	-3.066	-.376	-2.011	-.326	-1.283	-.282	-.587
.700	-.364	-3.137	-.347	-1.856	-.289	-1.039	-.208	-.443
.850	-.446	-2.712	-.359	-1.393	-.234	-.707	-.129	-.276
.950	-.713	-2.425	-.407	-.769	-.229	-.402	-.082	-.123

28 (C) CONCLUDED

TABLE
PRESSURE PROFILE ON WING

FORWARD NOZZLES DEFLECTED 90 DEGREES



BETA = 11.98 DEG Q = 910 N/SQ.M. (19.00 LBF/SQ.FT.) EFFECT. VELOC. RATIO = .201

	ETA = .250		ETA = .387		ETA = .524		ETA = .800	
X/C	CP		CP		CP		CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			-1.734		-.805		.793	
.010			-.186	-3.006				
.015					.377	-2.240		
.025			.366	-2.829	.290	-1.987	-.013	-.850
.050			.108	-2.273	.073	-1.706	-.095	-.850
.100			-.095	-1.841	-.188	-1.334	-.107	-.659
.150			-.216	-1.893	-.147	-1.486	-.280	-.793
.250	-.264		-.322	-1.898	-.270	-1.502	-.306	-.749
.400	-.383		-.353	-1.974	-.352	-1.392	-.300	-.724
.550	-.359	-2.458	-.354	-1.974	-.317	-1.292	-.266	-.569
.700	-.364	-2.573	-.320	-1.808	-.271	-1.028	-.182	-.437
.850	-.457	-2.410	-.344	-1.301	-.222	-.700	-.118	-.269
.950	-.723	-2.322	-.405	-.787	-.228	-.393	-.075	-.126

TABLE 28 (D)
PRESSURE PROFILE ON WING

FORWARD NOZZLES DEFLECTED 90 DEGREES.

BETA = -11.97 DEG Q = 226 N/SQ.M. (4.72 LBF/SQ.FT.) EFFECT. VELOC. RATIO = .100

	ETA = .250		ETA = .387		ETA = .524		ETA = .800	
X/C	CP		CP		CP		CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			-.449		-.763		.604	
.010			-3.720	-1.614				
.015					.841	-2.493		
.025			.178	-1.454	.604	-2.102	.409	-1.384
.050			.060	-1.202	.325	-1.530	.255	-1.112
.100			-.052	-1.244	.123	-1.370	.172	-.889
.150			-.114	-1.377	-.003	-1.363	.004	-.777
.250	-.094		-.149	-1.464	-.121	-1.498	-.107	-.853
.400	-.163		-.248	-2.005	-.166	-1.492	-.153	-.768
.550	-.126	-5.619	-.166	-2.678	-.163	-1.329	-.095	-.606
.700	-.106	-8.043	-.173	-2.404	-.119	-1.076	-.051	-.474
.850	-.139	-3.990	-.210	-1.529	-.102	-.690	-.007	-.217
.950	-.379	-2.259	-.312	-.913	-.163	-.366	.003	-.082

BETA = -8.03 DEG Q = 226 N/SQ.M. (4.71 LBF/SQ.FT.) EFFECT. VELOC. RATIO = .101

	ETA = .250		ETA = .387		ETA = .524		ETA = .800	
X/C	CP		CP		CP		CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			-.839		-1.293		.370	
.010			-3.431	-1.985				
.015					.838	-2.718		
.025			.293	-1.810	.705	-2.292	.475	-1.607
.050			.084	-1.530	.419	-1.670	.391	-1.160
.100			.014	-1.488	.195	-1.384	.160	-.951
.150			-.042	-1.677	.014	-1.537	.090	-.923
.250	-.091		-.133	-1.775	-.049	-1.680	-.098	-.925
.400	-.180		-.224	-2.161	-.176	-1.544	-.143	-.799
.550	-.149	-4.592	-.173	-2.445	-.149	-1.395	-.092	-.634
.700	-.146	-4.880	-.166	-2.428	-.129	-1.003	-.024	-.488
.850	-.197	-4.172	-.217	-1.602	-.109	-.684	.010	-.220
.950	-.441	-2.943	-.302	-1.030	-.187	-.356	.017	-.058

BETA = -3.97 DEG Q = 224 N/SQ.M. (4.68 LBF/SQ.FT.) EFFECT. VELOC. RATIO = .100

	ETA = .250		ETA = .387		ETA = .524		ETA = .800	
X/C	CP		CP		CP		CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			-1.903		-1.579		.377	
.010			-3.162	-2.958				
.015						-3.190		
.025			.447	-2.543	.679	-2.522	.644	-1.706
.050			.222	-2.015	.412	-2.135	.320	-1.396
.100			.039	-1.903	.257	-1.790	.130	-1.136
.150			-.059	-1.875	.039	-1.650	.095	-1.009
.250	-.151		-.179	-1.870	-.144	-1.726	-.130	-.980
.400	-.233		-.246	-2.262	-.199	-1.651	-.147	-.836
.550	-.216	-4.420	-.212	-2.514	-.188	-1.440	-.103	-.710
.700	-.199	-4.997	-.223	-2.610	-.168	-1.164	-.072	-.529
.850	-.284	-3.854	-.267	-1.590	-.147	-.707	-.014	-.277
.950	-.574	-3.179	-.448	-1.055	-.216	-.403	.003	-.107

TABLE 28 (D) CONTINUED
PRESSURE PROFILE ON WING

FORWARD NOZZLES DEFLECTED 90 DEGREES

BETA = .01 DEG Q = 224 N/SQ.M. (4.69 LBF/SQ.FT.) EFFECT. VELOC. RATIO = .100

X/C	ETA = .250 CP		ETA = .387 CP		ETA = .524 CP		ETA = .800 CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			-3.083		-2.226		.316	
.010			-2.816	-3.476				
.015					.948	-3.448		
.025			.604	-3.266	.745	-3.224	.534	-1.714
.050			.344	-2.416	.499	-2.247	.351	-1.454
.100			.127	-2.079	.260	-1.770	.190	-1.067
.150			.007	-2.184	.112	-1.896	.084	-1.060
.250	-.084		-.126	-2.030	-.063	-1.860	-.091	-1.029
.400	-.219		-.219	-2.152	-.185	-1.730	-.154	-.903
.550	-.215	-4.239	-.202	-2.272	-.168	-1.444	-.117	-.699
.700	-.222	-4.236	-.195	-2.326	-.154	-1.135	-.062	-.478
.850	-.301	-3.606	-.270	-1.472	-.141	-.685	-.008	-.263
.950	-.617	-2.752	-.409	-1.056	-.222	-.423	.026	-.086

BETA = 4.03 DEG Q = 226 N/SQ.M. (4.71 LBF/SQ.FT.) EFFECT. VELOC. RATIO = .101

X/C	ETA = .250 CP		ETA = .387 CP		ETA = .524 CP		ETA = .800 CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			-4.020		-2.721		-.010	
.010			-2.735	-4.656				
.015					1.031	-4.139		
.025			.738	-3.587	.835	-3.434	.619	-1.855
.050			.437	-2.609	.563	-2.365	.346	-1.449
.100			.207	-2.525	.227	-1.966	.207	-1.135
.150			.074	-2.281	.116	-2.029	.060	-1.198
.250	-.122		-.108	-2.158	-.045	-1.830	-.073	-1.047
.400	-.221		-.221	-2.341	-.167	-1.711	-.153	-.949
.550	-.204	-3.584	-.208	-2.368	-.170	-1.474	-.119	-.770
.700	-.197	-3.861	-.218	-2.087	-.150	-1.135	-.069	-.489
.850	-.357	-3.252	-.285	-1.494	-.147	-.722	-.031	-.285
.950	-.682	-2.669	-.411	-1.061	-.218	-.407	-.004	-.103

BETA = 8.01 DEG Q = 224 N/SQ.M. (4.67 LBF/SQ.FT.) EFFECT. VELOC. RATIO = .100

X/C	ETA = .250 CP		ETA = .387 CP		ETA = .524 CP		ETA = .800 CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			-6.247		-3.914		-.235	
.010			-2.864	-5.401				
.015					1.076	-4.097		
.025			.808	-4.408	.879	-3.569	.618	-2.053
.050			.470	-2.991	.583	-2.476	.378	-1.631
.100			.237	-2.758	.329	-1.983	.244	-1.158
.150			.068	-2.653	.153	-1.941	.068	-1.095
.250	-.080		-.087	-2.286	-.017	-1.934	-.080	-1.090
.400	-.223		-.213	-2.129	-.154	-1.770	-.158	-.923
.550	-.202	-3.554	-.195	-2.238	-.161	-1.463	-.120	-.715
.700	-.247	-3.605	-.219	-2.040	-.158	-1.128	-.069	-.500
.850	-.342	-3.038	-.281	-1.446	-.134	-.674	-.018	-.284
.950	-.735	-2.635	-.431	-1.005	-.216	-.394	-.004	-.093

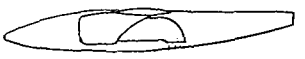
TABLE 28 (D) CONCLUDED
PRESSURE PROFILE ON WING

FORWARD NOZZLES DEFLECTED 90 DEGREES

BETA = 12.00 DEG Q = 223 N/SQ.M. (4.66 LBF/SQ.FT.) EFFECT. VELOC. RATIO = .100

	ETA = .250		ETA = .387		ETA = .524		ETA = .800	
X/C	CP		CP		CP		CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			-6.115		-4.914		-.198	
.010			-2.952	-5.896				
.015						-4.900		
.025			.889	-5.232	.946	-4.194	.670	-1.949
.050			.579	-3.354	.543	-2.613	.367	-1.659
.100			.282	-2.690	.360	-2.238	.226	-1.208
.150			.070	-2.613	.176	-2.062	.063	-1.208
.250	-.078		-.071	-2.240	-.050	-1.898	-.092	-1.049
.400	-.204		-.228	-1.973	-.184	-1.741	-.163	-.936
.550	-.214	-2.353	-.225	-2.305	-.160	-1.419	-.132	-.700
.700	-.266	-2.962	-.211	-1.843	-.166	-1.053	-.060	-.502
.850	-.406	-2.819	-.283	-1.487	-.156	-.731	-.026	-.273
.950	-.776	-2.606	-.433	-.967	-.218	-.430	-.006	-.108

TABLE 29 (A)
PRESSURE PROFILE ON WING



REAR NOZZLES DEFLECTED 90 DEGREES

BETA = -12.01 DEG Q = 2844 N/SQ.M. (59.40 LBF/SQ.FT.) EFFECT. VELOC. RATIO = ∞

	ETA = .250		ETA = .387		ETA = .524		ETA = .800	
X/C	CP		CP		CP		CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			.994		.993		.988	
.010			.313	.528				
.015					-.734	.036		
.025			-.828	.143	-.498	.001	-.403	-.134
.050			-.681	.061	-.471	-.083	-.320	-.243
.100			-.484	-.052	-.386	-.141	-.321	-.187
.150			-.499	-.244	-.486	-.231	-.284	-.249
.250	-.344		-.454	-.320	-.407	-.274	-.391	-.302
.400	-.387		-.421	-.312	-.420	-.285	-.337	-.289
.550	-.283	-.112	-.304	-.198	-.299	-.200	-.250	-.240
.700	-.211	-.053	-.204	-.170	-.208	-.163	-.167	-.159
.850	-.145	-.364	-.136	-.236	-.091	-.091	-.041	-.047
.950	-.097	-.136	-.086	-.140	-.020	-.012	.012	.032

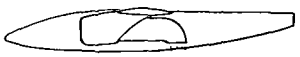
BETA = -8.01 DEG Q = 2850 N/SQ.M. (59.52 LBF/SQ.FT.) EFFECT. VELOC. RATIO = ∞

	ETA = .250		ETA = .387		ETA = .524		ETA = .800	
X/C	CP		CP		CP		CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			.965		.989		.973	
.010			.360	.425				
.015					-.348	-.103		
.025			-.837	.022	-.429	.023	-.249	-.172
.050			-.509	-.088	-.520	-.095	-.276	-.197
.100			-.432	-.191	-.423	-.225	-.298	-.215
.150			-.463	-.308	-.405	-.287	-.306	-.246
.250	-.328		-.414	-.353	-.408	-.275	-.374	-.335
.400	-.377		-.406	-.335	-.398	-.293	-.356	-.303
.550	-.298	-.206	-.313	-.241	-.297	-.239	-.246	-.246
.700	-.216	-.095	-.197	-.184	-.184	-.176	-.161	-.168
.850	-.158	-.334	-.139	-.238	-.076	-.091	-.046	-.048
.950	-.112	-.133	-.080	-.131	-.020	-.017	.015	.036

BETA = -4.01 DEG Q = 2851 N/SQ.M. (59.55 LBF/SQ.FT.) EFFECT. VELOC. RATIO = ∞

	ETA = .250		ETA = .387		ETA = .524		ETA = .800	
X/C	CP		CP		CP		CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			.781		.974		.975	
.010			.393	.078				
.015					-.155	-.090		
.025			-.502	-.157	-.447	-.280	-.240	-.180
.050			-.395	-.117	-.315	-.202	-.207	-.158
.100			-.415	-.303	-.335	-.230	-.259	-.277
.150			-.452	-.394	-.345	-.280	-.337	-.298
.250	-.393		-.414	-.365	-.413	-.358	-.380	-.370
.400	-.385		-.411	-.332	-.380	-.326	-.342	-.331
.550	-.292	-.211	-.300	-.253	-.286	-.262	-.257	-.249
.700	-.223	-.134	-.213	-.197	-.178	-.176	-.144	-.177
.850	-.161	-.285	-.137	-.231	-.073	-.093	-.042	-.057
.950	-.115	-.109	-.082	-.122	-.014	-.017	.016	.032

TABLE 29 (A) Continued
PRESSURE PROFILE ON WING



REAR NOZZLES DEFLECTED 90 DEGREES

BETA = 0.00 DEG Q = 2842 N/SQ.M. (59.36 LBF/SQ.FT.) EFFECT. VELOC. RATIO = ∞

	ETA = .250		ETA = .387		ETA = .524		ETA = .800	
X/C	CP		CP		CP		CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			.661		.938		.913	
.010			.405	-.199				
.015					-.193	-.433		
.025			-.360	-.377	-.357	-.342	-.361	-.255
.050			-.340	-.352	-.323	-.333	-.304	-.181
.100			-.320	-.397	-.417	-.267	-.317	-.213
.150			-.349	-.475	-.381	-.346	-.356	-.309
.250	-.319		-.371	-.411	-.360	-.355	-.360	-.346
.400	-.384		-.385	-.346	-.364	-.299	-.325	-.316
.550	-.306	-.222	-.308	-.247	-.277	-.253	-.256	-.248
.700	-.230	-.148	-.202	-.196	-.172	-.167	-.153	-.168
.850	-.176	-.305	-.137	-.225	-.074	-.101	-.050	-.062
.950	-.134	-.113	-.078	-.112	-.014	-.015	.017	.031

BETA = 4.01 DEG Q = 2850 N/SQ.M. (59.52 LBF/SQ.FT.) EFFECT. VELOC. RATIO = ∞

	ETA = .250		ETA = .387		ETA = .524		ETA = .800	
X/C	CP		CP		CP		CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			.324		.877		.926	
.010			.391	-.467				
.015					-.209	-.166		
.025			-.170	-.575	-.327	-.519	-.360	-.351
.050			-.305	-.558	-.326	-.333	-.388	-.251
.100			-.291	-.501	-.257	-.253	-.271	-.228
.150			-.350	-.469	-.321	-.380	-.289	-.334
.250	-.284		-.368	-.423	-.349	-.355	-.389	-.316
.400	-.357		-.375	-.336	-.358	-.356	-.328	-.335
.550	-.283	-.217	-.287	-.237	-.284	-.267	-.260	-.270
.700	-.222	-.139	-.197	-.202	-.179	-.165	-.155	-.181
.850	-.163	-.306	-.135	-.225	-.078	-.094	-.040	-.064
.950	-.121	-.104	-.075	-.111	-.010	-.013	.022	.033

BETA = 7.99 DEG Q = 2850 N/SQ.M. (59.52 LBF/SQ.FT.) EFFECT. VELOC. RATIO = ∞

	ETA = .250		ETA = .387		ETA = .524		ETA = .800	
X/C	CP		CP		CP		CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			.026		.669		.888	
.010			.372	-1.065				
.015					-.035	-.614		
.025			-.028	-1.012	-.160	-.624	-.461	-.142
.050			-.082	-.687	-.295	-.412	-.108	-.367
.100			-.181	-.623	-.239	-.385	-.227	-.349
.150			-.223	-.551	-.316	-.381	-.304	-.331
.250	-.258		-.341	-.462	-.337	-.359	-.325	-.428
.400	-.346		-.354	-.362	-.330	-.355	-.330	-.350
.550	-.280	-.216	-.276	-.245	-.271	-.265	-.238	-.262
.700	-.214	-.155	-.198	-.212	-.187	-.161	-.155	-.176
.850	-.163	-.327	-.124	-.221	-.074	-.102	-.038	-.051
.950	-.118	-.098	-.071	-.108	-.008	-.012	.020	.032

TABLE 29 (A) CONCLUDED
PRESSURE PROFILE ON WING

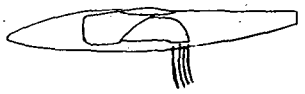
REAR NOZZLES DEFLECTED 90 DEGREES

BETA = 12.00 DEG Q = 2836 N/SQ.M. (59.23 LBF/SQ.FT.) EFFECT. VELOC. RATIO = ∞

	ETA = .250		ETA = .387		ETA = .524		ETA = .800	
X/C	CP		CP		CP		CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			-.875		.478		.829	
.010			.258	-1.660				
.015					.171	-.798		
.025			.183	-1.015	.055	-.639	-.269	-.386
.050			-.041	-.992	-.232	-.474	-.197	-.340
.100			-.172	-.684	-.133	-.484	-.209	-.329
.150			-.194	-.613	-.226	-.468	-.256	-.367
.250	-.235		-.288	-.455	-.278	-.357	-.285	-.378
.400	-.328		-.297	-.370	-.310	-.370	-.306	-.345
.550	-.266	-.233	-.241	-.255	-.245	-.259	-.220	-.270
.700	-.209	-.155	-.181	-.221	-.159	-.175	-.153	-.171
.850	-.149	-.322	-.121	-.225	-.067	-.099	-.025	-.055
.950	-.103	-.091	-.069	-.107	.015	-.004	.031	.034

TABLE 29 (B)
PRESSURE PROFILE ON WING

REAR NOZZLES DEFLECTED 90 DEGREES



BETA = -11.98 DEG Q = 2041 N/SQ.M. (42.62 LBF/SQ.FT.) EFFECT. VELOC. RATIO = .294

	ETA = .250		ETA = .387		ETA = .524		ETA = .800	
X/C	CP		CP		CP		CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			.788		.869		.730	
.010			.317	.822				
.015					-1.163	.513		
.025			-1.252	.510	-1.092	.213	-.754	.138
.050			-1.052	.316	-.831	.294	-.594	.119
.100			-.713	.067	-.701	.058	-.463	-.063
.150			-.662	-.011	-.662	-.058	-.486	-.205
.250	-.450		-.631	-.203	-.575	-.190	-.521	-.242
.400	-.514		-.580	-.297	-.522	-.265	-.479	-.285
.550	-.426	-.067	-.453	-.253	-.431	-.273	-.348	-.247
.700	-.381	-.168	-.355	-.360	-.313	-.279	-.206	-.184
.850	-.342	-.956	-.303	-.572	-.226	-.266	-.129	-.130
.950	-.339	-.548	-.285	-.431	-.177	-.208	-.056	-.050

BETA = -8.01 DEG Q = 2034 N/SQ.M. (42.49 LBF/SQ.FT.) EFFECT. VELOC. RATIO = .294

	ETA = .250		ETA = .387		ETA = .524		ETA = .800	
X/C	CP		CP		CP		CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			.970		.861		.927	
.010			.428	.775				
.015					-1.074	.402		
.025			-1.276	.327	-.984	.124	-.736	.058
.050			-.937	.162	-.844	.064	-.521	-.039
.100			-.626	.013	-.602	.028	-.449	-.054
.150			-.655	-.177	-.614	-.167	-.502	-.188
.250	-.446		-.615	-.297	-.565	-.227	-.469	-.293
.400	-.518		-.547	-.315	-.548	-.336	-.447	-.319
.550	-.437	-.187	-.433	-.311	-.416	-.327	-.350	-.273
.700	-.392	-.242	-.355	-.413	-.330	-.319	-.219	-.218
.850	-.355	-.989	-.316	-.619	-.235	-.297	-.119	-.135
.950	-.358	-.574	-.292	-.444	-.182	-.230	-.064	-.041

BETA = -3.97 DEG Q = 2031 N/SQ.M. (42.43 LBF/SQ.FT.) EFFECT. VELOC. RATIO = .294

	ETA = .250		ETA = .387		ETA = .524		ETA = .800	
X/C	CP		CP		CP		CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			.986		.945		.910	
.010			.579	.628				
.015					-.821	.257		
.025			-.953	.123	-.906	-.037	-.701	.011
.050			-.773	.057	-.635	.101	-.749	-.068
.100			-.536	-.006	-.569	-.150	-.495	-.132
.150			-.630	-.289	-.553	-.165	-.437	-.218
.250	-.427		-.534	-.385	-.542	-.322	-.505	-.251
.400	-.496		-.562	-.363	-.555	-.373	-.469	-.332
.550	-.445	-.207	-.448	-.369	-.410	-.371	-.344	-.303
.700	-.383	-.298	-.379	-.453	-.321	-.357	-.230	-.219
.850	-.371	-1.022	-.322	-.653	-.229	-.324	-.126	-.145
.950	-.382	-.601	-.305	-.471	-.189	-.232	-.066	-.050

TABLE 29 (B) CONTINUED
PRESSURE PROFILE ON WING

REAR NOZZLES DEFLECTED 90 DEGREES

BETA = .01 DEG Q = 2034 N/SQ.M. (42.48 LBF/SQ.FT.) EFFECT. VELOC. RATIO = .294

	ETA = .250		ETA = .387		ETA = .524		ETA = .800	
X/C	CP		CP		CP		CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			.907		.960		.911	
.010			.654	.177				
.015					-.638	.287		
.025			-.871	-.070	-1.017	-.125	-.514	.097
.050			-.579	-.190	-.566	-.122	-.575	-.031
.100			-.562	-.297	-.573	-.163	-.462	-.152
.150			-.565	-.372	-.555	-.287	-.520	-.233
.250	-.424		-.524	-.377	-.519	-.351	-.496	-.321
.400	-.486		-.549	-.412	-.505	-.379	-.431	-.360
.550	-.430	-.232	-.432	-.401	-.412	-.388	-.341	-.306
.700	-.390	-.355	-.349	-.495	-.302	-.374	-.211	-.250
.850	-.377	-1.047	-.329	-.676	-.224	-.337	-.123	-.165
.950	-.397	-.644	-.310	-.490	-.187	-.238	-.061	-.066

BETA = 4.00 DEG Q = 2040 N/SQ.M. (42.60 LBF/SQ.FT.) EFFECT. VELOC. RATIO = .294

	ETA = .250		ETA = .387		ETA = .524		ETA = .800	
X/C	CP		CP		CP		CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			.820		.918		.876	
.010			.657	.008				
.015					-.640	.029		
.025			-.599	-.200	-.669	-.234	-.481	-.010
.050			-.531	-.273	-.559	-.100	-.612	-.160
.100			-.501	-.389	-.527	-.226	-.416	-.147
.150			-.512	-.430	-.496	-.260	-.502	-.279
.250	-.386		-.473	-.415	-.485	-.365	-.488	-.304
.400	-.494		-.496	-.415	-.487	-.406	-.449	-.372
.550	-.428	-.226	-.433	-.427	-.410	-.420	-.339	-.300
.700	-.392	-.349	-.351	-.499	-.321	-.392	-.221	-.228
.850	-.378	-1.068	-.317	-.680	-.230	-.352	-.126	-.157
.950	-.402	-.655	-.301	-.495	-.185	-.240	-.061	-.058

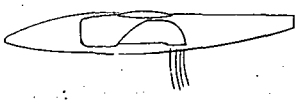
BETA = 8.02 DEG Q = 2038 N/SQ.M. (42.57 LBF/SQ.FT.) EFFECT. VELOC. RATIO = .294

	ETA = .250		ETA = .387		ETA = .524		ETA = .800	
X/C	CP		CP		CP		CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			.437		.903		.788	
.010			.627	-.477				
.015					-.591	-.126		
.025			-.323	-.365	-.478	-.235	-.576	-.174
.050			-.325	-.364	-.472	-.202	-.337	-.147
.100			-.312	-.583	-.402	-.216	-.294	-.174
.150			-.414	-.559	-.409	-.375	-.448	-.302
.250	-.346		-.510	-.515	-.464	-.388	-.489	-.402
.400	-.481		-.497	-.449	-.438	-.449	-.426	-.405
.550	-.421	-.229	-.402	-.374	-.382	-.431	-.336	-.317
.700	-.380	-.370	-.348	-.509	-.312	-.391	-.210	-.244
.850	-.373	-1.099	-.312	-.680	-.228	-.347	-.125	-.153
.950	-.385	-.629	-.301	-.503	-.176	-.240	-.055	-.051

TABLE 29 (B) CONCLUDED

PRESSURE PROFILE ON WING

REAR NOZZLES DEFLECTED 90 DEGREES



BETA = 12.02 DEG Q = 2031 N/SQ.M. (42.42 LBF/SQ.FT.) EFFECT. VELOC. RATIO = .293

	ETA = .250		ETA = .387		ETA = .524		ETA = .800	
X/C	CP		CP		CP		CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			.109		.832		.730	
.010			.574	-.901				
.015					-.028	-.218		
.025			-.148	-.578	-.373	-.260	-.616	-.254
.050			-.287	-.655	-.381	-.263	-.559	-.208
.100			-.228	-.585	-.381	-.286	-.391	-.160
.150			-.381	-.575	-.384	-.365	-.363	-.214
.250	-.313		-.410	-.560	-.400	-.380	-.400	-.313
.400	-.463		-.477	-.433	-.434	-.408	-.412	-.364
.550	-.402	-.214	-.392	-.422	-.371	-.410	-.325	-.331
.700	-.364	-.376	-.349	-.512	-.310	-.392	-.222	-.268
.850	-.368	-1.129	-.312	-.684	-.206	-.336	-.117	-.149
.950	-.382	-.638	-.295	-.493	-.173	-.229	-.050	-.043

TABLE 29 (C)
PRESSURE PROFILE ON WING

REAR NOZZLES DEFLECTED 90 DEGREES

BETA = -12.01 DEG Q = 912 N/SQ.M. (19.05 LBF/SQ.FT.) EFFECT. VELOC. RATIO = .197

	ETA = .250		ETA = .387		ETA = .524		ETA = .800	
X/C	CP		CP		CP		CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			.874		.741		.812	
.010			.306	.816				
.015					-1.187	.321		
.025			-1.503	.448	-1.009	.235	-.739	-.010
.050			-1.038	.235	-.895	.019	-.572	-.007
.100			-.776	-.010	-.774	-.053	-.534	-.117
.150			-.746	-.147	-.703	-.159	-.563	-.230
.250	-.539		-.696	-.346	-.648	-.310	-.555	-.352
.400	-.579		-.656	-.454	-.601	-.389	-.483	-.374
.550	-.508	-.348	-.523	-.443	-.497	-.429	-.416	-.339
.700	-.459	-.427	-.457	-.564	-.404	-.427	-.294	-.285
.850	-.461	-1.286	-.423	-.815	-.316	-.411	-.188	-.224
.950	-.498	-.965	-.409	-.619	-.280	-.347	-.131	-.154

BETA = -8.00 DEG Q = 908 N/SQ.M. (18.96 LBF/SQ.FT.) EFFECT. VELOC. RATIO = .196

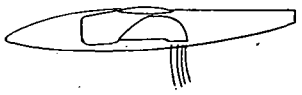
	ETA = .250		ETA = .387		ETA = .524		ETA = .800	
X/C	CP		CP		CP		CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			.955		.936		.785	
.010			.433	.703				
.015					-.919	.323		
.025			-1.329	.205	-1.226	.217	-.742	.016
.050			-.969	.067	-.794	-.074	-.643	-.104
.100			-.770	-.164	-.680	-.180	-.534	-.147
.150			-.749	-.227	-.642	-.218	-.515	-.241
.250	-.473		-.701	-.427	-.638	-.389	-.548	-.357
.400	-.565		-.629	-.524	-.601	-.475	-.496	-.402
.550	-.503	-.356	-.540	-.523	-.519	-.493	-.387	-.376
.700	-.466	-.492	-.455	-.646	-.406	-.485	-.286	-.312
.850	-.466	-1.334	-.451	-.878	-.312	-.447	-.193	-.238
.950	-.513	-1.003	-.445	-.667	-.309	-.387	-.135	-.143

BETA = -4.00 DEG Q = 905 N/SQ.M. (18.91 LBF/SQ.FT.) EFFECT. VELOC. RATIO = .196

	ETA = .250		ETA = .387		ETA = .524		ETA = .800	
X/C	CP		CP		CP		CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			.958		.947		.897	
.010			.401	.317				
.015					-.879	.119		
.025			-.999	.241	-.811	-.142	-.658	-.047
.050			-.708	.018	-.870	-.059	-.546	-.041
.100			-.600	-.195	-.665	-.205	-.482	-.229
.150			-.631	-.445	-.626	-.318	-.532	-.259
.250	-.473		-.607	-.536	-.588	-.387	-.569	-.422
.400	-.562		-.609	-.571	-.613	-.521	-.519	-.449
.550	-.510	-.381	-.517	-.558	-.488	-.539	-.421	-.399
.700	-.480	-.553	-.461	-.657	-.402	-.525	-.298	-.324
.850	-.477	-1.373	-.429	-.876	-.318	-.491	-.190	-.247
.950	-.541	-.955	-.428	-.677	-.294	-.385	-.140	-.140

TABLE 29 (C) CONTINUED
PRESSURE PROFILE ON WING

REAR NOZZLES DEFLECTED 90 DEGREES



BETA = .03 DEG Q = 911 N/SQ.M. (19.04 LBF/SQ.FT.) EFFECT. VELOC. RATIO = .196

X/C	ETA = .250 CP		ETA = .387 CP		ETA = .524 CP		ETA = .800 CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			.903		.941		.868	
.010			.372	.144				
.015					-.577	-.091		
.025			-.812	-.185	-.783	-.161	-.672	-.033
.050			-.600	-.252	-.613	-.278	-.505	-.133
.100			-.515	-.349	-.570	-.178	-.432	-.254
.150			-.593	-.472	-.553	-.418	-.461	-.349
.250	-.451		-.613	-.617	-.605	-.507	-.543	-.464
.400	-.568		-.589	-.612	-.603	-.577	-.497	-.489
.550	-.514	-.424	-.529	-.599	-.511	-.585	-.430	-.436
.700	-.474	-.595	-.465	-.725	-.415	-.553	-.301	-.346
.850	-.486	-1.421	-.434	-.926	-.321	-.519	-.207	-.255
.950	-.551	-.985	-.439	-.706	-.302	-.388	-.149	-.152

BETA = 4.00 DEG Q = 913 N/SQ.M. (19.07 LBF/SQ.FT.) EFFECT. VELOC. RATIO = .197

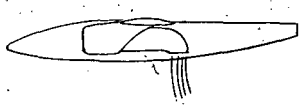
X/C	ETA = .250 CP		ETA = .387 CP		ETA = .524 CP		ETA = .800 CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			.756		.906		.865	
.010			.361	-.051				
.015					-.274	-.105		
.025			-.621	-.346	-.566	-.300	-.664	-.227
.050			-.510	-.369	-.559	-.215	-.541	-.251
.100			-.583	-.538	-.566	-.346	-.534	-.312
.150			-.595	-.590	-.576	-.396	-.464	-.402
.250	-.452		-.566	-.597	-.633	-.539	-.545	-.467
.400	-.578		-.581	-.652	-.575	-.602	-.520	-.472
.550	-.521	-.447	-.524	-.656	-.496	-.600	-.434	-.457
.700	-.490	-.634	-.472	-.754	-.427	-.568	-.301	-.358
.850	-.519	-1.452	-.441	-.949	-.326	-.539	-.209	-.271
.950	-.575	-.990	-.449	-.739	-.301	-.395	-.150	-.160

BETA = 8.02 DEG Q = 905 N/SQ.M. (18.90 LBF/SQ.FT.) EFFECT. VELOC. RATIO = .196

X/C	ETA = .250 CP		ETA = .387 CP		ETA = .524 CP		ETA = .800 CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			.370		.868		.864	
.010			.349	-.813				
.015					-.548	-.222		
.025			-.379	-.997	-.322	-.442	-.374	-.125
.050			-.362	-.527	-.548	-.323	-.402	-.268
.100			-.400	-.759	-.498	-.440	-.384	-.370
.150			-.491	-.628	-.499	-.538	-.463	-.423
.250	-.428		-.538	-.705	-.550	-.551	-.525	-.518
.400	-.553		-.573	-.686	-.526	-.632	-.502	-.535
.550	-.515	-.480	-.510	-.666	-.502	-.633	-.404	-.468
.700	-.502	-.682	-.459	-.791	-.428	-.605	-.317	-.377
.850	-.521	-1.529	-.448	-.973	-.327	-.547	-.198	-.279
.950	-.585	-1.056	-.454	-.766	-.304	-.404	-.140	-.164

TABLE 29 (C) CONCLUDED
PRESSURE PROFILE ON WING

REAR NOZZLES DEFLECTED 90 DEGREES

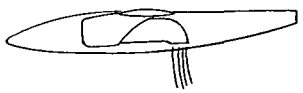


BETA = 12.00 DEG Q = 910 N/SQ.M. (19.00 LBF/SQ.FT.) EFFECT. VELOC. RATIO = .196

	ETA = .250		ETA = .387		ETA = .524		ETA = .800	
X/C	CP		CP		CP		CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			-.138		.803		.813	
.010			.283	-1.163				
.015					-.198	-.635		
.025			-.070	-.919	-.302	-.607	-.456	-.401
.050			-.259	-.851	-.475	-.449	-.420	-.316
.100			-.430	-.742	-.364	-.479	-.437	-.399
.150			-.363	-.773	-.408	-.584	-.461	-.479
.250	-.378		-.482	-.637	-.491	-.569	-.475	-.521
.400	-.516		-.547	-.740	-.523	-.678	-.481	-.505
.550	-.481	-.510	-.485	-.680	-.461	-.661	-.395	-.469
.700	-.485	-.720	-.445	-.817	-.414	-.615	-.313	-.395
.850	-.510	-1.557	-.436	-1.008	-.319	-.551	-.193	-.279
.950	-.595	-1.078	-.449	-.747	-.297	-.397	-.136	-.146

TABLE 29 (D)
PRESSURE PROFILE ON WING

REAR NOZZLES DEFLECTED 90 DEGREES



BETA = -11.98 DEG Q = 225 N/SQ.M. (4.69 LBF/SQ.FT.) EFFECT. VELOC. RATIO = .098

X/C	ETA = .250 CP		ETA = .387 CP		ETA = .524 CP		ETA = .800 CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			.873		.852		.852	
.010			-1.258	.452				
.015					-.851	-.059		
.025			-1.139	.109	-1.020	-.277	-.613	-.319
.050			-.971	-.024	-.795	-.270	-.683	-.389
.100			-.823	-.193	-.774	-.438	-.641	-.466
.150			-.809	-.529	-.746	-.543	-.613	-.564
.250	-.620		-.746	-.718	-.753	-.698	-.690	-.640
.400	-.701		-.776	-.769	-.803	-.732	-.674	-.688
.550	-.664	-.711	-.667	-.759	-.674	-.766	-.579	-.654
.700	-.623	-.759	-.623	-.868	-.606	-.735	-.525	-.558
.850	-.637	-1.500	-.613	-.987	-.538	-.664	-.426	-.494
.950	-.691	-1.089	-.650	-.874	-.538	-.609	-.406	-.474

BETA = -8.01 DEG Q = 227 N/SQ.M. (4.74 LBF/SQ.FT.) EFFECT. VELOC. RATIO = .098

X/C	ETA = .250 CP		ETA = .387 CP		ETA = .524 CP		ETA = .800 CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			.832		.853			
.010			-1.040	.278				
.015					-.673	-.368		
.025			-1.235	.035	-.846	-.395	-.631	-.368
.050			-1.006	-.236	-.624	-.368	-.659	-.479
.100			-.721	-.465	-.659	-.486	-.624	-.527
.150			-.791	-.645	-.714	-.659	-.624	-.652
.250	-.645		-.763	-.861	-.784	-.828	-.694	-.697
.400	-.683		-.761	-.885	-.747	-.855	-.670	-.717
.550	-.650	-.747	-.673	-.855	-.676	-.821	-.582	-.673
.700	-.619	-.835	-.629	-.908	-.586	-.784	-.505	-.582
.850	-.646	-1.584	-.636	-1.070	-.532	-.720	-.414	-.505
.950	-.724	-1.225	-.666	-.962	-.518	-.660	-.357	-.418

BETA = -3.98 DEG Q = 229 N/SQ.M. (4.79 LBF/SQ.FT.) EFFECT. VELOC. RATIO = .098

X/C	ETA = .250 CP		ETA = .387 CP		ETA = .524 CP		ETA = .800 CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			.724		.758		.820	
.010			-1.098	.098				
.015					-.576	-.191		
.025			-.734	-.418	-.665	-.748	-.583	-.603
.050			-.679	-.534	-.665	-.658	-.569	-.686
.100			-.658	-.754	-.631	-.686	-.672	-.693
.150			-.748	-.823	-.741	-.782	-.727	-.761
.250	-.638		-.775	-.973	-.734	-.853	-.734	-.806
.400	-.699		-.786	-.963	-.733	-.946	-.676	-.773
.550	-.666	-.763	-.696	-.923	-.669	-.899	-.589	-.696
.700	-.656	-.856	-.653	-.993	-.599	-.866	-.519	-.626
.850	-.679	-1.613	-.633	-1.106	-.539	-.776	-.433	-.586
.950	-.773	-1.229	-.683	-1.019	-.523	-.719	-.379	-.436

TABLE 29 (D) CONTINUED
PRESSURE PROFILE ON WING

REAR NOZZLES DEFLECTED 90 DEGREES

BETA = .02 DEG Q = 228 N/SQ.M. (4.76 LBF/SQ.FT.) EFFECT. VELOC. RATIO = .098

X/C	ETA = .250 CP UPPER	ETA = .250 CP LOWER	ETA = .387 CP UPPER	ETA = .387 CP LOWER	ETA = .524 CP UPPER	ETA = .524 CP LOWER	ETA = .800 CP UPPER	ETA = .800 CP LOWER
.000			.601		.656		.801	
.010			-1.134	-.374				
.015					-.498	-.678		
.025			-.761	-.844	-.768	-.733	-.699	-.561
.050			-.706	-.955	-.616	-.644	-.623	-.637
.100			-.678	-.775	-.678	-.727	-.630	-.651
.150			-.678	-.941	-.692	-.844	-.637	-.803
.250	-.630		-.699	-1.018	-.761	-.948	-.733	-.840
.400	-.720		-.757	-1.082	-.787	-.998	-.730	-.844
.550	-.686	-.854	-.710	-1.018	-.700	-.991	-.619	-.780
.700	-.680	-.948	-.676	-1.068	-.613	-.924	-.535	-.683
.850	-.710	-1.651	-.653	-1.169	-.576	-.850	-.465	-.582
.950	-.814	-1.333	-.700	-1.085	-.542	-.720	-.411	-.428

BETA = 4.00 DEG Q = 229 N/SQ.M. (4.79 LBF/SQ.FT.) EFFECT. VELOC. RATIO = .098

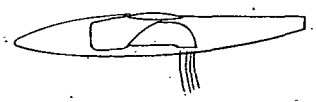
X/C	ETA = .250 CP UPPER	ETA = .250 CP LOWER	ETA = .387 CP UPPER	ETA = .387 CP LOWER	ETA = .524 CP UPPER	ETA = .524 CP LOWER	ETA = .800 CP UPPER	ETA = .800 CP LOWER
.000			.091		.696		.738	
.010			-1.188	-1.050				
.015					-.459	-.899		
.025			-.548	-1.215	-.445	-.768	-.603	-.837
.050			-.562	-1.057	-.645	-.872	-.562	-.858
.100			-.583	-.961	-.631	-.810	-.590	-.741
.150			-.645	-1.009	-.645	-.934	-.693	-.837
.250	-.665		-.741	-1.158	-.720	-1.012	-.713	-.915
.400	-.728		-.758	-1.128	-.795	-1.068	-.715	-.888
.550	-.708	-.902	-.708	-1.058	-.731	-1.035	-.651	-.801
.700	-.705	-.955	-.681	-1.128	-.658	-.975	-.568	-.725
.850	-.731	-1.715	-.688	-1.235	-.588	-.902	-.488	-.611
.950	-.818	-1.265	-.705	-1.128	-.575	-.721	-.421	-.455

BETA = 8.01 DEG Q = 225 N/SQ.M. (4.70 LBF/SQ.FT.) EFFECT. VELOC. RATIO = .098

X/C	ETA = .250 CP UPPER	ETA = .250 CP LOWER	ETA = .387 CP UPPER	ETA = .387 CP LOWER	ETA = .524 CP UPPER	ETA = .524 CP LOWER	ETA = .800 CP UPPER	ETA = .800 CP LOWER
.000			-.660		.033		.676	
.010			-1.220	-1.857				
.015					-.212	-1.024		
.025			-.212	-1.514	-.408	-1.262	-.569	-1.010
.050			-.485	-1.388	-.373	-1.038	-.569	-.968
.100			-.520	-1.339	-.583	-.996	-.520	-.947
.150			-.632	-1.374	-.597	-1.150	-.709	-.905
.250	-.646		-.688	-1.241	-.688	-1.183	-.765	-.980
.400	-.725		-.742	-1.187	-.729	-1.149	-.742	-.987
.550	-.712	-.970	-.708	-1.136	-.688	-1.105	-.630	-.878
.700	-.712	-1.034	-.685	-1.193	-.641	-1.051	-.569	-.800
.850	-.763	-1.695	-.691	-1.271	-.593	-.939	-.515	-.658
.950	-.858	-1.309	-.729	-1.054	-.586	-.729	-.454	-.505

TABLE 29 (D) Concluded
PRESSURE PROFILE ON WING

REAR NOZZLES DEFLECTED 90 DEGREES



BETA = 12.01 DEG Q = 226 N/SQ.M. (4.71 LBF/SQ.FT.) EFFECT. VELOC. RATIO = .098

	ETA = .250		ETA = .387		ETA = .524		ETA = .800	
X/C	CP		CP		CP		CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			-1.309		-.157		.535	
.010			-1.274	-2.322				
.015					-.031	-1.609		
.025			-.052	-1.931	-.240	-1.589	-.317	-1.232
.050			-.317	-1.742	-.373	-1.190	-.457	-1.009
.100			-.443	-1.414	-.478	-1.239	-.527	-.953
.150			-.569	-1.463	-.597	-1.197	-.583	-.988
.250	-.604		-.639	-1.311	-.673	-1.189	-.653	-1.040
.400	-.732		-.715	-1.226	-.725	-1.209	-.701	-1.023
.550	-.705	-1.006	-.701	-1.165	-.681	-1.155	-.623	-.931
.700	-.715	-1.094	-.677	-1.209	-.644	-1.043	-.556	-.833
.850	-.765	-1.741	-.688	-1.304	-.596	-.955	-.495	-.664
.950	-.874	-1.253	-.715	-1.019	-.583	-.738	-.440	-.525

TABLE 30 (A)

PRESSURE PROFILE ON WING

LIFT JET ALONE

ALPHA = .00 DEG Q = 693 N/SQ.M. (14.47 LBF/SQ.FT.) EFFECT. VELOC. RATIO = .099

	ETA = .250		ETA = .337		ETA = .524		ETA = .800	
X/C	CP		CP		CP		CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000	.55		.18		.68		.93	
.010	.35	-.75	-.96	-.58				
.015					.14	-.87		
.025	.009	-.705	-.091	-.669	-.082	-.696	-.016	-.471
.050	-.050	-.596	.022	-.587	-.068	-.466	-.112	-.428
.100	-.189	-.534	-.155	-.509	-.171	-.441	-.191	-.396
.150	-.207	-.507	-.198	-.509	-.184	-.489	-.159	-.423
.250	-.257	-.520	-.230	-.434	-.241	-.442	-.250	-.403
.400	-.232	-.387	-.270	-.439	-.225	-.366	-.263	-.344
.550	-.157	-.338	-.182	-.304	-1.206	-.302	-.189	-.289
.700	-.093	-.180	-.123	-.174	-.116	-.180	-.124	-.163
.850	-.022	-.061	-.019	-.072	-.002	-.076	-.015	-.063
.950	.050	.035	.038	.029	.056	.025	.035	.036

ALPHA = -.01 DEG Q = 1549 N/SQ.M. (32.36 LBF/SQ.FT.) EFFECT. VELOC. RATIO = .148

	ETA = .250		ETA = .387		ETA = .524		ETA = .800	
X/C	CP		CP		CP		CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000	.86		.22		.73		.91	
.010	.25	-.60	.09	-.57				
.015					.06	-.58		
.025	-.099	-.708	-.122	-.663	-.336	-.478	-.225	-.396
.050	-.163	-.521	-.112	-.464	-.082	-.444	-.242	-.411
.100	-.198	-.401	-.146	-.470	-.173	-.407	-.169	-.308
.150	-.268	-.545	-.219	-.489	-.271	-.451	-.257	-.392
.250	-.305	-.513	-.238	-.487	-.269	-.484	-.290	-.362
.400	-.290	-.427	-.270	-.456	-.276	-.418	-.289	-.365
.550	-.187	-.350	-.192	-.335	-.092	-.282	-.207	-.266
.700	-.110	-.198	-.112	-.198	-.124	-.181	-.117	-.160
.850	-.041	-.075	-.030	-.074	-.009	-.071	-.019	-.053
.950	.040	.034	.036	.036	.040	.030	.036	.040

ALPHA = -.00 DEG Q = 2746 N/SQ.M. (57.36 LBF/SQ.FT.) EFFECT. VELOC. RATIO = .196

	ETA = .250		ETA = .387		ETA = .524		ETA = .800	
X/C	CP		CP		CP		CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000	.79		.39		.82		.98	
.010	.14	-.27	.49	-.17				
.015					.02	-.20		
.025	-.414	-.595	-.143	-.390	-.188	-.419	-.274	-.240
.050	-.202	-.316	-.221	-.456	-.184	-.363	-.251	-.290
.100	-.229	-.350	-.230	-.380	-.227	-.325	-.198	-.255
.150	-.273	-.450	-.265	-.432	-.260	-.373	-.255	-.324
.250	-.312	-.488	-.295	-.425	-.310	-.433	-.338	-.358
.400	-.284	-.427	-.315	-.432	-.317	-.386	-.301	-.347
.550	-.204	-.318	-.204	-.307	.275	-.292	-.218	-.260
.700	-.128	-.178	-.109	-.170	-.114	-.175	-.109	-.153
.850	-.045	-.060	-.037	-.069	-.025	-.059	-.025	-.056
.950	.039	.033	.032	.031	.035	.027	.033	.045

TABLE 30 (A) – CONCLUDED
PRESSURE PROFILE ON WING

LIFT JET ALONE

ALPHA = 0.00 DEG Q = 1272 N/SQ.M. (26.57 LBF/SQ.FT.) EFFECT. VELOC. RATIO = .205

	ETA = .250		ETA = .387		ETA = .524		ETA = .800	
X/C	CP		CP		CP		CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000	.82		.48		.89		.97	
.010	.14	-.39	-.02	-.34				
.015					-.13	-.49		
.025	-.376	-.658	-.221	-.669	-.138	-.489	-.304	-.490
.050	-.132	-.291	-.117	-.341	-.216	-.213	-.252	-.220
.100	-.216	-.422	-.187	-.366	-.199	-.308	-.211	-.294
.150	-.291	-.419	-.278	-.427	-.244	-.385	-.274	-.260
.250	-.310	-.450	-.302	-.433	-.292	-.389	-.289	-.317
.400	-.261	-.399	-.299	-.395	-.279	-.379	-.293	-.320
.550	-.186	-.330	-.189	-.289	-.648	-.278	-.195	-.223
.700	-.115	-.177	-.112	-.161	-.124	-.152	-.120	-.143
.850	-.031	-.053	-.020	-.059	-.001	-.056	-.001	-.040
.950	.046	.038	.041	.055	.051	.044	.056	.065

ALPHA = -.00 DEG Q = 2844 N/SQ.M. (59.39 LBF/SQ.FT.) EFFECT. VELOC. RATIO = .315

	ETA = .250		ETA = .387		ETA = .524		ETA = .800	
X/C	CP		CP		CP		CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000	.89		.73		.96		.96	
.010	-.08	-.04	.45	-.10				
.015					-.05	-.33		
.025	-.457	-.423	-.239	-.333	-.130	-.229	-.296	-.260
.050	-.218	-.243	-.269	-.226	-.309	-.225	-.196	-.192
.100	-.251	-.320	-.271	-.287	-.251	-.289	-.226	-.309
.150	-.298	-.406	-.261	-.355	-.276	-.326	-.259	-.333
.250	-.304	-.449	-.301	-.392	-.318	-.353	-.321	-.355
.400	-.296	-.434	-.309	-.426	-.312	-.358	-.305	-.321
.550	-.212	-.323	-.230	-.307	.015	-.266	-.216	-.246
.700	-.129	-.178	-.117	-.170	-.121	-.148	-.101	-.140
.850	-.045	-.067	-.043	-.059	-.015	-.054	-.012	-.044
.950	.037	.038	.031	.047	.043	.041	.043	.055

TABLE 30 (B)
PRESSURE PROFILE ON WING

LIFT JET ALONE

ALPHA = 10.02 DEG Q = 695 N/SQ.M. (14.51 LBF/SQ.FT.) EFFECT. VELOC. RATIO = .100

	ETA = .250		ETA = .387		ETA = .524		ETA = .800	
X/C	CP		CP		CP		CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000	-1.14		.09		-.50		-1.69	
.010	-2.94	.93	-.62	.95				
.015					-2.57	.88		
.025	-2.341	.742	-2.313	.751	-2.347	.783	-2.477	.790
.050	-1.361	.549	-1.463	.515	-1.436	.547	-1.255	.622
.100	-.987	.264	-.981	.323	-1.062	.427	-.987	.386
.150	-.865	.146	-.903	.153	-.928	.166	-.879	.171
.250	-.647	.033	-.704	.022	-.718	.049	-.688	.030
.400	-.508	-.127	-.538	-.061	-.547	-.050	-.514	-.059
.550	-.322	-.099	-.343	-.077	-1.553	-.050	-.338	-.059
.700	-.173	-.048	-.187	-.032	-.193	-.032	-.182	-.041
.850	-.047	.005	-.061	.012	-.049	-.010	-.072	.002
.950	.041	.045	.031	.055	.030	.035	.013	.036

ALPHA = 10.00 DEG Q = 1552 N/SQ.M. (32.41 LBF/SQ.FT.) EFFECT. VELOC. RATIO = .150

	ETA = .250		ETA = .387		ETA = .524		ETA = .800	
X/C	CP		CP		CP		CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000	-1.14		.02		-1.24		-2.31	
.010	-3.38	.97	.26	.94				
.015					-3.73	.94		
.025	-2.906	.796	-2.653	.771	-2.959	.786	-2.526	.834
.050	-1.460	.577	-1.528	.580	-1.576	.607	-1.662	.639
.100	-1.148	.336	-1.103	.332	-1.148	.401	-1.128	.400
.150	-.945	.136	-.991	.208	-.997	.216	-.943	.245
.250	-.738	-.007	-.772	.033	-.795	.044	-.770	.100
.400	-.553	-.112	-.582	-.085	-.587	-.061	-.560	-.046
.550	-.374	-.138	-.385	-.095	-.199	-.076	-.371	-.072
.700	-.218	-.070	-.220	-.039	-.226	-.043	-.210	-.040
.850	-.085	-.018	-.085	-.019	-.077	-.014	-.084	-.009
.950	.021	.029	.014	.042	.011	.027	.003	.036

ALPHA = 10.01 DEG Q = 2753 N/SQ.M. (57.50 LBF/SQ.FT.) EFFECT. VELOC. RATIO = .199

	ETA = .250		ETA = .387		ETA = .524		ETA = .800	
X/C	CP		CP		CP		CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000	-1.43		-.27		-1.84		-2.69	
.010	-3.60	.96	.56	.94				
.015					-3.50	.97		
.025	-3.163	.892	-2.805	.836	-2.896	.868	-2.346	.875
.050	-1.564	.637	-1.756	.641	-1.780	.678	-1.672	.709
.100	-1.200	.372	-1.200	.396	-1.213	.490	-1.168	.407
.150	-.986	.216	-1.104	.261	-1.034	.338	-1.003	.270
.250	-.751	.068	-.823	.067	-.838	.111	-.840	.090
.400	-.571	-.071	-.626	-.035	-.623	-.030	-.587	-.036
.550	-.374	-.081	-.412	-.079	.204	-.050	-.394	-.056
.700	-.219	-.033	-.237	-.037	-.233	-.025	-.225	-.028
.850	-.077	-.011	-.093	-.008	-.074	-.002	-.090	-.007
.950	.029	.033	.009	.040	.022	.027	.001	.029

TABLE 30 (B) – CONCLUDED

PRESSURE PROFILE ON WING

LIFT JET ALONE

ALPHA = 9.75 DEG Q = 1270 N/SQ.M. (26.52 LBF/SQ.FT.) EFFECT. VELOC. RATIO = .206

	ETA = .250		ETA = .387		ETA = .524		ETA = .800	
X/C	CP		CP		CP		CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000	-1.16		-.21		-1.35		-2.66	
.010	-3.26	.98	.07	.96				
.015					-2.99	.96		
.025	-2.787	.864	-2.985	.849	-2.876	.840	-2.628	.849
.050	-1.478	.599	-1.507	.606	-1.558	.651	-1.599	.680
.100	-1.049	.346	-1.144	.378	-1.091	.443	-1.127	.409
.150	-.912	.210	-.953	.200	-.990	.281	-.910	.254
.250	-.701	.010	-.750	.091	-.743	.088	-.743	.103
.400	-.531	-.083	-.584	-.075	-.581	-.040	-.537	-.017
.550	-.346	-.100	-.368	-.077	-.770	-.037	-.349	-.054
.700	-.198	-.035	-.202	-.028	-.208	-.019	-.195	-.032
.850	-.062	-.001	-.066	.000	-.059	.012	-.070	-.005
.950	.039	.046	.034	.048	.029	.052	.013	.034

ALPHA = 10.01 DEG Q = 2843 N/SQ.M. (59.38 LBF/SQ.FT.) EFFECT. VELOC. RATIO = .316

	ETA = .250		ETA = .387		ETA = .524		ETA = .800	
X/C	CP		CP		CP		CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000	-2.05		-.34		-1.92		-2.97	
.010	-3.58	.95	.34	.93				
.015					-3.57	.98		
.025	-3.237	.895	-2.748	.867	-2.712	.866	-2.314	.872
.050	-1.638	.684	-1.776	.700	-1.796	.708	-1.767	.704
.100	-1.230	.444	-1.260	.451	-1.302	.495	-1.204	.485
.150	-1.006	.289	-1.054	.300	-1.078	.298	-.979	.294
.250	-.771	.077	-.833	.126	-.828	.127	-.805	.137
.400	-.587	-.066	-.632	-.039	-.627	.004	-.583	-.006
.550	-.380	-.085	-.395	-.056	-.094	-.028	-.384	-.049
.700	-.219	-.049	-.223	-.034	-.227	-.020	-.213	-.032
.850	-.075	-.013	-.081	-.009	-.077	.003	-.081	-.005
.950	.028	.036	.018	.042	.020	.030	.006	.028

TABLE 31 (A)

PRESSURE PROFILE ON WING

FORWARD NOZZLES DEFLECTED 45 DEGREES

ALPHA = .01 DEG Q = 231 N/SQ.M. (4.82 LBF/SQ.FT.) EFFECT. VELOC. RATIO = .099

	ETA = .250		ETA = .387		ETA = .524		ETA = .800	
X/C	CP		CP		CP		CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			-3.924		-2.462		.666	
.010			-.605	-4.129				
.015					1.103	-2.961		
.025			.857	-3.289	.912	-2.708	.543	-1.028
.050			.604	-2.531	.666	-1.725	.454	-.666
.100			.406	-2.346	.406	-1.417	.317	-.489
.150			.283	-2.620	.290	-1.479	.242	-.461
.250	.153		.147	-2.588	.160	-1.376	.126	-.459
.400	.100		.057	-3.419	.067	-1.254	.087	-.360
.550	.130	-2.313	.117	-2.137	.097	-.959	.134	-.244
.700	.157	-.499	.124	-1.095	.117	-.638	.183	-.115
.850	.183	-.022	.100	-.436	.140	-.280	.230	.067
.950	.173	.137	.044	-.188	.100	-.058	.239	.180

ALPHA = .01 DEG Q = 510 N/SQ.M. (10.65 LBF/SQ.FT.) EFFECT. VELOC. RATIO = .148

	ETA = .250		ETA = .387		ETA = .524		ETA = .800	
X/C	CP		CP		CP		CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			-1.723		-.892		.917	
.010			.076	-2.750				
.015					.744	-2.107		
.025			.450	-2.094	.351	-1.634	.190	-.743
.050			.187	-1.813	.299	-1.250	.085	-.666
.100			.076	-1.748	.153	-1.130	-.008	-.508
.150			-.026	-1.583	-.023	-1.241	-.038	-.496
.250	-.103		-.125	-2.088	-.168	-1.231	-.153	-.516
.400	-.143		-.153	-2.491	-.158	-1.192	-.180	-.456
.550	-.104	-3.703	-.111	-2.088	-.110	-.962	-.089	-.391
.700	-.044	-1.797	-.080	-1.334	-.074	-.693	-.017	-.236
.850	-.015	-.243	-.077	-.642	-.032	-.414	.024	-.081
.950	-.005	-.098	-.093	-.305	-.026	-.170	.076	.037

ALPHA = 0.00 DEG Q = 911 N/SQ.M. (19.02 LBF/SQ.FT.) EFFECT. VELOC. RATIO = .197

	ETA = .250		ETA = .387		ETA = .524		ETA = .800	
X/C	CP		CP		CP		CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			-1.081		.045		.971	
.010			.370	-2.045				
.015					.464	-1.527		
.025			.153	-1.782	.294	-1.514	.059	-.661
.050			.087	-1.527	.040	-1.143	.014	-.543
.100			-.043	-1.430	-.015	-.975	-.078	-.417
.150			-.123	-1.610	-.074	-.956	-.116	-.493
.250	-.166		-.207	-1.766	-.211	-1.049	-.226	-.514
.400	-.241		-.254	-2.064	-.239	-1.083	-.209	-.459
.550	-.163	-3.030	-.189	-1.836	-.180	-.915	-.152	-.370
.700	-.117	-2.592	-.148	-1.331	-.140	-.677	-.106	-.234
.850	-.084	-.798	-.137	-.722	-.087	-.408	.003	-.090
.950	-.103	-.278	-.157	-.382	-.062	-.195	.048	.028

TABLE 31 (A) – CONCLUDED

PRESSURE PROFILE ON WING

FORWARD NOZZLES DEFLECTED 45 DEGREES

ALPHA = .01 DEG Q = 2032 N/SQ.M. (42.43 LBF/SQ.FT.) EFFECT. VELOC. RATIO = .295

	ETA = .250		ETA = .387		ETA = .524		ETA = .800	
X/C	CP		CP		CP		CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			-.195		.743		.970	
.010			.596	-1.284				
.015					.240	-1.042		
.025			-.052	-1.314	.024	-.996	-.167	-.477
.050			-.065	-1.009	-.042	-.745	-.069	-.330
.100			-.245	-1.059	-.218	-.745	-.179	-.378
.150			-.256	-1.205	-.242	-.778	-.206	-.451
.250	-.232		-.330	-1.240	-.276	-.878	-.306	-.487
.400	-.303		-.325	-1.485	-.321	-.861	-.296	-.428
.550	-.234	-1.889	-.256	-1.410	-.245	-.761	-.208	-.341
.700	-.182	-1.502	-.206	-1.134	-.190	-.567	-.118	-.214
.850	-.189	-1.686	-.163	-.732	-.087	-.350	-.028	-.094
.950	-.296	-1.048	-.176	-.376	-.073	-.157	.020	.021

ALPHA = .02 DEG Q = 2846 N/SQ.M. (59.44 LBF/SQ.FT.) EFFECT. VELOC. RATIO = .396

	ETA = .250		ETA = .387		ETA = .524		ETA = .800	
X/C	CP		CP		CP		CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			.095		.615		.995	
.010			.723	-.769				
.015					.122	-.774		
.025			-.198	-.836	-.105	-.749	-.177	-.335
.050			-.153	-.827	-.164	-.568	-.217	-.364
.100			-.221	-.921	-.247	-.504	-.226	-.322
.150			-.327	-.548	-.250	-.658	-.256	-.400
.250	-.256		-.331	-1.048	-.366	-.707	-.313	-.406
.400	-.321		-.354	-1.119	-.308	-.705	-.311	-.385
.550	-.260	-1.326	-.261	-1.043	-.255	-.607	-.239	-.302
.700	-.199	-1.319	-.184	-.849	-.166	-.422	-.110	-.190
.850	-.200	-1.312	-.155	-.584	-.069	-.267	-.027	-.075
.950	-.331	-1.073	-.141	-.283	-.035	-.100	.029	.031

ALPHA = .03 DEG Q = 2849 N/SQ.M. (59.50 LBF/SQ.FT.) EFFECT. VELOC. RATIO = .521

	ETA = .250		ETA = .387		ETA = .524		ETA = .800	
X/C	CP		CP		CP		CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			.308		.855		.998	
.010			.726	-.558				
.015					.032	-.721		
.025			-.212	-.742	-.084	-.655	-.162	-.360
.050			-.241	-.575	-.195	-.440	-.208	-.311
.100			-.217	-.670	-.202	-.463	-.212	-.323
.150			-.300	-.831	-.271	-.521	-.244	-.355
.250	-.253		-.345	-.867	-.317	-.596	-.290	-.378
.400	-.316		-.325	-.907	-.335	-.600	-.302	-.354
.550	-.245	-1.016	-.249	-.818	-.231	-.488	-.218	-.272
.700	-.186	-1.017	-.157	-.663	-.138	-.343	-.097	-.164
.850	-.183	-.999	-.112	-.454	-.047	-.198	-.013	-.056
.950	-.279	-.929	-.083	-.188	-.002	-.048	.046	.050

TABLE 31 (B)

PRESSURE PROFILE ON WING

FORWARD NOZZLES DEFLECTED 45 DEGREES

ALPHA = 10.04 DEG Q = 233 N/SQ.M. (4.87 LBF/SQ.FT.) EFFECT. VELOC. RATIO = .100

	ETA = .250		ETA = .387		ETA = .524		ETA = .800	
X/C	CP		CP		CP		CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000							-.152	
.010			-.321	.240				
.015					-.707	.335		
.025			-.598	-.172	-.774	-.044	-1.153	.855
.050			-.355	-.470	-.510	.105	-.734	.767
.100			-.254	-.977	-.321	-.254	-.389	.450
.150			-.260	-1.464	-.281	-.524	-.382	.240
.250	.017		-.145	-2.286	-.260	-.703	-.308	.097
.400	.002		-.093	-2.977	-.139	-.847	-.152	.012
.550	.074	-1.794	.012	-1.922	-.018	-.732	.021	.044
.700	.156	.018	.103	-.945	.107	-.503	.146	.107
.850	.215	.189	.139	-.332	.172	-.192	.231	.195
.950	.244	.274	.136	-.044	.175	.074	.284	.257

ALPHA = 10.05 DEG Q = 514 N/SQ.M. (10.73 LBF/SQ.FT.) EFFECT. VELOC. RATIO = .148

	ETA = .250		ETA = .387		ETA = .524		ETA = .800	
X/C	CP		CP		CP		CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000					.648		-1.389	
.010			.397	.903				
.015					-1.827	.737		
.025			-1.444	.299	-1.714	.440	-1.858	.906
.050			-1.051	-.042	-1.287	.311	-1.143	.664
.100			-.671	-.364	-.778	-.017	-.830	.378
.150			-.640	-.784	-.738	-.303	-.744	.219
.250	-.215		-.527	-1.397	-.579	-.576	-.600	.001
.400	-.314		-.417	-2.104	-.482	-.800	-.414	-.084
.550	-.194	-3.420	-.271	-1.819	-.305	-.741	-.241	-.102
.700	-.112	-1.199	-.172	-1.184	-.172	-.606	-.096	-.054
.850	-.039	-.072	-.117	-.580	-.072	-.372	.007	.007
.950	-.014	.007	-.097	-.298	-.038	-.152	.084	.062

ALPHA = 10.04 DEG Q = 908 N/SQ.M. (18.96 LBF/SQ.FT.) EFFECT. VELOC. RATIO = .197

	ETA = .250		ETA = .387		ETA = .524		ETA = .800	
X/C	CP		CP		CP		CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			.908		-.043		-1.813	
.010			.646					
.015					-2.380	.804		
.025			-1.858	.469	-2.122	.637	-2.441	.868
.050			-1.277	.162	-1.355	.348	-1.518	.639
.100			-.883	-.255	-1.008	.037	-1.075	.391
.150			-.817	-.633	-.859	-.199	-.855	.200
.250	-.402		-.636	-1.154	-.708	-.493	-.701	.042
.400	-.419		-.526	-1.787	-.565	-.708	-.516	-.094
.550	-.294	-2.992	-.357	-1.654	-.397	-.699	-.332	-.124
.700	-.190	-2.203	-.245	-1.196	-.258	-.588	-.183	-.088
.850	-.127	-.246	-.186	-.674	-.135	-.385	-.055	-.028
.950	-.097	-.123	-.174	-.350	-.085	-.193	.029	.030

TABLE 31 (B) – CONCLUDED

PRESSURE PROFILE ON WING

FORWARD NOZZLES DEFLECTED 45 DEGREES

ALPHA = 10.00 DEG Q = 2037 N/SQ.M. (42.55 LBF/SQ.FT.) EFFECT. VELOC. RATIO = .295

	ETA = .250		ETA = .387		ETA = .524		ETA = .800	
X/C	CP		CP		CP		CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			.549		-.430		-2.439	
.010			.839					
.015					-2.936	.966		
.025			-2.272	.604	-2.504	.760	-2.392	.907
.050			-1.447	.231	-1.590	.515	-1.591	.718
.100			-.983	-.027	-1.142	.200	-1.145	.450
.150			-.886	-.322	-.987	-.039	-.943	.256
.250	-.449		-.693	-.790	-.793	-.284	-.745	.073
.400	-.461		-.556	-1.282	-.607	-.499	-.548	-.046
.550	-.330	-2.000	-.383	-1.288	-.405	-.505	-.354	-.071
.700	-.224	-2.007	-.262	-1.007	-.244	-.421	-.185	-.051
.850	-.162	-1.029	-.178	-.631	-.119	-.266	-.057	-.003
.950	-.161	-.457	-.158	-.317	-.045	-.123	.036	.055

ALPHA = 10.03 DEG Q = 2841 N/SQ.M. (59.35 LBF/SQ.FT.) EFFECT. VELOC. RATIO = .391

	ETA = .250		ETA = .387		ETA = .524		ETA = .800	
X/C	CP		CP		CP		CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			.517		-1.190		-3.106	
.010			.893					
.015					-3.048	1.005		
.025			-2.581	.756	-3.053	.855	-2.363	.956
.050			-1.436	.533	-1.631	.635	-1.631	.721
.100			-1.018	.186	-1.133	.315	-1.200	.494
.150			-.879	-.113	-1.006	.130	-.899	.352
.250	-.428		-.694	-.506	-.756	-.133	-.755	.133
.400	-.430		-.545	-.895	-.588	-.331	-.539	.028
.550	-.305	-1.299	-.351	-.907	-.371	-.339	-.328	-.029
.700	-.203	-1.335	-.225	-.729	-.216	-.282	-.161	.009
.850	-.150	-1.227	-.130	-.489	-.072	-.162	-.028	.042
.950	-.187	-.629	-.088	-.212	.011	-.036	.067	.088

ALPHA = 9.97 DEG Q = 2843 N/SQ.M. (59.39 LBF/SQ.FT.) EFFECT. VELOC. RATIO = .517

	ETA = .250		ETA = .387		ETA = .524		ETA = .800	
X/C	CP		CP		CP		CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			.340		-.983		-2.842	
.010			.895					
.015					-3.028			
.025			-2.532	.858	-2.944	.930	-2.254	.989
.050			-1.481	.673	-1.681	.732	-1.554	.778
.100			-.957	.311	-1.114	.445	-1.054	.552
.150			-.844	.042	-.900	.271	-.883	.366
.250	-.397		-.649	-.288	-.713	.035	-.683	.185
.400	-.371		-.487	-.641	-.518	-.161	-.485	.055
.550	-.246	-.977	-.291	-.648	-.307	-.186	-.274	.037
.700	-.144	-.585	-.155	-.501	-.143	-.122	-.108	.062
.850	-.083	-.969	-.047	-.312	.004	-.038	.022	.096
.950	-.089	-.487	.014	-.089	.091	.057	.116	.139

TABLE 32 (A)

PRESSURE PROFILE ON WING

REAR NOZZLES DEFLECTED 45 DEGREES

ALPHA = 0.00 DEG Q = 228 N/SQ.M. (4.75 LBF/SQ.FT.) EFFECT. VELOC. RATIO = .097

	ETA = .250		ETA = .387		ETA = .524		ETA = .800	
X/C	CP		CP		CP		CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			-.201		.311		.463	
.010			-4.038	-1.074				
.015					-.264	-1.039		
.025			-.638	-.991	-.405	-1.109	-.409	-.908
.050			-.556	-1.019	-.631	-.762	-.555	-.769
.100			-.596	-1.039	-.562	-.859	-.562	-.755
.150			-.679	-1.088	-.624	-.880	-.652	-.790
.250	-.679		-.749	-1.029	-.745	-.872	-.742	-.825
.400	-.737		-.744	-.972	-.727	-.915	-.724	-.811
.550	-.704	-.828	-.724	-.888	-.684	-.852	-.637	-.734
.700	-.697	-.892	-.684	-.939	-.647	-.784	-.576	-.657
.850	-.714	-1.509	-.670	-1.040	-.583	-.724	-.516	-.570
.950	-.791	-1.154	-.680	-.919	-.549	-.617	-.442	-.455

ALPHA = .01 DEG Q = 512 N/SQ.M. (10.69 LBF/SQ.FT.) EFFECT. VELOC. RATIO = .146

	ETA = .250		ETA = .387		ETA = .524		ETA = .800	
X/C	CP		CP		CP		CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			.321		.691		.749	
.010			-1.299	-.452				
.015					-.246	-.403		
.025			-.369	-.778	-.464	-.704	-.461	-.554
.050			-.452	-.640	-.418	-.378	-.437	-.477
.100			-.467	-.652	-.520	-.498	-.532	-.403
.150			-.566	-.674	-.554	-.575	-.492	-.443
.250	-.483		-.615	-.749	-.557	-.646	-.550	-.600
.400	-.573		-.588	-.674	-.588	-.610	-.570	-.546
.550	-.519	-.541	-.522	-.591	-.513	-.564	-.483	-.504
.700	-.497	-.577	-.473	-.639	-.449	-.522	-.400	-.422
.850	-.489	-1.143	-.441	-.755	-.352	-.453	-.277	-.312
.950	-.531	-.804	-.416	-.579	-.295	-.341	-.224	-.215

ALPHA = .01 DEG Q = 908 N/SQ.M. (18.96 LBF/SQ.FT.) EFFECT. VELOC. RATIO = .195

	ETA = .250		ETA = .387		ETA = .524		ETA = .800	
X/C	CP		CP		CP		CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			.455		.719		.627	
.010			-.319	-.072				
.015					-.327	-.251		
.025			-.511	-.687	-.466	-.614	-.364	-.201
.050			-.433	-.517	-.378	-.421	-.365	-.295
.100			-.432	-.576	-.430	-.428	-.378	-.369
.150			-.468	-.629	-.465	-.489	-.418	-.406
.250	-.426		-.560	-.645	-.518	-.555	-.515	-.491
.400	-.523		-.540	-.605	-.517	-.530	-.467	-.481
.550	-.469	-.422	-.463	-.490	-.443	-.484	-.378	-.403
.700	-.423	-.481	-.412	-.522	-.365	-.423	-.293	-.310
.850	-.411	-1.029	-.355	-.660	-.261	-.351	-.194	-.219
.950	-.446	-.710	-.335	-.480	-.215	-.247	-.125	-.119

TABLE 32 (A) – CONCLUDED

PRESSURE PROFILE ON WING

REAR NOZZLES DEFLECTED 45 DEGREES

ALPHA = -.01 DEG Q = 2041 N/SQ.M. (42.63 LBF/SQ.FT.) EFFECT. VELOC. RATIO = .292

	ETA = .250		ETA = .387		ETA = .524		ETA = .800	
X/C	CP		CP		CP		CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			.668		.884		.900	
.010			.394	-.057				
.015					-.395	-.398		
.025			-.416	-.588	-.683	-.452	-.396	-.271
.050			-.439	-.432	-.420	-.288	-.354	-.288
.100			-.403	-.475	-.395	-.342	-.395	-.281
.150			-.466	-.531	-.385	-.430	-.392	-.375
.250	-.393		-.472	-.543	-.481	-.459	-.459	-.424
.400	-.465		-.490	-.488	-.465	-.463	-.429	-.393
.550	-.410	-.351	-.416	-.405	-.399	-.406	-.347	-.330
.700	-.363	-.381	-.325	-.407	-.291	-.320	-.222	-.242
.850	-.342	-.682	-.280	-.524	-.206	-.262	-.136	-.149
.950	-.356	-.547	-.253	-.371	-.152	-.165	-.069	-.053

ALPHA = .01 DEG Q = 2850 N/SQ.M. (59.51 LBF/SQ.FT.) EFFECT. VELOC. RATIO = .391

	ETA = .250		ETA = .387		ETA = .524		ETA = .800	
X/C	CP		CP		CP		CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			.617		.916		.921	
.010			.549	.082				
.015					-.300	-.416		
.025			-.617	-.489	-.564	-.171	-.378	-.150
.050			-.400	-.333	-.418	-.320	-.419	-.220
.100			-.366	-.422	-.368	-.301	-.338	-.272
.150			-.421	-.565	-.421	-.385	-.366	-.373
.250	-.345		-.453	-.530	-.435	-.408	-.444	-.360
.400	-.428		-.465	-.464	-.465	-.421	-.392	-.373
.550	-.383	-.307	-.374	-.352	-.355	-.344	-.316	-.302
.700	-.319	-.315	-.294	-.335	-.253	-.269	-.182	-.197
.850	-.283	-.757	-.250	-.449	-.161	-.204	-.096	-.109
.950	-.284	-.438	-.206	-.303	-.103	-.111	-.033	-.016

ALPHA = .00 DEG Q = 2837 N/SQ.M. (59.26 LBF/SQ.FT.) EFFECT. VELOC. RATIO = .510

	ETA = .250		ETA = .387		ETA = .524		ETA = .800	
X/C	CP		CP		CP		CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			.648		.867		.907	
.010			.567	-.300				
.015					-.245	-.342		
.025			-.476	-.550	-.446	-.317	-.400	-.263
.050			-.427	-.448	-.291	-.273	-.350	-.176
.100			-.362	-.459	-.487	-.316	-.380	-.291
.150			-.452	-.518	-.385	-.353	-.365	-.302
.250	-.344		-.438	-.495	-.407	-.438	-.420	-.378
.400	-.422		-.442	-.404	-.425	-.355	-.395	-.348
.550	-.347	-.294	-.357	-.317	-.339	-.332	-.301	-.292
.700	-.295	-.280	-.260	-.311	-.229	-.245	-.175	-.186
.850	-.251	-.639	-.206	-.382	-.134	-.175	-.088	-.101
.950	-.241	-.358	-.164	-.252	-.075	-.083	-.024	-.004

TABLE 32 (B)

PRESSURE PROFILE ON WING

REAR NOZZLES DEFLECTED 45 DEGREES

ALPHA = 10.00 DEG Q = 228 N/SQ.M. (4.75 LBF/SQ.FT.) EFFECT. VELOC. RATIO = .098

	ETA = .250		ETA = .387		ETA = .524		ETA = .800	
X/C	CP		CP		CP		CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			-1.104		-1.976		-3.174	
.010			-1.845	.406				
.015					-3.922	.441		
.025			-3.562	.351	-3.466	.344	-3.438	.364
.050			-1.976	.136	-2.108	.171	-1.956	.136
.100			-1.595	-.155	-1.692	-.058	-1.623	-.093
.150			-1.388	-.252	-1.547	-.217	-1.492	-.252
.250	-1.000		-1.221	-.479	-1.318	-.432	-1.284	-.442
.400	-1.022		-1.123	-.697	-1.122	-.610	-1.090	-.566
.550	-.945	-.888	-.982	-.757	-.972	-.683	-.906	-.606
.700	-.881	-1.086	-.895	-.868	-.855	-.727	-.767	-.606
.850	-.881	-1.714	-.831	-1.066	-.747	-.727	-.653	-.583
.950	-.952	-1.405	-.834	-.935	-.663	-.660	-.583	-.549

ALPHA = 10.02 DEG Q = 513 N/SQ.M. (10.72 LBF/SQ.FT.) EFFECT. VELOC. RATIO = .146

	ETA = .250		ETA = .387		ETA = .524		ETA = .800	
X/C	CP		CP		CP		CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			-.822		-2.544		-3.597	
.010			-.321	.572				
.015					-4.469	.710		
.025			-3.588	.667	-3.984	.646	-2.983	.643
.050			-2.139	.495	-2.234	.477	-1.998	.440
.100			-1.476	.259	-1.691	.238	-1.605	.201
.150			-1.341	.041	-1.399	.026	-1.359	.017
.250	-.825		-1.159	-.210	-1.245	-.142	-1.166	-.160
.400	-.841		-.977	-.393	-1.001	-.316	-.920	-.292
.550	-.751	-.509	-.782	-.434	-.789	-.423	-.721	-.350
.700	-.664	-.670	-.672	-.569	-.636	-.442	-.548	-.349
.850	-.626	-1.341	-.596	-.813	-.509	-.465	-.428	-.343
.950	-.652	-1.036	-.570	-.703	-.442	-.420	-.344	-.310

ALPHA = 10.01 DEG Q = 906 N/SQ.M. (18.93 LBF/SQ.FT.) EFFECT. VELOC. RATIO = .195

	ETA = .250		ETA = .387		ETA = .524		ETA = .800	
X/C	CP		CP		CP		CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			-1.041		-2.973		-3.988	
.010			.127	.695				
.015					-5.179	.821		
.025			-2.962	.796	-3.228	.789	-2.773	.775
.050			-2.008	.588	-2.290	.610	-1.982	.586
.100			-1.474	.349	-1.533	.376	-1.497	.348
.150			-1.292	.184	-1.448	.203	-1.372	.169
.250	-.742		-1.036	-.061	-1.140	-.005	-1.128	-.054
.400	-.765		-.878	-.269	-.907	-.232	-.837	-.192
.550	-.676	-.381	-.705	-.328	-.710	-.276	-.633	-.217
.700	-.585	-.535	-.575	-.457	-.550	-.326	-.461	-.237
.850	-.542	-1.217	-.504	-.713	-.420	-.361	-.327	-.238
.950	-.564	-.966	-.458	-.607	-.339	-.325	-.239	-.205

TABLE 32 (B) – CONCLUDED

PRESSURE PROFILE ON WING

REAR NOZZLES DEFLECTED 45 DEGREES

ALPHA = 10.00 DEG Q = 2040 N/SQ.M. (42.60 LBF/SQ.FT.) EFFECT. VELOC. RATIO = .292

	ETA = .250		ETA = .387		ETA = .524		ETA = .800	
X/C	CP		CP		CP		CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			-.890		-2.499		-4.021	
.010			.231	.796				
.015					-4.964	.914		
.025			-2.911	.690	-3.302	.885	-2.878	.855
.050			-1.941	.689	-2.235	.732	-2.052	.678
.100			-1.378	.475	-1.591	.453	-1.484	.418
.150			-1.213	.292	-1.354	.312	-1.216	.228
.250	-.682		-.981	.042	-1.048	.102	-1.002	.072
.400	-.695		-.805	-.143	-.843	-.071	-.758	-.073
.550	-.573	-.247	-.612	-.187	-.611	-.152	-.528	-.138
.700	-.481	-.412	-.470	-.303	-.439	-.197	-.366	-.143
.850	-.423	-1.032	-.383	-.539	-.300	-.228	-.226	-.134
.950	-.420	-.816	-.330	-.435	-.212	-.205	-.135	-.098

ALPHA = 10.05 DEG Q = 2846 N/SQ.M. (59.43 LBF/SQ.FT.) EFFECT. VELOC. RATIO = .390

	ETA = .250		ETA = .387		ETA = .524		ETA = .800	
X/C	CP		CP		CP		CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			-.956		-3.152		-4.196	
.010			.321	.857				
.015					-4.608	.940		
.025			-3.248	.901	-3.220	.914	-2.976	.870
.050			-1.945	.727	-2.123	.718	-1.990	.708
.100			-1.343	.507	-1.528	.542	-1.420	.449
.150			-1.204	.321	-1.273	.341	-1.181	.281
.250	-.632		-.958	.075	-1.032	.125	-.984	.095
.400	-.649		-.767	-.084	-.794	-.028	-.709	-.043
.550	-.529	-.180	-.568	-.117	-.565	-.103	-.511	-.097
.700	-.429	-.353	-.425	-.219	-.395	-.127	-.329	-.092
.850	-.361	-.891	-.316	-.442	-.240	-.158	-.184	-.090
.950	-.344	-.705	-.255	-.354	-.155	-.140	-.094	-.060

ALPHA = 10.01 DEG Q = 2844 N/SQ.M. (59.39 LBF/SQ.FT.) EFFECT. VELOC. RATIO = .515

	ETA = .250		ETA = .387		ETA = .524		ETA = .800	
X/C	CP		CP		CP		CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			-.942		-2.997		-4.219	
.010			.189	.853				
.015					-4.609	.928		
.025			-2.730	.917	-3.048	.908	-2.935	.885
.050			-1.909	.751	-2.089	.737	-1.868	.698
.100			-1.333	.500	-1.464	.519	-1.358	.496
.150			-1.145	.305	-1.229	.332	-1.165	.316
.250	-.612		-.903	.119	-.958	.143	-.933	.085
.400	-.616		-.720	-.076	-.756	-.018	-.697	-.044
.550	-.488	-.163	-.529	-.099	-.531	-.082	-.480	-.076
.700	-.386	-.296	-.384	-.169	-.356	-.092	-.302	-.087
.850	-.318	-.785	-.271	-.368	-.205	-.119	-.167	-.070
.950	-.290	-.600	-.206	-.284	-.117	-.097	-.076	-.043

TABLE 33 (A)
PRESSURE PROFILE ON WING

FORWARD NOZZLES DEFLECTED 90 DEGREES

ALPHA = -.02 DEG Q = 230 N/SQ.M. (4.79 LBF/SQ.FT.) EFFECT. VELOC. RATIO = .101

X/C	ETA = .250 CP		ETA = .387 CP		ETA = .524 CP		ETA = .800 CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			-3.155		-2.056		.320	
.010			.182	-3.423				
.015					1.027	-2.894		
.025			.649	-2.818	.842	-2.818	.629	-1.637
.050			.430	-2.207	.560	-1.939	.430	-1.205
.100			.224	-1.878	.368	-1.651	.320	-.868
.150			.141	-1.671	.203	-1.610	.176	-.896
.250	.038		.031	-1.653	.045	-1.534	.038	-.798
.400	-.089		-.082	-1.653	-.066	-1.424	-.046	-.731
.550	-.072	-3.128	-.066	-1.886	-.045	-1.161	.001	-.558
.700	-.069	-3.744	-.076	-1.836	-.036	-.944	.051	-.359
.850	-.176	-2.931	-.142	-1.214	-.022	-.468	.091	-.142
.950	-.468	-2.406	-.269	-.748	-.079	-.275	.117	.011

ALPHA = -.04 DEG Q = 512 N/SQ.M. (10.69 LBF/SQ.FT.) EFFECT. VELOC. RATIO = .151

X/C	ETA = .250 CP		ETA = .337 CP		ETA = .524 CP		ETA = .800 CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			-1.232		-.826		.748	
.010			.360	-1.953				
.015					.622	-2.397		
.025			.148	-2.560	.376	-1.931	.197	-1.124
.050			.052	-1.787	.123	-1.667	.129	-.869
.100			-.037	-1.630	.026	-1.389	-.049	-.749
.150			-.130	-1.737	-.096	-1.442	-.117	-.801
.250	-.176		-.222	-1.831	-.231	-1.515	-.179	-.825
.400	-.282		-.298	-2.205	-.271	-1.539	-.235	-.713
.550	-.250	-4.196	-.252	-2.466	-.235	-1.388	-.177	-.592
.700	-.225	-4.656	-.237	-2.351	-.198	-1.085	-.116	-.437
.850	-.305	-3.606	-.267	-1.525	-.159	-.737	-.050	-.234
.950	-.582	-2.742	-.353	-.879	-.182	-.391	-.010	-.073

ALPHA = -.04 DEG Q = 912 N/SQ.M. (19.05 LBF/SQ.FT.) EFFECT. VELOC. RATIO = .202

X/C	ETA = .250 CP		ETA = .387 CP		ETA = .524 CP		ETA = .800 CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			-.547		.387		.978	
.010			.461	-1.243				
.015					.183	-1.580		
.025			-.051	-1.445	-.033	-1.544	-.103	-.609
.050			-.170	-1.145	-.054	-1.271	-.096	-.574
.100			-.218	-1.363	-.194	-1.088	-.211	-.571
.150			-.303	-1.478	-.244	-1.200	-.248	-.645
.250	-.260		-.351	-1.524	-.370	-1.281	-.336	-.675
.400	-.374		-.413	-1.759	-.355	-1.297	-.298	-.614
.550	-.324	-3.295	-.317	-2.027	-.298	-1.191	-.245	-.510
.700	-.285	-3.784	-.289	-1.920	-.252	-.938	-.176	-.383
.850	-.357	-3.024	-.305	-1.345	-.188	-.649	-.076	-.207
.950	-.613	-2.321	-.360	-.747	-.178	-.360	-.030	-.066

TABLE 33 (A) – CONCLUDED

PRESSURE PROFILE ON WING

FORWARD NOZZLES DEFLECTED 90 DEGREES

ALPHA = -.01 DEG Q = 2040 N/SQ.M. (42.61 LBF/SQ.FT.) EFFECT. VELOC. RATIO = .303

	ETA = .250		ETA = .387		ETA = .524		ETA = .800	
X/C	CP		CP		CP		CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			.659		.955		.966	
.010			.750	-.099				
.015					-.284	-.486		
.025			-.574	-.764	-.535	-.585	-.512	-.172
.050			-.479	-.753	-.511	-.557	-.498	-.336
.100			-.423	-.659	-.410	-.629	-.355	-.271
.150			-.510	-1.083	-.480	-.744	-.419	-.403
.250	-.362		-.493	-1.225	-.454	-.884	-.477	-.457
.400	-.466		-.521	-1.305	-.487	-.976	-.420	-.505
.550	-.413	-1.602	-.411	-1.253	-.392	-.877	-.321	-.422
.700	-.385	-1.556	-.327	-1.130	-.281	-.714	-.205	-.306
.850	-.403	-1.592	-.313	-.877	-.203	-.500	-.113	-.193
.950	-.610	-1.447	-.304	-.517	-.175	-.291	-.057	-.070

ALPHA = -.02 DEG Q = 2847 N/SQ.M. (59.46 LBF/SQ.FT.) EFFECT. VELOC. RATIO = .405

	ETA = .250		ETA = .387		ETA = .524		ETA = .800	
X/C	CP		CP		CP		CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			.933		.956		.937	
.010			.855	.312				
.015					-.704	-.010		
.025			-.755	-.247	-.780	-.218	-.566	.038
.050			-.645	-.362	-.555	-.220	-.506	-.167
.100			-.477	-.495	-.458	-.364	-.417	-.173
.150			-.549	-.711	-.516	-.507	-.451	-.319
.250	-.380		-.513	-.913	-.519	-.658	-.469	-.379
.400	-.469		-.503	-.965	-.510	-.768	-.421	-.414
.550	-.422	-.992	-.399	-.905	-.385	-.688	-.312	-.353
.700	-.383	-1.074	-.333	-.806	-.287	-.567	-.193	-.262
.850	-.424	-1.158	-.300	-.686	-.193	-.406	-.097	-.161
.950	-.573	-1.093	-.278	-.426	-.140	-.239	-.035	-.045

ALPHA = -.09 DEG Q = 2849 N/SQ.M. (59.51 LBF/SQ.FT.) EFFECT. VELOC. RATIO = .529

	ETA = .250		ETA = .387		ETA = .524		ETA = .800	
X/C	CP		CP		CP		CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			.981		.955		.913	
.010			.874	.539				
.015					-.693	.036		
.025			-.726	-.161	-.684	-.079	-.576	-.023
.050			-.656	-.178	-.569	-.163	-.534	-.141
.100			-.501	-.417	-.517	-.182	-.434	-.195
.150			-.516	-.557	-.526	-.456	-.342	-.266
.250	-.360		-.456	-.755	-.470	-.554	-.488	-.337
.400	-.445		-.481	-.826	-.471	-.643	-.394	-.369
.550	-.385	-.787	-.388	-.750	-.366	-.594	-.301	-.337
.700	-.343	-.826	-.303	-.685	-.261	-.482	-.165	-.229
.850	-.349	-.807	-.256	-.573	-.163	-.342	-.083	-.133
.950	-.452	-.812	-.224	-.331	-.115	-.188	-.015	-.023

TABLE 33 (B)
PRESSURE PROFILE ON WING

FORWARD NOZZLES DEFLECTED 90 DEGREES

ALPHA = 10.00 DEG Q = 228 N/SQ.M. (4.75 LBF/SQ.FT.) EFFECT. VELOC. RATIO = .101

	ETA = .250		ETA = .387		ETA = .524		ETA = .800	
X/C	CP		CP		CP		CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			1.330		1.433		1.371	
.010			1.717	.138				
.015					-.070	-.457		
.025			-.180	-.270	-.139	-.409	-.492	.332
.050			-.125	-.340	-.187	-.312	-.326	.166
.100			-.076	-.811	-.104	-.527	-.160	.021
.150			-.104	-.859	-.146	-.707	-.111	-.083
.250	.076		-.083	-1.337	-.111	-.917	-.139	-.196
.400	.026		-.078	-2.018	-.078	-1.136	-.078	-.259
.550	.097	-5.550	.039	-2.488	.019	-.971	.046	-.199
.700	.113	-5.412	.113	-2.220	.113	-.837	.211	-.071
.850	.100	-3.664	.097	-1.240	.184	-.357	.308	.060
.950	-.108	-1.959	.006	-.454	.174	-.038	.338	.285

ALPHA = 10.01 DEG Q = 510 N/SQ.M. (10.65 LBF/SQ.FT.) EFFECT. VELOC. RATIO = .151

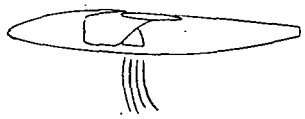
	ETA = .250		ETA = .337		ETA = .524		ETA = .800	
X/C	CP		CP		CP		CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			1.237		1.011		-.036	
.010			1.394	.641				
.015					-1.677	.495		
.025			-1.498	.180	-1.467	.103	-1.523	.683
.050			-1.025	-.163	-1.062	-.064	-1.359	.486
.100			-.688	-.589	-.747	-.416	-.800	.183
.150			-.642	-.793	-.738	-.661	-.707	-.030
.250	-.293		-.552	-1.339	-.589	-.966	-.593	-.225
.400	-.346		-.448	-2.017	-.472	-1.188	-.434	-.331
.550	-.238	-5.465	-.301	-2.431	-.334	-1.156	-.259	-.316
.700	-.168	-5.372	-.220	-2.152	-.210	-.963	-.121	-.244
.850	-.190	-3.581	-.219	-1.349	-.120	-.608	-.006	-.121
.950	-.340	-1.453	-.273	-.695	-.103	-.307	.054	-.014

ALPHA = 10.04 DEG Q = 913 N/SQ.M. (19.06 LBF/SQ.FT.) EFFECT. VELOC. RATIO = .203

	ETA = .250		ETA = .387		ETA = .524		ETA = .800	
X/C	CP		CP		CP		CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			.869		.160		-2.166	
.010								
.015					-2.819	.869		
.025			-2.065	.550	-2.372	.543	-2.529	.861
.050			-1.357	.189	-1.507	.295	-1.564	.628
.100			-.984	-.210	-1.130	-.037	-1.108	.332
.150			-.895	-.547	-.987	-.379	-.961	.135
.250	-.423		-.733	-.957	-.802	-.682	-.773	-.087
.400	-.467		-.576	-1.664	-.623	-.915	-.575	-.241
.550	-.358	-3.455	-.422	-1.828	-.437	-.930	-.378	-.255
.700	-.270	-2.953	-.312	-1.653	-.302	-.798	-.218	-.216
.850	-.249	-2.280	-.254	-1.074	-.180	-.539	-.096	-.133
.950	-.403	-1.284	-.277	-.525	-.143	-.290	-.017	-.042

TABLE 33 (B) – CONCLUDED
PRESSURE PROFILE ON WING

FORWARD NOZZLES DEFLECTED 90 DEGREES



ALPHA = 10.18 DEG Q = 2029 N/SQ.M. (42.39 LBF/SQ.FT.) EFFECT. VELOC. RATIO = .302

	ETA = .250		ETA = .387		ETA = .524		ETA = .800	
X/C	CP		CP		CP		CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			-.757		-3.238		-4.354	
.010			.829	.990				
.015					-5.117			
.025			-3.351	.953	-3.559	.943	-3.072	.985
.050			-2.201	.690	-2.407	.717	-2.073	.761
.100			-1.448	.276	-1.655	.334	-1.533	.467
.150			-1.257	-.098	-1.371	.104	-1.233	.268
.250	-.635		-.978	-.591	-1.063	-.253	-1.005	.021
.400	-.649		-.777	-1.044	-.835	-.585	-.745	-.138
.550	-.535	-1.437	-.585	-1.103	-.591	-.677	-.505	-.211
.700	-.428	-1.447	-.446	-1.038	-.429	-.620	-.337	-.193
.850	-.414	-1.383	-.357	-.810	-.284	-.510	-.194	-.162
.950	-.452	-1.045	-.309	-.497	-.220	-.324	-.103	-.098

ALPHA = 10.23 DEG Q = 2847 N/SQ.M. (59.45 LBF/SQ.FT.) EFFECT. VELOC. RATIO = .409

	ETA = .250		ETA = .387		ETA = .524		ETA = .800	
X/C	CP		CP		CP		CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			-1.631		-3.440		-4.510	
.010			.696	.862				
.015					-4.510			
.025			-3.467		-4.091		-3.121	.975
.050			-2.283	.817	-2.539	.839	-2.354	.799
.100			-1.622	.508	-1.706	.508	-1.610	.549
.150			-1.306	.212	-1.487	.258	-1.306	.327
.250	-.652		-1.040	-.288	-1.105	-.095	-1.014	.123
.400	-.672		-.803	-.785	-.863	-.418	-.771	-.087
.550	-.561	-1.041	-.597	-.883	-.612	-.521	-.519	-.168
.700	-.459	-1.004	-.448	-.802	-.416	-.528	-.341	-.161
.850	-.418	-.935	-.343	-.677	-.278	-.448	-.196	-.139
.950	-.437	-.731	-.293	-.449	-.206	-.317	-.093	-.088

ALPHA = 10.15 DEG Q = 2839 N/SQ.M. (59.29 LBF/SQ.FT.) EFFECT. VELOC. RATIO = .536

	ETA = .250		ETA = .387		ETA = .524		ETA = .800	
X/C	CP		CP		CP		CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			.006		-.480		-1.596	
.010			.654					
.015					-1.480			
.025			-1.480	.572	-1.236	.903	-1.614	.889
.050			-1.077	.770	-1.148	.703	-1.938	.696
.100			-1.427	.464	-1.380	.442	-1.595	.450
.150			-1.462	.190	-1.199	.221	-1.600	.273
.250	-.714		-1.289	-.237	-1.302	-.060	-1.230	.062
.400	-.759		-1.078	-.654	-.972	-.344	-.761	-.103
.550	-.537	-.695	-.666	-.741	-.603	-.448	-.579	-.167
.700	-.444	-.741	-.490	-.658	-.605	-.462	-.463	-.179
.850	-.398	-.792	-.374	-.595	-.354	-.418	-.273	-.177
.950	-.405	-.646	-.421	-.446	-.357	-.364	-.225	-.154

TABLE 34 (A)

PRESSURE PROFILE ON WING

FORWARD NOZZLES DEFLECTED 90 DEGREES WITH FLAPS

ALPHA = -.01 DEG Q = 228 N/SQ.M. (4.76 LBF/SQ.FT.) EFFECT. VELOC. RATIO = .101

X/C	ETA = .250 CP		ETA = .387 CP		ETA = .524 CP		ETA = .800 CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			-1.538		-1.434		.239	
.010			-2.602	-3.203				
.015					.550	-3.273		
.025			.066	-2.671	.391	-2.664	.211	-1.538
.050			-.003	-1.994	-.024	-1.938	.004	-1.330
.100			-.183	-1.552	-.127	-1.759	-.155	-1.150
.150			-.314	-2.049	-.217	-1.800	-.238	-1.130
.250	-.411		-.452	-1.589	-.432	-1.648	-.390	-1.115
.400	-.508		-.512	-2.271	-.495	-1.537	-.448	-.971
.550	-.495	-5.555	-.525	-2.643	-.498	-1.266	-.388	-.847
.700	-.495	-4.362	-.522	-1.909	-.495	-.843	-.351	-.636
.850	-.569	-3.209	-.652	-1.279	-.528	-.388	-.284	-.435
.950	-.676	-1.527	-.589	-.971	-.639	-.066	-.244	-.317

ALPHA = .00 DEG Q = 514 N/SQ.M. (10.74 LBF/SQ.FT.) EFFECT. VELOC. RATIO = .152

X/C	ETA = .250 CP		ETA = .387 CP		ETA = .524 CP		ETA = .800 CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			-.771		-.455		.819	
.010			-.955	-2.122				
.015					.182	-2.073		
.025			-.351	-2.119	.087	-2.021	-.262	-1.047
.050			-.244	-1.625	-.137	-1.647	-.391	-.762
.100			-.385	-1.592	-.422	-1.457	-.363	-.857
.150			-.474	-1.904	-.426	-1.622	-.391	-.860
.250	-.474		-.532	-2.009	-.511	-1.579	-.443	-.830
.400	-.559		-.616	-2.408	-.605	-1.505	-.497	-.813
.550	-.531	-5.632	-.562	-2.457	-.552	-1.250	-.427	-.714
.700	-.503	-5.047	-.548	-1.971	-.537	-.795	-.354	-.568
.850	-.542	-2.629	-.586	-1.124	-.562	-.228	-.297	-.378
.950	-.660	-1.165	-.622	-.637	-.657	.009	-.234	-.242

ALPHA = .01 DEG Q = 910 N/SQ.M. (19.00 LBF/SQ.FT.) EFFECT. VELOC. RATIO = .202

X/C	ETA = .250 CP		ETA = .387 CP		ETA = .524 CP		ETA = .800 CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			.187		.608		.911	
.010			-.206	-.707				
.015					-.094	-1.205		
.025			-.350	-1.335	-.360	-.904	-.599	-.591
.050			-.495	-1.088	-.409	-1.093	-.397	-.485
.100			-.473	-1.280	-.430	-1.088	-.494	-.533
.150			-.573	-1.434	-.499	-1.148	-.485	-.682
.250	-.471		-.587	-1.575	-.591	-1.214	-.532	-.718
.400	-.580		-.616	-1.960	-.585	-1.196	-.516	-.656
.550	-.532	-4.139	-.567	-2.007	-.520	-.988	-.451	-.579
.700	-.506	-3.782	-.542	-1.461	-.516	-.573	-.373	-.450
.850	-.522	-2.122	-.548	-.725	-.517	-.096	-.255	-.313
.950	-.618	-1.092	-.565	-.381	-.601	.167	-.201	-.187

TABLE 34 (a) - CONCLUDED
PRESSURE PROFILE ON WING

FORWARD NOZZLES DEFLECTED 90 DEGREES WITH FLAPS

ALPHA = -.01 DEG Q = 2039 N/SQ.M. (42.58 LBF/SQ.FT.) EFFECT. VELOC. RATIO = .303

	ETA = .250		ETA = .387		ETA = .524		ETA = .800	
X/C	CP		CP		CP		CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			.737		.917		.832	
.010			.458	-.160				
.015					-.687	-.421		
.025			-.798	-.573	-.807	-.185	-.658	-.068
.050			-.692	-.608	-.676	-.492	-.538	-.151
.100			-.694	-.743	-.697	-.568	-.553	-.273
.150			-.620	-.957	-.697	-.665	-.594	-.381
.250	-.491		-.635	-1.082	-.654	-.819	-.591	-.501
.400	-.600		-.749	-1.354	-.652	-.800	-.526	-.489
.550	-.536	-2.082	-.559	-1.266	-.560	-.633	-.429	-.410
.700	-.458	-1.817	-.515	-.889	-.506	-.313	-.319	-.317
.850	-.506	-1.561	-.498	-.369	-.483	.101	-.231	-.234
.950	-.514	-1.079	-.494	-.079	-.528	.339	-.154	-.131

ALPHA = .00 DEG Q = 2842 N/SQ.M. (59.37 LBF/SQ.FT.) EFFECT. VELOC. RATIO = .404

	ETA = .250		ETA = .387		ETA = .524		ETA = .800	
X/C	CP		CP		CP		CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			.927		.850		.644	
.010			.620	.299				
.015					-1.075	-.035		
.025			-.933	-.321	-.978	-.143	-.886	.001
.050			-.870	-.238	-.775	-.163	-.650	-.122
.100			-.651	-.498	-.700	-.246	-.510	-.149
.150			-.729	-.755	-.648	-.501	-.609	-.290
.250	-.487		-.679	-.854	-.676	-.595	-.603	-.388
.400	-.623		-.635	-.983	-.649	-.620	-.533	-.422
.550	-.536	-1.337	-.553	-.916	-.539	-.468	-.427	-.339
.700	-.488	-1.338	-.496	-.608	-.492	-.177	-.294	-.262
.850	-.483	-1.142	-.466	-.189	-.458	.201	-.206	-.185
.950	-.514	-1.004	-.451	.080	-.469	.411	-.131	-.098

ALPHA = .00 DEG Q = 2845 N/SQ.M. (59.42 LBF/SQ.FT.) EFFECT. VELOC. RATIO = .530

	ETA = .250		ETA = .387		ETA = .524		ETA = .800	
X/C	CP		CP		CP		CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			.941		.867		.723	
.010			.637	.484				
.015					-1.054	.310		
.025			-.960	.076	-1.209	.037	-.971	.060
.050			-.867	-.083	-.828	-.058	-.554	-.022
.100			-.656	-.235	-.709	-.182	-.638	-.122
.150			-.755	-.507	-.681	-.328	-.571	-.252
.250	-.485		-.681	-.675	-.694	-.477	-.607	-.391
.400	-.580		-.642	-.770	-.676	-.505	-.524	-.358
.550	-.528	-.978	-.542	-.705	-.546	-.369	-.421	-.330
.700	-.478	-1.024	-.490	-.450	-.485	-.094	-.302	-.249
.850	-.470	-.948	-.451	-.044	-.445	.258	-.202	-.180
.950	-.486	-.588	-.432	.214	-.461	.447	-.126	-.096

TABLE 34 (B)

PRESSURE PROFILE ON WING

FORWARD NOZZLES DEFLECTED 90 DEGREES WITH FLAPS

ALPHA = 10.00 DEG Q = 227 N/SQ.M. (4.74 LBF/SQ.FT.) EFFECT. VELOC. RATIO = .101

	ETA = .250		ETA = .337		ETA = .524		ETA = .800	
X/C	CP		CP		CP		CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			.293		.349		-.311	
.010			-1.395	-.513				
.015					-1.492	-.846		
.025			-1.694	-1.228	-1.750	-1.319	-2.007	-.513
.050			-1.485	1.131	-1.652	-1.194	-1.569	-.735
.100			-1.319	-1.624	-1.458	-1.472	-1.423	-.902
.150			-1.347	-2.021	-1.355	-1.673	-1.388	-1.013
.250	-1.117		-1.284	-2.402	-1.374	-1.792	-1.360	-1.236
.400	-1.162		-1.247	-3.072	-1.307	-1.964	-1.253	-1.253
.550	-1.088	-7.542	-1.149	-3.359	-1.186	-1.755	-1.078	-1.179
.700	-1.041	-8.088	-1.088	-2.520	-1.122	-1.344	-.954	-1.071
.850	-1.031	-2.584	-1.068	-1.587	-1.061	-.782	-.849	-.984
.950	-1.092	-1.381	-1.115	-1.152	-1.102	-.522	-.795	-.839

ALPHA = 10.01 DEG Q = 512 N/SQ.M. (10.70 LBF/SQ.FT.) EFFECT. VELOC. RATIO = .152

	ETA = .250		ETA = .387		ETA = .524		ETA = .800	
X/C	CP		CP		CP		CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			.639		.218		-1.514	
.010			-.099	.448				
.015					-2.584	.138		
.025			-2.251	-.124	-2.387	-.114	-2.636	.331
.050			-1.803	-.468	-1.913	-.391	-1.882	.085
.100			-1.320	-.889	-1.477	-.674	-1.510	-.210
.150			-1.280	-1.206	-1.320	-.997	-1.304	-.450
.250	-.905		-1.188	-1.719	-1.181	-1.235	-1.181	-.628
.400	-.984		-1.075	-2.473	-1.103	-1.401	-1.008	-.714
.550	-.886	-6.398	-.926	-2.551	-.954	-1.285	-.835	-.731
.700	-.811	-5.237	-.851	-1.987	-.855	-.911	-.689	-.661
.850	-.793	-1.758	-.822	-1.048	-.808	-.404	-.574	-.571
.950	-.817	-.965	-.632	-.696	-.848	-.130	-.501	-.509

ALPHA = 10.00 DEG Q = 909 N/SQ.M. (18.98 LBF/SQ.FT.) EFFECT. VELOC. RATIO = .202

	ETA = .250		ETA = .337		ETA = .524		ETA = .800	
X/C	CP		CP		CP		CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			.086		-.888		-3.133	
.010			.177					
.015					-3.269	.622		
.025			-2.904	.421	-3.118	.418	-3.277	.640
.050			-1.819	.036	-2.140	.159	-2.088	.392
.100			-1.514	-.368	-1.689	-.227	-1.613	.099
.150			-1.372	-.713	-1.465	-.500	-1.436	-.085
.250	-.841		-1.174	-1.213	-1.256	-.788	-1.186	-.317
.400	-.905		-1.016	-1.829	-1.094	-.958	-.983	-.456
.550	-.804	-4.020	-.857	-1.888	-.910	-.877	-.760	-.480
.700	-.726	-3.193	-.751	-1.392	-.780	-.602	-.601	-.465
.850	-.672	-1.408	-.702	-.623	-.707	-.133	-.469	-.407
.950	-.675	-.617	-.696	-.249	-.733	.106	-.373	-.353

TABLE 34 (B) – CONCLUDED

PRESSURE PROFILE ON WING

FORWARD NOZZLES DEFLECTED 90 DEGREES WITH FLAPS

ALPHA = 10.00 DEG Q = 2037 N/SQ.M. (42.55 LBF/SQ.FT.) EFFECT. VELOC. RATIO = .303

	ETA = .250		ETA = .387		ETA = .524		ETA = .800	
X/C	CP		CP		CP		CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			-1.128		-3.664		-5.406	
.010			.346	.776				
.015					-5.467	.884		
.025			-3.894	.845	-4.103	.863	-3.467	.855
.050			-2.541	.580	-2.692	.575	-2.507	.640
.100			-1.702	.193	-1.913	.265	-1.762	.404
.150			-1.516	-.162	-1.640	.013	-1.477	.162
.250	-.872		-1.240	-.646	-1.337	-.317	-1.227	-.041
.400	-.902		-1.065	-1.113	-1.116	-.541	-.953	-.229
.550	-.775	-1.740	-.844	-1.121	-.878	-.497	-.724	-.292
.700	-.678	-1.765	-.721	-.791	-.743	-.291	-.545	-.284
.850	-.634	-1.265	-.657	-.287	-.659	.108	-.393	-.280
.950	-.636	-.645	-.628	-.059	-.637	.333	-.292	-.244

ALPHA = 10.03 DEG Q = 2843 N/SQ.M. (59.38 LBF/SQ.FT.) EFFECT. VELOC. RATIO = .406

	ETA = .250		ETA = .387		ETA = .524		ETA = .800	
X/C	CP		CP		CP		CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			-.274		-.940		-2.018	
.010			.443	.868				
.015					-1.630	.907		
.025			-1.977	.859	-1.791	.810	-2.090	.802
.050			-1.999	.668	-1.488	.605	-1.984	.609
.100			-2.042	.325	-1.830	.335	-2.043	.366
.150			-1.729	.054	-1.745	.116	-1.804	.182
.250	-1.064		-1.726	-.404	-1.655	-.137	-1.544	-.024
.400	-.958		-1.146	-.797	-1.423	-.354	-1.034	-.183
.550	-.787	-1.192	-1.017	-.788	-1.069	-.344	-.808	-.235
.700	-.671	-1.234	-.711	-.522	-.848	-.121	-.619	-.245
.850	-.576	-.920	-.638	-.113	-.736	.235	-.472	-.257
.950	-.523	-.462	-.572	.155	-.649	.436	-.351	-.252

ALPHA = 10.00 DEG Q = 2846 N/SQ.M. (59.44 LBF/SQ.FT.) EFFECT. VELOC. RATIO = .532

	ETA = .250		ETA = .387		ETA = .524		ETA = .800	
X/C	CP		CP		CP		CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			-.226		-.740		-1.739	
.010			.393	.860				
.015					-1.486	.902		
.025			-1.713	.870	-1.495	.823	-1.907	.794
.050			-1.605	.687	-1.626	.621	-1.807	.612
.100			-1.868	.392	-1.716	.371	-1.774	.380
.150			-1.539	.101	-1.576	.171	-1.860	.216
.250	-1.026		-1.558	-.307	-1.510	-.093	-1.626	.008
.400	-1.009		-1.427	-.643	-1.418	-.280	-1.133	-.147
.550	-.817	-.927	-1.062	-.654	-1.065	-.252	-.845	-.207
.700	-.727	-.930	-.783	-.417	-.912	-.059	-.690	-.215
.850	-.593	-.652	-.626	-.036	-.772	.280	-.486	-.228
.950	-.500	-.397	-.558	.217	-.647	.470	-.340	-.238

TABLE 35 (A)

PRESSURE PROFILE ON WING

REAR NOZZLES DEFLECTED 90 DEGREES

ALPHA = .00 DEG C = 228 N/SQ.M. (4.75 LBF/SQ.FT.) EFFECT. VELOC. RATIO = .098

	ETA = .250		ETA = .387		ETA = .524		ETA = .800	
X/C	CP		CP		CP		CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			.297		.595		.650	
.010			-2.881	-.728				
.015					-.388	-.755		
.025			-.672	-1.011	-.495	-.908	-.561	-.534
.050			-.741	-.852	-.658	-.790	-.631	-.693
.100			-.610	-.915	-.693	-.845	-.638	-.721
.150			-.707	-.998	-.693	-.866	-.679	-.825
.250	-.658		-.728	-1.101	-.776	-.953	-.693	-.876
.400	-.752		-.765	-1.071	-.772	-1.017	-.682	-.883
.550	-.712	-.886	-.725	-1.031	-.705	-1.004	-.638	-.823
.700	-.695	-.960	-.688	-1.074	-.661	-.937	-.574	-.755
.850	-.732	-1.628	-.692	-1.178	-.601	-.853	-.507	-.651
.950	-.809	-1.272	-.759	-1.068	-.551	-.722	-.440	-.500

ALPHA = .02 DEG C = 510 N/SQ.M. (10.65 LBF/SQ.FT.) EFFECT. VELOC. RATIO = .147

	ETA = .250		ETA = .387		ETA = .524		ETA = .800	
X/C	CP		CP		CP		CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			.651		.824		.796	
.010			-.776	-.084				
.015					-.504	-.161		
.025			-.754	-.257	-.640	-.482	-.692	-.217
.050			-.671	-.498	-.739	-.439	-.665	-.368
.100			-.634	-.603	-.628	-.507	-.640	-.343
.150			-.637	-.649	-.637	-.522	-.563	-.516
.250	-.560		-.668	-.762	-.665	-.632	-.600	-.572
.400	-.639		-.681	-.812	-.657	-.747	-.572	-.590
.550	-.604	-.601	-.601	-.775	-.595	-.744	-.487	-.571
.700	-.583	-.770	-.571	-.881	-.526	-.711	-.421	-.500
.850	-.604	-1.538	-.551	-1.069	-.431	-.654	-.307	-.397
.950	-.680	-1.158	-.556	-.905	-.397	-.530	-.261	-.276

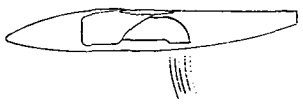
ALPHA = 0.00 DEG C = 908 N/SQ.M. (18.96 LBF/SQ.FT.) EFFECT. VELOC. RATIO = .196

	ETA = .250		ETA = .387		ETA = .524		ETA = .800	
X/C	CP		CP		CP		CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			.841		.868		.751	
.010			.006	.258				
.015					-.634	-.028		
.025			-.642	-.225	-.757	-.127	-.651	-.103
.050			-.610	-.303	-.735	-.221	-.599	-.266
.100			-.617	-.461	-.580	-.358	-.539	-.280
.150			-.620	-.565	-.615	-.374	-.533	-.389
.250	-.466		-.608	-.610	-.596	-.495	-.575	-.463
.400	-.593		-.646	-.644	-.600	-.595	-.542	-.489
.550	-.551	-.448	-.551	-.640	-.521	-.589	-.435	-.447
.700	-.512	-.633	-.490	-.739	-.445	-.574	-.321	-.375
.850	-.523	-1.425	-.464	-.948	-.352	-.530	-.232	-.286
.950	-.585	-1.000	-.462	-.729	-.327	-.415	-.167	-.177

TABLE 35 (A) – CONCLUDED

PRESSURE PROFILE ON WING

REAR NOZZLES DEFLECTED 90 DEGREES



ALPHA = .00 DEG Q = 2033 N/SQ.M. (42.45 LBF/SQ.FT.) EFFECT. VELOC. RATIO = .293

	ETA = .250		ETA = .387		ETA = .524		ETA = .800	
X/C	CP		CP		CP		CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			.905		.919		.853	
.010			.614	.128				
.015					-.666	-.062		
.025			-.781	.094	-.912	-.064	-.780	-.045
.050			-.587	-.202	-.595	-.137	-.554	-.080
.100			-.528	-.270	-.617	-.234	-.563	-.184
.150			-.577	-.350	-.570	-.241	-.491	-.244
.250	-.430		-.567	-.463	-.551	-.390	-.517	-.359
.400	-.518		-.565	-.451	-.548	-.421	-.468	-.356
.550	-.462	-.262	-.468	-.434	-.451	-.424	-.379	-.336
.700	-.414	-.382	-.397	-.521	-.348	-.431	-.257	-.259
.850	-.406	-1.070	-.360	-.702	-.257	-.371	-.156	-.179
.950	-.430	-.704	-.337	-.514	-.215	-.271	-.090	-.084

ALPHA = 0.00 DEG Q = 2840 N/SQ.M. (59.31 LBF/SQ.FT.) EFFECT. VELOC. RATIO = .395

	ETA = .250		ETA = .387		ETA = .524		ETA = .800	
X/C	CP		CP		CP		CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			.864		.943		.827	
.010			.732	.256				
.015					-.766	.207		
.025			-.715	-.166	-.833	-.123	-.637	.013
.050			-.574	-.127	-.588	-.103	-.532	-.128
.100			-.528	-.248	-.502	-.170	-.435	-.122
.150			-.531	-.318	-.556	-.273	-.476	-.254
.250	-.392		-.527	-.346	-.513	-.318	-.475	-.314
.400	-.487		-.504	-.354	-.495	-.367	-.432	-.339
.550	-.412	-.173	-.416	-.324	-.399	-.325	-.337	-.280
.700	-.361	-.268	-.344	-.389	-.290	-.308	-.212	-.206
.850	-.341	-.852	-.291	-.547	-.208	-.273	-.124	-.128
.950	-.338	-.481	-.261	-.400	-.150	-.187	-.054	-.036

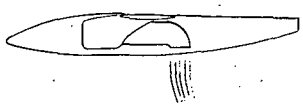
ALPHA = -.02 DEG Q = 2847 N/SQ.M. (59.47 LBF/SQ.FT.) EFFECT. VELOC. RATIO = .516

	ETA = .250		ETA = .387		ETA = .524		ETA = .800	
X/C	CP		CP		CP		CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			.812		.947		.794	
.010			.728	.210				
.015					-.442	.090		
.025			-.618	-.115	-.821	-.105	-.646	.120
.050			-.566	-.123	-.528	-.055	-.433	-.111
.100			-.511	-.245	-.457	-.079	-.452	-.165
.150			-.502	-.320	-.478	-.215	-.444	-.262
.250	-.371		-.486	-.306	-.485	-.294	-.454	-.323
.400	-.454		-.495	-.312	-.472	-.327	-.421	-.307
.550	-.387	-.128	-.390	-.260	-.372	-.284	-.319	-.260
.700	-.331	-.176	-.315	-.297	-.255	-.240	-.184	-.180
.850	-.290	-.688	-.249	-.441	-.163	-.199	-.100	-.097
.950	-.268	-.364	-.206	-.309	-.104	-.123	-.028	-.012

TABLE 35 (B)

PRESSURE PROFILE ON WING

REAR NOZZLES DEFLECTED 90 DEGREES



ALPHA = 10.00 DEG Q = 226 N/SQ.M. (4.72 LBF/SQ.FT.) EFFECT. VELOC. RATIO = .098

	ETA = .250		ETA = .387		ETA = .524		ETA = .800	
X/C	CP		CP		CP		CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			-.121		-1.090		-2.548	
.010			-1.007	.588				
.015					-3.426	.925		
.025			-3.217	.834	-3.345	.785	-3.109	.799
.050			-1.725	.611	-1.943	.583	-1.557	.569
.100			-1.174	.318	-1.306	.346	-1.216	.304
.150			-1.027	.081	-1.104	.151	-1.062	.109
.250	-.560		-.867	-.202	-.944	-.168	-.895	-.107
.400	-.638		-.743	-.455	-.787	-.367	-.692	-.266
.550	-.540	-.587	-.597	-.523	-.611	-.472	-.536	-.307
.700	-.489	-.814	-.509	-.672	-.486	-.489	-.381	-.293
.850	-.506	-1.493	-.476	-.851	-.368	-.452	-.259	-.236
.950	-.607	-1.236	-.506	-.712	-.374	-.381	-.205	-.168

ALPHA = 9.99 DEG Q = 514 N/SQ.M. (10.73 LBF/SQ.FT.) EFFECT. VELOC. RATIO = .148

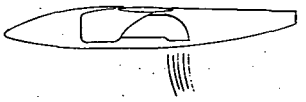
	ETA = .250		ETA = .387		ETA = .524		ETA = .800	
X/C	CP		CP		CP		CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			-.963		-2.804		-3.856	
.010			-.040	.890				
.015					-4.895	.979		
.025			-3.479	.970	-3.566	.948	-3.120	.884
.050			-2.206	.733	-2.245	.736	-2.117	.721
.100			-1.405	.466	-1.736	.473	-1.537	.411
.150			-1.282	.255	-1.414	.249	-1.256	.212
.250	-.693		-1.034	-.096	-1.092	-.019	-1.028	.015
.400	-.749		-.871	-.355	-.901	-.245	-.807	-.169
.550	-.640	-.518	-.688	-.480	-.706	-.408	-.593	-.255
.700	-.585	-.837	-.590	-.688	-.551	-.490	-.423	-.260
.850	-.584	-1.662	-.527	-.993	-.443	-.498	-.303	-.246
.950	-.664	-1.317	-.532	-.832	-.401	-.413	-.231	-.202

ALPHA = 9.99 DEG Q = 909 N/SQ.M. (18.98 LBF/SQ.FT.) EFFECT. VELOC. RATIO = .196

	ETA = .250		ETA = .387		ETA = .524		ETA = .800	
X/C	CP		CP		CP		CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			-1.434		-3.702		-5.105	
.010			.268	.761				
.015					-5.419	.965		
.025			-3.265	.581	-3.947	.936	-3.102	.924
.050			-2.168	.815	-2.540	.811	-2.220	.718
.100			-1.559	.579	-1.694	.572	-1.556	.497
.150			-1.294	.345	-1.441	.334	-1.290	.300
.250	-.716		-1.080	.042	-1.158	.094	-1.113	.073
.400	-.754		-.875	-.264	-.945	-.189	-.823	-.125
.550	-.649	-.369	-.659	-.366	-.709	-.329	-.607	-.210
.700	-.575	-.664	-.580	-.601	-.543	-.393	-.420	-.206
.850	-.555	-1.566	-.505	-.924	-.420	-.447	-.289	-.206
.950	-.599	-1.197	-.487	-.770	-.366	-.379	-.201	-.164

TABLE 35 (B) – CONCLUDED
PRESSURE PROFILE ON WING

REAR NOZZLES DEFLECTED 90 DEGREES



ALPHA = 9.98 DEG Q = 2031 N/SQ.M. (42.41 LBF/SQ.FT.) EFFECT. VELOC. RATIO = .294

	ETA = .250		ETA = .387		ETA = .524		ETA = .800	
X/C	CP		CP		CP		CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			-.291		-.878		-1.880	
.010			.333	.916				
.015					-1.374	.973		
.025			-1.817	.947	-1.638	.904	-1.654	.867
.050			-1.747	.802	-1.519	.738	-1.862	.688
.100			-1.973	.576	-1.671	.522	-1.680	.447
.150			-1.770	.397	-1.629	.362	-1.708	.284
.250	-.790		-1.373	.135	-1.383	.148	-1.279	.086
.400	-.733		-.907	-.086	-1.039	-.063	-.956	-.069
.550	-.605	-.089	-.635	-.189	-.757	-.180	-.574	-.150
.700	-.499	-.316	-.545	-.384	-.542	-.256	-.467	-.166
.850	-.447	-1.158	-.426	-.691	-.512	-.342	-.308	-.187
.950	-.451	-.879	-.406	-.583	-.357	-.355	-.276	-.180

ALPHA = 9.99 DEG Q = 2848 N/SQ.M. (59.48 LBF/SQ.FT.) EFFECT. VELOC. RATIO = .395

	ETA = .250		ETA = .387		ETA = .524		ETA = .800	
X/C	CP		CP		CP		CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			-.069		-.618		-1.674	
.010			.445	.946				
.015					-1.483	.973		
.025			-1.652	.943	-1.332	.895	-1.571	.859
.050			-1.672	.802	-1.376	.745	-1.388	.689
.100			-1.441	.555	-1.493	.542	-1.623	.458
.150			-1.434	.423	-1.499	.379	-1.580	.292
.250	-.777		-1.326	.179	-1.304	.186	-1.223	.127
.400	-.732		-.915	-.019	-1.076	-.006	-1.123	-.038
.550	-.560	.002	-.697	-.058	-.757	-.101	-.660	-.107
.700	-.461	-.191	-.436	-.267	-.565	-.178	-.472	-.125
.850	-.385	-.538	-.385	-.569	-.408	-.269	-.284	-.144
.950	-.339	-.698	-.341	-.463	-.355	-.297	-.256	-.183

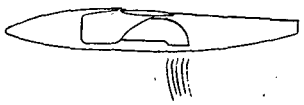
ALPHA = 10.00 DEG Q = 2845 N/SQ.M. (59.43 LBF/SQ.FT.) EFFECT. VELOC. RATIO = .517

	ETA = .250		ETA = .387		ETA = .524		ETA = .800	
X/C	CP		CP		CP		CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			.020		-.397		-1.460	
.010			.400	.948				
.015					-1.285	.972		
.025			-1.145	.939	-1.249	.893	-1.504	.848
.050			-1.466	.797	-1.244	.731	-1.434	.675
.100			-1.496	.588	-1.227	.542	-1.472	.460
.150			-1.676	.423	-1.342	.385	-1.458	.300
.250	-.669		-1.261	.194	-1.242	.200	-1.264	.119
.400	-.702		-.775	.023	-1.037	.035	-.823	-.025
.550	-.558	.065	-.662	-.044	-.736	-.052	-.578	-.089
.700	-.436	-.083	-.429	-.131	-.685	-.115	-.471	-.107
.850	-.347	-.754	-.494	-.449	-.395	-.205	-.292	-.118
.950	-.307	-.554	-.307	-.409	-.334	-.239	-.216	-.136

TABLE 36 (A)

PRESSURE PROFILE ON WING

REAR NOZZLES DEFLECTED 90 DEGREES WITH FLAPS



ALPHA = .02 DEG Q = 227 N/SQ.M. (4.74 LBF/SQ.FT.) EFFECT. VELOC. RATIO = .098

	ETA = .250		ETA = .387		ETA = .524		ETA = .800	
X/C	CP		CP		CP		CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			.649		.670		.614	
.010			-1.595	.114				
.015					-1.144	-.039		
.025			-1.074	-.310	-1.102	-.442	-1.040	-.261
.050			-.949	-.372	-1.053	-.407	-.908	-.525
.100			-.859	-.491	-.831	-.491	-.769	-.560
.150			-.901	-.609	-.928	-.567	-.782	-.588
.250	-.734		-.901	-.623	-.928	-.545	-.838	-.656
.400	-.875		-.916	-.596	-.919	-.602	-.801	-.683
.550	-.855	-.535	-.882	-.498	-.858	-.481	-.720	-.623
.700	-.855	-.515	-.872	-.350	-.872	-.296	-.616	-.582
.850	-.912	-.616	-.919	-.138	-.865	.014	-.549	-.515
.950	-1.040	-.552	-.976	.051	-1.017	.125	-.481	-.474

ALPHA = .04 DEG Q = 512 N/SQ.M. (10.69 LBF/SQ.FT.) EFFECT. VELOC. RATIO = .147

	ETA = .250		ETA = .387		ETA = .524		ETA = .800	
X/C	CP		CP		CP		CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			.804		.509		.497	
.010			-.666	.691				
.015					-1.469	.358		
.025			-1.245	.192	-1.362	.232	-1.306	.066
.050			-1.125	.054	-1.186	.149	-.913	-.008
.100			-.985	-.094	-.857	.008	-.789	-.196
.150			-.913	-.269	-.943	-.177	-.774	-.303
.250	-.673		-.934	-.328	-.919	-.270	-.793	-.386
.400	-.815		-.886	-.299	-.899	-.334	-.738	-.432
.550	-.805	-.161	-.836	-.189	-.814	-.225	-.611	-.416
.700	-.811	-.129	-.811	-.112	-.812	-.091	-.478	-.368
.850	-.914	-.314	-.874	.054	-.848	.194	-.387	-.344
.950	-1.069	-.358	-.950	.211	-.986	.311	-.332	-.298

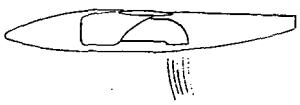
ALPHA = -.02 DEG Q = 907 N/SQ.M. (18.95 LBF/SQ.FT.) EFFECT. VELOC. RATIO = .196

	ETA = .250		ETA = .387		ETA = .524		ETA = .800	
X/C	CP		CP		CP		CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			.864		.488		.215	
.010			-.185	.696				
.015					-1.557	.510		
.025			-1.931	.305	-1.435	.359	-1.178	.260
.050			-1.248	.164	-1.293	.197	-1.055	.008
.100			-.959	.027	-.971	.036	-.779	-.098
.150			-.878	-.131	-.900	-.113	-.720	-.167
.250	-.662		-.848	-.104	-.871	-.150	-.760	-.272
.400	-.827		-.912	-.167	-.849	-.197	-.684	-.343
.550	-.790	-.069	-.797	-.090	-.774	-.133	-.567	-.340
.700	-.774	.018	-.776	-.004	-.749	.020	-.445	-.292
.850	-.884	-.146	-.833	.153	-.790	.268	-.361	-.258
.950	-1.064	-.201	-.877	.305	-.919	.380	-.278	-.202

TABLE 36 (A) – CONCLUDED

PRESSURE PROFILE ON WING

REAR NOZZLES DEFLECTED 90 DEGREES WITH FLAPS



ALPHA = -.01 DEG Q = 2035 N/SQ.M. (42.50 LBF/SQ.FT.) EFFECT. VELOC. RATIO = .294

	ETA = .250		ETA = .387		ETA = .524		ETA = .800	
X/C	CP		CP		CP		CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			.910		.572		.046	
.010			.514	.740				
.015					-1.596	.582		
.025			-1.293	.333	-1.545	.401	-1.003	.202
.050			-1.099	.260	-1.014	.236	-.867	.097
.100			-.770	.092	-.807	.110	-.703	.057
.150			-.849	.004	-.750	.071	-.700	-.116
.250	-.576		-.773	-.088	-.811	-.087	-.648	-.207
.400	-.696		-.768	-.033	-.763	-.092	-.588	-.243
.550	-.676	.092	-.704	.041	-.665	-.012	-.476	-.224
.700	-.671	.177	-.665	.130	-.633	.146	-.375	-.192
.850	-.742	.047	-.690	.272	-.640	.372	-.272	-.160
.950	-.900	.027	-.720	.422	-.719	.485	-.168	-.106

ALPHA = .01 DEG Q = 2844 N/SQ.M. (59.41 LBF/SQ.FT.) EFFECT. VELOC. RATIO = .393

	ETA = .250		ETA = .387		ETA = .524		ETA = .800	
X/C	CP		CP		CP		CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			.901		.678		.169	
.010			.718	.791				
.015					-1.510	.496		
.025			-1.217	.393	-1.425	.414	-1.021	.291
.050			-.998	.244	-1.063	.246	-.827	.142
.100			-.765	.155	-.785	.136	-.640	.046
.150			-.797	-.015	-.752	.047	-.616	-.083
.250	-.536		-.731	-.017	-.716	-.017	-.612	-.170
.400	-.656		-.732	.020	-.713	-.030	-.550	-.218
.550	-.616	.147	-.647	.106	-.599	.044	-.432	-.186
.700	-.600	.240	-.605	.199	-.552	.196	-.316	-.143
.850	-.651	.140	-.598	.339	-.546	.413	-.219	-.112
.950	-.791	.123	-.606	.476	-.595	.535	-.129	-.059

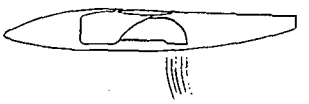
ALPHA = 0.00 DEG Q = 2846 N/SQ.M. (59.44 LBF/SQ.FT.) EFFECT. VELOC. RATIO = .515

	ETA = .250		ETA = .387		ETA = .524		ETA = .800	
X/C	CP		CP		CP		CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			.954		.566		.387	
.010			.776	.687				
.015					-1.092	.506		
.025			-1.132	.303	-1.434	.340	-1.125	.364
.050			-.943	.242	-.927	.216	-.749	.172
.100			-.641	.126	-.736	.151	-.645	.032
.150			-.691	-.029	-.796	.067	-.581	-.101
.250	-.511		-.695	-.043	-.664	-.033	-.577	-.162
.400	-.623		-.681	.012	-.673	-.016	-.510	-.187
.550	-.572	.140	-.584	.138	-.559	.086	-.409	-.168
.700	-.555	.255	-.533	.225	-.522	.224	-.293	-.129
.850	-.590	.191	-.523	.365	-.479	.444	-.188	-.084
.950	-.706	.167	-.509	.508	-.518	.557	-.099	-.028

TABLE 36 (B)

PRESSURE PROFILE ON WING

REAR NOZZLES DEFLECTED 90 DEGREES WITH FLAPS



ALPHA = 9.58 DEG Q = 224 N/SQ.M. (4.69 LBF/SQ.FT.) EFFECT. VELOC. RATIO = .098

	ETA = .250		ETA = .387		ETA = .524		ETA = .800	
X/C	CP		CP		CP		CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			-1.904		-3.646		-4.432	
.010				.441				
.015					-5.752	.680		
.025			-4.207	.708	-4.299	.680	-3.814	.623
.050			-2.831	.581	-2.578	.532	-2.361	.476
.100			-1.729	.342	-2.003	.286	-1.799	.167
.150			-1.588	.153	-1.722	.111	-1.532	-.009
.250	-.992		-1.350	-.142	-1.455	-.106	-1.336	-.232
.400	-1.079		-1.219	-.276	-1.263	-.225	-1.124	-.419
.550	-1.018	-.371	-1.093	-.256	-1.075	-.256	-.923	-.484
.700	-.981	-.375	-1.021	-.201	-1.021	-.167	-.766	-.508
.850	-1.021	-.603	-1.032	-.140	-1.018	.078	-.651	-.514
.950	-1.083	-.623	-1.059	.112	-1.103	.187	-.548	-.511

ALPHA = 10.03 DEG Q = 514 N/SQ.M. (10.73 LBF/SQ.FT.) EFFECT. VELOC. RATIO = .148

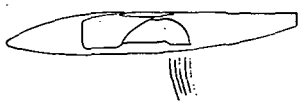
	ETA = .250		ETA = .387		ETA = .524		ETA = .800	
X/C	CP		CP		CP		CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			-1.177		-1.733		-2.929	
.010				.544				
.015					-2.183	.857		
.025			-2.683	.881	-2.260	.850	-2.683	.820
.050			-2.671	.774	-2.177	.722	-2.625	.651
.100			-2.723	.627	-2.245	.522	-2.625	.412
.150			-2.585	.418	-2.312	.350	-2.530	.228
.250	-1.122		-2.318	.148	-2.153	.157	-2.153	-.007
.400	-1.209		-1.303	-.004	-1.956	-.001	-1.317	-.184
.550	-1.089	-.308	-1.147	.020	-1.434	-.023	-.961	-.277
.700	-1.016	-.063	-1.047	.044	-1.184	.039	-.771	-.314
.850	-1.065	-.362	-1.074	.090	-1.116	.271	-.597	-.362
.950	-1.104	-.345	-1.147	.276	-1.227	.329	-.497	-.387

ALPHA = 10.10 DEG Q = 904 N/SQ.M. (18.89 LBF/SQ.FT.) EFFECT. VELOC. RATIO = .196

	ETA = .250		ETA = .387		ETA = .524		ETA = .800	
X/C	CP		CP		CP		CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			-1.186		-1.635		-2.775	
.010				.582				
.015					-2.031	.685		
.025			-2.386	.929	-2.005	.857	-2.421	.857
.050			-2.381	.857	-1.970	.802	-2.439	.704
.100			-2.439	.671	-2.015	.601	-2.407	.476
.150			-2.397	.526	-1.972	.432	-2.379	.310
.250	-1.223		-2.285	.288	-2.076	.265	-2.167	.098
.400	-1.258		-1.872	.131	-1.892	.115	-1.755	-.057
.550	-1.138	.172	-1.312	.136	-1.615	.098	-.980	-.160
.700	-1.022	.117	-1.052	.147	-1.290	.153	-.731	-.218
.850	-1.050	-.186	-1.043	.205	-1.100	.356	-.557	-.278
.950	-1.088	-.169	-1.111	.314	-1.148	.424	-.412	-.310

TABLE 36 (B) — CONCLUDED
PRESSURE PROFILE ON WING

REAR NOZZLES DEFLECTED 90 DEGREES WITH FLAPS



ALPHA = 10.12 DEG Q = 2039 N/SQ.M. (42.58 LBF/SQ.FT.) EFFECT. VELOC. RATIO = .295

	ETA = .250		ETA = .387		ETA = .524		ETA = .800	
X/C	CP		CP		CP		CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			-.965		-1.373		-2.482	
.010				.655				
.015					-1.687	.927		
.025			-2.096	.968	-1.719	.946	-2.123	.888
.050			-1.961	.896	-1.674	.842	-2.155	.756
.100			-1.996	.738	-1.695	.672	-2.104	.544
.150			-1.995	.597	-1.742	.527	-2.163	.388
.250	-1.234		-2.010	.410	-1.750	.360	-2.154	.172
.400	-1.273		-1.821	.260	-1.786	.228	-1.743	.015
.550	-1.036	.336	-1.473	.271	-1.529	.205	-1.244	-.082
.700	-.930	.333	-1.235	.279	-1.295	.275	-.747	-.125
.850	-.892	.107	-.916	.371	-1.083	.462	-.489	-.177
.950	-.905	.080	-.917	.476	-1.002	.545	-.338	-.206

ALPHA = 10.02 DEG Q = 2843 N/SQ.M. (59.38 LBF/SQ.FT.) EFFECT. VELOC. RATIO = .393

	ETA = .250		ETA = .387		ETA = .524		ETA = .800	
X/C	CP		CP		CP		CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			-.696		-1.185		-2.211	
.010				.721				
.015					-1.507	.946		
.025			-1.695	.976	-1.501	.950	-1.971	.904
.050			-1.570	.910	-1.487	.849	-1.892	.766
.100			-1.648	.734	-1.553	.669	-1.943	.544
.150			-1.820	.632	-1.567	.531	-1.988	.379
.250	-1.143		-1.904	.453	-1.638	.399	-1.968	.216
.400	-1.209		-1.693	.306	-1.659	.274	-1.716	.044
.550	-.963	.381	-1.386	.213	-1.517	.263	-1.125	-.037
.700	-.848	.396	-1.147	.353	-1.155	.327	-.695	-.074
.850	-.766	.240	-.954	.439	-1.027	.506	-.451	-.123
.950	-.751	.187	-.867	.562	-.903	.587	-.291	-.149

ALPHA = 9.98 DEG Q = 2838 N/SQ.M. (59.27 LBF/SQ.FT.) EFFECT. VELOC. RATIO = .514

	ETA = .250		ETA = .387		ETA = .524		ETA = .800	
X/C	CP		CP		CP		CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			-.493		-.993		-2.022	
.010				.788				
.015					-1.374	.954		
.025			-1.496	.914	-1.391	.944	-1.817	.894
.050			-1.668	.904	-1.274	.847	-1.797	.752
.100			-1.609	.748	-1.462	.664	-1.786	.543
.150			-1.546	.611	-1.532	.550	-1.771	.384
.250	-1.122		-1.721	.443	-1.553	.406	-1.823	.209
.400	-1.194		-1.566	.335	-1.544	.281	-1.687	.062
.550	-.970	.403	-1.333	.352	-1.396	.284	-1.135	-.011
.700	-.810	.429	-.979	.388	-1.117	.351	-.751	-.043
.850	-.675	.306	-.885	.484	-.970	.534	-.429	-.091
.950	-.634	.267	-.672	.598	-.820	.607	-.260	-.123

TABLE 37 (A)

PRESSURE PROFILE ON WING

FORWARD NOZZLES DEFLECTED 90 DEGREES WITH LIFT JET

ALPHA = -.01 DEG Q = 227 N/SQ.M. (4.74 LBF/SQ.FT.) EFFECT. VELOC. RATIO = .094

X/C	ETA = .250 CP		ETA = .387 CP		ETA = .524 CP		ETA = .800 CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			-4.541		-4.410		-1.295	
.010			-2.197	-4.868				
.015					.945	-4.777		
.025			.578	-4.451	.786	-4.292	.647	-2.807
.050			.335	-3.002	.425	-2.648	.439	-2.343
.100			.057	-2.502	.185	-2.190	.196	-1.420
.150			-.040	-2.558	.051	-2.266	.030	-1.358
.250	-.213		-.227	-2.325	-.130	-2.015	-.158	-1.387
.400	-.304		-.311	-2.856	-.274	-1.992	-.294	-1.138
.550	-.280	-5.344	-.300	-2.789	-.274	-1.706	-.270	-.916
.700	-.290	-5.250	-.321	-2.785	-.257	-1.373	-.213	-.684
.850	-.368	-4.251	-.368	-1.834	-.314	-.916	-.196	-.472
.950	-.596	-2.395	-.569	-1.319	-.411	-.667	-.220	-.297

ALPHA = .01 DEG Q = 513 N/SQ.M. (10.71 LBF/SQ.FT.) EFFECT. VELOC. RATIO = .142

X/C	ETA = .250 CP		ETA = .387 CP		ETA = .524 CP		ETA = .800 CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			-2.222		-1.635		.007	
.010			-.941	-3.332				
.015					.546	-3.120		
.025			.014	-2.674	.340	-2.533	.119	-1.338
.050			.073	-2.210	.119	-1.952	-.130	-1.199
.100			-.127	-2.186	-.063	-1.789	-.133	-.984
.150			-.265	-2.244	-.228	-1.722	-.228	-.960
.250	-.284		-.361	-2.296	-.336	-1.914	-.314	-.925
.400	-.391		-.421	-2.722	-.407	-1.880	-.383	-.924
.550	-.349	-5.388	-.377	-2.939	-.382	-1.688	-.318	-.765
.700	-.322	-5.408	-.373	-2.467	-.349	-1.325	-.264	-.613
.850	-.382	-3.101	-.415	-1.652	-.322	-.918	-.199	-.404
.950	-.607	-2.328	-.552	-1.103	-.358	-.566	-.164	-.252

ALPHA = .00 DEG Q = 911 N/SQ.M. (19.03 LBF/SQ.FT.) EFFECT. VELOC. RATIO = .189

X/C	ETA = .250 CP		ETA = .387 CP		ETA = .524 CP		ETA = .800 CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			-.452		-.051		.943	
.010			-.283	-1.599				
.015					.162	-1.457		
.025			-.113	-1.386	-.049	-1.391	-.127	-.639
.050			-.191	-1.269	-.293	-1.222	-.222	-.784
.100			-.282	-1.661	-.279	-1.295	-.305	-.594
.150			-.395	-1.609	-.342	-1.380	-.387	-.693
.250	-.325		-.439	-1.758	-.361	-1.361	-.413	-.773
.400	-.434		-.456	-2.050	-.440	-1.463	-.420	-.720
.550	-.389	-3.212	-.395	-2.158	-.408	-1.305	-.337	-.621
.700	-.354	-3.073	-.357	-1.834	-.336	-1.052	-.266	-.502
.850	-.408	-2.748	-.366	-1.272	-.281	-.722	-.169	-.309
.950	-.622	-1.992	-.431	-.737	-.269	-.444	-.130	-.171

TABLE 37 (A) – CONCLUDED

PRESSURE PROFILE ON WING

FORWARD NOZZLES DEFLECTED 90 DEGREES WITH LIFT JET

ALPHA = .00 DEG Q = 2035 N/SQ.M. (42.51 LBF/SQ.FT.) EFFECT. VELOC. RATIO = .282

	ETA = .250		ETA = .387		ETA = .524		ETA = .800	
X/C	CP		CP		CP		CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			.720		.905		.967	
.010			.394	-.415				
.015					-.382	-.466		
.025			-.612	-.759	-.615	-.425	-.527	-.206
.050			-.537	-.664	-.500	-.585	-.561	-.186
.100			-.556	-.921	-.471	-.643	-.453	-.319
.150			-.542	-1.062	-.528	-.826	-.477	-.437
.250	-.416		-.538	-1.227	-.581	-.964	-.504	-.501
.400	-.515		-.558	-1.344	-.522	-1.051	-.462	-.544
.550	-.468	-1.582	-.463	-1.376	-.443	-.942	-.373	-.468
.700	-.434	-1.752	-.394	-1.262	-.341	-.773	-.246	-.368
.850	-.484	-1.885	-.370	-.965	-.258	-.574	-.152	-.253
.950	-.703	-1.746	-.379	-.613	-.231	-.351	-.102	-.125

ALPHA = -.01 DEG Q = 2846 N/SQ.M. (59.44 LBF/SQ.FT.) EFFECT. VELOC. RATIO = .386

	ETA = .250		ETA = .347		ETA = .524		ETA = .800	
X/C	CP		CP		CP		CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			.833		.965		.910	
.010			.599	.152				
.015					-.625	.019		
.025			-.807	-.335	-.712	-.362	-.649	-.159
.050			-.654	-.448	-.627	-.232	-.657	-.149
.100			-.526	-.675	-.548	-.502	-.464	-.268
.150			-.590	-.856	-.563	-.619	-.469	-.370
.250	-.424		-.572	-1.004	-.552	-.731	-.491	-.398
.400	-.506		-.557	-1.081	-.520	-.827	-.459	-.466
.550	-.452	-1.184	-.442	-1.069	-.418	-.765	-.346	-.411
.700	-.417	-1.302	-.379	-.934	-.322	-.638	-.227	-.317
.850	-.444	-1.316	-.341	-.787	-.230	-.458	-.140	-.211
.950	-.642	-1.407	-.328	-.489	-.184	-.284	-.074	-.092

ALPHA = .00 DEG Q = 2841 N/SQ.M. (59.33 LBF/SQ.FT.) EFFECT. VELOC. RATIO = .523

	ETA = .250		ETA = .387		ETA = .524		ETA = .800	
X/C	CP		CP		CP		CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			.989		.939		.830	
.010			.635	.325				
.015					-.745	.247		
.025			-.955	-.127	-.801	-.117	-.611	.108
.050			-.748	-.132	-.705	-.165	-.685	-.114
.100			-.571	-.474	-.616	-.238	-.478	-.151
.150			-.575	-.638	-.565	-.458	-.484	-.291
.250	-.395		-.547	-.805	-.529	-.572	-.506	-.346
.400	-.489		-.520	-.881	-.507	-.656	-.433	-.402
.550	-.429	-.915	-.441	-.837	-.395	-.624	-.339	-.356
.700	-.384	-.970	-.341	-.718	-.303	-.532	-.211	-.269
.850	-.354	-.962	-.296	-.608	-.198	-.397	-.122	-.173
.950	-.493	-.756	-.261	-.379	-.152	-.231	-.054	-.060

TABLE 37 (B)

PRESSURE PROFILE ON WING

FORWARD NOZZLES DEFLECTED 90 DEGREES WITH LIFT JET

ALPHA = 10.02 DEG Q = 227 N/SQ.M. (4.74 LBF/SQ.FT.) EFFECT. VELOC. RATIO = .095

	ETA = .250		ETA = .387		ETA = .524		ETA = .800	
X/C	CP		CP		CP		CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			.391		.765		.911	
.010			-.553	-.802				
.015					-.310	-1.087		
.025			-.587	-1.198	-.540	-1.177	-.768	-.365
.050			-.546	-1.350	-.511	-1.184	-.657	-.490
.100			-.407	-1.565	-.476	-1.329	-.601	-.553
.150			-.518	-1.808	-.393	-1.482	-.664	-.670
.250	-.393		-.490	-2.138	-.559	-1.771	-.615	-.826
.400	-.440		-.514	-3.113	-.540	-1.902	-.524	-.830
.550	-.352	-9.000	-.440	-3.439	-.453	-1.717	-.423	-.702
.700	-.275	-9.095	-.396	-2.982	-.382	-1.428	-.319	-.584
.850	-.298	-2.176	-.362	-1.566	-.329	-.877	-.194	-.446
.950	-.443	-1.452	-.463	-.984	-.319	-.544	-.160	-.214

ALPHA = 10.01 DEG Q = 514 N/SQ.M. (10.73 LBF/SQ.FT.) EFFECT. VELOC. RATIO = .142

	ETA = .250		ETA = .387		ETA = .524		ETA = .800	
X/C	CP		CP		CP		CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			.948		.923		-.334	
.010			.117	.206				
.015					-1.491	-.129		
.025			-1.607	-.310	-1.534	-.233	-1.755	.325
.050			-1.163	-.558	-1.175	-.558	-1.482	.153
.100			-.865	-.979	-.963	-.730	-.991	-.169
.150			-.856	-1.322	-.899	-1.025	-.939	-.350
.250	-.546		-.748	-1.852	-.794	-1.314	-.828	-.518
.400	-.594		-.696	-2.534	-.679	-1.547	-.656	-.628
.550	-.484	-5.648	-.549	-2.939	-.530	-1.456	-.515	-.610
.700	-.409	-5.695	-.464	-2.408	-.420	-1.242	-.366	-.527
.850	-.393	-2.591	-.439	-1.471	-.338	-.867	-.262	-.411
.950	-.551	-1.510	-.491	-.949	-.340	-.537	-.209	-.282

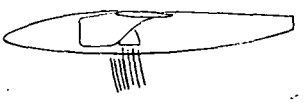
ALPHA = 10.03 DEG Q = 910 N/SQ.M. (19.00 LBF/SQ.FT.) EFFECT. VELOC. RATIO = .189

	ETA = .250		ETA = .387		ETA = .524		ETA = .800	
X/C	CP		CP		CP		CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			.794		.135		-1.815	
.010			.355	.886				
.015					-2.522	.595		
.025			-2.231	.356	-2.328	.377	-2.698	.689
.050			-1.517	-.051	-1.677	.055	-1.550	.384
.100			-1.124	-.432	-1.351	-.244	-1.299	.152
.150			-1.067	-.856	-1.098	-.560	-1.084	-.054
.250	-.591		-.857	-1.413	-.953	-.906	-.960	-.311
.400	-.627		-.741	-1.893	-.811	-1.099	-.743	-.428
.550	-.519	-3.201	-.571	-2.039	-.595	-1.140	-.547	-.450
.700	-.428	-3.036	-.462	-1.710	-.456	-.989	-.385	-.388
.850	-.412	-2.261	-.410	-1.155	-.349	-.710	-.268	-.314
.950	-.545	-1.401	-.419	-.700	-.312	-.475	-.195	-.221

TABLE 37 (B) – CONCLUDED

PRESSURE PROFILE ON WING

FORWARD NOZZLES DEFLECTED 90 DEGREES WITH LIFT JET



ALPHA = 10.00 DEG Q = 2035 N/SQ.M. (42.50 LBF/SQ.FT.) EFFECT. VELOC. RATIO = .282

	ETA = .250		ETA = .387		ETA = .524		ETA = .800	
X/C	CP		CP		CP		CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			-.572		-2.205		-4.533	
.010			.532	.953				
.015					-4.842	.980		
.025			-3.215	.667	-3.485	.891	-3.003	.904
.050			-2.214	.516	-2.370	.568	-2.200	.651
.100			-1.495	.126	-1.635	.240	-1.526	.376
.150			-1.269	-.241	-1.418	-.012	-1.293	.191
.250	-.687		-1.055	-.740	-1.119	-.384	-1.051	-.045
.400	-.718		-.841	-1.257	-.899	-.696	-.797	-.240
.550	-.591	-1.616	-.648	-1.259	-.658	-.774	-.570	-.295
.700	-.497	-1.679	-.512	-1.127	-.485	-.699	-.401	-.274
.850	-.480	-1.622	-.435	-.893	-.358	-.586	-.268	-.237
.950	-.536	-1.148	-.387	-.574	-.282	-.405	-.164	-.167

ALPHA = 10.02 DEG Q = 2846 N/SQ.M. (59.43 LBF/SQ.FT.) EFFECT. VELOC. RATIO = .385

	ETA = .250		ETA = .387		ETA = .524		ETA = .800	
X/C	CP		CP		CP		CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			-1.699		-3.714		-4.558	
.010			.508	.826				
.015					-4.558	.978		
.025			-3.499	.943	-3.884	.932	-3.258	.920
.050			-2.314	.764	-2.504	.728	-2.253	.736
.100			-1.586	.346	-1.885	.397	-1.530	.461
.150			-1.360	-.012	-1.464	.208	-1.282	.241
.250	-.695		-1.053	-.475	-1.162	-.201	-1.066	.020
.400	-.733		-.856	-.902	-.895	-.472	-.809	-.180
.550	-.588	-1.210	-.656	-1.052	-.643	-.620	-.568	-.232
.700	-.501	-1.251	-.501	-.939	-.474	-.588	-.394	-.211
.850	-.461	-1.053	-.395	-.768	-.335	-.505	-.251	-.194
.950	-.501	-.849	-.391	-.512	-.247	-.360	-.151	-.135

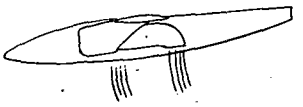
ALPHA = 10.03 DEG Q = 2846 N/SQ.M. (59.45 LBF/SQ.FT.) EFFECT. VELOC. RATIO = .526

	ETA = .250		ETA = .387		ETA = .524		ETA = .800	
X/C	CP		CP		CP		CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			-.070		-.632		-1.875	
.010			.462	.960				
.015					-1.571	.954		
.025			-1.603	.910	-1.469	.842	-1.542	.833
.050			-1.633	.717	-1.328	.638	-1.775	.634
.100			-1.468	.469	-1.309	.387	-1.876	.393
.150			-1.395	.114	-1.628	.149	-1.658	.203
.250	-.792		-1.427	-.320	-1.362	-.138	-1.257	-.001
.400	-.737		-.833	-.742	-1.090	-.417	-.878	-.177
.550	-.621	-.812	-.697	-.848	-.801	-.539	-.645	-.232
.700	-.528	-.901	-.648	-.821	-.598	-.557	-.458	-.244
.850	-.485	-.900	-.395	-.767	-.431	-.492	-.294	-.236
.950	-.504	-.764	-.357	-.495	-.351	-.431	-.336	-.213

TABLE 38 'A)

PRESSURE PROFILE ON WING

REAR NOZZLES DEFLECTED 90 DEGREES WITH LIFT JET



ALPHA = -.02 DEG Q = 222 N/SQ.M. (4.65 LBF/SQ.FT.) EFFECT. VELOC. RATIO = .092

	ETA = .250		ETA = .387		ETA = .524		ETA = .800	
X/C	CP UPPER	CP LOWER	CP UPPER	CP LOWER	CP UPPER	CP LOWER	CP UPPER	CP LOWER
.000			-1.104		-.360		.851	
.010			-2.974	-1.642				
.015					.716	-1.735		
.025			.489	-1.359	.556	-1.572	.433	-.934
.050			.284	-1.303	.312	-1.140	.270	-.523
.100			.213	-.842	.270	-.658	.220	-.438
.150			.036	-.856	.100	-.715	.128	-.488
.250	-.035		-.021	-.713	.029	-.607	-.042	-.486
.400	-.119		-.105	-.565	-.078	-.607	-.019	-.421
.550	-.098	-.390	-.067	-.497	-.047	-.407	.022	-.332
.700	-.102	-.483	-.040	-.442	-.009	-.315	.091	-.191
.850	-.143	-.795	-.050	-.486	.042	-.208	.128	-.026
.950	-.243	-.596	-.078	-.291	.073	-.064	.142	.104

ALPHA = -.02 DEG Q = 516 N/SQ.M. (10.78 LBF/SQ.FT.) EFFECT. VELOC. RATIO = .140

	ETA = .250		ETA = .387		ETA = .524		ETA = .800	
X/C	CP UPPER	CP LOWER	CP UPPER	CP LOWER	CP UPPER	CP LOWER	CP UPPER	CP LOWER
.000			-.612		.813			
.010			-.801	-.786				
.015					.166	-.551		
.025			-.130	-.670	-.133	-.590	-.200	-.392
.050			-.175	-.673	-.130	-.487	-.169	-.298
.100			-.169	-.511	-.221	-.496	-.276	-.328
.150			-.279	-.603	-.252	-.535	-.224	-.352
.250	-.230		-.301	-.731	-.294	-.573	-.282	-.436
.400	-.342		-.355	-.660	-.361	-.575	-.277	-.407
.550	-.306	-.506	-.312	-.552	-.294	-.519	-.195	-.367
.700	-.289	-.580	-.262	-.607	-.229	-.451	-.133	-.260
.850	-.306	-1.095	-.253	-.725	-.160	-.358	-.034	-.133
.950	-.377	-.704	-.252	-.488	-.112	-.198	.018	-.000

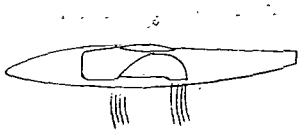
ALPHA = -.02 DEG Q = 908 N/SQ.M. (18.96 LBF/SQ.FT.) EFFECT. VELOC. RATIO = .185

	ETA = .250		ETA = .387		ETA = .524		ETA = .800	
X/C	CP UPPER	CP LOWER	CP UPPER	CP LOWER	CP UPPER	CP LOWER	CP UPPER	CP LOWER
.000			.520		.973			
.010			-.094	-.119				
.015					-.320	-.268		
.025			-.351	-.553	-.233	-.389	-.358	-.122
.050			-.310	-.490	-.414	-.384	-.327	-.167
.100			-.291	-.421	-.343	-.370	-.241	-.271
.150			-.358	-.632	-.377	-.494	-.365	-.339
.250	-.291		-.405	-.696	-.360	-.498	-.349	-.408
.400	-.402		-.446	-.642	-.452	-.552	-.366	-.404
.550	-.363	-.529	-.366	-.569	-.344	-.510	-.278	-.334
.700	-.333	-.580	-.322	-.602	-.295	-.458	-.182	-.249
.850	-.355	-1.182	-.290	-.754	-.191	-.381	-.067	-.148
.950	-.432	-.816	-.302	-.539	-.163	-.244	-.022	-.032

TABLE 38 (A) – CONCLUDED

PRESSURE PROFILE ON WING

REAR NOZZLES DEFLECTED 90 DEGREES WITH LIFT JET



ALPHA = -.00 DEG Q = 2034 N/SQ.M. (42.48 LBF/SQ.FT.) EFFECT. VELOC. RATIO = .278

	ETA = .250		ETA = .387		ETA = .524		ETA = .800	
X/C	CP		CP		CP		CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			.769		.995		.986	
.010			.526	-.044				
.015					-.310	-.153		
.025			-.489	-.319	-.641	-.145	-.529	-.013
.050			-.477	-.343	-.477	-.172	-.381	-.064
.100			-.364	-.402	-.456	-.276	-.400	-.191
.150			-.495	-.524	-.444	-.318	-.419	-.281
.250	-.350		-.478	-.594	-.453	-.421	-.427	-.354
.400	-.449		-.494	-.515	-.467	-.448	-.408	-.345
.550	-.391	-.389	-.396	-.444	-.378	-.417	-.301	-.288
.700	-.352	-.476	-.322	-.521	-.282	-.375	-.179	-.216
.850	-.346	-1.032	-.298	-.674	-.188	-.317	-.087	-.122
.950	-.390	-.722	-.284	-.476	-.152	-.207	-.022	-.026

ALPHA = -.04 DEG Q = 2837 N/SQ.M. (59.26 LBF/SQ.FT.) EFFECT. VELOC. RATIO = .379

	ETA = .250		ETA = .387		ETA = .524		ETA = .800	
X/C	CP		CP		CP		CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			.913		1.002		.898	
.010			.685	.185				
.015					-.575	.043		
.025			-.657	-.136	-.595	-.161	-.598	-.034
.050			-.515	-.223	-.448	-.052	-.474	-.069
.100			-.404	-.294	-.456	-.203	-.398	-.172
.150			-.475	-.398	-.468	-.236	-.392	-.208
.250	-.347		-.484	-.479	-.490	-.364	-.431	-.292
.400	-.441		-.478	-.440	-.455	-.376	-.394	-.309
.550	-.368	-.325	-.371	-.361	-.357	-.340	-.291	-.259
.700	-.326	-.424	-.292	-.405	-.245	-.287	-.167	-.170
.850	-.300	-.858	-.249	-.554	-.159	-.247	-.077	-.089
.950	-.313	-.658	-.223	-.380	-.110	-.149	-.006	.007

ALPHA = -.13 DEG Q = 2849 N/SQ.M. (59.50 LBF/SQ.FT.) EFFECT. VELOC. RATIO = .512

	ETA = .250		ETA = .387		ETA = .524		ETA = .800	
X/C	CP		CP		CP		CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			.939		.991		.953	
.010			.704	.160				
.015					-.553	.195		
.025			-.670	-.082	-.637	-.093	-.544	.009
.050			-.462	-.098	-.521	-.038	-.453	.075
.100			-.436	-.256	-.415	-.130	-.390	-.162
.150			-.448	-.338	-.499	-.222	-.406	-.198
.250	-.337		-.439	-.376	-.471	-.272	-.431	-.262
.400	-.416		-.426	-.363	-.430	-.310	-.359	-.273
.550	-.343	-.265	-.339	-.279	-.328	-.241	-.268	-.220
.700	-.282	-.341	-.254	-.302	-.224	-.211	-.134	-.134
.850	-.256	-.704	-.210	-.442	-.121	-.162	-.045	-.055
.950	-.270	-.554	-.168	-.288	-.062	-.077	.023	.032

TABLE 38 (B)

PRESSURE PROFILE ON WING

REAR NOZZLES DEFLECTED 90 DEGREES WITH LIFT JET

ALPHA = 9.97 DEG Q = 226 N/SQ.M. (4.72 LBF/SQ.FT.) EFFECT. VELOC. RATIO = .093

	ETA = .250		ETA = .387		ETA = .524		ETA = .800	
X/C	CP		CP		CP		CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000					.712		-.041	
.010			-.313					
.015					-1.658			
.025			-1.198	.831	-1.631	.775	-1.359	.914
.050			-.940	.614	-.940	.517	-.933	.733
.100			-.446	.287	-.595	.426	-.571	.426
.150			-.445	.203	-.473	.217	-.522	.287
.250	-.097		-.334	-.144	-.410	.012	-.383	.073
.400	-.154		-.245	-.178	-.292	-.036	-.225	.035
.550	-.083	-.140	-.120	-.161	-.130	-.147	-.083	.066
.700	-.039	-.276	-.039	-.221	-.022	-.100	.062	.089
.850	-.039	-.749	-.002	-.279	.056	-.015	.154	.154
.950	-.086	-.502	-.039	-.235	.100	.002	.201	.201

ALPHA = 9.99 DEG Q = 510 N/SQ.M. (10.65 LBF/SQ.FT.) EFFECT. VELOC. RATIO = .139

	ETA = .250		ETA = .387		ETA = .524		ETA = .800	
X/C	CP		CP		CP		CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			.407		-.928		-2.156	
.010			.314					
.015					-2.879			
.025			-2.622	.870	-2.466	.889	-2.446	.954
.050			-1.537	.611	-1.621	.623	-1.531	.710
.100			-1.061	.364	-1.170	.345	-1.188	.407
.150			-.907	.113	-1.030	.129	-.932	.224
.250	-.477		-.746	-.143	-.823	-.055	-.814	.032
.400	-.524		-.659	-.272	-.677	-.242	-.576	-.094
.550	-.443	-.450	-.477	-.375	-.486	-.313	-.393	-.161
.700	-.378	-.624	-.384	-.495	-.354	-.342	-.230	-.127
.850	-.359	-1.281	-.333	-.741	-.241	-.307	-.121	-.083
.950	-.428	-.982	-.310	-.569	-.193	-.251	-.047	-.032

ALPHA = 9.99 DEG Q = 908 N/SQ.M. (18.96 LBF/SQ.FT.) EFFECT. VELOC. RATIO = .186

	ETA = .250		ETA = .387		ETA = .524		ETA = .800	
X/C	CP		CP		CP		CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			-.444		-1.805		-3.620	
.010			.454					
.015					-4.140			
.025			-3.724	.521	-3.500	.949	-2.816	.973
.050			-1.826	.713	-1.836	.713	-1.869	.725
.100			-1.262	.407	-1.439	.470	-1.328	.480
.150			-1.086	.232	-1.231	.251	-1.073	.277
.250	-.589		-.916	-.071	-.963	-.035	-.944	.059
.400	-.640		-.749	-.314	-.771	-.206	-.677	-.093
.550	-.536	-.475	-.575	-.357	-.578	-.293	-.475	-.169
.700	-.458	-.691	-.457	-.505	-.420	-.341	-.304	-.146
.850	-.455	-1.341	-.387	-.777	-.303	-.345	-.175	-.116
.950	-.516	-1.107	-.370	-.603	-.239	-.275	-.090	-.068

TABLE 38 (B) – CONCLUDED

PRESSURE PROFILE ON WING

REAR NOZZLES DEFLECTED 90 DEGREES WITH LIFT JET

ALPHA = 9.99 DEG Q = 2039 N/SQ.M. (42.58 LBF/SQ.FT.) EFFECT. VELOC. RATIO = .279

	ETA = .250		ETA = .387		ETA = .524		ETA = .800	
X/C	CP		CP		CP		CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			-1.093		-3.037		-4.462	
.010			.556	.647				
.015					-4.806	1.012		
.025			-3.069	1.003	-3.656	1.003	-3.112	.988
.050			-2.126	.819	-2.395	.825	-2.129	.784
.100			-1.448	.571	-1.621	.570	-1.493	.533
.150			-1.269	.377	-1.335	.378	-1.203	.354
.250	-.638		-.977	.090	-1.065	.166	-.985	.141
.400	-.676		-.789	-.136	-.825	-.067	-.730	-.024
.550	-.556	-.285	-.593	-.221	-.603	-.184	-.502	-.102
.700	-.477	-.511	-.462	-.375	-.432	-.234	-.324	-.095
.850	-.440	-1.142	-.372	-.660	-.286	-.270	-.181	-.089
.950	-.463	-.955	-.337	-.511	-.213	-.215	-.094	-.055

ALPHA = 9.99 DEG Q = 2842 N/SQ.M. (59.36 LBF/SQ.FT.) EFFECT. VELOC. RATIO = .379

	ETA = .250		ETA = .387		ETA = .524		ETA = .800	
X/C	CP		CP		CP		CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			-1.444		-3.922		-1.656	
.010			.527	.804				
.015					-4.242	.955		
.025			-3.304	1.014	-3.756	1.004	-1.875	.894
.050			-2.146	.879	-2.386	.853	-1.603	.713
.100			-1.429	.622	-1.651	.620	-1.695	.487
.150			-1.235	.416	-1.385	.447	-1.529	.315
.250	-.642		-.966	.155	-1.055	.214	-1.080	.134
.400	-.648		-.786	-.060	-.809	.026	-.721	-.021
.550	-.537	-.293	-.576	-.112	-.576	-.074	-.487	-.089
.700	-.435	-.486	-.429	-.239	-.398	-.133	-.308	-.095
.850	-.386	-.937	-.324	-.502	-.247	-.176	-.302	-.121
.950	-.391	-.812	-.265	-.391	-.162	-.154	-.151	-.108

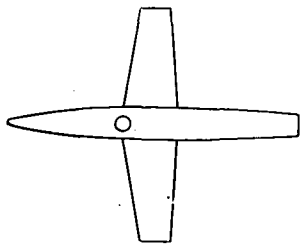
ALPHA = 9.98 DEG Q = 2846 N/SQ.M. (59.45 LBF/SQ.FT.) EFFECT. VELOC. RATIO = .515

	ETA = .250		ETA = .387		ETA = .524		ETA = .800	
X/C	CP		CP		CP		CP	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			.053		-.378		-1.356	
.010			.474	.956				
.015					-1.308	1.005		
.025			-1.407	.969	-1.303	.916	-1.648	.886
.050			-1.487	.802	-1.170	.757	-1.657	.709
.100			-1.406	.582	-1.205	.551	-1.414	.491
.150			-1.469	.436	-1.256	.398	-1.311	.326
.250	-.709		-1.235	.166	-1.249	.200	-1.135	.144
.400	-.687		-.955	-.029	-1.007	.034	-.752	-.002
.550	-.547	-.186	-.642	-.057	-.875	-.064	-.519	-.065
.700	-.401	-.352	-.468	-.172	-.590	-.105	-.418	-.068
.850	-.320	-.796	-.366	-.419	-.361	-.178	-.316	-.104
.950	-.344	-.668	-.274	-.355	-.299	-.223	-.155	-.108

TABLE 39 (A)

PRESSURE PROFILE ON FUSELAGE

LIFT JET ALONE



ALPHA = .00 DEG
EFF. VEL. RATIO = .099

	CP		
Y/YMAX =	0.0	0.5	0.8
X/L			
.130	-.059		
.219	-.104		
.286	-.152		
.331	-.157		
.353	-.047	-.257	-.394
.375	.035		
.397		-4.301	-1.419
.420	-2.289	-2.010	.153
.442	-1.774	-.647	-1.521
.464	.040	-.464	-.824
.487	-.464	.159	-.754
.509	-1.248	-.539	-.743
.531	-.432	-.405	-.512
.554	-.404		
.576	-.437		
.598	-.492	-.410	-.329
.620	-.473		
.643	-.356	-.318	-.233
.665	-.357	-.315	-.280
.687	-.245		
.710	-.286		
.732	-.223		

ALPHA = -.01 DEG
EFF. VEL. RATIO = .148

	CP		
	0.0	0.5	0.8
	-.055		
	-.069		
	-.106		
	-.056		
	.154	-.101	-.326
	.246		
		-4.247	-1.542
	-1.497	-2.066	-.124
	-1.775	-.719	-1.242
	-.299	-.532	-1.007
	-.513	-.422	-.822
	-.465	-.542	-.710
	-.515	-.465	-.563
	-.504		
	-.548		
	-.421	-.434	-.376
	-.511		
	-.434	-.418	-.312
	-.439	-.395	-.300
	-.409		
	-.366		
	-.411		

ALPHA = -.00 DEG
EFF. VEL. RATIO = .196

	CP		
	0.0	0.5	0.8
	-.057		
	-.036		
	-.040		
	.045		
	.256	-.012	-.203
	.369		
		-4.262	-1.489
	-1.067	-1.818	-1.266
	-1.424	-.822	-1.181
	-.205	-.516	-.742
	-.508	-1.429	-.416
	-.677	-.394	-.500
	-.444	-.425	-.428
	-.546		
	-.512		
	-.407	-.434	-.268
	-.404		
	-.367	-.240	-.252
	-.311	-.288	-.237
	-.381		
	-.277		
	-.295		

ALPHA = 0.00 DEG
EFF. VEL. RATIO = .205

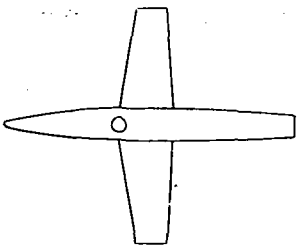
	CP		
Y/YMAX =	0.0	0.5	0.8
X/L			
.130	-.039		
.219	-.020		
.286	-.036		
.331	.030		
.353	.238	-.021	-.207
.375	.315		
.397		-4.930	-1.690
.420	-2.369	-2.304	-2.386
.442	-1.863	-.778	-1.322
.464	.514	-.547	-.930
.487	-.702	-1.398	-.638
.509	-.372	-.354	-.428
.531	-.635	-.387	-.492
.554	-.496		
.576	-.406		
.598	-.411	-.372	-.306
.620	-.453		
.643	-.321	-.356	-.253
.665	-.289	-.285	-.305
.687	-.342		
.710	-.265		
.732	-.284		

ALPHA = -.00 DEG
EFF. VEL. RATIO = .315

	CP		
	0.0	0.5	0.8
	-.055		
	-.023		
	.020		
	.152		
	.361	.140	-.043
	.478		
		-4.135	-1.353
	-1.322	-2.462	-1.746
	-1.580	-.875	-1.166
	-1.272	-.620	-.846
	-.509	-.629	-.544
	-.379	-.420	-.447
	-.373	-.386	-.391
	-.416		
	-.307		
	-.361	-.304	-.241
	-.310		
	-.254	-.240	-.190
	-.260	-.245	-.151
	-.226		
	-.250		
	-.195		

TABLE 39 (B)
PRESSURE PROFILE ON FUSELAGE

LIFT JET ALONE



ALPHA =10.02 DEG
EFF. VEL. RATIO = .100

	CP	
0.0	0.5	0.8
.070		
.003		
-.076		
-.087		
.030	-.169	-.316
.013		
	-4.038	-1.073
-.038	-1.255	.463
-.988	-.394	-.816
.018	-.292	-.715
-.372	-.319	-.576
-.415	-.383	-.394
-.335	-.404	-.372
-.313		
-.353		
-.385	-.340	-.240
-.382		
-.295	-.351	-.267
-.293	-.331	-.233
-.252		
-.249		
-.248		

ALPHA =10.03 DEG
EFF. VEL. RATIO = .150

	CP	
0.0	0.5	0.8
.066		
.042		
-.001		
.021		
.206	-.079	-.303
.227		
	-4.461	-1.444
-.687	-2.136	.241
-1.456	-.701	-1.326
-.075	-.631	-.845
-.449	-1.561	-.708
-.368	-.411	-.660
-.979	-.433	-.483
-.519		
-.512		
-.525	-.473	-.308
-.500		
-.484	-.448	-.269
-.481	-.414	-.301
-.493		
-.429		
-.388		

ALPHA =10.01 DEG
EFF. VEL. RATIO = .199

Y/YMAX =	CP		
X/L	0.0	0.5	0.8
.150	.079		
.219	.082		
.286	.080		
.331	.131		
.353	.318	.055	-.253
.375	.370		
.397		-3.756	-1.259
.420	-1.579	-1.633	-2.993
.442	-1.142	-.563	-1.139
.464	-1.162	-.486	-.651
.487	-.547	-.392	-.455
.509	-.624	-.436	-.380
.531	-.690	-.469	-.332
.554	-.600		
.576	-.517		
.598	-.458	-.476	-.273
.620	-.484		
.643	-.465	-.416	-.248
.665	-.569	-.364	-.239
.687	-.442		
.710	-.392		
.732	-.401		

ALPHA = 9.75 DEG
EFF. VEL. RATIO = .206

	CP	
0.0	0.5	0.8
.088		
.090		
.074		
.110		
.265	.007	-.256
.299		
	-5.468	-1.645
-2.388	-2.277	-1.074
-1.680	-.684	-1.012
-2.148	-.550	-.763
-.324	-.517	-.465
-1.132	-.362	-.345
-.515	-.254	-.280
-.430		
-.504		
-.451	-.381	-.188
-.388		
-.409	-.268	-.216
-.282	-.217	-.164
-.305		
-.299		
-.262		

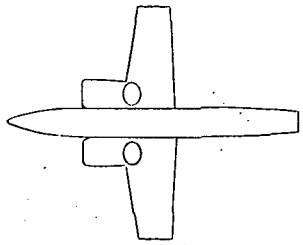
ALPHA =10.01 DEG
EFF. VEL. RATIO = .316

	CP	
0.0	0.5	0.8
.091		
.116		
.144		
.247		
.438	.195	-.149
.483		
	-4.495	-1.281
-1.059	-2.529	-2.428
-1.257	-.731	-.914
-.833	-.430	-.652
-.380	-.380	-.429
-.775	-.276	-.358
-.244	-.178	-.260
-.225		
-.226		
-.120	-.130	-.106
-.172		
-.167	-.056	-.046
-.126	-.115	-.075
-.114		
-.120		
-.117		

TABLE 40 (A)

PRESSURE PROFILE ON FUSELAGE

FORWARD NOZZLES DEFLECTED 45 DEGREES



ALPHA = .01 DEG
EFF. VEL. RATIO = .099

	CP		
Y/YMAX =	0.0	0.5	0.8
X/L			
.130	.210		
.219	.259		
.286	.299		
.331	.306		
.353	.319	.299	.266
.375	.346		
.397		.411	.362
.420	.378	.427	.443
.442	.475	.491	.523
.464	.507	.459	.491
.487	.346	.298	.121
.509	.153	.009	-1.330
.531	-.170	-.653	-2.120
.554	-.104		
.576	.223		
.598	.438	.299	-.171
.620	.445		
.643	.461	.299	.210
.665	.445	.313	.299
.687	.428		
.710	.445		
.732	.432		

ALPHA = .01 DEG
EFF. VEL. RATIO = .148

	CP		
Y/YMAX =	0.0	0.5	0.8
X/L			
.130	.070		
.219	.092		
.286	.104		
.331	.130		
.353	.130	.142	.092
.375	.150		
.397		.172	.172
.420	.230	.230	.230
.442	.296	.266	.296
.464	.354	.164	.150
.487	.077	.157	-.274
.509	-.113	-.128	-.668
.531	-.076	-.215	-1.018
.554	-.057		
.576	.097		
.598	.217	.138	-.012
.620	.277		
.643	.272	.157	-.160
.665	.253	.114	-.096
.687	.247		
.710	.207		
.732	.189		

ALPHA = 0.00 DEG
EFF. VEL. RATIO = .197

	CP		
Y/YMAX =	0.0	0.5	0.8
X/L			
.130	.037		
.219	.055		
.286	.056		
.331	.065		
.353	.079	.055	-.002
.375	.113		
.397		.126	.130
.420	.142	.179	.158
.442	.224	.183	.216
.464	.265	.113	.134
.487	.020	.044	-.193
.509	-.185	-.156	-.818
.531	-.164	-.328	-1.156
.554	-.151		
.576	-.006		
.598	.129	.032	-.031
.620	.258		
.643	.242	.134	-.177
.665	.247	.140	-.245
.687	.233		
.710	.233		
.732	.222		

ALPHA = .01 DEG
EFF. VEL. RATIO = .295

	CP		
Y/YMAX =	0.0	0.5	0.8
X/L			
.130	.002		
.219	.022		
.286	-.017		
.331	-.016		
.353	.047	.003	-.053
.375	.058		
.397		.076	.111
.420	.111	.146	.148
.442	.172	.153	.195
.464	.172	.115	.043
.487	.049	.040	-.271
.509	-.205	-.065	-.621
.531	-.094	-.216	-.908
.554	-.132		
.576	-.089		
.598	.020	-.197	-.381
.620	.151		
.643	.226	.061	.059
.665	.195	.082	-.023
.687	.188		
.710	.146		
.732	.123		

ALPHA = .02 DEG
EFF. VEL. RATIO = .396

	CP		
Y/YMAX =	0.0	0.5	0.8
X/L			
.130	.002		
.219	.010		
.286	-.025		
.331	-.023		
.353	.032	-.025	-.072
.375	.027		
.397		.063	.083
.420	.078	.124	.125
.442	.160	.135	.135
.464	.133	.083	-.035
.487	.036	.016	-.267
.509	-.304	-.063	-.442
.531	-.050	-.173	-.585
.554	-.091		
.576	-.064		
.598	-.008	-.189	-.559
.620	.107		
.643	.175	.034	.118
.665	.198	.120	-.041
.687	.168		
.710	.137		
.732	.130		

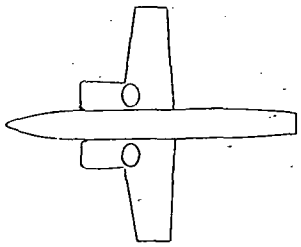
ALPHA = .03 DEG
EFF. VEL. RATIO = .521

	CP		
Y/YMAX =	0.0	0.5	0.8
X/L			
.130	-.001		
.219	.002		
.286	-.033		
.331	-.047		
.353	-.001	-.039	-.090
.375	.011		
.397		.037	.049
.420	.049	.087	.079
.442	.109	.108	.116
.464	.121	.041	-.049
.487	.016	.019	-.223
.509	-.203	-.088	-.319
.531	-.054	-.123	-.362
.554	-.078		
.576	-.054		
.598	-.002	-.053	-.203
.620	.092		
.643	.158	.089	.108
.665	.205	.111	-.058
.687	.162		
.710	.135		
.732	.113		

TABLE 40 (B)

PRESSURE PROFILE ON FUSELAGE

FORWARD NOZZLES DEFLECTED 45 DEGREES



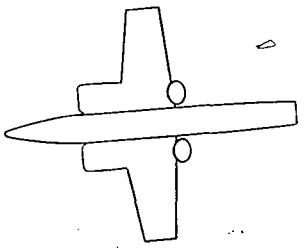
	ALPHA =10.04 DEG EFF. VEL. RATIO = .100			ALPHA =10.05 DEG EFF. VEL. RATIO = .148			ALPHA =10.04 DEG EFF. VEL. RATIO = .197		
		CP			CP			CP	
Y/YMAX =	0.0	0.5	0.8	0.0	0.5	0.8	0.0	0.5	0.8
X/L									
.130	.438			.270			.222		
.219	.520			.325			.258		
.286	.592			.353			.274		
.331	.595			.350			.242		
.353	.598	.572	.578	.380	.349	.349	.294	.270	.247
.375	.560			.372			.310		
.397		.592	.544		.379	.379		.319	.323
.420	.656	.640	.576	.386	.401	.444	.306	.364	.331
.442	.672	.672	.736	.451	.430	.480	.396	.388	.429
.464	.704	.640	.720	.487	.357	.379	.429	.319	.335
.487	.544	.624	.321	.263	.169	-.055	.237	.245	.106
.509	-.094	.177	-.971	.003	.053	-.468	.044	.065	-.501
.531	.050	-.540	-2.153	.046	-.106	-.722	.028	-.050	-.951
.554	-.024			.064			.020		
.576	.310			.141			.095		
.598	.530	.366	-.037	.264	.193	.015	.180	.017	.017
.620	.507			.279			.254		
.643	.490	.310	.123	.282	.096	-.164	.243	.116	-.186
.665	.457	.307	.300	.230	.046	-.164	.227	.111	-.182
.687	.464			.229			.207		
.710	.356			.223			.187		
.732	.444			.220			.156		

	ALPHA =10.00 DEG EFF. VEL. RATIO = .295			ALPHA =10.03 DEG EFF. VEL. RATIO = .391			ALPHA = 9.97 DEG EFF. VEL. RATIO = .517		
		CP			CP			CP	
Y/YMAX =	0.0	0.5	0.8	0.0	0.5	0.8	0.0	0.5	0.8
X/L									
.130	.174			.179			.214		
.219	.224			.229			.270		
.286	.224			.218			.241		
.331	.217			.194			.214		
.353	.261	.214	.204	.239	.208	.186	.260	.220	.211
.375	.233			.216			.218		
.397		.259	.283		.253	.277		.239	.266
.420	.277	.314	.316	.274	.287	.284	.282	.270	.266
.442	.372	.312	.374	.334	.301	.339	.329	.270	.306
.464	.398	.277	.252	.305	.262	.173	.337	.239	.198
.487	.215	.219	-.053	.202	.198	-.062	.172	.198	.030
.509	.122	.120	-.307	.150	.104	-.282	.113	.125	-.082
.531	.098	.018	-.594	.109	.007	-.396	.101	.044	-.174
.554	.057			.070			.064		
.576	.101			.073			.078		
.598	.148	-.076	-.286	.113	-.000	-.250	.117	.096	.066
.620	.206			.189			.174		
.643	.242	-.041	.084	.221	.050	.154	.216	.160	.150
.665	.215	-.002	.004	.240	.051	.107	.233	.119	.063
.687	.172			.223			.244		
.710	.110			.166			.210		
.732	.078			.108			.183		

TABLE 41 (A)

PRESSURE PROFILE ON FUSELAGE

REAR NOZZLES DEFLECTED 45 DEGREES



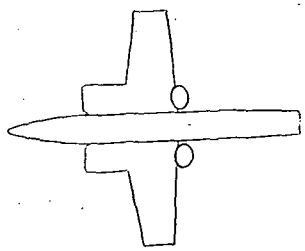
	ALPHA = 0.00 DEG EFF. VEL. RATIO = .097			ALPHA = .01 DEG EFF. VEL. RATIO = .146			ALPHA = .01 DEG EFF. VEL. RATIO = .195		
	CP			CP			CP		
Y/YMAX =	0.0	0.5	0.8	0.0	0.5	0.8	0.0	0.5	0.8
X/L									
.130	-.543			-.273			-.176		
.219	-.526			-.267			-.161		
.286	-.486			-.270			-.210		
.331	-.509			-.316			-.232		
.353	-.519	-.506	-.566	-.297	-.315	-.380	-.224	-.246	-.299
.375	-.554			-.307			-.224		
.397		-.603	-.620		-.329	-.337		-.236	-.273
.420	-.587	-.587	-.554	-.344	-.286	-.300	-.277	-.187	-.183
.442	-.587	-.538	-.603	-.315	-.264	-.322	-.216	-.130	-.199
.464	-.571	-.603	-.603	-.307	-.307	-.293	-.204	-.167	-.183
.487	-.603	-.603	-.587	-.264	-.242	-.271	-.134	-.134	-.142
.509	-.538	-.620	-.587	-.257	-.249	-.235	-.146	-.101	-.101
.531	-.636	-.636	-.620	-.271	-.249	-.235	-.097	-.142	-.077
.554	-.680			-.292			-.090		
.576	-.737			-.321			-.088		
.598	-.864	-.894	-1.035	-.427	-.423	-.524	-.101	-.096	-.165
.620	-1.005			-.557			-.165		
.643	-1.149	-1.324	-1.777	-.762	-.960	-1.721	-.285	-.306	-.777
.665	-1.129	-1.227	-.941	-.900	-1.145	-1.494	-.423	-.508	-1.061
.687	-1.009			-.909			-.557		
.710	-.720			-.732			-.599		
.732	-.552			-.517			-.544		

	ALPHA = -.01 DEG EFF. VEL. RATIO = .292			ALPHA = .01 DEG EFF. VEL. RATIO = .391			ALPHA = .00 DEG EFF. VEL. RATIO = .510		
	CP			CP			CP		
Y/YMAX =	0.0	0.5	0.8	0.0	0.5	0.8	0.0	0.5	0.8
X/L									
.130	-.107			-.095			-.082		
.219	-.097			-.070			-.074		
.286	-.174			-.144			-.168		
.331	-.207			-.201			-.206		
.353	-.172	-.168	-.279	-.155	-.179	-.293	-.166	-.221	-.289
.375	-.181			-.181			-.181		
.397		-.187	-.229		-.172	-.208		-.176	-.210
.420	-.230	-.139	-.130	-.201	-.112	-.113	-.207	-.127	-.122
.442	-.172	-.070	-.138	-.155	-.057	-.117	-.154	-.066	-.129
.464	-.136	-.096	-.119	-.124	-.074	-.091	-.117	-.079	-.096
.487	-.041	-.041	-.052	-.028	-.037	-.037	-.045	-.034	-.037
.509	-.032	.008	.008	-.026	.013	.020	-.050	-.000	.023
.531	.026	.043	.048	.040	.047	.071	.018	.032	.057
.554	.055			.074			.054		
.576	.092			.116			.092		
.598	.123	.156	.188	.147	.180	.227	.116	.141	.189
.620	.134			.165			.132		
.643	.104	.134	.211	.147	.128	.167	.115	.100	.074
.665	.049	.039	-.276	.103	.065	-.062	.087	.049	-.003
.687	-.047			.039			.049		
.710	-.152			-.032			-.005		
.732	-.236			-.081			-.039		

TABLE 41 (B)

PRESSURE PROFILE ON FUSELAGE

REAR NOZZLES DEFLECTED 45 DEGREES



ALPHA =10.00 DEG
EFF. VEL. RATIO = .098

	CP		
Y/YMAX =	0.0	0.5	0.8
X/L			
.130	-.448		
.219	-.391		
.286	-.341		
.331	-.401		
.353	-.401	-.435	-.442
.375	-.444		
.397		-.526	-.558
.420	-.558	-.493	-.493
.442	-.558	-.477	-.591
.464	-.575	-.558	-.624
.487	-.558	-.542	-.591
.509	-.607	-.542	-.624
.531	-.607	-.624	-.640
.554	-.649		
.576	-.720		
.598	-.834	-.864	-1.032
.620	-.985		
.643	-1.183	-1.414	-1.974
.665	-1.277	-1.411	-1.277
.687	-1.129		
.710	-.928		
.732	-.713		

ALPHA =10.02 DEG
EFF. VEL. RATIO = .146

	CP		
	0.0	0.5	0.8
	-.169		
	-.111		
	-.133		
	-.163		
	-.179	-.200	-.215
	-.201		
		-.245	-.288
	-.288	-.223	-.223
	-.266	-.187	-.283
	-.332	-.237	-.317
	-.201	-.201	-.295
	-.259	-.201	-.266
	-.223	-.208	-.230
	-.222		
	-.269		
	-.351	-.339	-.482
	-.486		
	-.690	-.836	-1.691
	-.881	-1.159	-1.812
	-.994		
	-.858		
	-.710		

ALPHA =10.01 DEG
EFF. VEL. RATIO = .195

	CP		
	0.0	0.5	0.8
	-.053		
	-.022		
	-.031		
	-.092		
	-.071	-.107	-.130
	-.094		
		-.139	-.184
	-.196	-.110	-.106
	-.172	-.049	-.176
	-.151	-.098	-.155
	-.053	-.053	-.139
	-.123	-.053	-.082
	-.032	-.036	-.053
	-.019		
	-.008		
	-.021	-.012	-.074
	-.066		
	-.143	-.145	-.364
	-.287	-.304	-.779
	-.488		
	-.581		
	-.593		

ALPHA =10.00 DEG
EFF. VEL. RATIO = .292

	CP		
Y/YMAX =	0.0	0.5	0.8
X/L			
.130	.041		
.219	.075		
.286	.034		
.331	-.029		
.353	.010	-.040	-.060
.375	-.030		
.397		-.048	-.101
.420	-.097	-.015	-.023
.442	-.081	.034	-.065
.464	-.054	.005	-.059
.487	.043	.052	-.010
.509	.001	.076	.048
.531	.094	.096	.105
.554	.122		
.576	.157		
.598	.178	.215	.241
.620	.185		
.643	.145	.178	.256
.665	.080	.070	-.188
.687	-.026		
.710	-.134		
.732	-.232		

ALPHA =10.05 DEG
EFF. VEL. RATIO = .390

	CP		
	0.0	0.5	0.8
	.060		
	.097		
	.047		
	-.011		
	.020	-.027	-.066
	-.028		
		-.042	-.090
	-.083	-.016	-.026
	-.050	.049	-.055
	-.050	.008	-.041
	.044	.057	.021
	.010	.082	.072
	.102	.111	.120
	.131		
	.169		
	.189	.230	.287
	.210		
	.186	.186	.214
	.146	.124	.017
	.074		
	-.004		
	-.070		

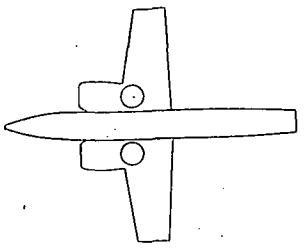
ALPHA =10.01 DEG
EFF. VEL. RATIO = .515

	CP		
	0.0	0.5	0.8
	.068		
	.089		
	.043		
	-.013		
	.030	-.018	-.059
	-.024		
		-.050	-.089
	-.081	-.013	-.025
	-.057	.043	-.050
	-.045	.001	-.043
	.040	.044	.008
	.014	.072	.073
	.089	.096	.120
	.096		
	.128		
	.150	.178	.250
	.161		
	.143	.140	.151
	.114	.085	.030
	.082		
	.037		
	.008		

TABLE 42 (A)

PRESSURE PROFILE ON FUSELAGE

FORWARD NOZZLES DEFLECTED 90 DEGREES



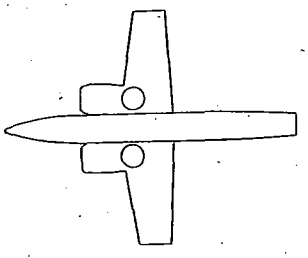
	ALPHA = -.02 DEG EFF. VEL. RATIO = .101			ALPHA = -.04 DEG EFF. VEL. RATIO = .151			ALPHA = -.04 DEG EFF. VEL. RATIO = .202		
	CP			CP			CP		
Y/YMAX =	0.0	0.5	0.8	0.0	0.5	0.8	0.0	0.5	0.8
X/L									
.130	.264			.178			.161		
.219	.250			.235			.246		
.286	.230			.250			.266		
.331	.037			.166			.270		
.353	-.059	-.086	.004	.102	.102	.142	.254	.238	.264
.375	-1.132			.339			.278		
.397		.181	-1.488		-.359	-.926		.144	.421
.420	-1.262	-.062	-1.780	-.156	-1.261	-1.784	.315	-.705	-.232
.442	-2.266	-3.109	-4.033	-1.901	-2.031	-4.045	-.656	-1.194	-3.401
.464	-4.471	-2.574	-2.753	-5.855	-2.787	-2.947	-2.614	-3.116	-4.454
.487	-1.278	-.581	-6.059	-1.850	-1.379	-5.783	-2.104	-1.977	-2.430
.509	-.403	-.970	-6.059	-4.445	-.367	-5.775	-1.170	-2.544	-4.751
.531	-3.871	-.030	-2.445	-.679	-1.377	-2.678	-.729	-2.259	-3.009
.554	-.598			-.728			-.830		
.576	-.785			-.477			-.543		
.598	-.275	-1.227	-1.950	-.238	-.724	-1.203	-.251	-.211	-.773
.620	-.349			-.262			-.053		
.643	-.036	-.182	-.545	-.026	-.047	-.904	.017	-.196	-.955
.665	.177	.097	-.498	.090	.071	-.373	.030	-.062	-.578
.687	.317			.148			.119		
.710	.397			.196			.134		
.732	.427			.236			.162		

	ALPHA = -.01 DEG EFF. VEL. RATIO = .303			ALPHA = -.02 DEG EFF. VEL. RATIO = .405			ALPHA = -.09 DEG EFF. VEL. RATIO = .529		
	CP			CP			CP		
Y/YMAX =	0.0	0.5	0.8	0.0	0.5	0.8	0.0	0.5	0.8
X/L									
.130	.124			.100			.088		
.219	.261			.220			.196		
.286	.321			.292			.243		
.331	.385			.360			.315		
.353	.436	.430	.412	.432	.412	.409	.401	.384	.367
.375	.367			.396			.413		
.397		.491	.318		.404	.446		.447	.504
.420	.471	.228	.338	.464	.315	.447	.556	.404	.506
.442	.383	-.025	-.271	.409	.148	.030	.473	.225	-.239
.464	-.021	-.962	-2.007	-.002	-.539	-1.468	-.134	-.370	-.997
.487	-1.229	-1.247	-1.721	-.678	-.678	-1.313	-.501	-.474	-1.188
.509	-1.679	-1.338	-1.692	-.899	-1.044	-1.238	-.698	-.925	-1.088
.531	-1.265	-1.497	-1.905	-1.037	-1.235	-1.191	-.806	-.919	-.940
.554	-1.128			-.877			-.798		
.576	-.877			-.791			-.673		
.598	-.623	-.675	-.866	-.634	-.751	-.810	-.538	-.726	-.776
.620	-.379			-.491			-.417		
.643	-.222	-.234	-.736	-.422	-.444	-.511	-.353	-.439	-.420
.665	-.112	-.157	-.944	-.270	-.239	-.451	-.239	-.260	-.326
.687	-.081			-.171			-.184		
.710	-.042			-.100			-.107		
.732	-.009			-.047			-.031		

TABLE 42 (B)

PRESSURE PROFILE ON FUSELAGE

FORWARD NOZZLES DEFLECTED 90 DEGREES



ALPHA =10.00 DEG
EFF. VEL. RATIO = .101

	CP		
Y/YMAX =	0.0	0.5	0.8
X/L			
.130	.583		
.219	.614		
.286	.563		
.331	.362		
.353	.221	.224	.365
.375	-.347		
.397		-.674	-.968
.420	-.249	-1.573	-3.306
.442	-3.535	-3.044	-3.551
.464	-3.878	-3.061	-5.186
.487	-1.622	-2.178	2.706
.509	-3.829	-2.260	-5.529
.531	-3.142	-.690	-3.110
.554	-.222		
.576	-.306		
.598	-.145	-.491	-.770
.620	-.051		
.643	.180	.164	-.300
.665	.415	.258	.385
.687	.620		
.710	.634		
.732	.634		

ALPHA =10.01 DEG
EFF. VEL. RATIO = .151

	CP		
	0.0	0.5	0.8
	.406		
	.473		
	.470		
	.374		
	.289	.296	.370
	.031		
		-.013	-.494
	-.042	-1.705	-1.464
	-1.026	-2.645	-4.439
	-2.208	-3.455	-3.185
	-2.682	-3.076	-6.445
	-1.821	-2.798	-5.708
	-.363	-1.726	-2.346
	-.514		
	-.207		
	-.132	-.267	-1.321
	-.060		
	.081	.078	-.341
	.156	.076	-.252
	.277		
	.293		
	.322		

ALPHA =10.04 DEG
EFF. VEL. RATIO = .203

	CP		
	0.0	0.5	0.8
	.350		
	.438		
	.487		
	.478		
	.472	.446	.477
	.430		
		.520	.479
	.255	-.010	.002
	.304	-.866	-3.271
	-3.059	-2.439	-4.652
	-2.093	-2.252	-2.121
	-1.632	-2.582	-4.363
	-1.269	-1.665	-3.483
	-.786		
	-.447		
	-.180	-.314	-1.327
	-.026		
	-.092	-.165	-1.254
	-.051	-.096	-.836
	.005		
	.066		
	.123		

ALPHA =10.18 DEG
EFF. VEL. RATIO = .302

	CP		
Y/YMAX =	0.0	0.5	0.8
X/L			
.130	.312		
.219	.440		
.286	.532		
.331	.571		
.353	.616	.600	.610
.375	.764		
.397		.668	.610
.420	.683	.542	.569
.442	.513	.225	.027
.464	.056	-.541	-1.771
.487	-1.003	-.959	-1.749
.509	-1.201	-1.166	-1.588
.531	-1.258	-1.436	-1.581
.554	-1.227		
.576	-1.062		
.598	-.799	-.958	-1.142
.620	-.587		
.643	-.401	-.433	-.469
.665	-.334	-.294	-.666
.687	-.190		
.710	-.137		
.732	-.088		

ALPHA =10.23 DEG
EFF. VEL. RATIO = .409

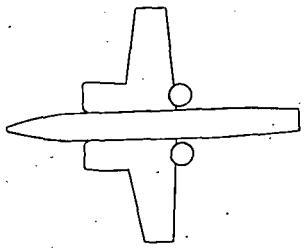
	CP		
	0.0	0.5	0.8
	.275		
	.401		
	.480		
	.538		
	.589	.572	.590
	.561		
		.572	.715
	.742	.595	.503
	.694	.349	.208
	.170	-.249	-1.121
	-.529	-.610	-1.209
	-.879	-.962	-1.282
	-.937	-1.122	-1.271
	-.987		
	-.882		
	-.712	-.841	-1.001
	-.510		
	-.404	-.494	-.463
	-.338	-.383	-.365
	-.283		
	-.215		
	-.139		

ALPHA =10.15 DEG
EFF. VEL. RATIO = .536

	CP		
	0.0	0.5	0.8
	.263		
	.391		
	.451		
	.506		
	.558	.554	.561
	.556		
		.602	.725
	.697	.570	.562
	.632	.347	.199
	.195	-.187	-.819
	-.308	-.361	-.915
	-.610	-.797	-1.026
	-.703	-.999	-1.021
	-.816		
	-.769		
	-.645	-.741	-.834
	-.489		
	-.392	-.502	-.445
	-.345	-.356	-.279
	-.208		
	-.168		
	-.091		

TABLE 43 (A)
PRESSURE PROFILE ON FUSELAGE

REAR NOZZLES DEFLECTED 90 DEGREES



ALPHA = .00 DEG
EFF. VEL. RATIO = .098

	CP		
Y/YMAX =	0.0	0.5	0.8
X/L			
.130	-.356		
.219	-.359		
.286	-.359		
.331	-.423		
.353	-.423	-.463	-.490
.375	-.522		
.397		-.522	-.587
.420	-.554	-.571	-.636
.442	-.538	-.489	-.587
.464	-.652	-.652	-.669
.487	-.750	-.718	-.685
.509	-.930	-.865	-.734
.531	-1.257	-1.159	-.946
.554	-1.761		
.576	-2.459		
.598	-2.711	-2.741	-3.275
.620	-1.758		
.643	-1.235	-1.956	-1.990
.665	-1.557	-1.574	-1.423
.687	-1.003		
.710	-.839		
.732	-.731		

ALPHA = .02 DEG
EFF. VEL. RATIO = .147

	CP		
Y/YMAX =	0.0	0.5	0.8
X/L			
.130	-.151		
.219	-.123		
.286	-.121		
.331	-.169		
.353	-.150	-.195	-.241
.375	-.189		
.397		-.218	-.262
.420	-.233	-.189	-.204
.442	-.262	-.160	-.277
.464	-.269	-.255	-.277
.487	-.284	-.335	-.291
.509	-.401	-.364	-.342
.531	-.656	-.576	-.474
.554	-1.181		
.576	-2.011		
.598	-2.866	-3.670	-4.397
.620	-2.702		
.643	-1.368	-1.462	-2.396
.665	-1.109	-1.089	-1.541
.687	-.977		
.710	-.778		
.732	-.652		

ALPHA = 0.00 DEG
EFF. VEL. RATIO = .196

	CP		
Y/YMAX =	0.0	0.5	0.8
X/L			
.130	-.082		
.219	-.045		
.286	-.061		
.331	-.095		
.353	-.049	-.095	-.146
.375	-.080		
.397		-.109	-.158
.420	-.154	-.060	-.039
.442	-.117	.010	-.109
.464	-.097	-.043	-.113
.487	-.039	-.039	-.084
.509	-.080	-.080	-.084
.531	-.240	-.199	-.154
.554	-.577		
.576	-1.227		
.598	-2.098	-2.993	-4.544
.620	-2.563		
.643	-2.019	-2.082	-2.245
.665	-1.030	-.872	-1.471
.687	-.679		
.710	-.642		
.732	-.587		

ALPHA = .00 DEG
EFF. VEL. RATIO = .293

	CP		
Y/YMAX =	0.0	0.5	0.8
X/L			
.130	-.038		
.219	-.009		
.286	-.040		
.331	-.057		
.353	-.007	-.061	-.118
.375	-.016		
.397		-.022	-.060
.420	-.066	.044	.051
.442	-.009	.121	.020
.464	.049	.117	.064
.487	.174	.156	.113
.509	.179	.190	.187
.531	.172	.203	.214
.554	.063		
.576	-.170		
.598	-.605	-.694	-1.909
.620	-1.068		
.643	-1.314	-1.800	-2.626
.665	-1.213	-1.748	-1.769
.687	-.916		
.710	-.675		
.732	-.560		

ALPHA = 0.00 DEG
EFF. VEL. RATIO = .395

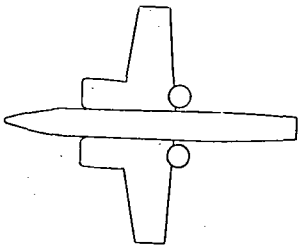
	CP		
Y/YMAX =	0.0	0.5	0.8
X/L			
.130	-.027		
.219	-.005		
.286	-.043		
.331	-.075		
.353	-.010	-.079	-.131
.375	-.015		
.397		-.011	-.057
.420	-.054	.054	.060
.442	.011	.137	.046
.464	.062	.143	.105
.487	.214	.210	.185
.509	.255	.266	.266
.531	.284	.302	.323
.554	.247		
.576	.139		
.598	-.119	-.117	-.944
.620	-.499		
.643	-.849	-1.171	-1.846
.665	-.948	-1.236	-1.621
.687	-.852		
.710	-.687		
.732	-.491		

ALPHA = -.02 DEG
EFF. VEL. RATIO = .516

	CP		
Y/YMAX =	0.0	0.5	0.8
X/L			
.130	-.039		
.219	-.017		
.286	-.058		
.331	-.092		
.353	-.010	-.090	-.150
.375	-.037		
.397		-.022	-.071
.420	-.067	.038	.036
.442	.006	.128	.044
.464	.059	.132	.100
.487	.211	.205	.187
.509	.219	.273	.285
.531	.305	.324	.357
.554	.306		
.576	.266		
.598	.078	.192	-.267
.620	-.226		
.643	-.574	-.917	-1.515
.665	-.727	-.951	-1.317
.687	-.714		
.710	-.637		
.732	-.506		

TABLE 43 (B)
PRESSURE PROFILE ON FUSELAGE

REAR NOZZLES DEFLECTED 90 DEGREES



ALPHA =10.00 DEG
EFF. VEL. RATIO = .098

	CP		
Y/YMAX =	0.0	0.5	0.8
X/L			
.130	.126		
.219	.109		
.286	.129		
.331	.058		
.353	.038	.014	-.002
.375	-.025		
.397		-.124	-.157
.420	-.206	-.157	-.141
.442	-.256	-.157	-.322
.464	-.305	-.272	-.354
.487	-.387	-.404	-.404
.509	-.470	-.535	-.453
.531	-.963	-.848	-.700
.554	-1.543		
.576	-2.340		
.598	-2.793	-3.178	-3.682
.620	-2.306		
.643	-.833	-1.658	-1.904
.665	-.458	-.745	-.965
.687	-.455		
.710	-.357		
.732	-.289		

ALPHA = 9.99 DEG
EFF. VEL. RATIO = .148

	CP		
	0.0	0.5	0.8
	.131		
	.179		
	.175		
	.100		
	.124	.093	.051
	.064		
		.007	-.080
	-.080	.007	-.008
	-.095	.021	-.138
	-.131	-.073	-.204
	-.117	-.109	-.233
	-.225	-.218	-.247
	-.522	-.435	-.385
	-1.050		
	-1.982		
	-3.057	-4.093	-5.346
	-3.237		
	-2.010	-1.604	-2.122
	-.851	-.912	-1.093
	-.662		
	-.518		
	-.412		

ALPHA = 9.99 DEG
EFF. VEL. RATIO = .196

	CP		
	0.0	0.5	0.8
	.131		
	.184		
	.192		
	.151		
	.157	.121	.094
	.125		
		.097	.006
	-.010	.097	.109
	.023	.170	-.010
	.015	.097	-.026
	.101	.137	-.022
	.039	.039	-.026
	-.108	-.075	-.084
	-.437		
	-1.067		
	-1.950	-2.863	-4.569
	-2.502		
	-2.069	-2.395	-2.319
	-1.088	-.807	-1.307
	-.613		
	-.550		
	-.461		

ALPHA = 9.98 DEG
EFF. VEL. RATIO = .294

	CP		
Y/YMAX =	0.0	0.5	0.8
X/L			
.130	.139		
.219	.200		
.286	.185		
.331	.151		
.353	.202	.149	.119
.375	.163		
.397		.152	.092
.420	.083	.194	.191
.442	.112	.266	.119
.464	.105	.244	.147
.487	.291	.299	.192
.509	.170	.306	.242
.531	.268	.295	.282
.554	.171		
.576	-.083		
.598	-.469	-.665	-2.005
.620	-.945		
.643	-1.310	-2.004	-3.042
.665	-1.314	-1.892	-1.561
.687	-.976		
.710	-.561		
.732	-.384		

ALPHA = 9.99 DEG
EFF. VEL. RATIO = .395

	CP		
	0.0	0.5	0.8
	.145		
	.186		
	.184		
	.141		
	.190	.138	.116
	.165		
		.158	.101
	.098	.198	.194
	.136	.279	.151
	.177	.262	.208
	.335	.333	.242
	.316	.356	.323
	.375	.396	.383
	.325		
	.220		
	-.040	-.098	-.923
	-.456		
	-.847	-1.291	-2.101
	-1.004	-1.408	-1.759
	-.941		
	-.776		
	-.576		

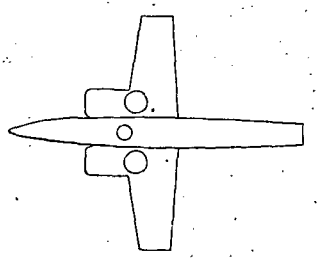
ALPHA =10.00 DEG
EFF. VEL. RATIO = .517

	CP		
	0.0	0.5	0.8
	.143		
	.180		
	.165		
	.123		
	.175	.124	.083
	.148		
		.137	.089
	.095	.182	.183
	.135	.261	.154
	.153	.252	.200
	.315	.312	.272
	.325	.362	.358
	.391	.418	.431
	.394		
	.328		
	.146	.221	-.265
	-.182		
	-.558	-.951	-1.509
	-.749	-1.056	-1.428
	-.767		
	-.695		
	-.571		

TABLE 44 (A)

PRESSURE PROFILE ON FUSELAGE

FORWARD NOZZLES DEFLECTED 90 DEGREES WITH LIFT JET



ALPHA = -.01 DEG
EFF. VEL. RATIO = .094

	CP		
Y/YMAX =	0.0	0.5	0.8
X/L			
.130	-.169		
.219	-.166		
.286	-.156		
.331	-.250		
.353	-.196	-.408	-.448
.375	3.178		
.397		-7.841	-.718
.420	-1.946	.068	-5.680
.442	-.866	-.375	-2.421
.464	.068	-2.028	-.391
.487	-.473	-.555	-2.716
.509	-1.013	-.669	-.293
.531	-1.292	-.882	-2.798
.554	-2.122		
.576	-1.665		
.598	-.811	-.845	-1.214
.620	-.219		
.643	.130	.026	-.028
.665	.107	.171	.076
.687	.288		
.710	.352		
.732	.332		

ALPHA = .01 DEG
EFF. VEL. RATIO = .142

	CP		
	0.0	0.5	0.8
	.040		
	.093		
	.083		
	.087		
	.157	-.024	-.129
	.363		
		-8.630	-.166
	-2.429	-7.752	-8.543
	-4.670	-3.393	-7.281
	-9.515	-4.532	-4.626
	-1.675	-1.174	-5.315
	-2.240	-3.814	-3.872
	-.637	-1.341	-4.220
	-2.119		
	-1.836		
	-1.106	-1.283	-2.363
	-.503		
	-.337	-.143	-.563
	-.142	-.078	-.332
	.030		
	.042		
	.161		

ALPHA = .00 DEG
EFF. VEL. RATIO = .189

	CP		
	0.0	0.5	0.8
	.073		
	.164		
	.220		
	.233		
	.351	.196	.099
	.137		
		-4.668	.006
	-1.435	-.006	-6.681
	-1.500	-2.349	-5.497
	-2.051	-2.280	-5.280
	-2.345	-1.904	-2.835
	-1.827	-2.398	-3.105
	-1.325	-2.509	-1.651
	-1.621		
	-1.414		
	-1.038	-1.278	-2.524
	-.595		
	-.295	-.363	-1.107
	-.115	-.104	-.817
	-.106		
	.002		
	-.001		

ALPHA = .00 DEG
EFF. VEL. RATIO = .282

	CP		
Y/YMAX =	0.0	0.5	0.8
X/L			
.130	.079		
.219	.215		
.286	.283		
.331	.408		
.353	.501	.378	.265
.375	.383		
.397		-1.933	-.208
.420	.092	-2.461	-1.286
.442	-1.251	-.736	-1.695
.464	-.445	-1.439	-2.439
.487	-1.586	-1.370	-2.052
.509	-1.531	-1.845	-2.205
.531	-1.385	-1.034	-1.646
.554	-1.329		
.576	-1.174		
.598	-1.032	-.928	-1.155
.620	-.799		
.643	-.632	-.745	-1.620
.665	-.507	-.682	-1.633
.687	-.348		
.710	-.303		
.732	-.230		

ALPHA = -.01 DEG
EFF. VEL. RATIO = .386

	CP		
	0.0	0.5	0.8
	.084		
	.210		
	.315		
	.432		
	.556	.428	.338
	.532		
		-1.149	.019
	-.131	-1.243	-1.511
	-.488	-.151	-1.145
	-.451	-.497	-1.803
	-1.008	-.956	-1.409
	-1.174	-1.222	-1.508
	-1.131	-1.755	-1.667
	-1.191		
	-1.076		
	-.904	-.565	-.803
	-.702		
	-.573	-.589	-.923
	-.457	-.584	-1.093
	-.390		
	-.334		
	-.271		

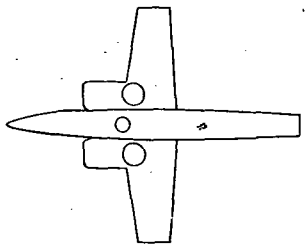
ALPHA = .00 DEG
EFF. VEL. RATIO = .523

	CP		
	0.0	0.5	0.8
	.075		
	.181		
	.282		
	.436		
	.568	.467	.385
	.740		
		-.873	.079
	.075	-.493	-1.010
	-.553	-.289	-1.648
	-1.271	-1.029	-1.194
	-.657	-1.101	-1.097
	-.937	-1.177	-1.302
	-.717	-.781	-1.285
	-1.035		
	-.901		
	-.778	-.886	-.842
	-.625		
	-.502	-.509	-.553
	-.370	-.367	-.453
	-.318		
	-.252		
	-.217		

TABLE 44 (B)

PRESSURE PROFILE ON FUSELAGE

FORWARD NOZZLES DEFLECTED 90 DEGREES WITH LIFT JET



ALPHA =10.02 DEG
EFF. VEL. RATIO = .095

	CP		
Y/YMAX =	0.0	0.5	0.8
X/L			
.130	.038		
.219	.038		
.286	.008		
.331	-.137		
.353	-.130	-.271	-.359
.375	-.381		
.397		-6.373	-2.067
.420	-3.213	-3.131	-1.265
.442	-5.506	-2.575	-4.441
.464	-1.707	-2.509	-2.067
.487	-.233	-.430	-2.525
.509	-3.393	-2.804	-4.867
.531	-1.740	-1.609	-3.901
.554	-2.200		
.576	-1.649		
.598	-.876	-.785	-1.414
.620	-.331		
.643	-.086	-.042	-.046
.665	-.032	.092	-.012
.687	.220		
.710	.200		
.732	.203		

ALPHA =10.01 DEG
EFF. VEL. RATIO = .142

	CP		
	0.0	0.5	0.8
	.116		
	.179		
	.200		
	.127		
	.185	-.002	-.106
	.059		
		-5.509	-2.005
	-.614	-7.146	-9.753
	-4.474	-2.540	-2.635
	-2.207	-1.194	-3.873
	-1.512	-2.005	-4.278
	-1.556	-1.925	-3.453
	-2.721	-3.504	-2.917
	-2.279		
	-2.049		
	-1.053	-1.472	-2.596
	-.604		
	-.291	-.266	-.834
	-.134	-.064	-.367
	-.134		
	-.012		
	.019		

ALPHA =10.03 DEG
EFF. VEL. RATIO = .189

	CP		
	0.0	0.5	0.8
	.184		
	.281		
	.332		
	.326		
	.404	.247	.156
	.507		
		-3.989	-1.385
	-1.099	-6.290	-4.099
	-2.481	-2.109	-1.512
	-4.451	-2.791	-3.494
	-2.211	-1.851	-4.018
	-1.953	-3.637	-3.609
	-2.951	-1.532	-2.325
	-1.303		
	-1.428		
	-1.190	-1.315	-2.653
	-.725		
	-.458	-.337	-1.641
	-.263	-.282	-1.142
	-.169		
	-.082		
	.006		

ALPHA =10.00 DEG
EFF. VEL. RATIO = .282

	CP		
Y/YMAX =	0.0	0.5	0.8
X/L			
.130	.236		
.219	.352		
.286	.452		
.331	.518		
.353	.601	.484	.396
.375	.345		
.397		-.444	-.320
.420	.163	-1.950	-2.076
.442	-.651	-.645	-2.182
.464	.280	-.750	-1.844
.487	-1.402	-1.559	-1.974
.509	-1.561	-1.682	-1.702
.531	-.965	-2.638	-2.139
.554	-1.464		
.576	-1.403		
.598	-1.212	-1.262	-1.362
.620	-1.006		
.643	-.923	-.877	-1.636
.665	-.824	-.765	-1.753
.687	-.648		
.710	-.518		
.732	-.447		

ALPHA =10.02 DEG
EFF. VEL. RATIO = .385

	CP		
	0.0	0.5	0.8
	.232		
	.360		
	.461		
	.548		
	.639	.525	.449
	.552		
		-.546	.228
	-.023	-1.402	-.126
	-.301	-.105	-1.195
	-1.200	-.821	-1.632
	-.886	-.956	-1.676
	-1.287	-1.404	-1.487
	-1.583	-1.159	-1.568
	-1.182		
	-1.074		
	-.923	-.958	-1.021
	-.774		
	-.685	-.710	-.958
	-.624	-.654	-1.074
	-.603		
	-.511		
	-.458		

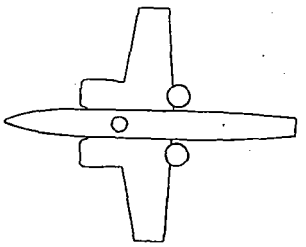
ALPHA =10.03 DEG
EFF. VEL. RATIO = .526

	CP		
	0.0	0.5	0.8
	.211		
	.343		
	.463		
	.562		
	.669	.582	.505
	.618		
		-.941	.156
	.131	-.653	-.907
	-.299	.163	-1.504
	-1.189	-.316	-1.269
	-.647	-.430	-1.182
	-.990	-1.188	-1.333
	-1.023	-1.369	-1.094
	-1.095		
	-.998		
	-.901	-.892	-.920
	-.722		
	-.515	-.455	-.533
	-.372	-.376	-.490
	-.311		
	-.289		
	-.287		

TABLE 45 (A)

PRESSURE PROFILE ON FUSELAGE

REAR NOZZLES DEFLECTED 90 DEGREES WITH LIFT JET



ALPHA = -.02 DEG
EFF. VEL. RATIO = .092

	CP		
Y/YMAX =	0.0	0.5	0.8
X/L			
.130	.053		
.219	.018		
.286	-.057		
.331	-.164		
.353	-.085	-.483	-.644
.375	-.193		
.397		-5.711	-1.715
.420	-4.373	-1.698	.158
.442	-.076	-.627	-.978
.464	-.026	-.276	-.962
.487	-.243	.142	-1.229
.509	-.377	-.326	-.611
.531	-.711	-.544	-.644
.554	-1.011		
.576	-1.471		
.598	-1.612	-1.691	-2.240
.620	-.856		
.643	-.668	-.702	-.887
.665	-.482	-.592	-.719
.687	-.379		
.710	-.208		
.732	-.173		

ALPHA = -.02 DEG
EFF. VEL. RATIO = .140

	CP		
	0.0	0.5	0.8
	.065		
	.047		
	-.015		
	-.065		
	.056	-.181	-.450
	.038		
		-4.622	-1.295
	-1.417	-1.784	-5.177
	-2.296	-.646	-1.071
	-.618	-.315	-.747
	-.452	-.474	-.517
	-.502	-.430	-.445
	-.661	-.560	-.495
	-1.089		
	-1.669		
	-2.203	-2.753	-3.424
	-1.849		
	-1.073	-1.184	-1.636
	-.856	-1.064	-1.107
	-.789		
	-.607		
	-.452		

ALPHA = -.02 DEG
EFF. VEL. RATIO = .185

	CP		
	0.0	0.5	0.8
	.040		
	.058		
	.013		
	-.021		
	.127	-.116	-.418
	.199		
		-4.983	-1.596
	-2.620	-1.419	-.424
	-.346	-.588	-.817
	-1.030	-.240	-.580
	-.371	-.563	-.477
	-.354	-.281	-.342
	-.506	-.346	-.313
	-.799		
	-1.424		
	-2.136	-2.955	-4.231
	-2.310		
	-1.514	-1.598	-2.007
	-1.021	-1.163	-1.176
	-.880		
	-.620		
	-.553		

ALPHA = -.00 DEG
EFF. VEL. RATIO = .278

	CP		
Y/YMAX =	0.0	0.5	0.8
X/L			
.130	.045		
.219	.088		
.286	.071		
.331	.108		
.353	.300	.040	-.217
.375	.376		
.397		-4.691	-1.513
.420	-1.330	-1.800	-1.575
.442	-.937	-.559	-.666
.464	-.738	-.240	-.474
.487	-.299	.083	-.281
.509	-.116	-.345	-.098
.531	-.081	-.145	-.034
.554	-.209		
.576	-.457		
.598	-1.020	-1.220	-2.959
.620	-1.514		
.643	-1.655	-2.156	-1.993
.665	-1.323	-1.394	-1.294
.687	-.865		
.710	-.677		
.732	-.514		

ALPHA = -.04 DEG
EFF. VEL. RATIO = .379

	CP		
	0.0	0.5	0.8
	.054		
	.107		
	.119		
	.225		
	.414	.190	-.049
	.517		
		-4.071	-1.204
	-1.115	-2.097	-2.057
	-1.278	-.618	-.793
	-.940	-.190	-.460
	.053	.062	-.173
	.074	.083	.071
	.163	.152	.180
	.177		
	.104		
	-.199	-.207	-1.390
	-.697		
	-1.103	-1.356	-1.735
	-1.121	-1.418	-1.458
	-.820		
	-.554		
	-.398		

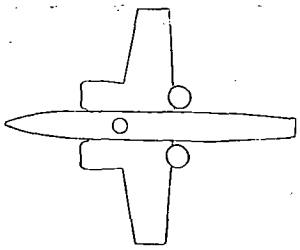
ALPHA = -.13 DEG
EFF. VEL. RATIO = .512

	CP		
	0.0	0.5	0.8
	.055		
	.112		
	.175		
	.324		
	.510	.337	.099
	.643		
		-.857	-.506
	-.461	-.732	-.677
	-.534	-.694	-.574
	-.538	-.643	-.473
	-.469	-.301	-.346
	-.256	-.262	-.173
	-.121	-.089	-.030
	.015		
	.103		
	-.028	.063	-.295
	-.274		
	-.781	-1.144	-1.510
	-.920	-1.178	-1.306
	-.798		
	-.594		
	-.399		

TABLE 45 (B)

PRESSURE PROFILE ON FUSELAGE

REAR NOZZLES DEFLECTED 90 DEGREES WITH LIFT JET



ALPHA = 9.97 DEG
EFF. VEL. RATIO = .093

	CP		
Y/YMAX =	0.0	0.5	0.8
X/L			
.130	.350		
.219	.302		
.286	.221		
.331	.086		
.353	.137	-.130	-.292
.375	.093		
.397		-3.923	-1.289
.420	.258	-.746	.258
.442	.406	.093	-.384
.464	-1.042	.110	-.318
.487	.077	.274	-.466
.509	-.071	.061	-.236
.531	-.417	-.400	-.318
.554	-.809		
.576	-1.268		
.598	-1.451	-1.721	-2.106
.620	-.897		
.643	-.501	-.511	-1.126
.665	-.349	-.353	-.687
.687	-.113		
.710	-.099		
.732	.060		

ALPHA = 9.99 DEG
EFF. VEL. RATIO = .139

	CP		
	0.0	0.5	0.8
	.209		
	.192		
	.137		
	.040		
	.136	-.140	-.296
	.110		
		-5.166	-1.467
	-.525	-1.773	.992
	-2.722	-.248	-1.022
	-1.846	-.175	-.657
	-.387	-.095	-.606
	-.452	-.438	-.482
	-.671	-.540	-.518
	-1.127		
	-1.795		
	-2.352	-2.930	-3.714
	-1.946		
	-.968	-1.175	-1.787
	-.842	-.995	-1.182
	-.670		
	-.514		
	-.433		

ALPHA = 9.99 DEG
EFF. VEL. RATIO = .186

	CP	
0.0	0.5	0.8
.194		
.213		
.150		
.103		
.212	-.031	-.254
.227		
	-4.430	-1.551
-2.677	-1.563	-.318
-1.121	-.527	-.871
-1.198	-.203	-.449
-.219	-.187	-.420
-.318	-.187	-.269
-.473	-.416	-.310
-.814		
-1.403		
-2.259	-3.024	-4.423
-2.425		
-1.581	-1.606	-2.190
-1.013	-1.048	-1.341
-.797		
-.659		
-.492		

ALPHA = 9.99 DEG
EFF. VEL. RATIO = .279

	CP		
Y/YMAX =	0.0	0.5	0.8
X/L			
.130	.200		
.219	.253		
.286	.250		
.331	.238		
.353	.387	.142	-.086
.375	.400		
.397		-4.432	-1.305
.420	-1.373	-1.595	-3.177
.442	-.897	-.525	-.652
.464	.028	-.273	-.344
.487	-.295	-.295	-.287
.509	-.255	-.408	-.087
.531	-.152	-.090	-.110
.554	-.256		
.576	-.629		
.598	-1.083	-1.374	-3.037
.620	-1.714		
.643	-1.777	-2.252	-2.115
.665	-1.386	-1.542	-1.384
.687	-.885		
.710	-.718		
.732	-.595		

ALPHA = 9.99 DEG
EFF. VEL. RATIO = .379

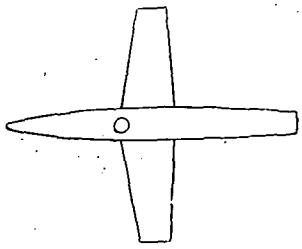
	CP	
0.0	0.5	0.8
.196		
.276		
.314		
.377		
.526	.317	.062
.533		
	-3.663	-.978
-1.008	-1.721	-1.377
-.898	-.667	-.671
-.743	-.336	-.426
-.070	-.188	-.133
.022	-.002	-.059
.129	.180	.121
.177		
.121		
-.155	-.242	-1.487
-.723		
-1.204	-1.486	-1.876
-1.247	-1.613	-1.539
-1.022		
-.646		
-.478		

ALPHA = 9.98 DEG
EFF. VEL. RATIO = .515

	CP	
0.0	0.5	0.8
.209		
.290		
.356		
.465		
.630	.457	.223
.651		
	-.651	-.395
-.393	-.474	-.491
-.351	-.477	-.454
-.417	-.457	-.420
-.326	-.590	-.317
-.202	-.215	-.180
-.108	-.135	-.095
-.049		
.005		
-.029	-.003	-.317
-.301		
-.687	-1.076	-1.283
-.889	-1.128	-1.200
-.855		
-.672		
-.496		

TABLE 46
PRESSURE PROFILE ON WING

LIFT JET ALONE



ALPHA = .00 DEG THRUST = 361 N (81 LBF)

X/C	ETA = .250 CP' X 100 UPPER	ETA = .250 CP' X 100 LOWER	ETA = .387 CP' X 100 UPPER	ETA = .387 CP' X 100 LOWER	ETA = .524 CP' X 100 UPPER	ETA = .524 CP' X 100 LOWER	ETA = .800 CP' X 100 UPPER	ETA = .800 CP' X 100 LOWER
.000	-.04		-.04		-.04		-.03	
.010	-.01	-.03	-1.83	-.04				
.015					-.02	-.03		
.025	-.013	-.031	-.009	-.029	-.013	-.031	-.013	-.107
.050	-.011	-.036	-.007	-.031	-.011	-.047	-.016	-.038
.100	-.016	-.038	-.002	-.034	-.013	-.045	-.009	-.054
.150	-.018	-.034	-.011	-.040	-.009	-.029	-.011	-.146
.250	-.020	-.036	-.016	-.030	-.096	-.050	-.016	-.035
.400	-.022	-.039	-.018	-.040	-.017	-.053	-.015	-.037
.550	-.024	-.035	-.022	-.039	-2.226	-.052	-.017	-.035
.700	-.027	-.034	-.021	-.045	-.023	-.039	-.016	-.030
.850	-.035	-.035	-.022	-.033	-.028	-.030	-.018	-.040
.950	-.035	-.038	-.025	-.034	-.033	-.033	-.024	-.020

ALPHA = 9.99 DEG THRUST = 353 N (79 LBF)

X/C	ETA = .250 CP' X 100 UPPER	ETA = .250 CP' X 100 LOWER	ETA = .387 CP' X 100 UPPER	ETA = .387 CP' X 100 LOWER	ETA = .524 CP' X 100 UPPER	ETA = .524 CP' X 100 LOWER	ETA = .800 CP' X 100 UPPER	ETA = .800 CP' X 100 LOWER
.000	-.03		-.03		-.02		-.02	
.010	-.02	-.01	-1.85	-.02				
.015					-.02	-.02		
.025	-.007	-.018	-.014	-.023	-.023	-.018	-.044	-.069
.050	-.014	-.016	-.009	-.023	-.014	-.021	-.014	-.041
.100	-.007	-.021	-.011	-.027		-.016	-.014	-.053
.150	-.016	-.016	-.009	-.018	-.002	-.007		-.030
.250	-.007	-.018	-.007	-.021	-.011	-.022	-.005	-.022
.400	-.008	-.036	-.010	-.012	-.008	-.017	-.011	-.021
.550	-.009	-.018	-.011	-.019	-3.042	-.016	-.011	-.016
.700	-.011	-.019	-.016	-.018	-.013	-.017	-.009	-.023
.850	-.007	-.014	-.013	-.023	-.012	-.019	-.008	-.022
.950	-.012	-.024	-.012	-.018	-.012	-.017	-.012	-.017

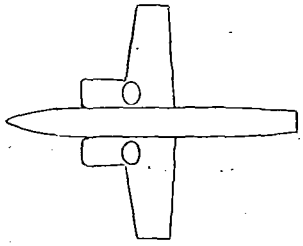


TABLE 47

PRESSURE PROFILE ON WING

FORWARD NOZZLES DEFLECTED 45 DEGREES

ALPHA = .02 DEG THRUST = 638 N (143 LBF)

	ETA = .250		ETA = .387		ETA = .524		ETA = .800	
X/C	CP' X 100		CP' X 100		CP' X 100		CP' X 100	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			-1.281		-1.247		-.485	
.010			-1.274	-1.112				
.015					-.047	-1.119		
.025			-.034	-1.132	-.027	-1.052	-.020	-.553
.050			-.034	-1.092	-.040	-.930	-.040	-.425
.100			-.061	-.964	-.061	-.721	-.057	-.310
.150			-.081	-.890	-.074	-.553	-.088	-.297
.250	-.115		-.088	-.954	-.088	-.484	-.108	-.268
.400	-.114		-.111	-1.026	-.105	-.441	-.114	-.242
.550	-.114	-.657	-.118	-.640	-.114	-.372	-.127	-.212
.700	-.124	-.333	-.134	-.438	-.121	-.307	-.127	-.189
.850	-.127	-.199	-.144	-.337	-.137	-.258	-.124	-.163
.950	-.140	-.163	-.176	-.281	-.160	-.225	-.127	-.154

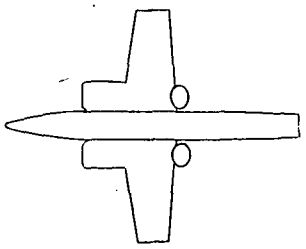
ALPHA = 9.98 DEG THRUST = 638 N (143 LBF)

	ETA = .250		ETA = .387		ETA = .524		ETA = .800	
X/C	CP' X 100		CP' X 100		CP' X 100		CP' X 100	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			-.992		-.600		-.054	
.010			-2.192	-.890				
.015					-.034	-.553		
.025			-.027	-.762	-.040	-.499	-.101	-.229
.050			-.040	-.654	-.061	-.526	-.108	-.202
.100			-.054	-.600	-.088	-.371	-.128	-.202
.150			-.088	-.654	-.108	-.371	-.121	-.196
.250	-.108		-.115	-.726	-.115	-.383	-.135	-.199
.400	-.111		-.144	-.814	-.141	-.353	-.144	-.193
.550	-.114	-.575	-.141	-.536	-.147	-.320	-.134	-.177
.700	-.128	-.275	-.137	-.353	-.137	-.255	-.128	-.173
.850	-.134	-.180	-.128	-.258	-.167	-.213	-.124	-.157
.950	-.141	-.157	-.144	-.209	-.154	-.173	-.124	-.144

TABLE 48

PRESSURE PROFILE ON WING

REAR NOZZLES DEFLECTED 45 DEGREES



ALPHA = -.00 DEG THRUST = 652 N (147 LBF)

	ETA = .250		ETA = .387		ETA = .524		ETA = .800	
X/C	CP' X 100		CP' X 100		CP' X 100		CP' X 100	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			-.316		-.297		-.099	
.010			-4.470	-.356				
.015					-.020	-.336		
.025			-.053	-.310	-.040	-.310	-.059	-.237
.050			-.086	-.283	-.066	-.283	-.073	-.224
.100			-.092	-.244	-.066	-.218	-.079	-.191
.150			-.092	-.244	-.105	-.211	-.105	-.185
.250	-.132		-.112	-.240	-.112	-.208	-.112	-.179
.400	-.141		-.150	-.233	-.141	-.208	-.128	-.179
.550	-.144	-.240	-.137	-.220	-.141	-.198	-.128	-.160
.700	-.147	-.272	-.150	-.220	-.134	-.179	-.118	-.157
.850	-.173	-.438	-.157	-.243	-.134	-.169	-.144	-.134
.950	-.211	-.345	-.173	-.220	-.144	-.160	-.125	-.121

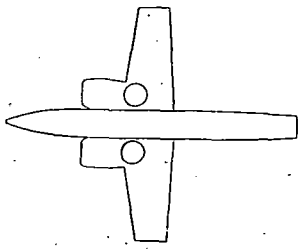
ALPHA = 9.99 DEG THRUST = 653 N (147 LBF)

	ETA = .250		ETA = .387		ETA = .524		ETA = .800	
X/C	CP' X 100		CP' X 100		CP' X 100		CP' X 100	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			-.007		-.013		-.040	
.010			-2.972	-.072				
.015					-.158	-.053		
.025			-.171	-.099	-.152	-.079	-.165	-.046
.050			-.138	-.092	-.145	-.079	-.158	-.086
.100			-.145	-.138	-.138	-.112	-.138	-.099
.150			-.132	-.125	-.138	-.112	-.125	-.105
.250	-.105		-.125	-.153	-.138	-.141	-.138	-.121
.400	-.131		-.147	-.179	-.134	-.153	-.137	-.125
.550	-.134	-.211	-.150	-.182	-.131	-.150	-.125	-.125
.700	-.147	-.259	-.141	-.185	-.125	-.147	-.118	-.115
.850	-.163	-.364	-.150	-.204	-.125	-.150	-.115	-.112
.950	-.201	-.339	-.166	-.201	-.134	-.134	-.109	-.109

TABLE 49

PRESSURE PROFILE ON WING

FORWARD NOZZLES DEFLECTED 90 DEGREES



ALPHA = -.01 DEG THRUST = 609 N (137 LBF)

X/C	ETA = .250 CP' X 100 UPPER	LOWER	ETA = .387 CP' X 100 UPPER	LOWER	ETA = .524 CP' X 100 UPPER	LOWER	ETA = .800 CP' X 100 UPPER	LOWER
.000			-.219		-.367		-.636	
.010			-1.264	-.360				
.015					-.085	-.184		
.025			-.092	-.226	-.042	-.240	-.014	-.346
.050			-.049	-.219	-.049	-.530	-.028	-.155
.100			-.042	-.466	-.042	-.290	-.042	-.593
.150			-.014	-.198	-.042	-.473	-.071	-.982
.250	-.078		-.085	-.277	-.071	-.216	-.042	-.178
.400	-.134		-.092	-.455	-.062	-.387	-.058	-.267
.550	-.147	-.192	-.099	-.181	-.099	-.541	-.092	-.264
.700	-.161	-.216	-.151	-.407	-.140	-.185	-.099	-.250
.850	-.157	-.284	-.116	-.216	-.168	-.209	-.113	-.202
.950	-.202	-.192	-.154	-.236	-.140	-.157	-.147	-.168

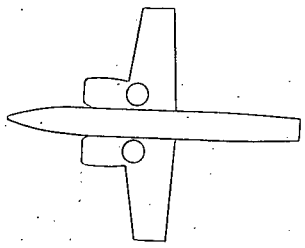
ALPHA = 10.00 DEG THRUST = 607 N (136 LBF)

X/C	ETA = .250 CP' X 100 UPPER	LOWER	ETA = .387 CP' X 100 UPPER	LOWER	ETA = .524 CP' X 100 UPPER	LOWER	ETA = .800 CP' X 100 UPPER	LOWER
.000			-.099		-.234		-.340	
.010			-1.028	-.099				
.015					-.177	-.312		
.025			-.142	-.255	-.227	-.270	-.270	-.333
.050			-.078	-.347	-.078	-.156	-.142	-.433
.100			-.043	-.227	-.050	-.262	-.099	-.355
.150			-.050	-.078	-.099	-.184	-.078	-.142
.250	-.064		-.071	-.217	-.071	-.278	-.085	-.158
.400	-.058		-.034	-.100	-.031	-.364	-.055	-.151
.550	-.055	-.323	-.031	-.117	-.038	-.144	-.052	-.117
.700	-.079	-.254	-.062	-.082	-.031	-.213	-.045	-.168
.850	-.086	-.113	-.052	-.113	-.055	-.131	-.065	-.144
.950	-.110	-.100	-.065	-.137	-.058	-.113	-.082	-.144

TABLE 50

PRESSURE PROFILE ON WING

FORWARD NOZZLES DEFLECTED 90 DEGREES WITH FLAPS



ALPHA = [illegible] DEG THRUST = 607 N (136 LBF)

	ETA = .250		ETA = .387		ETA = .524		ETA = .800	
X/C	CP X 100		CP X 100		CP X 100		CP X 100	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			-.191		-.333		-.574	
.010			-3.431	-.199				
.015					-.255	-.376		
.025			-.078	-.156	-.106	-.199	-.163	-.553
.050			-.043	-.227	-.121	-.206	-.057	-.666
.100			-.021	-.362	-.035	-.262	-.007	-.652
.150			-.028	-.184	-.028	-.312	-.043	-.404
.250	-.028		-.014	-.378	-.035	-.505	-.021	-.598
.400	-.072		-.072	-.234	-.045	-.584	-.079	-.306
.550	-.058	-.148	-.055	-.598	-.072	-.347	-.055	-.124
.700	-.079	-.089	-.069	-.206	-.069	-.113	-.048	-.137
.850	-.107	-.079	-.076	-.144	-.079	-.141	-.072	-.158
.950	-.137	-.162	-.151	-.158	-.110	-.158	-.082	-.141

ALPHA = 10.00 DEG THRUST = 607 N (137 LBF)

	ETA = .250		ETA = .387		ETA = .524		ETA = .800	
X/C	CP X 100		CP X 100		CP X 100		CP X 100	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			-.191		-.220		-.411	
.010			-2.995	-.184				
.015					-.142	-.135		
.025			-.106	-.269	-.149	-.149	-.205	-.135
.050			-.021	-.113	-.071	-.191	-.071	-.531
.100			-.064	-.305	-.113	-.220	-.099	-.220
.150			-.050	-.163	-.042	-.205	-.064	-.205
.250	-.057		-.042	-.213	-.042	-.206	-.064	-.165
.400	-.051		-.056	-.093	-.051	-.161	-.045	-.158
.550	-.055	-.206	-.062	-.113	-.072	-.168	-.051	-.154
.700	-.055	-.134	-.058	-.124	-.065	-.158	-.058	-.113
.850	-.065	-.096	-.065	-.178	-.058	-.172	-.069	-.175
.950	-.093	-.079	-.120	-.137	-.076	-.172	-.058	-.196

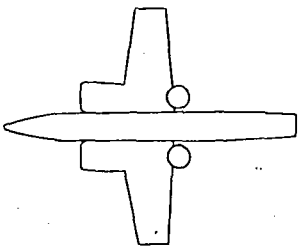


TABLE 51

PRESSURE PROFILE ON WING

REAR NOZZLES DEFLECTED 90 DEGREES

ALPHA = 0.00 DEG THRUST = 646 N (145 LBF)

	ETA = .250		ETA = .387		ETA = .524		ETA = .800	
X/C	CP' X 100		CP' X 100		CP' X 100		CP' X 100	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			-.207		-.247		-.393	
.010			-3.411	-.187				
.015					-.133	-.247		
.025			-.073	-.207	-.160	-.320	-.073	-.360
.050			-.027	-.220	-.067	-.286	-.027	-.373
.100			-.033	-.253	-.033	-.286	-.020	-.333
.150			-.067	-.273	-.053	-.326	-.047	-.393
.250	-.047		-.027	-.284	-.013	-.317	-.033	-.436
.400	-.061		-.042	-.220	-.036	-.349	-.045	-.362
.550	-.068	-.388	-.058	-.258	-.045	-.255	-.058	-.207
.700	-.087	-.455	-.065	-.191	-.078	-.339	-.048	-.191
.850	-.120	-.352	-.090	-.249	-.081	-.165	-.084	-.155
.950	-.171	-.287	-.135	-.126	-.107	-.226	-.090	-.132

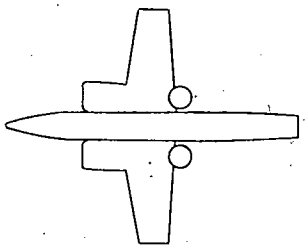
ALPHA = 10.00 DEG THRUST = 641 N (144 LBF)

	ETA = .250		ETA = .387		ETA = .524		ETA = .800	
X/C	CP' X 100		CP' X 100		CP' X 100		CP' X 100	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			-.087		-.081		-.161	
.010			-2.650	-.094				
.015					-.134	-.094		
.025			-.074	-.081	-.121	-.161	-.134	-.221
.050			-.114	-.148	-.067	-.154	-.107	-.288
.100			-.114	-.148	-.087	-.174	-.060	-.154
.150			-.047	-.127	-.067	-.221	-.060	-.141
.250	-.047		-.067	-.234	-.060	-.244	-.060	-.205
.400	-.042		-.062	-.127	-.039	-.137	-.059	-.195
.550	-.049	-.276	-.059	-.094	-.059	-.166	-.039	-.111
.700	-.075	-.309	-.065	-.111	-.055	-.166	-.042	-.205
.850	-.075	-.267	-.068	-.114	-.068	-.176	-.049	-.368
.950	-.159	-.283	-.078	-.133	-.081	-.198	-.104	-.185

TABLE 52

PRESSURE PROFILE ON WING

REAR NOZZLES DEFLECTED 90 DEGREES WITH FLAPS



ALPHA = -.01 DEG THRUST = 643 N (145 LBF)

X/C	ETA = .250 CP' X 100		ETA = .387 CP' X 100		ETA = .524 CP' X 100		ETA = .800 CP' X 100	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			-.201		-.174		-.248	
.010			-2.677	-.114				
.015					-.161	-.268		
.025			-.100	-.181	-.080	-.281	-.127	-.301
.050			-.054	-.254	-.007	-.261	-.027	-.348
.100			-.033	-.221	-.020	-.181	-.040	-.415
.150			-.027	-.214	-.033	-.241	-.007	-.415
.250	-.060		-.040	-.259	-.040	-.276	-.060	-.373
.400	-.088		-.049	-.295	-.062	-.334	-.045	-.334
.550	-.094	-.204	-.084	-.247	-.055	-.234	-.065	-.315
.700	-.088	-.448	-.091	-.126	-.078	-.279	-.088	-.201
.850	-.114	-.438	-.094	-.165	-.097	-.272	-.101	-.123
.950	-.156	-.402	-.143	-.237	-.126	-.224	-.104	-.143

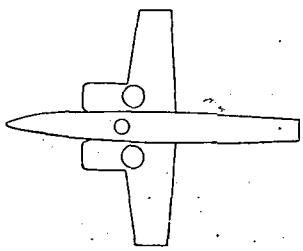
ALPHA = 9.96 DEG THRUST = 639 N (144 LBF)

X/C	ETA = .250 CP' X 100		ETA = .387 CP' X 100		ETA = .524 CP' X 100		ETA = .800 CP' X 100	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			-.081		-.121		-.222	
.010			-.578	-.087				
.015					-.101	-.182		
.025			-.061	-.101	-.087	-.202	-.074	-.182
.050			-.020	-.087	-.081	-.155	-.061	-.148
.100			-.020	-.168	-.081	-.161	-.020	-.168
.150			-.013	-.101	-.061	-.195	-.034	-.188
.250	-.054		-.020	-.111	-.020	-.091	-.027	-.238
.400	-.055		-.065	-.130	-.010	-.170	-.052	-.228
.550	-.049	-.342	-.020	-.137	-.029	-.117	-.016	-.117
.700	-.052	-.313	-.029	-.078	-.026	-.150	-.010	-.170
.850	-.081	-.238	-.029	-.078	-.052	-.117	-.049	-.143
.950	-.095	-.228	-.065	-.091	-.075	-.130	-.068	-.143

TABLE 53

PRESSURE PROFILE ON WING

FORWARD NOZZLES DEFLECTED 90 DEGREES WITH LIFT JET



ALPHA = .01 DEG THRUST = 830 N (187 LBF)

X/C	ETA = .250 CP' X 100		ETA = .387 CP' X 100		ETA = .524 CP' X 100		ETA = .800 CP' X 100	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			-.172		-.271		-.302	
.010			-2.628	-.142				
.015					-.160	-.228		
.025			-.049	-.123	-.105	-.302	-.197	-.357
.050			-.043	-.172	-.062	-.345	-.129	-.406
.100			-.086	-.283	-.058	-.185	-.099	-.628
.150			-.062	-.191	-.062	-.166	-.068	-.696
.250	-.012		-.037	-.185	-.043	-.218	-.031	-.349
.400	-.042		-.027	-.230	-.045	-.161	-.051	-.227
.550	-.051	-.069	-.045	-.065	-.048	-.152	-.045	-.149
.700	-.057	-.098	-.060	-.122	-.048	-.146	-.057	-.221
.850	-.057	-.081	-.060	-.092	-.063	-.230	-.069	-.385
.950	-.078	-.125	-.104	-.119	-.095	-.152	-.110	-.286

ALPHA = 10.00 DEG THRUST = 824 N (185 LBF)

X/C	ETA = .250 CP' X 100		ETA = .387 CP' X 100		ETA = .524 CP' X 100		ETA = .800 CP' X 100	
	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			-.081		-.143		-.198	
.010			-2.205	-.105				
.015					-.167	-.124		
.025			-.099	-.093	-.118	-.167	-.205	-.415
.050			-.074	-.074	-.068	-.217	-.211	-.601
.100			-.068	-.093	-.105	-.149	-.087	-.304
.150			-.081	-.087	-.087	-.285	-.050	-.143
.250	-.031		-.068	-.186	-.081	-.150	-.062	-.652
.400	-.069		-.048	-.129	-.072	-.303	-.066	-.180
.550	-.051	-.084	-.054	-.150	-.042	-.162	-.069	-.147
.700	-.057	-.111	-.081	-.201	-.051	-.355	-.087	-.129
.850	-.072	-.084	-.087	-.261	-.069	-.448	-.081	-.288
.950	-.069	-.102	-.078	-.264	-.087	-.370	-.093	-.207

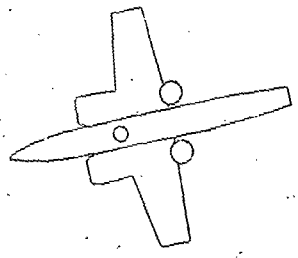


TABLE 5H

PRESSURE PROFILE ON WING

REAR NOZZLES DEFLECTED 90 DEGREES WITH LIFT JET

ALPHA = -.02 DEG THRUST = 860 N (193 LBF)

	ETA = .250		ETA = .387		ETA = .524		ETA = .800	
	CP' X 100		CP' X 100		CP' X 100		CP' X 100	
X/C	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			-.117	-.172	-.220		-.446	
.010			-3.594					
.015				-.143	-.166	-.202	-.143	-.339
.025			-.131	-.232	-.160	-.265	-.053	-.267
.050			-.089	-.196	-.071	-.250	-.071	-.333
.100			-.071	-.250	-.071	-.327	-.083	-.309
.150	-.036		-.071	-.216	-.053	-.178	-.059	-.374
.250	-.043	-.205	-.053	-.228	-.053	-.268	-.049	-.181
.400	-.035	-.308	-.052	-.184	-.066	-.288	-.046	-.204
.550	-.058	-.363	-.037	-.161	-.049	-.259	-.052	-.228
.700	-.069	-.314	-.049	-.130	-.058	-.170	-.060	-.135
.850	-.176	?	-.061	-.193	-.066	-.164	-.086	-.144
.950			-.072		-.115	-.155		

ALPHA = 9.97 DEG THRUST = 852 N (192 LBF)

	ETA = .250		ETA = .387		ETA = .524		ETA = .800	
	CP' X 100		CP' X 100		CP' X 100		CP' X 100	
X/C	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER	UPPER	LOWER
.000			-.162	-.120	-.156		-.264	
.010			-2.237					
.015				-.198	-.180	-.192	-.168	-.288
.025			-.138	-.234	-.114	-.120	-.102	-.342
.050			-.042	-.234	-.114	-.282	-.072	-.318
.100			-.048	-.114	-.078	-.216	-.096	-.384
.150	-.034		-.048	-.087	-.060	-.192	-.066	-.206
.250	-.041	-.183	-.048	-.090	-.042	-.285	-.055	-.174
.400	-.055	-.253	-.049	-.148	-.064	-.230	-.052	-.157
.550	-.055	-.297	-.061	-.157	-.061	-.238	-.079	-.276
.700	-.070	-.270	-.055	-.183	-.061	-.224	-.081	-.227
.850	-.093		-.096	-.186	-.079	-.201	-.116	-.291
.950			-.061		-.102	-.186		

GEOMETRY

Wing:

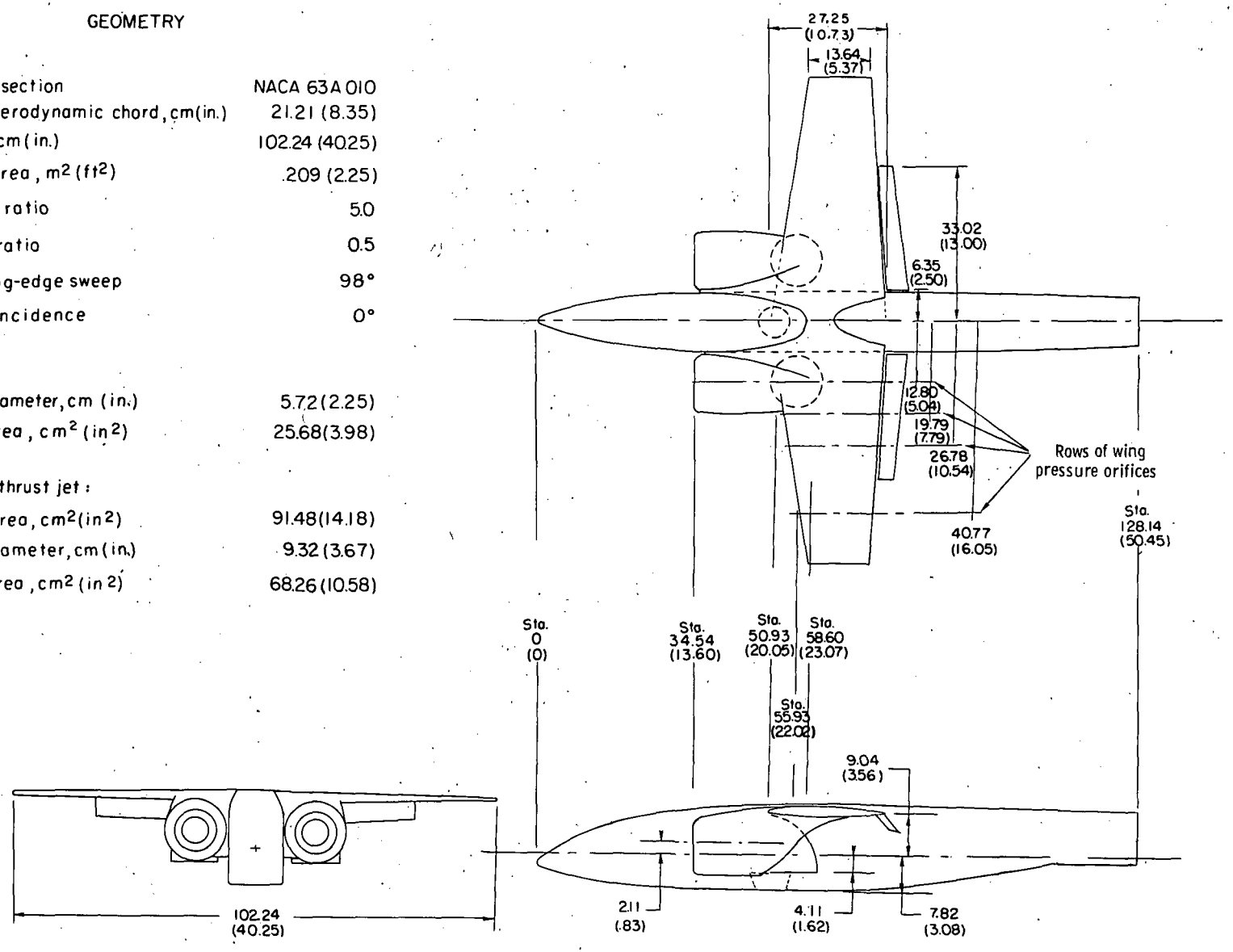
Airfoil section	NACA 63A 010
Mean aerodynamic chord, cm(in.)	21.21 (8.35)
Span, cm(in.)	102.24 (40.25)
Wing area, m^2 (ft^2)	.209 (2.25)
Aspect ratio	5.0
Taper ratio	0.5
Leading-edge sweep	98°
Wing incidence	0°

Lift jet:

Exit diameter, cm (in.)	5.72 (2.25)
Exit area, cm^2 (in^2)	25.68 (3.98)

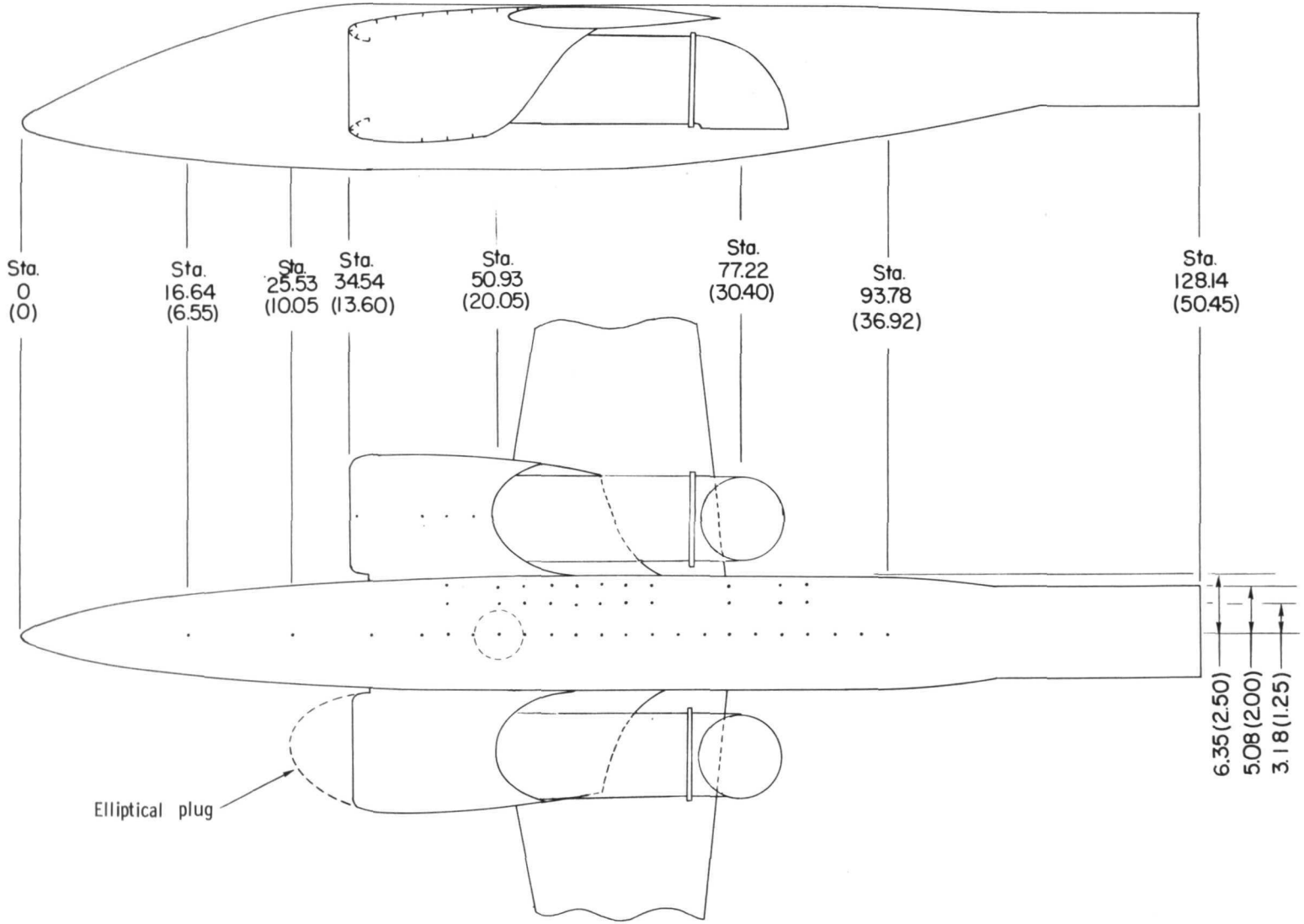
Vectored thrust jet:

Inlet area, cm^2(in^2)	91.48 (14.18)
Exit diameter, cm (in.)	9.32 (3.67)
Exit area, cm^2 (in^2)	68.26 (10.58)



(a) Front vectored-thrust configuration with lift jet and flaps.

Figure 1.- Details of model. All dimensions are in centimeters (inches) unless otherwise noted.



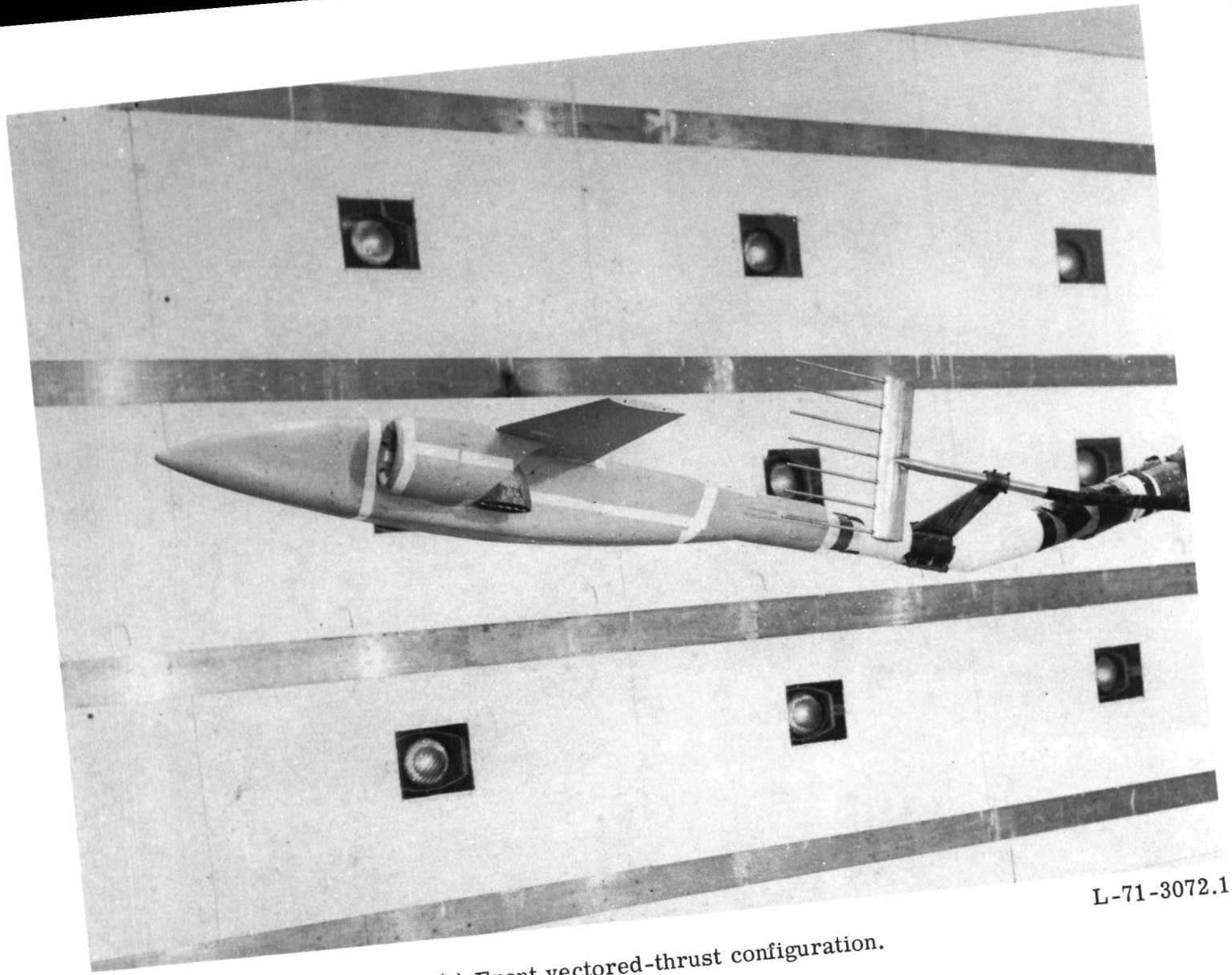
(b) Rear vectored-thrust configuration.

Figure 1.- Concluded.

L-71-3220

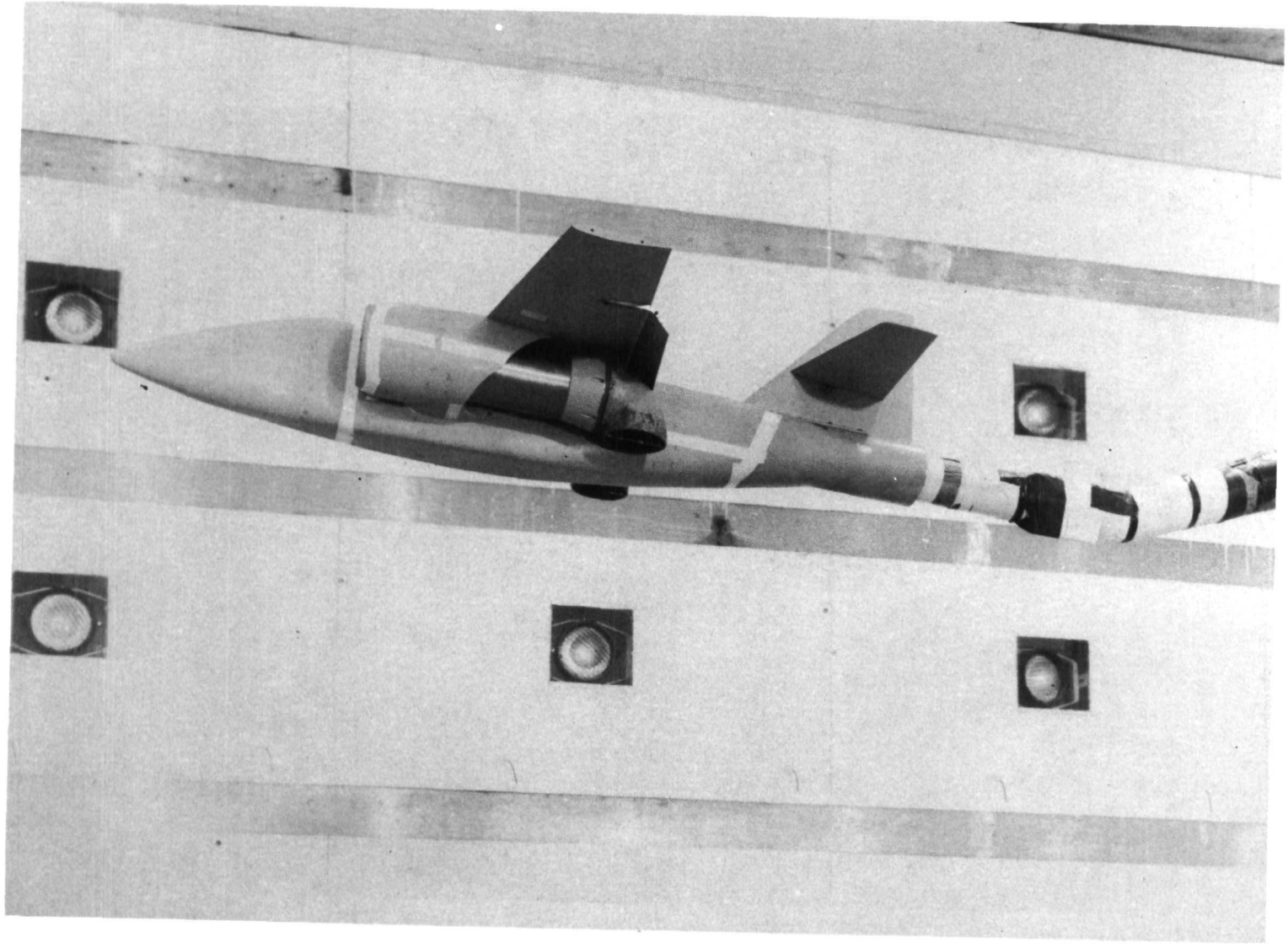
(a) Lift-jet configuration.

Figure 2.- Model in the V/STOL tunnel.



(b) Front vectored-thrust configuration.

Figure 2.- Continued.



L-71-3208

(c) Rear vectored-thrust configuration with flaps.

Figure 2.- Concluded.

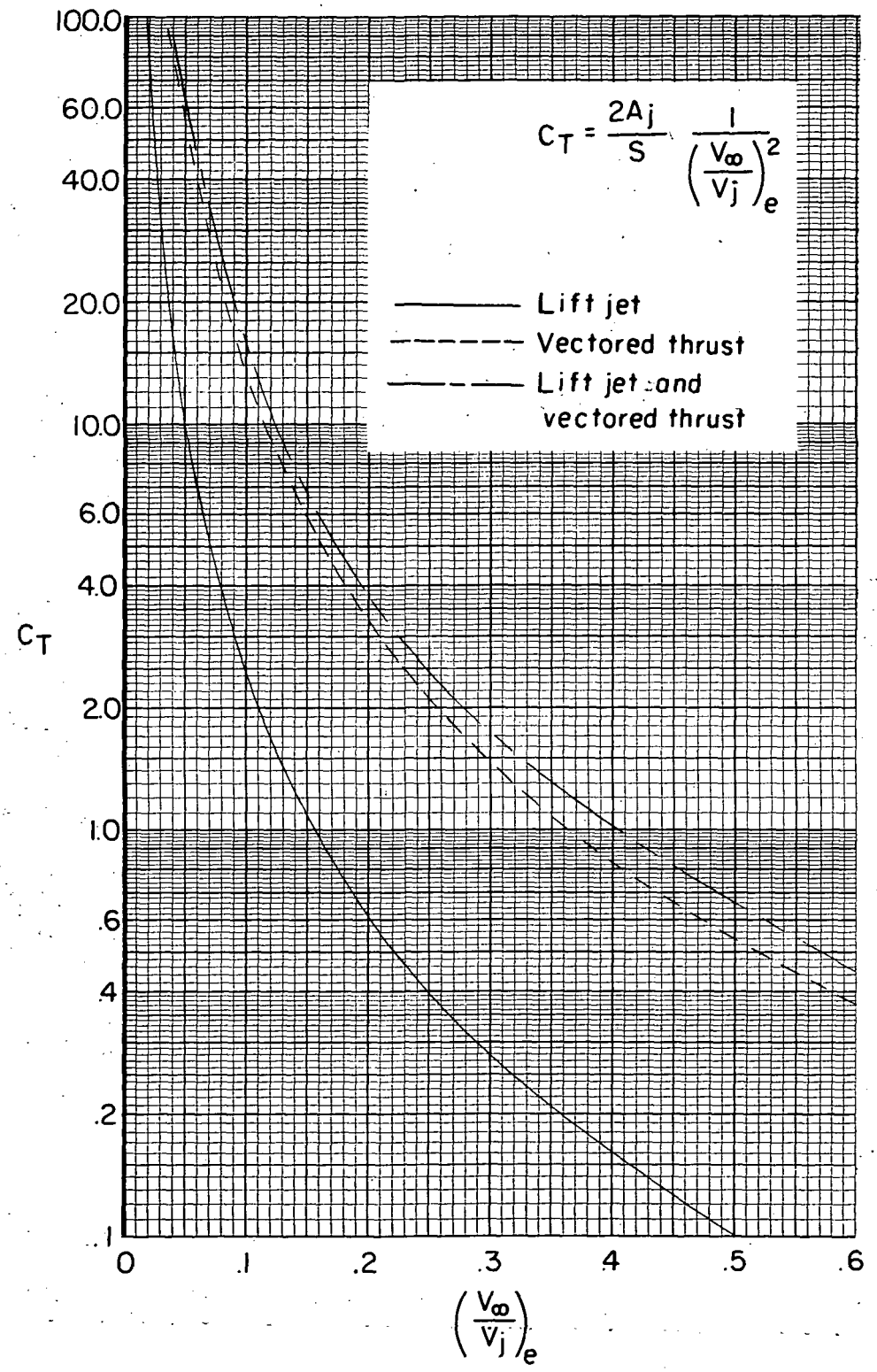
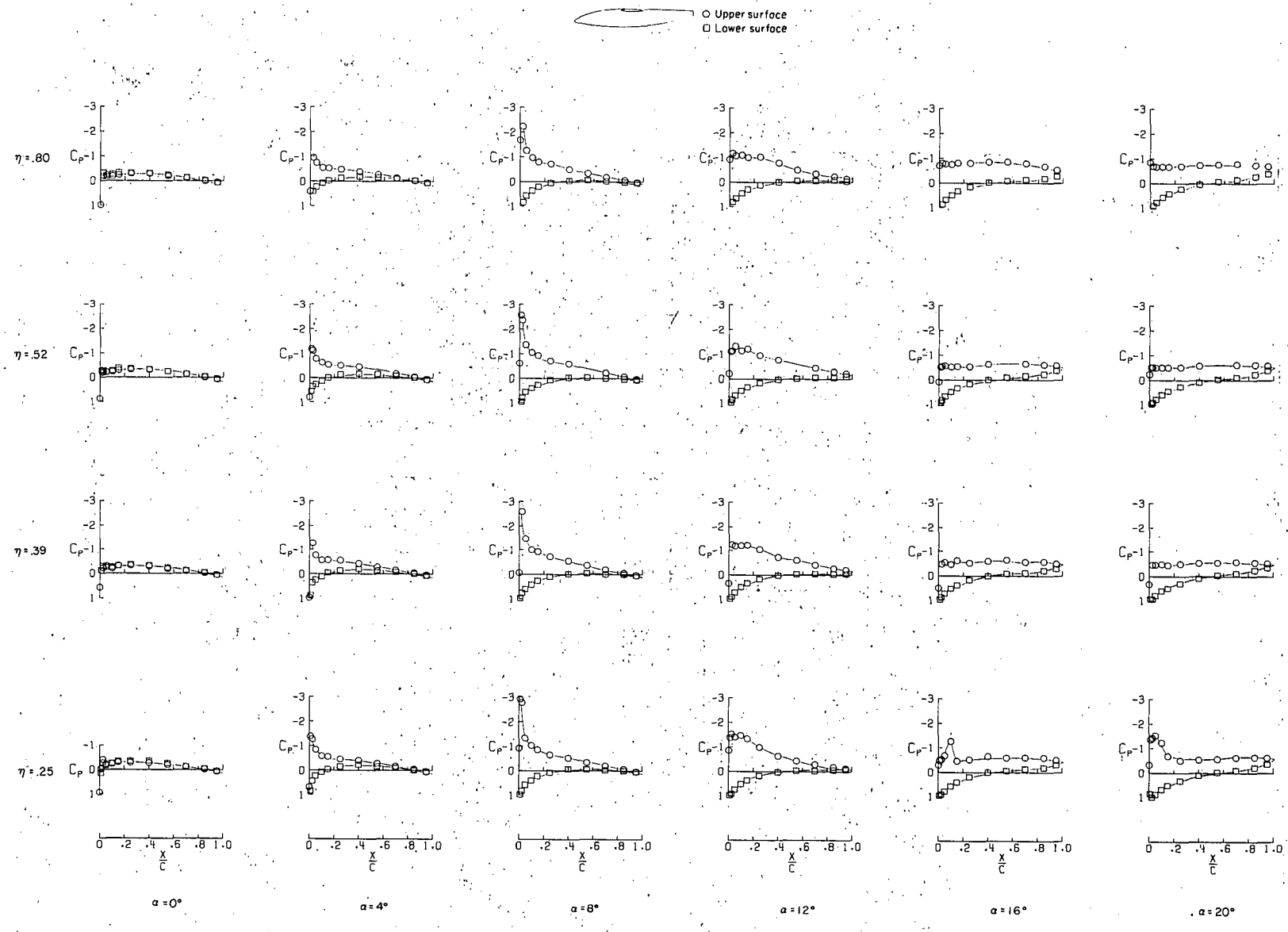
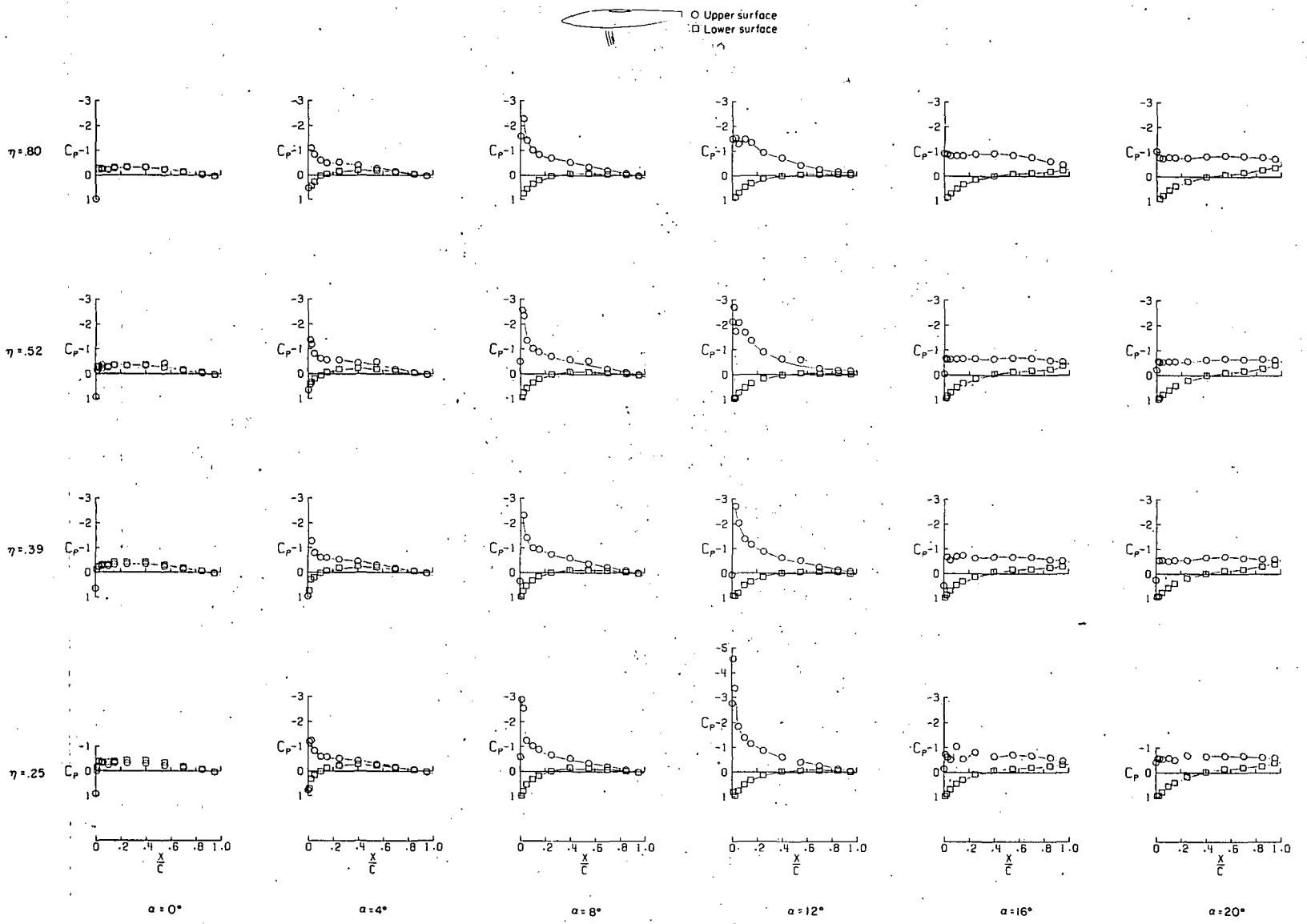


Figure 3.- Variation of thrust coefficient with effective velocity ratio.



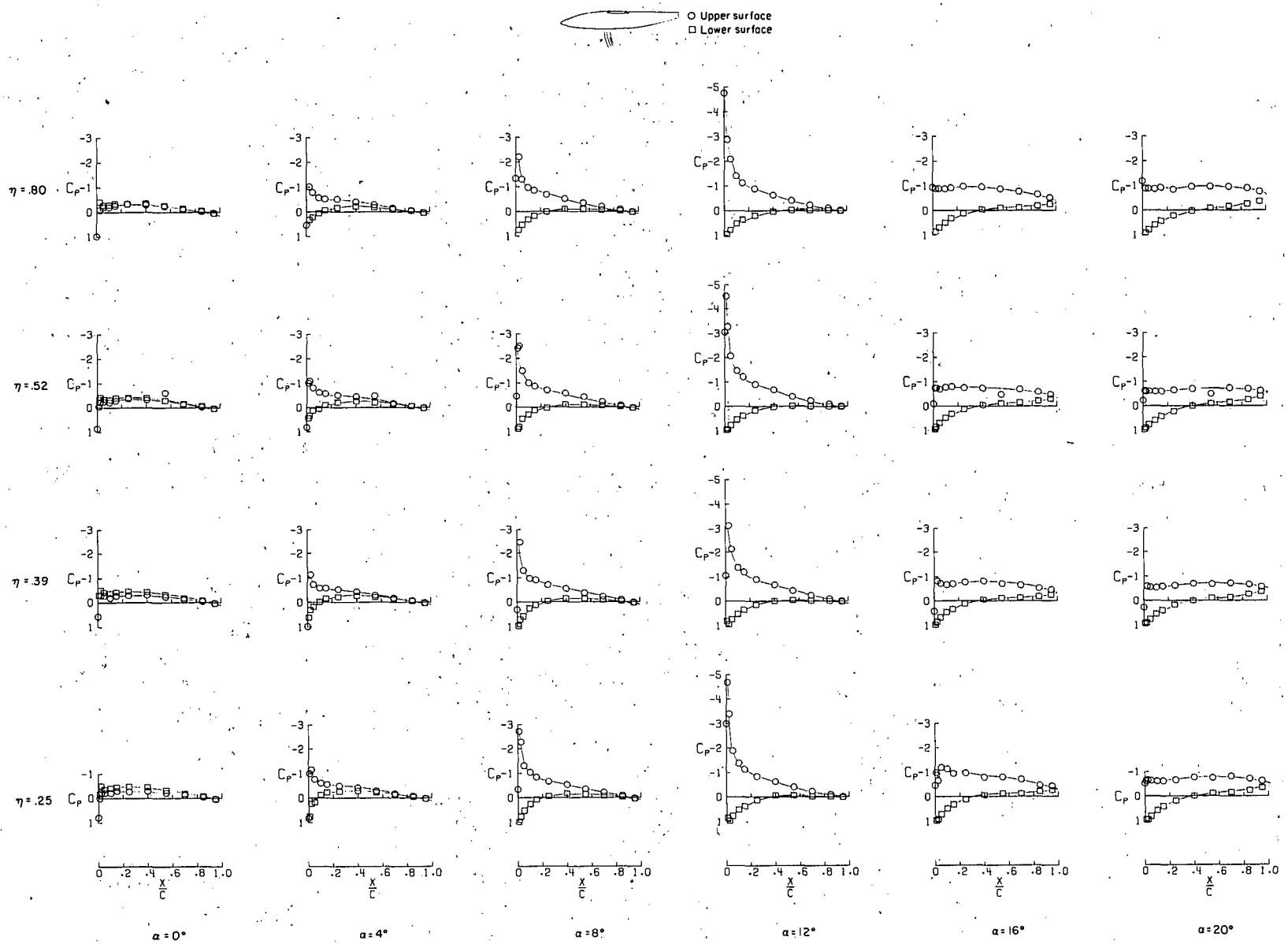
(a) $\left(V_\infty/V_j\right)_e = \infty$.

Figure 4.- Pressure distributions on the wing for the lift-jet configuration.



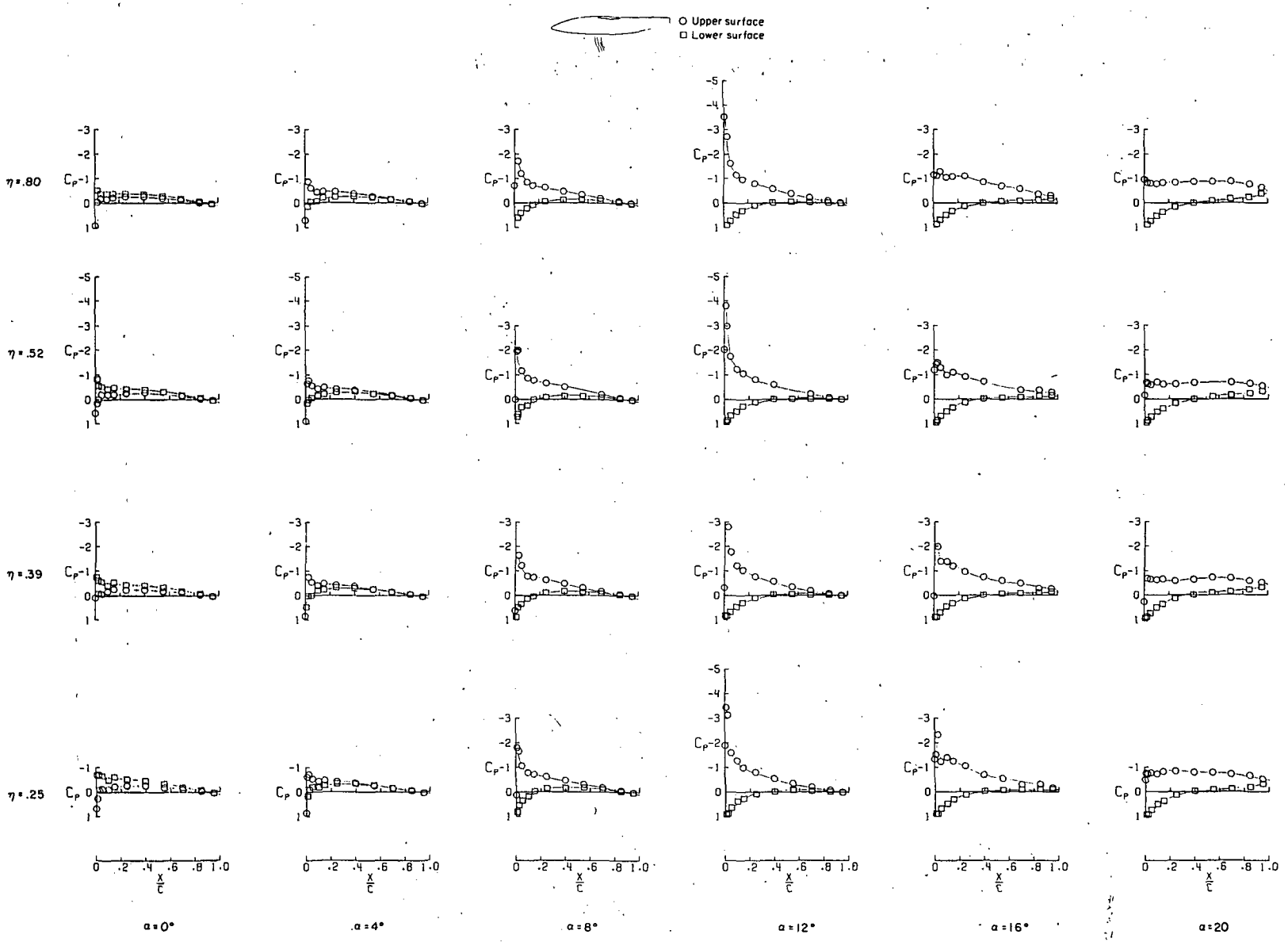
(b) $\left(V_\infty/V_j\right)_e \approx 0.3$.

Figure 4.- Continued.



(c) $(V_\infty/V_j)_e \approx 0.2$.

Figure 4.- Continued.



(d) $\left(V_\infty/V_j\right)_e \approx 0.1$.

Figure 4.- Concluded.

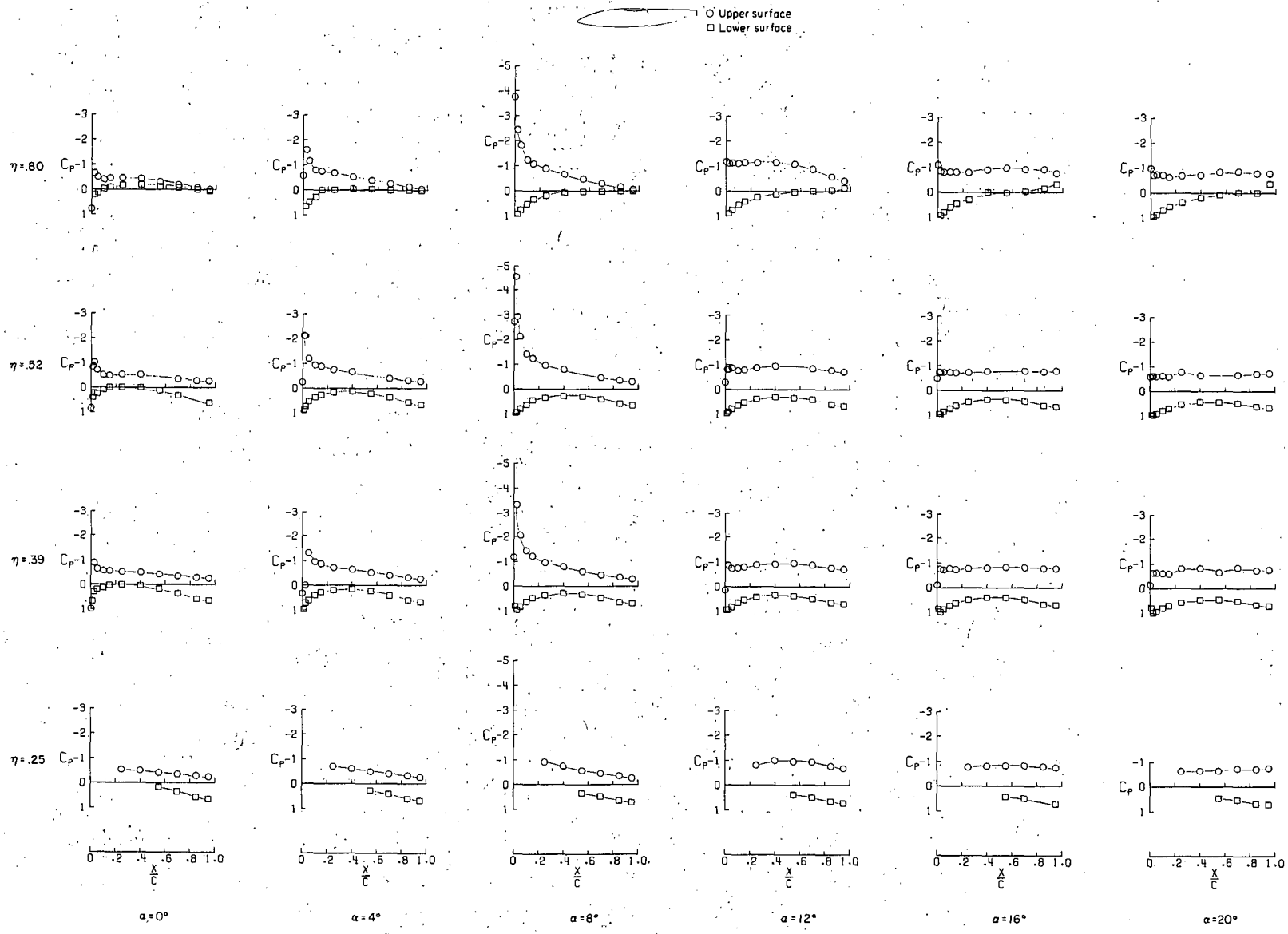
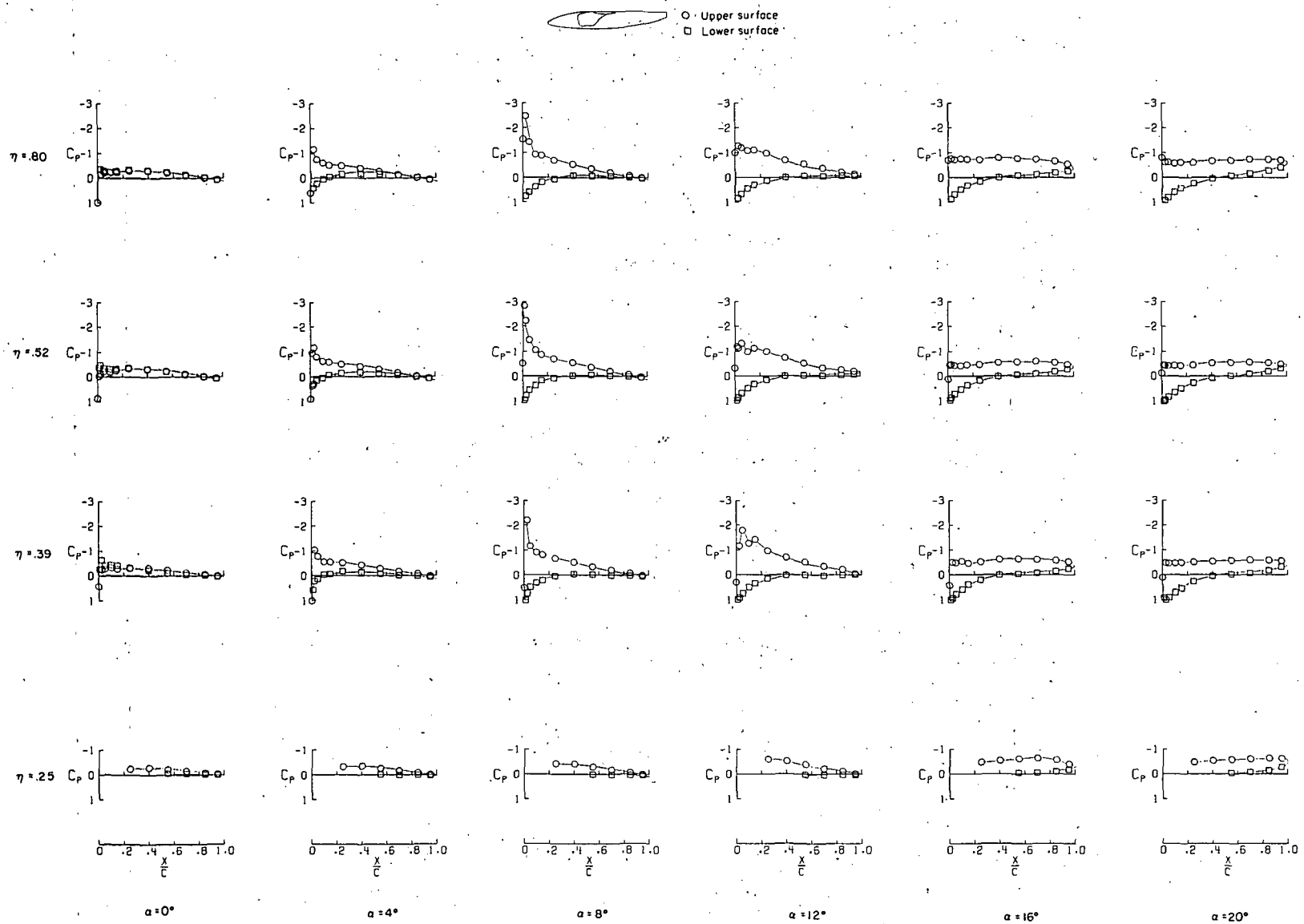
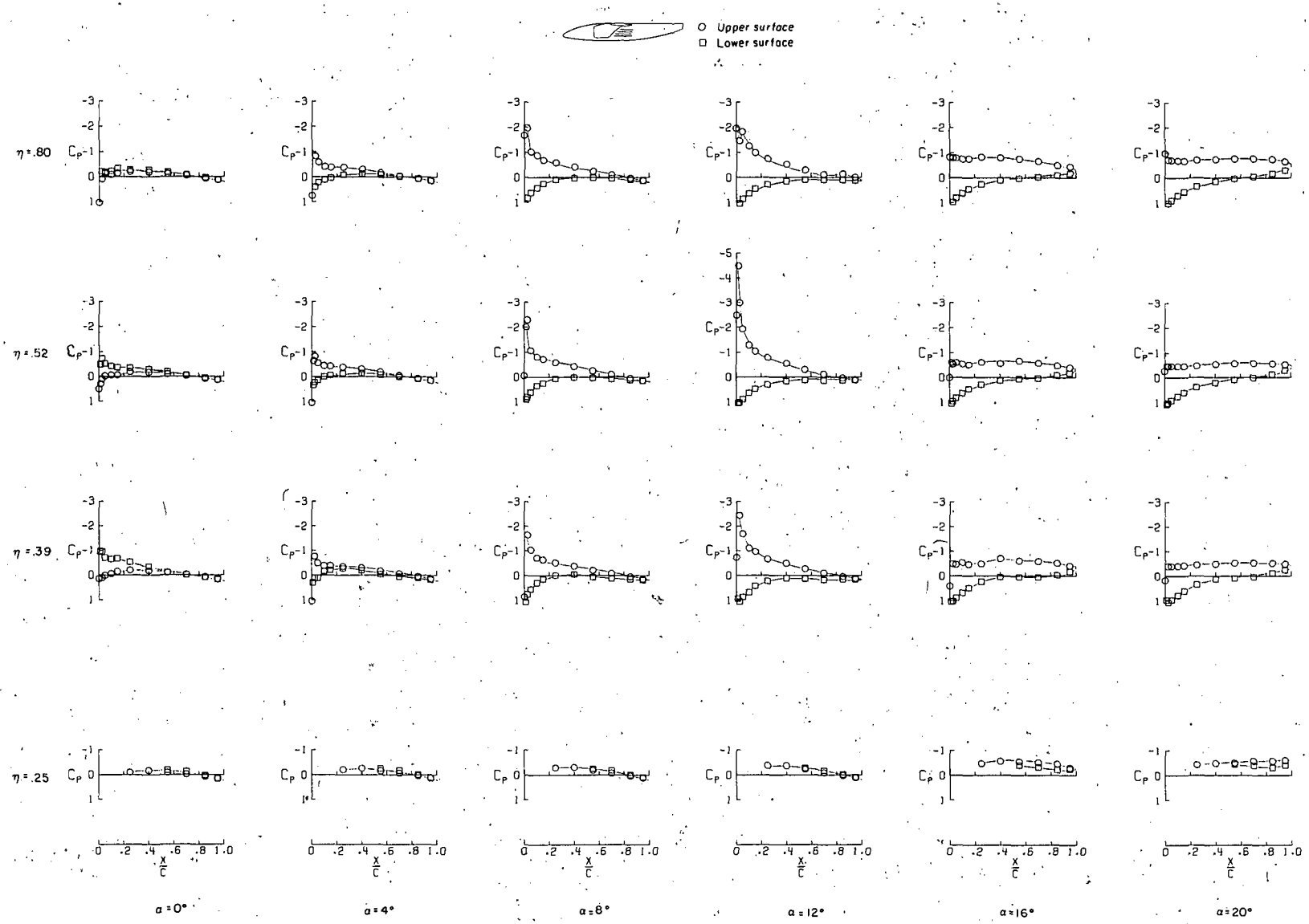


Figure 5.- Pressure distributions on the wing with flaps for the lift-jet configuration. $\left(V_\infty/V_j\right)_e \approx \infty$.



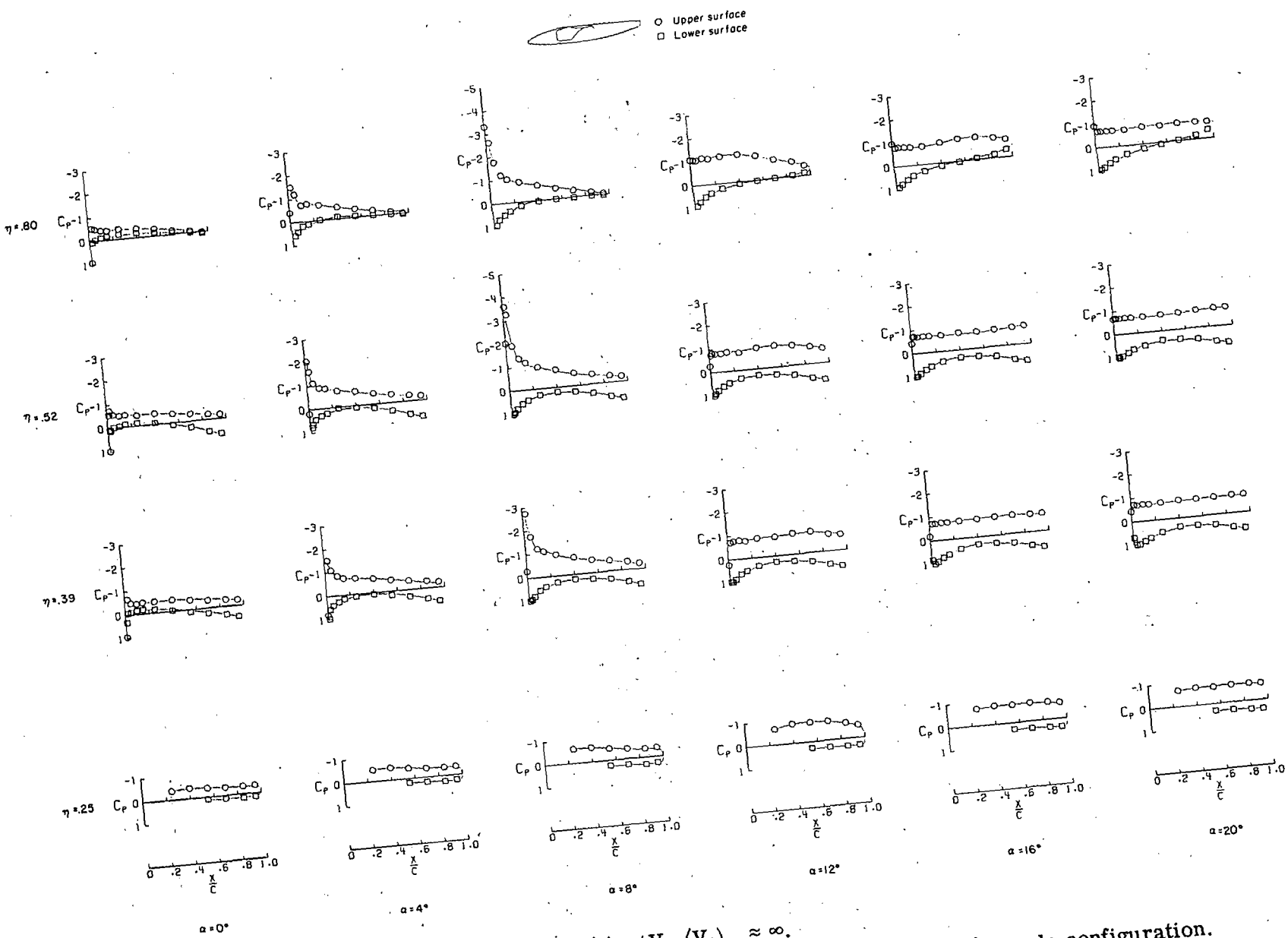
(a) $\left(V_\infty/V_j\right)_e \approx \infty$.

Figure 6.- Pressure distributions on the wing for the 0° nozzle configuration.



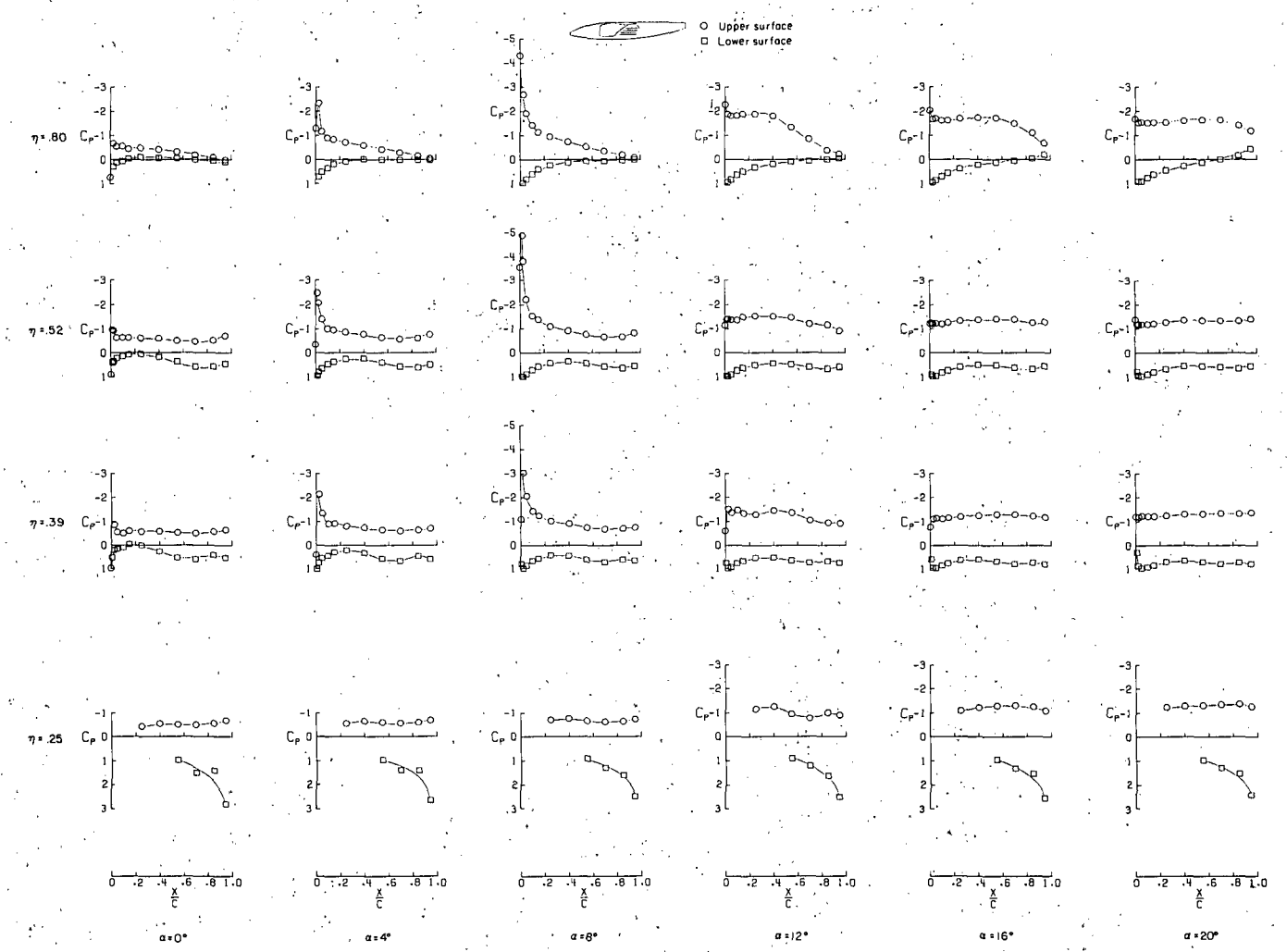
(b) $\left(V_\infty/V_j\right)_e \approx 0.3$.

Figure 6.- Concluded.



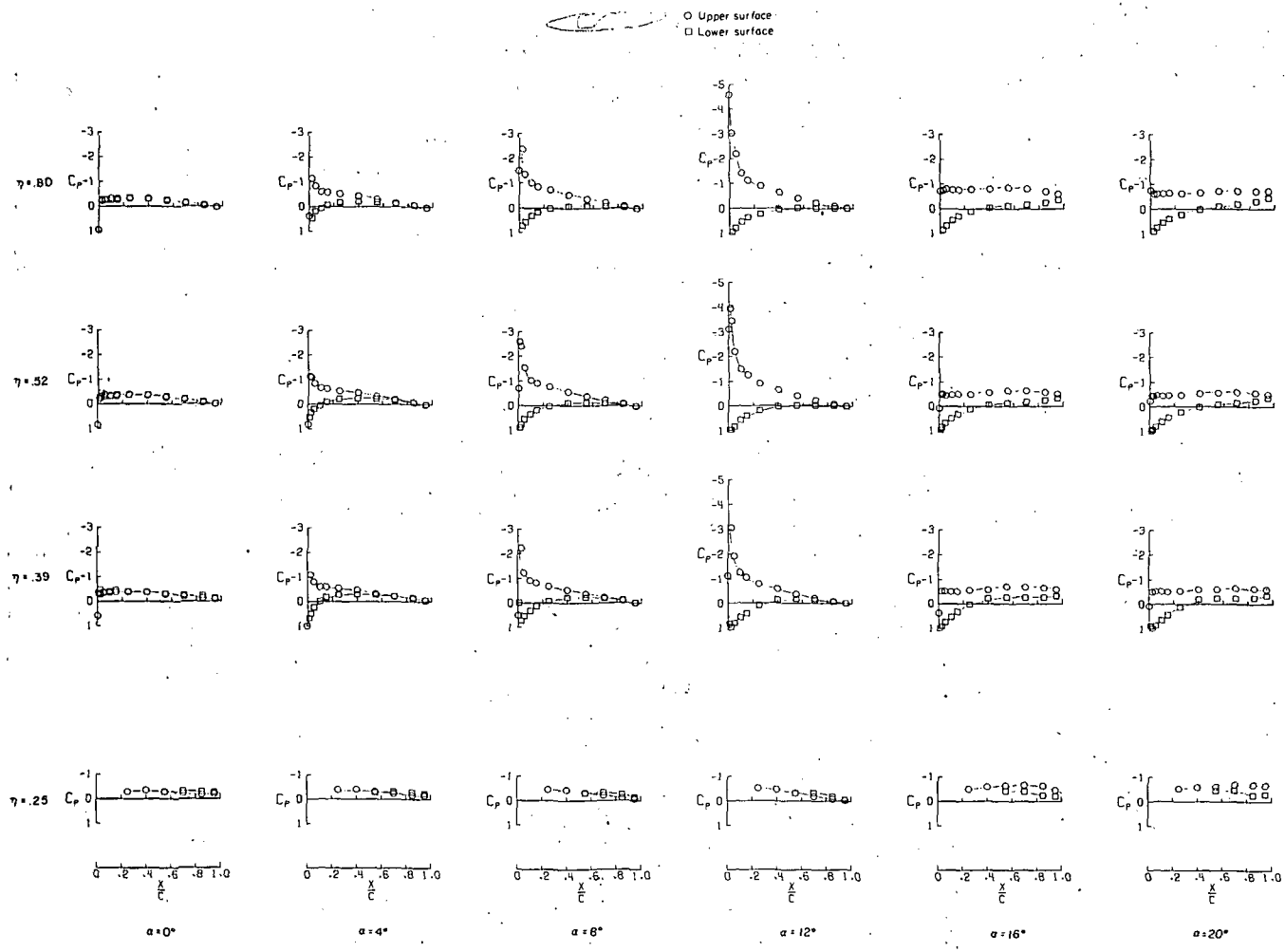
(a) $\left(V_\infty/V_j\right)_e \approx \infty$.

Figure 7.- Pressure distributions on the wing with flaps for the forward 0° nozzle configuration.



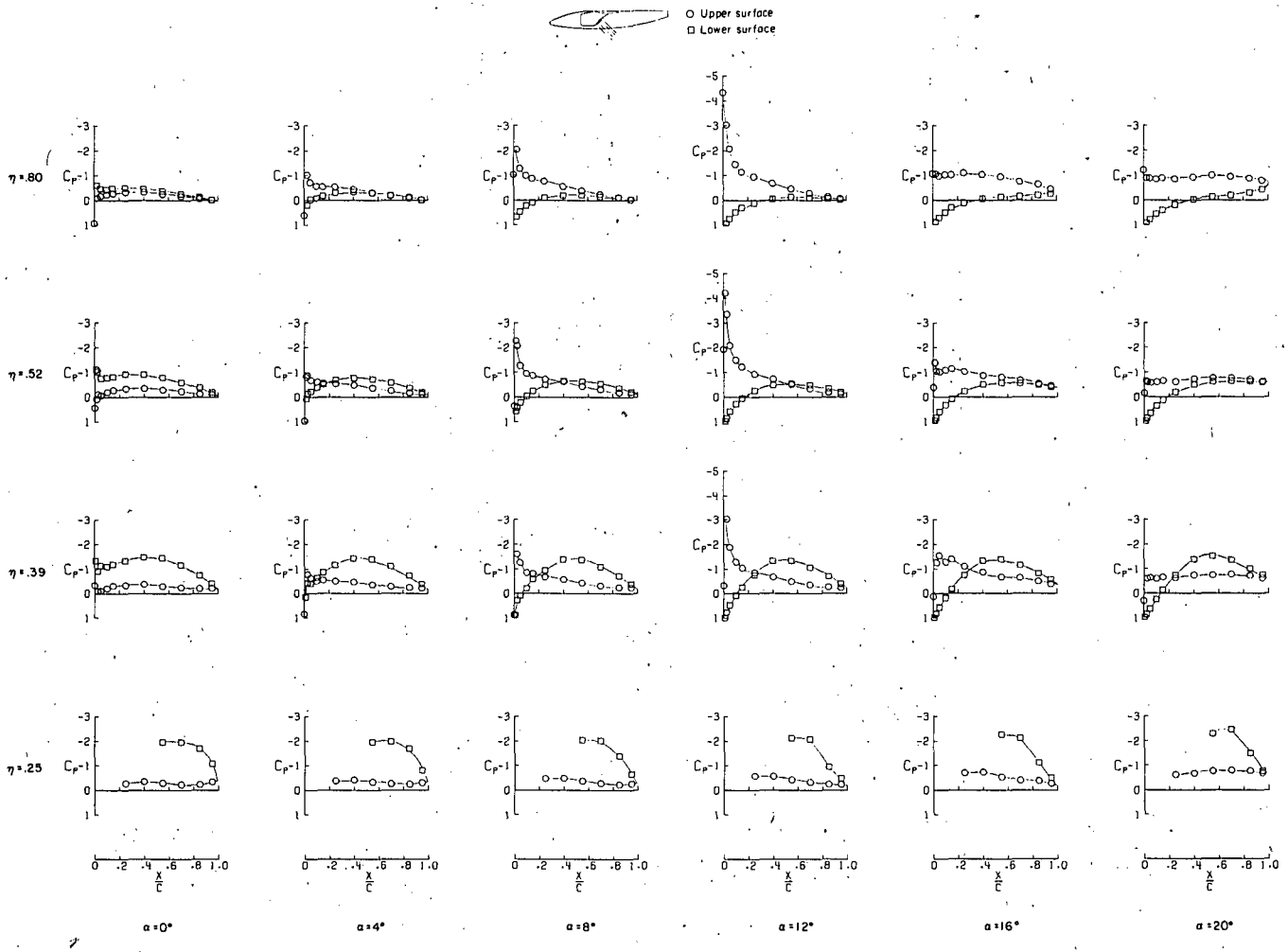
(b) $\left(V_\infty/V_j\right)_e \approx 0.3$.

Figure 7.- Concluded.



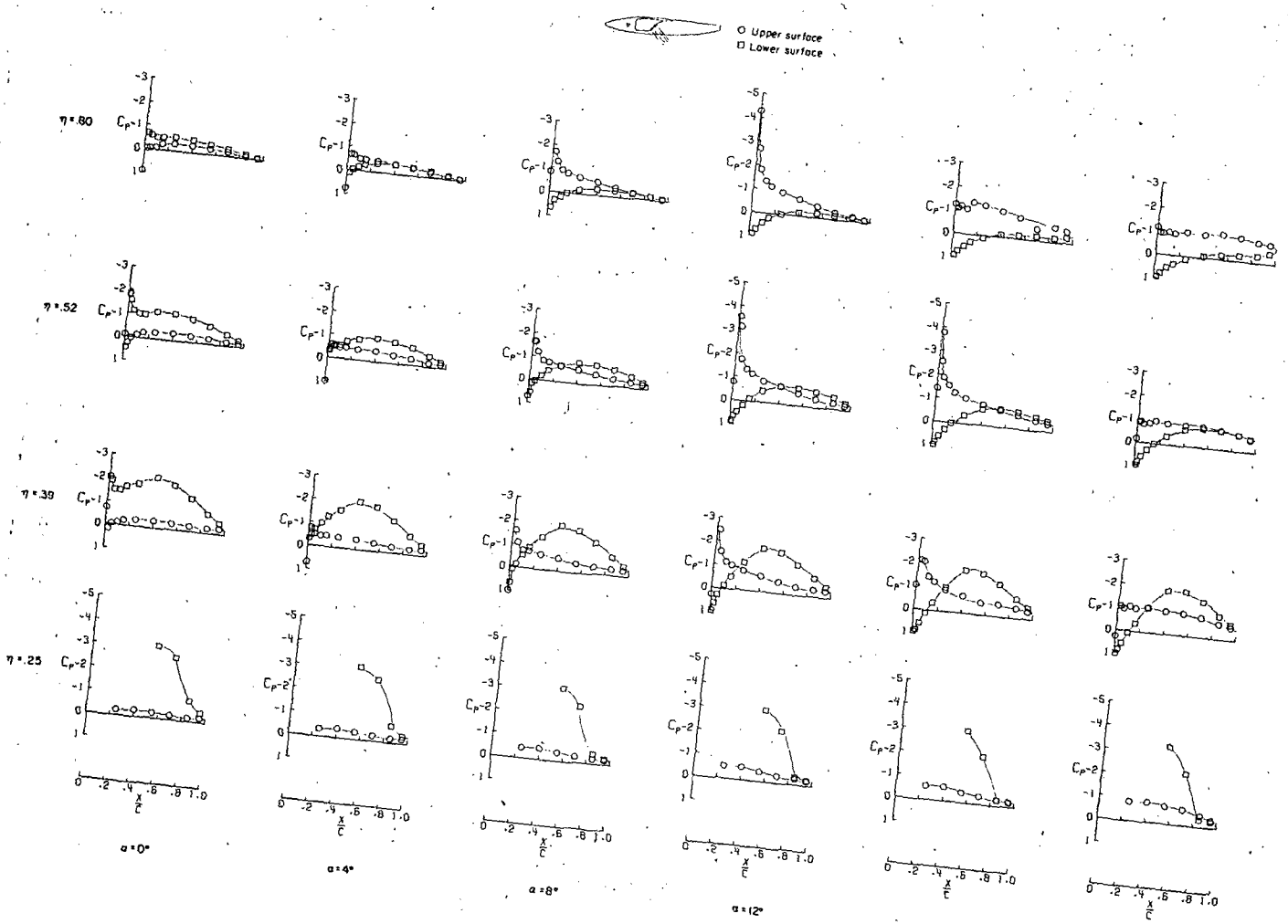
(a) $\left(V_\infty/V_j\right)_e \approx \infty$.

Figure 8.- Pressure distributions on the wing for the forward 45° nozzle configuration.



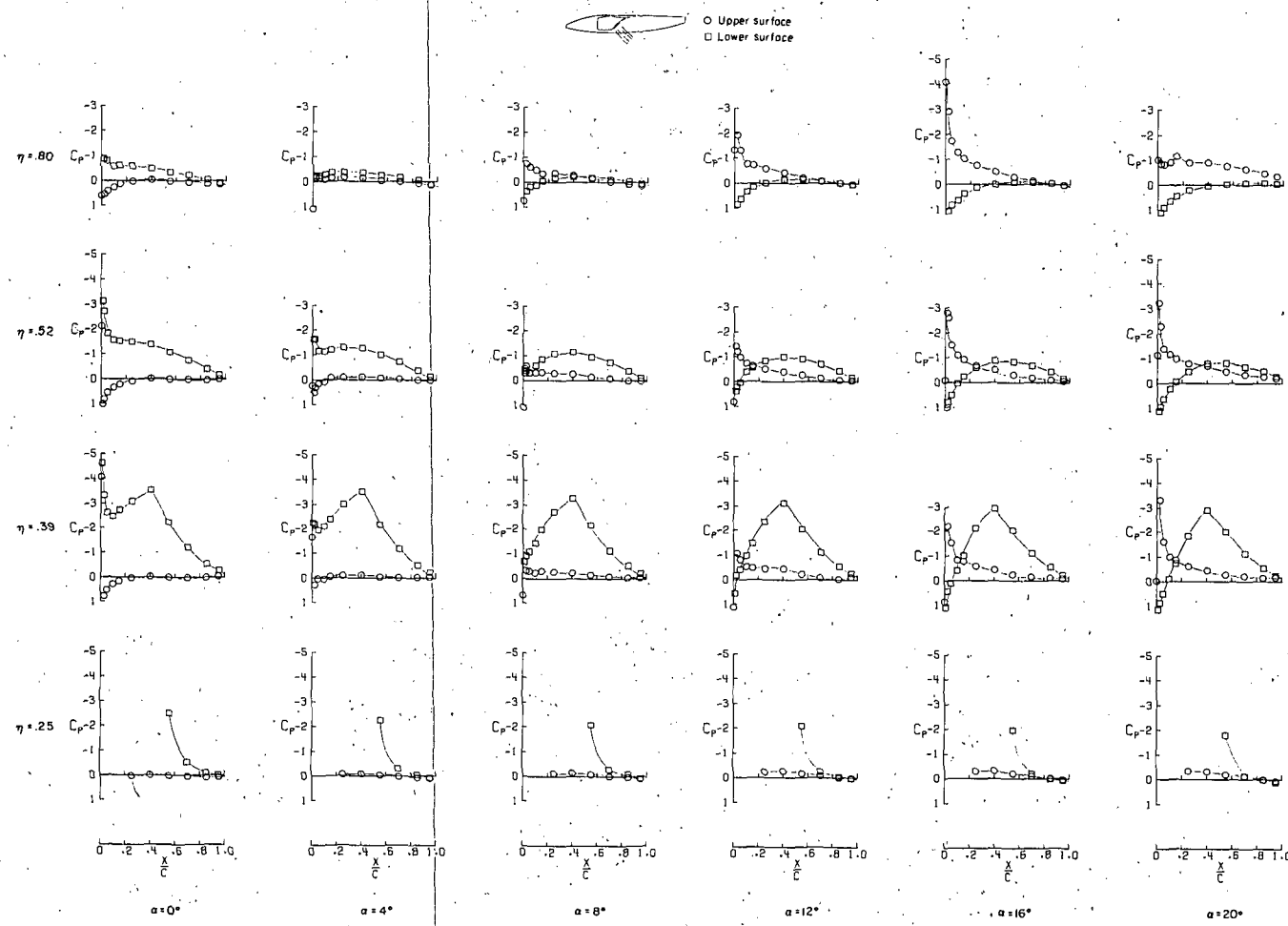
(b) $(V_\infty/V_j)_e \approx 0.3$.

Figure 8.- Continued.



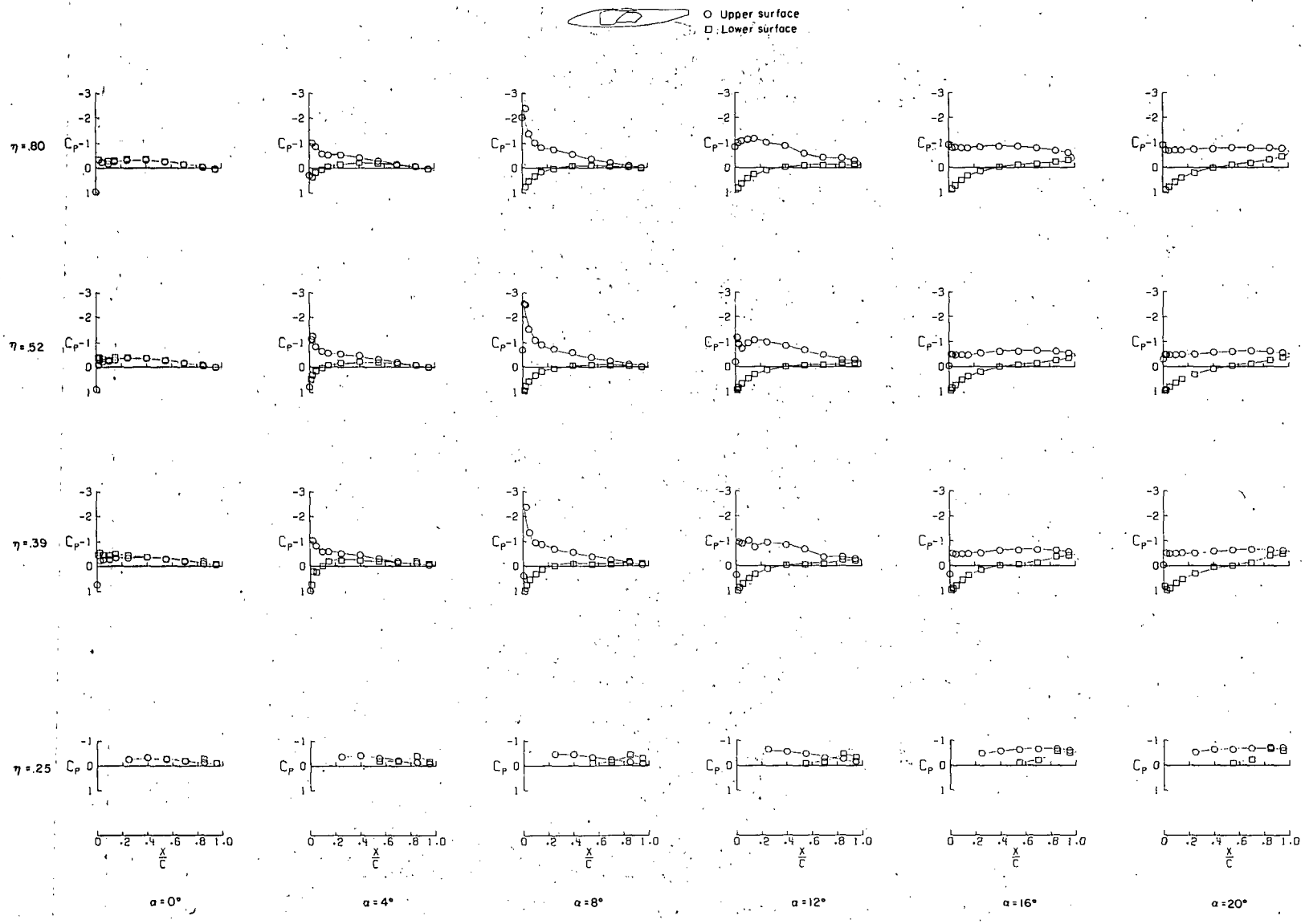
(c) $(V_\infty/V_j)_e \approx 0.2$.

Figure 8.- Continued.



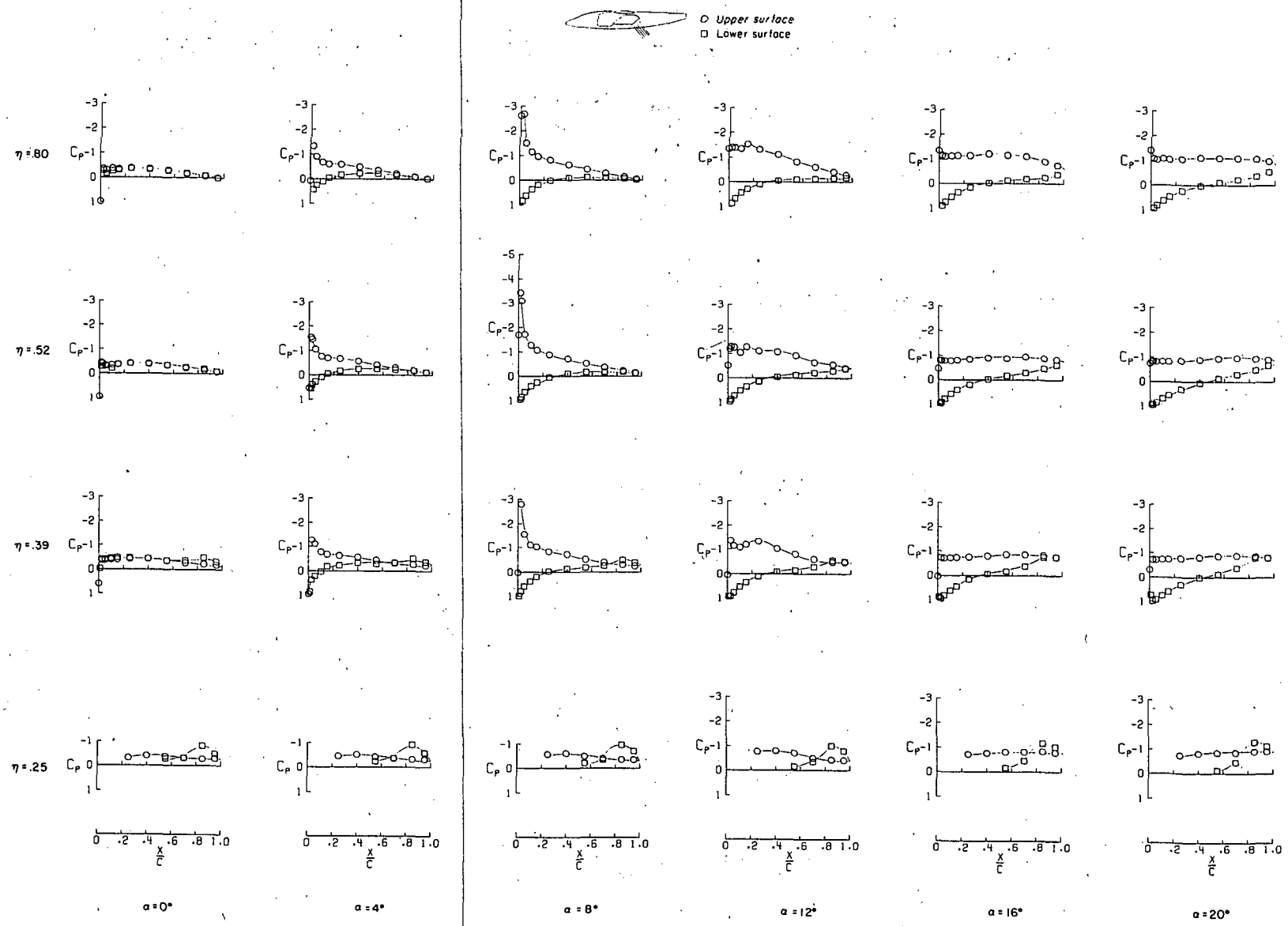
(d) $\left(V_\infty/V_j\right)_e \approx 0.1$.

Figure 8.- Concluded.



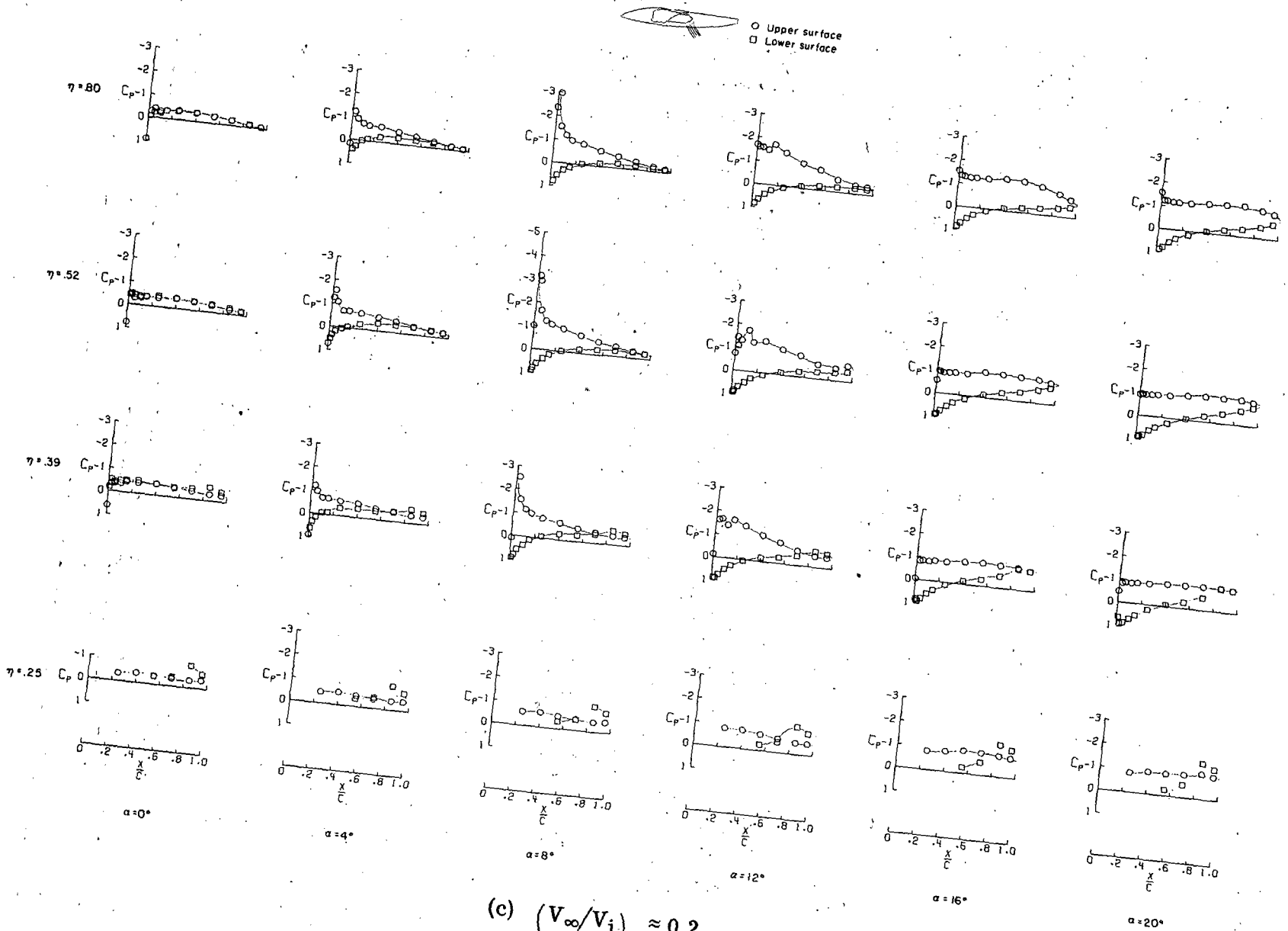
(a) $\left(V_\infty/V_j\right)_e \approx \infty$.

Figure 9.- Pressure distributions on the wing for the rear 45° nozzle configuration.



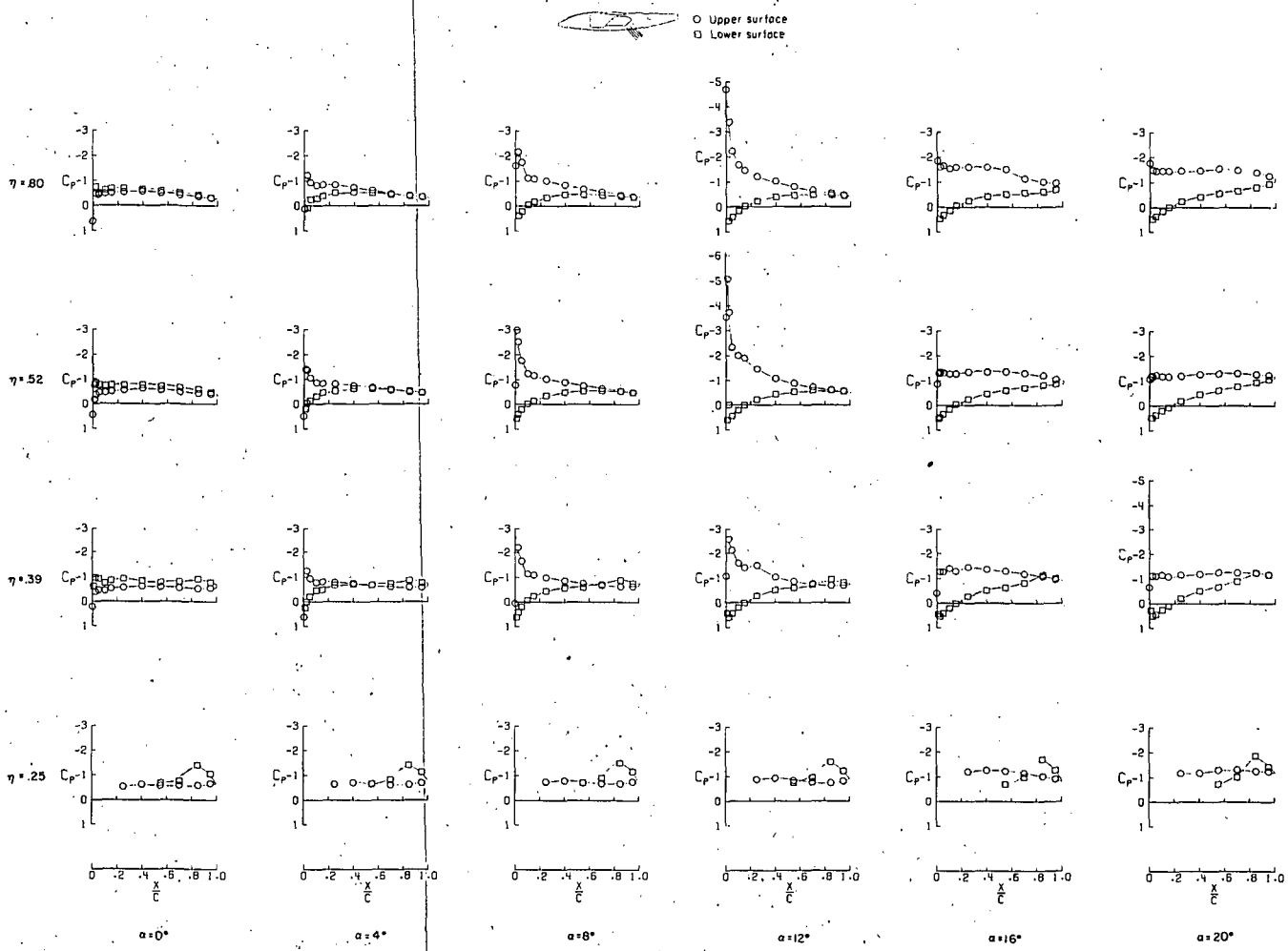
(b) $(V_\infty/V_j)_e \approx 0.3$.

Figure 9.- Continued.



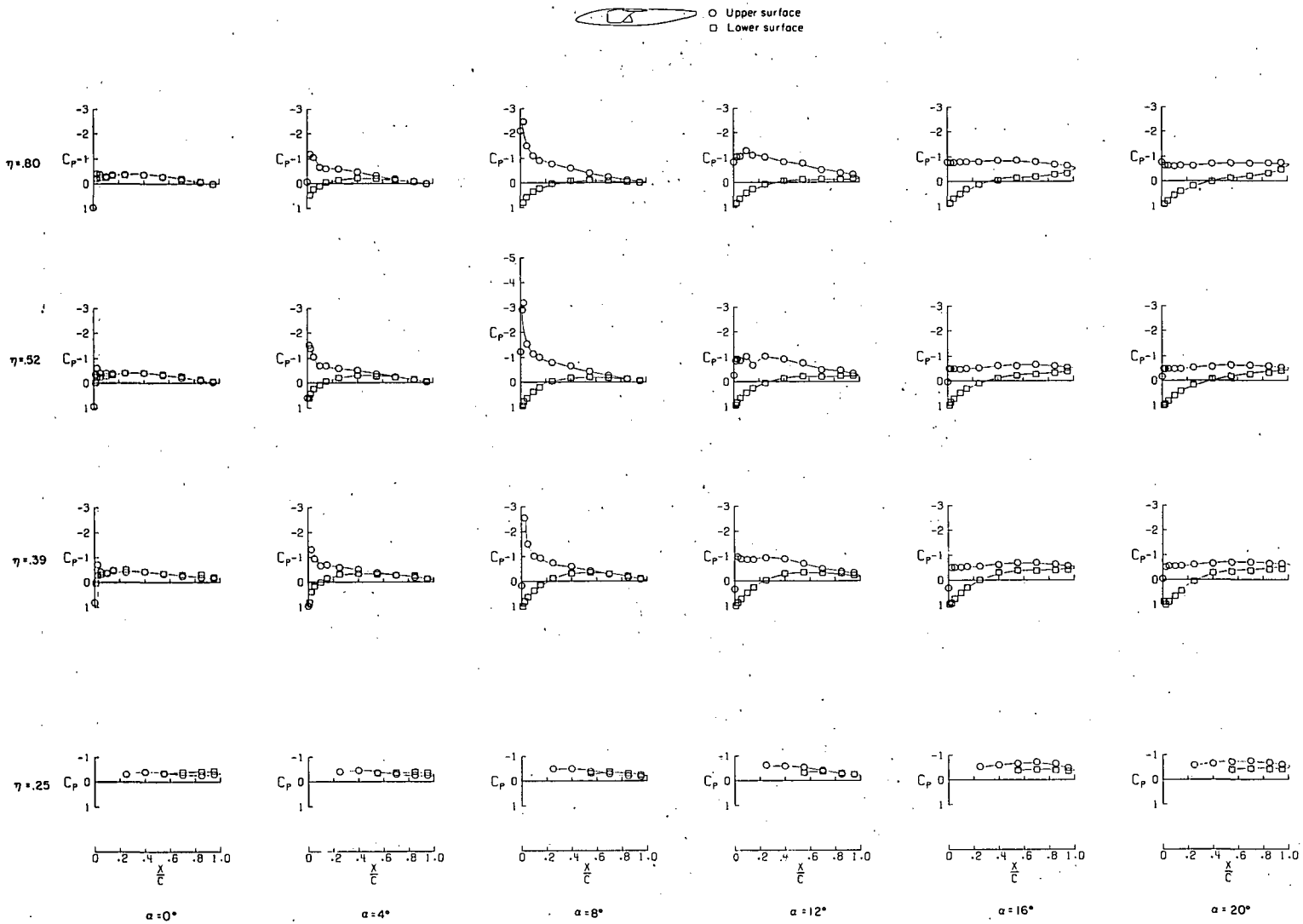
(c) $\left(V_\infty/V_j\right)_e \approx 0.2$.

Figure 9.- Continued.



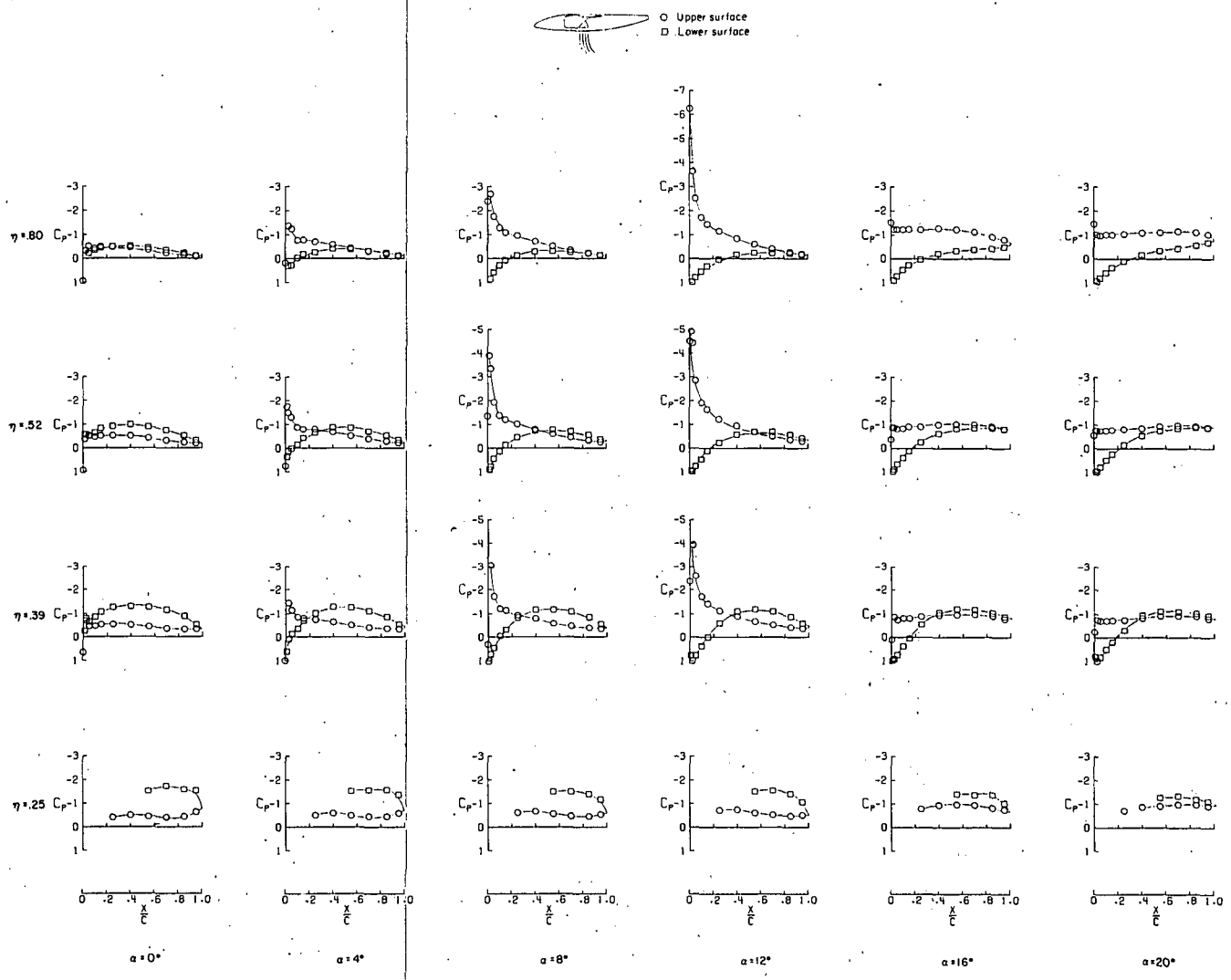
(d) $\left(V_\infty / V_j\right)_e \approx 0.1$.

Figure 9.- Concluded.



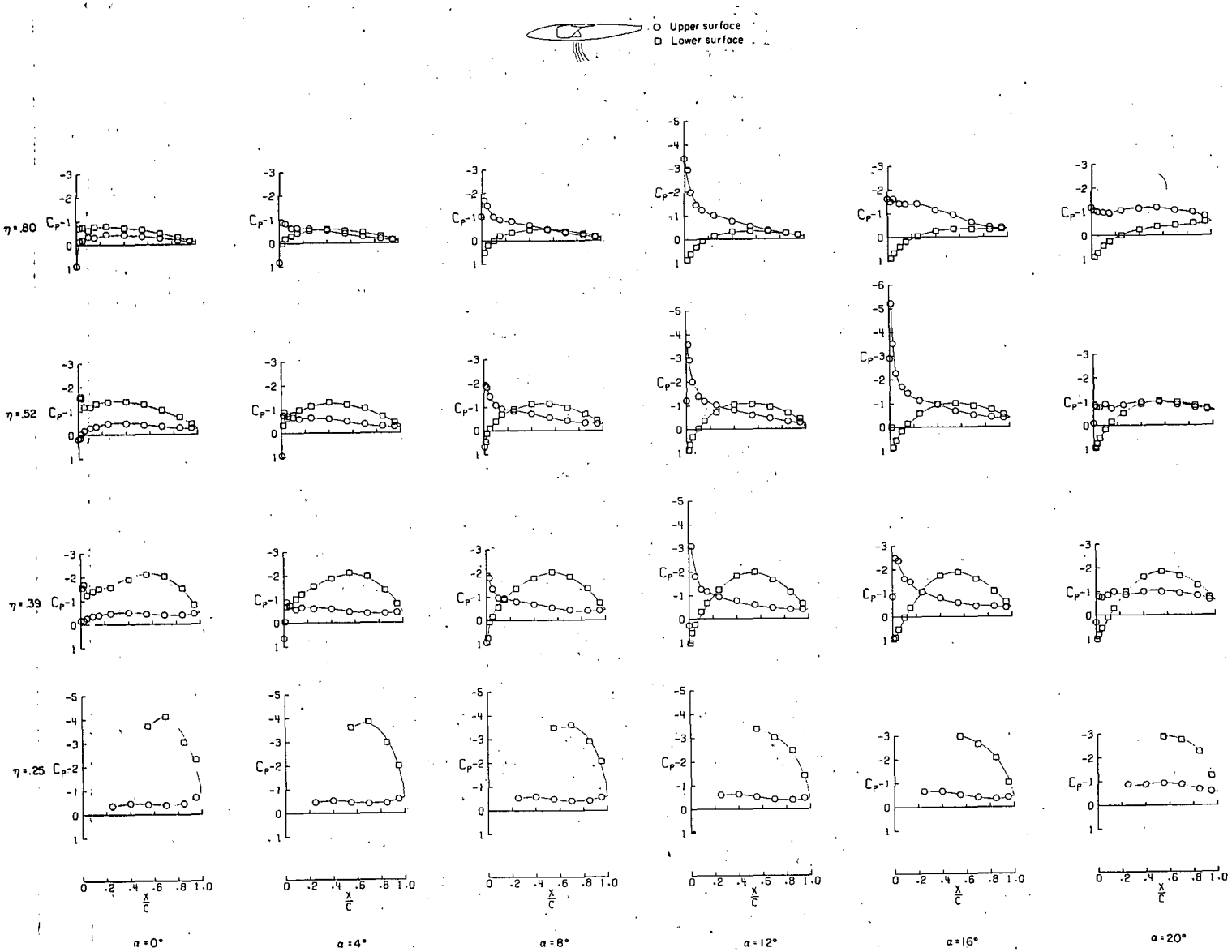
(a) $\left(V_\infty/V_j\right)_e \approx \infty$.

Figure 10.- Pressure distributions on the wing for the forward 90° nozzle configuration.



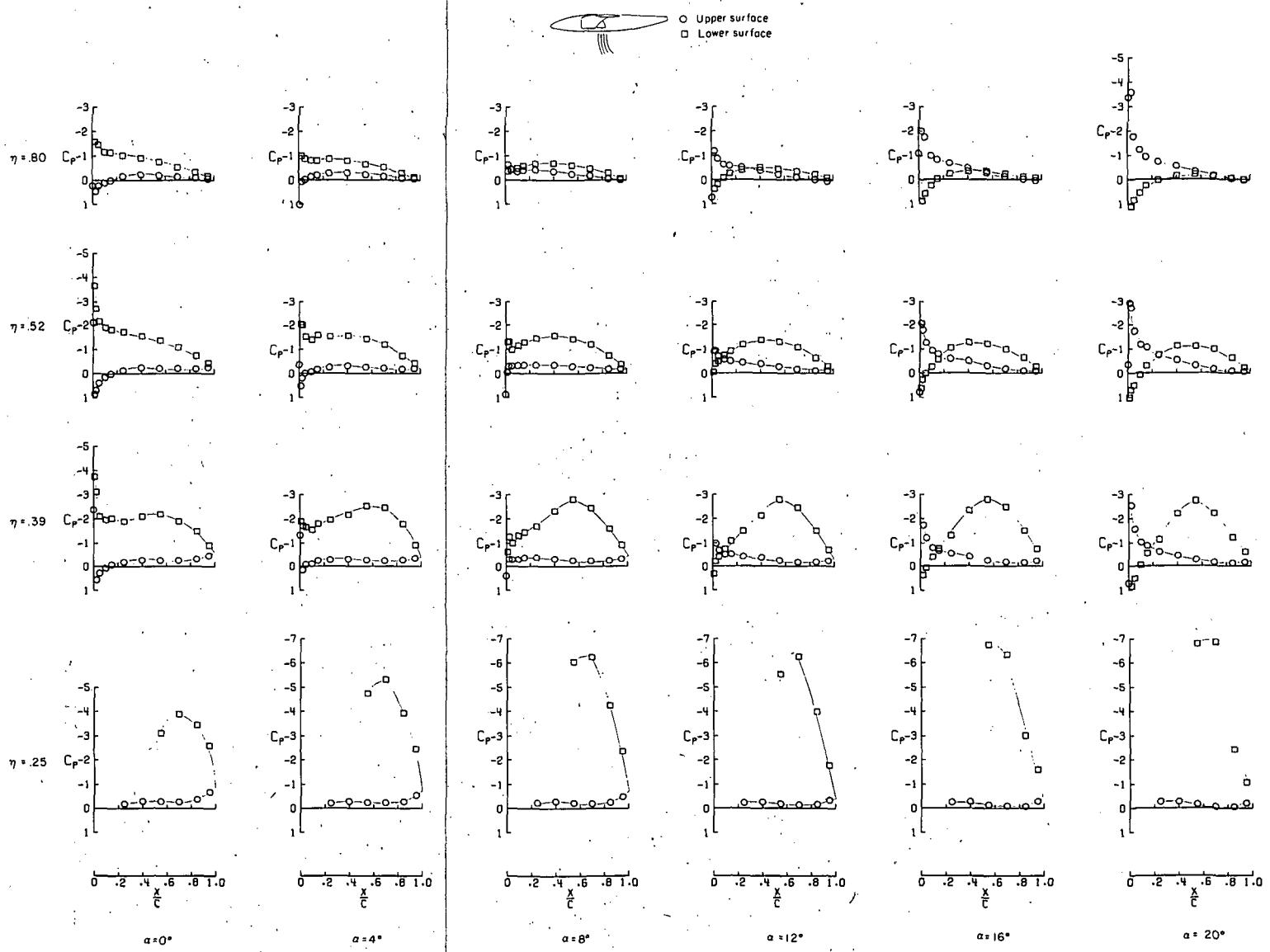
(b) $\left(V_\infty/V_j\right)_e \approx 0.3$.

Figure 10.- Continued.



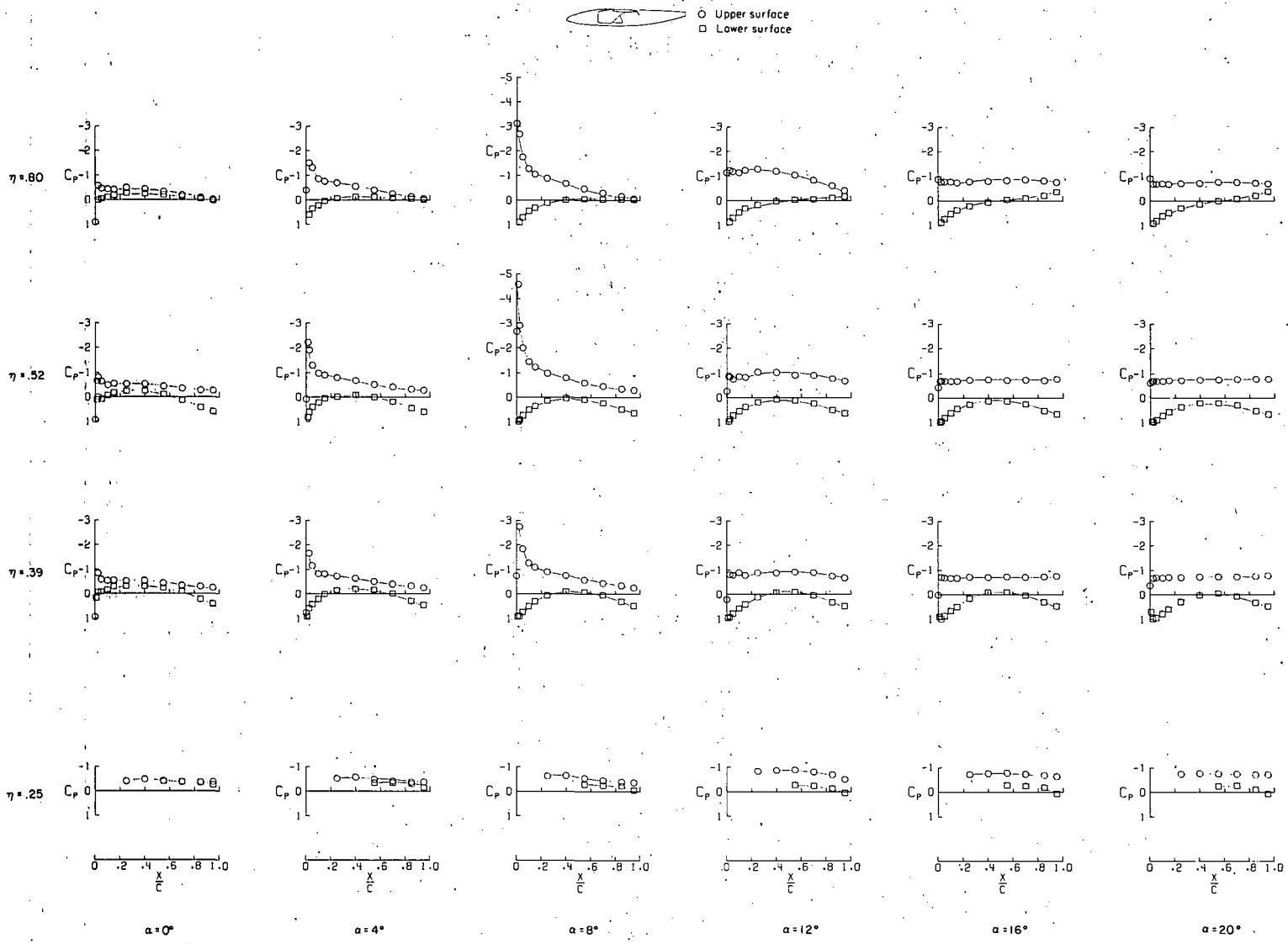
(c) $\left(V_\infty/V_j\right)_e \approx 0.2$.

Figure 10.- Continued.



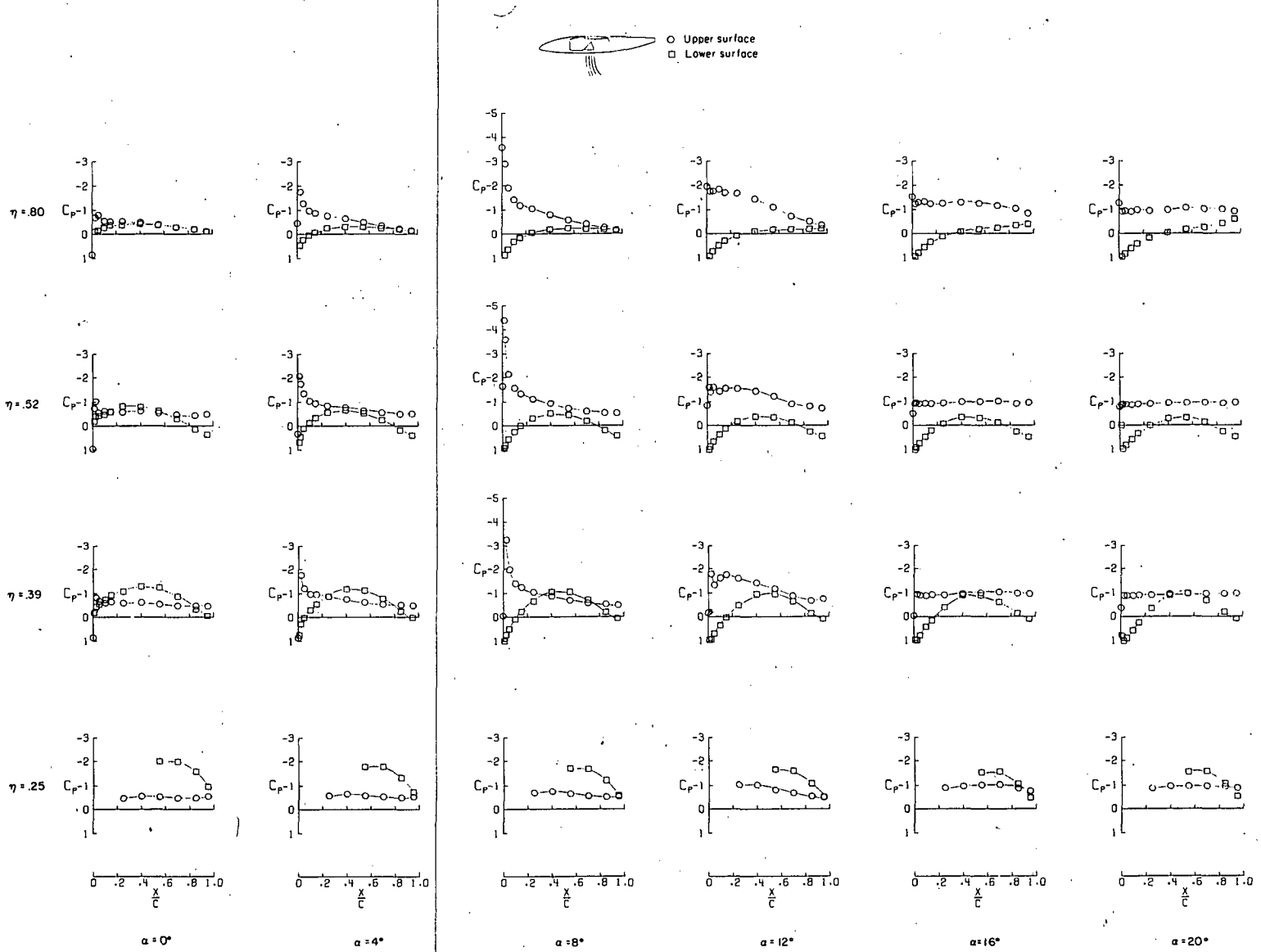
(d) $\left(V_\infty/V_j\right)_e \approx 0.1$.

Figure 10.- Concluded.



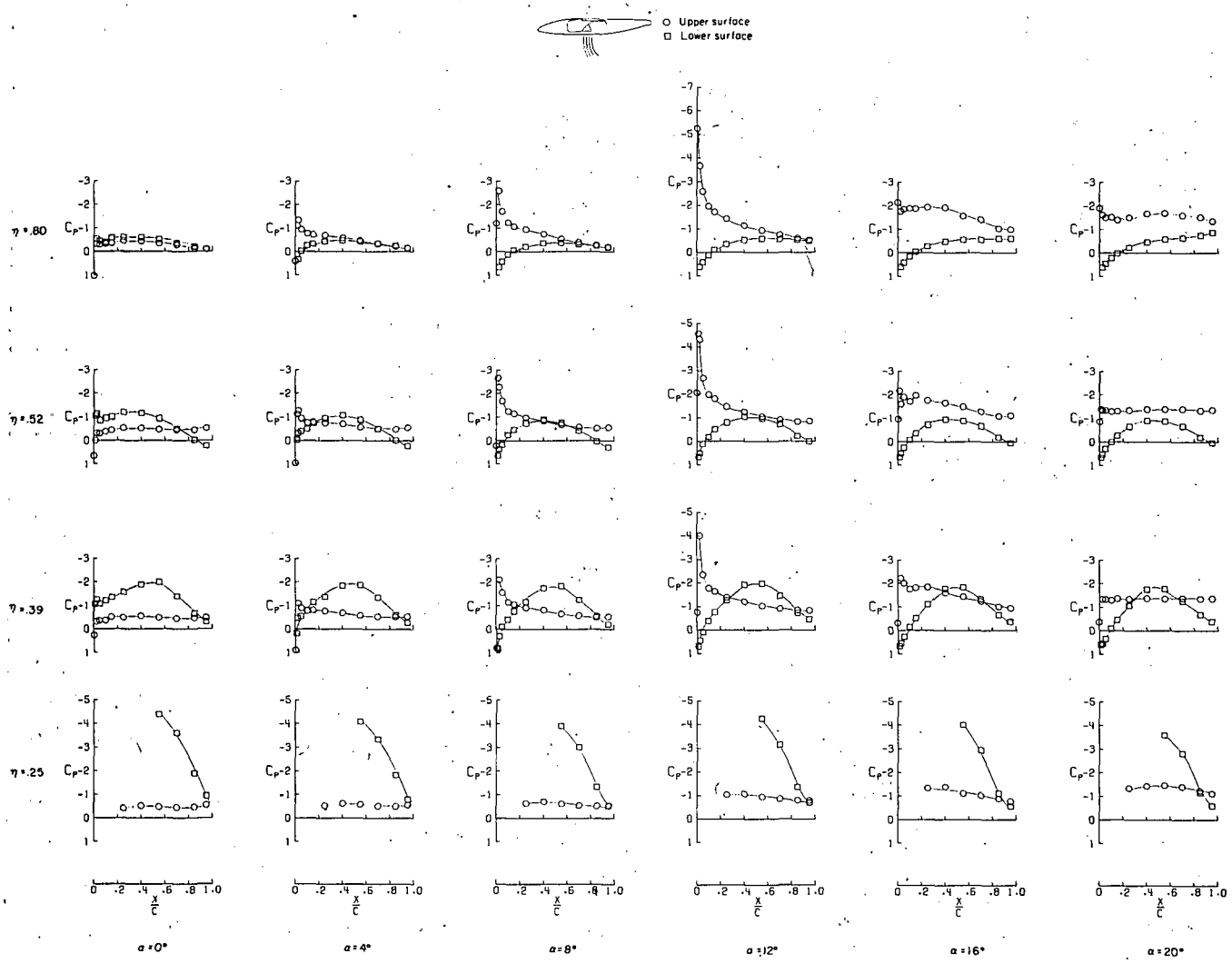
(a) $(V_\infty/V_j)_e \approx \infty$.

Figure 11.- Pressure distributions on the wing with flaps for the forward 90° nozzle configuration.



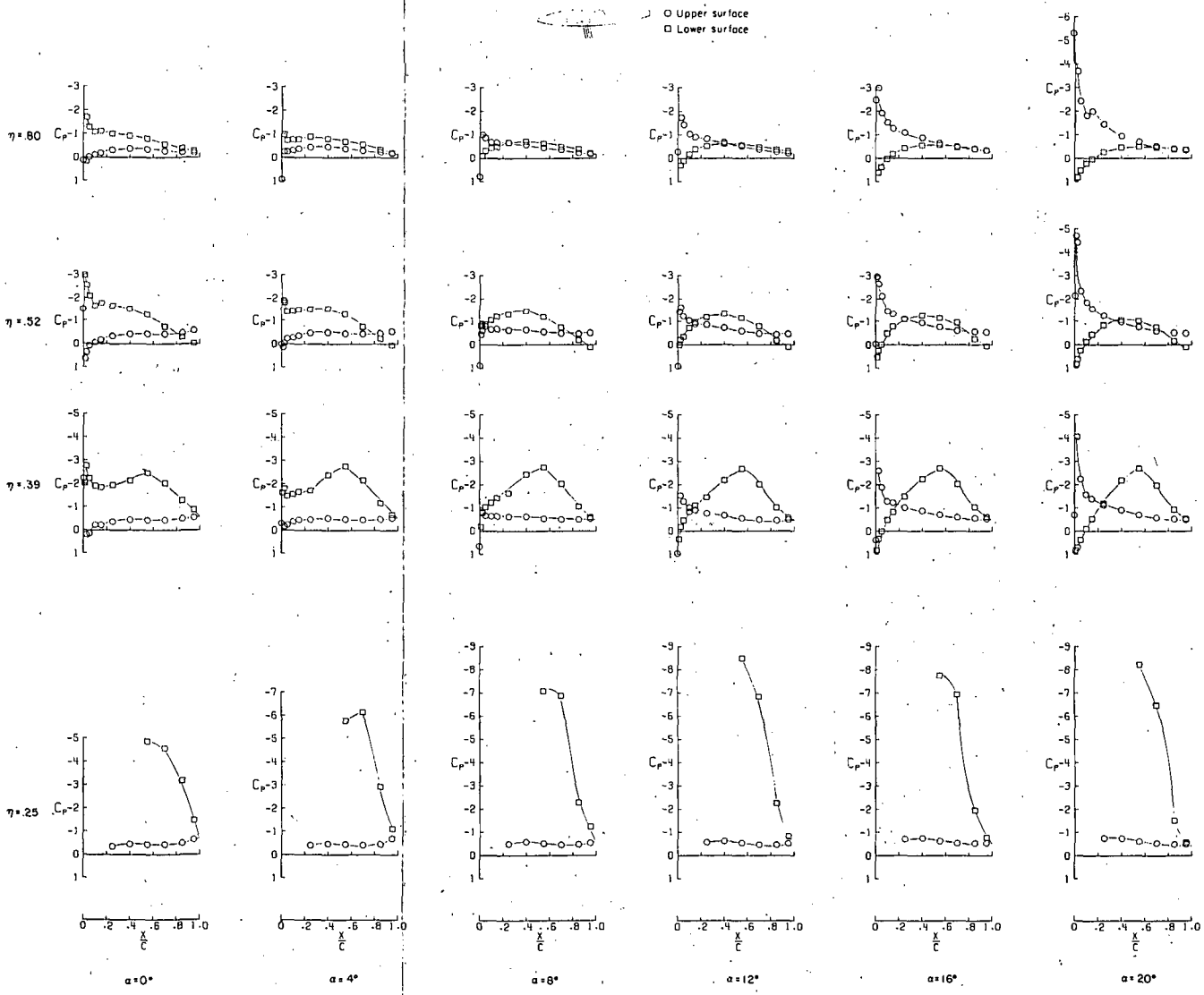
(b) $\left(V_\infty/V_j\right)_e \approx 0.3$.

Figure 11.- Continued.



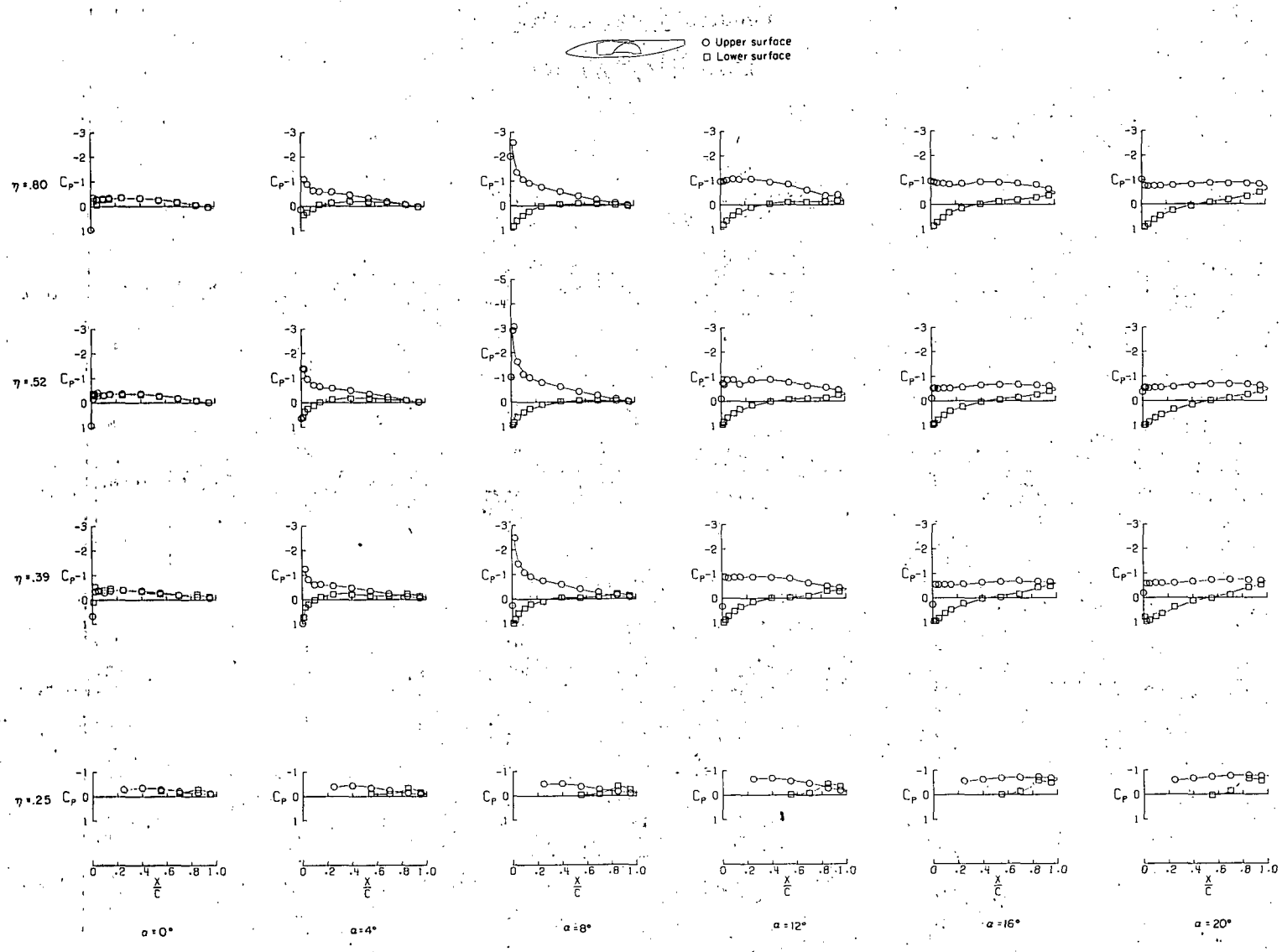
(c) $\left(V_\infty/V_j\right)_e \approx 0.2$.

Figure 11.- Continued.



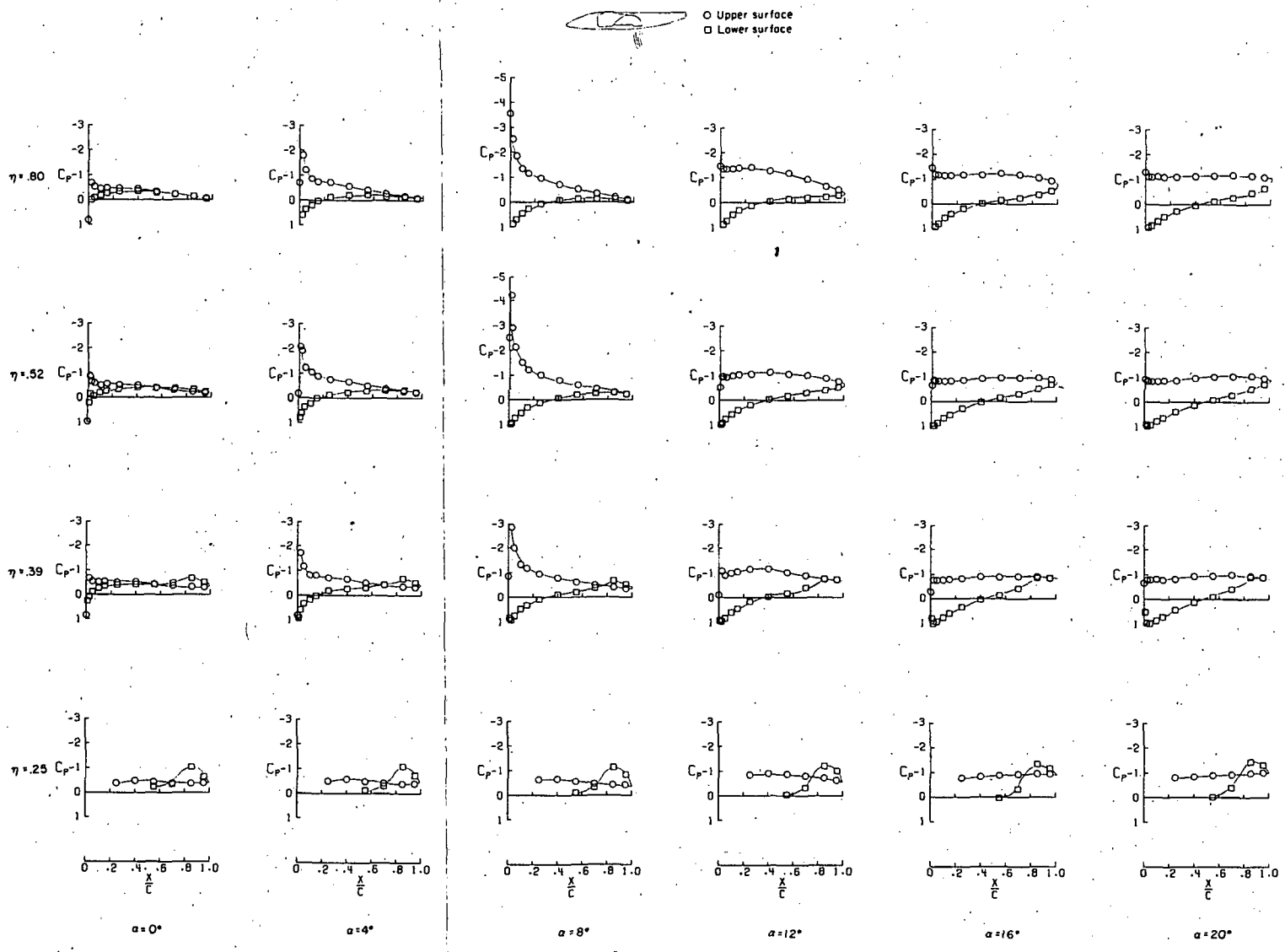
(d) $\left(V_\infty/V_j\right)_e \approx 0.1$.

Figure 11.- Concluded.



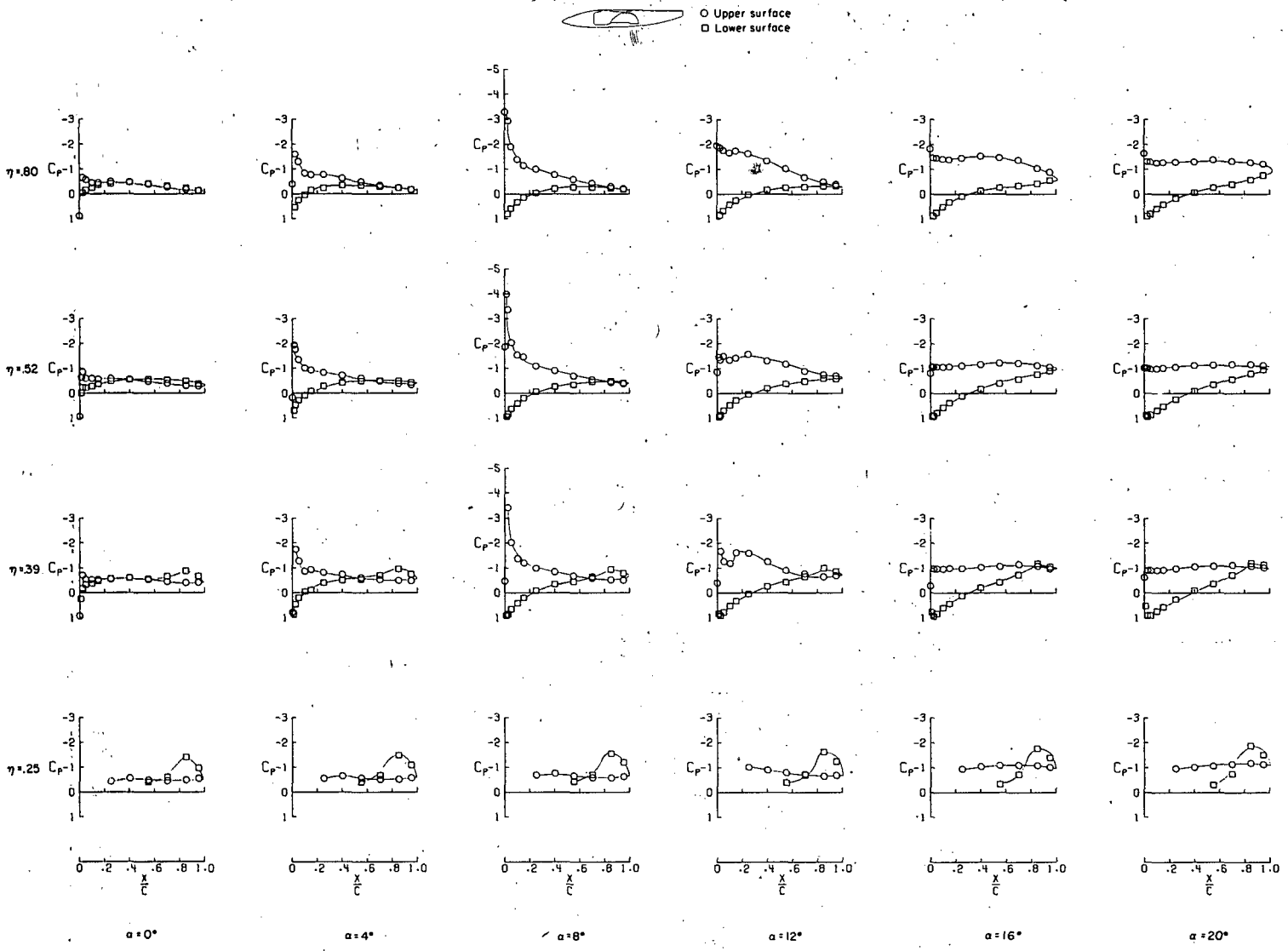
(a) $\left(V_\infty/V_j\right)_e \approx \infty$.

Figure 12.- Pressure distributions on the wing for the rear 90° nozzle configuration.



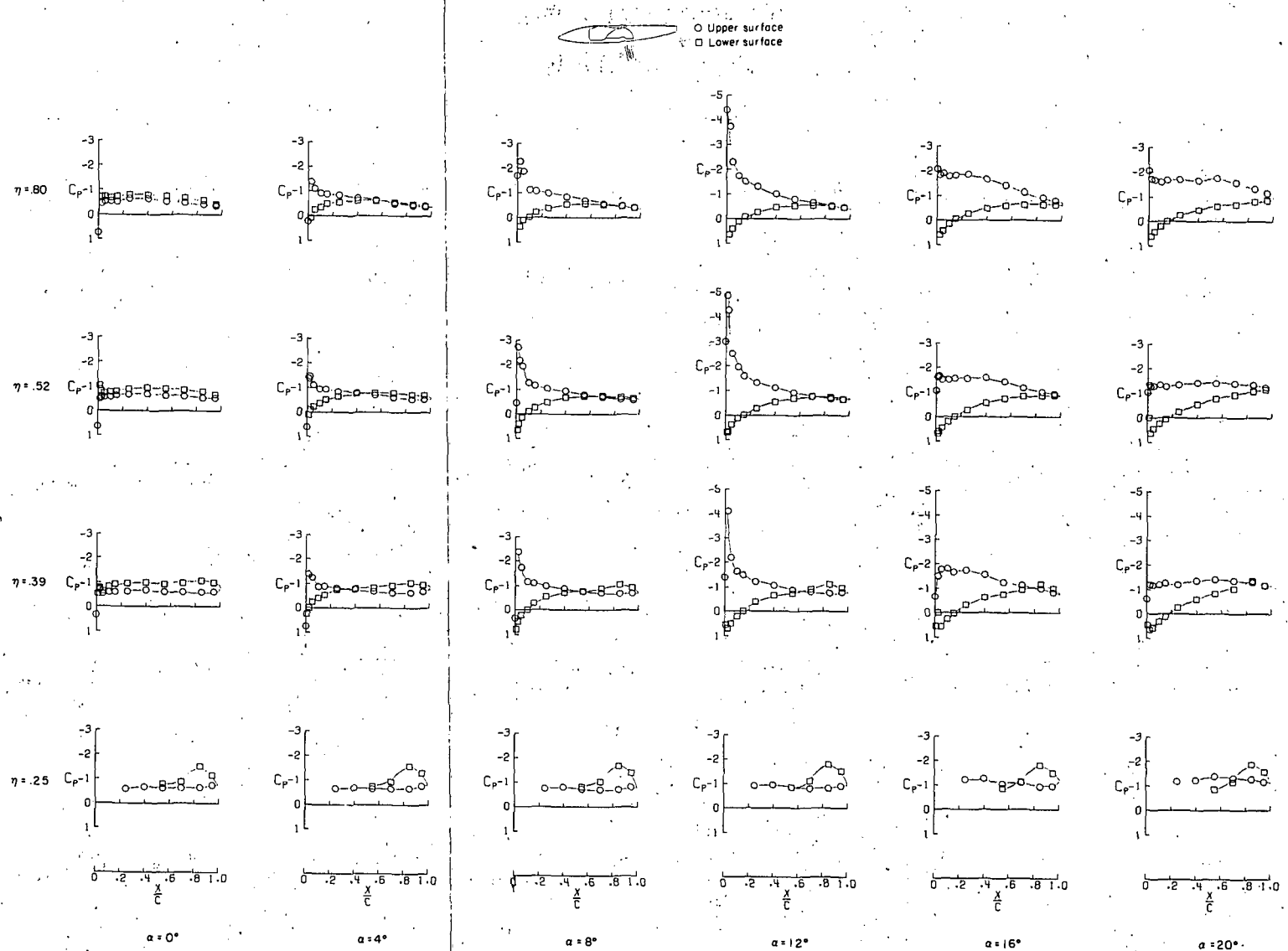
(b) $\left(V_\infty/V_j\right)_e \approx 0.3$.

Figure 12.- Continued.



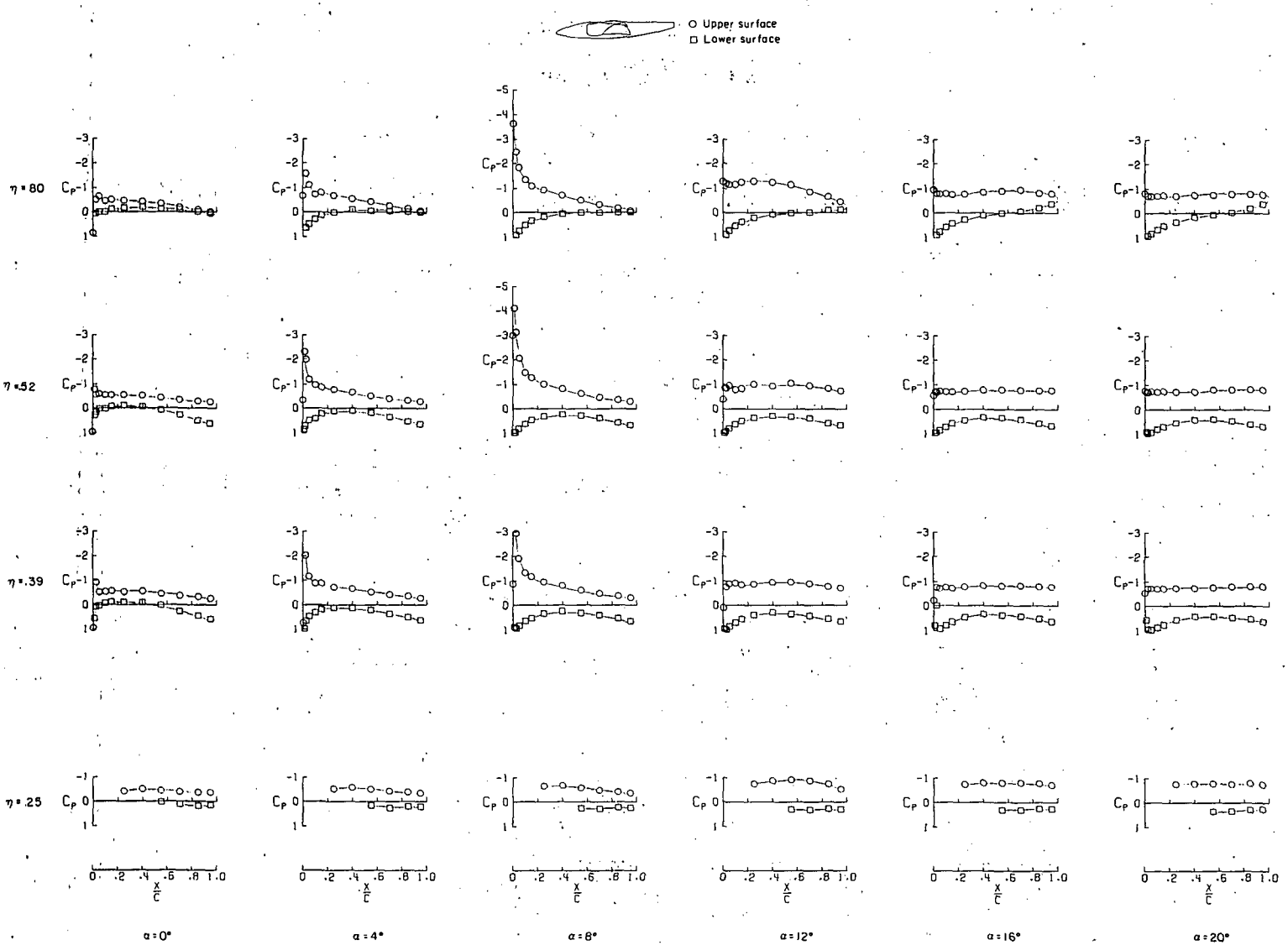
(c) $\left(V_\infty/V_j\right)_e \approx 0.2$.

Figure 12.- Continued.



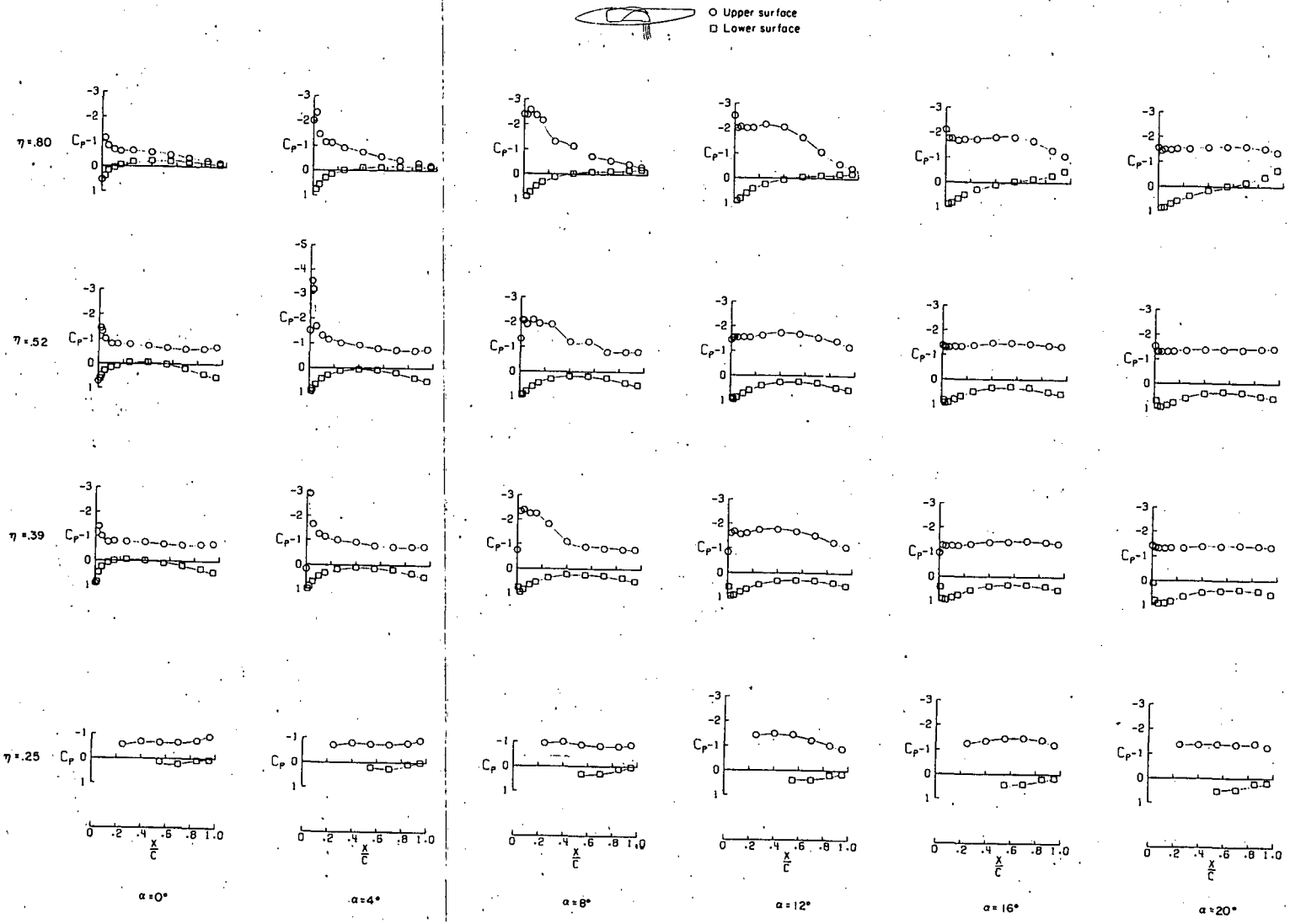
(d) $\left(V_\infty/V_j\right)_e \approx 0.1$.

Figure 12.- Concluded.



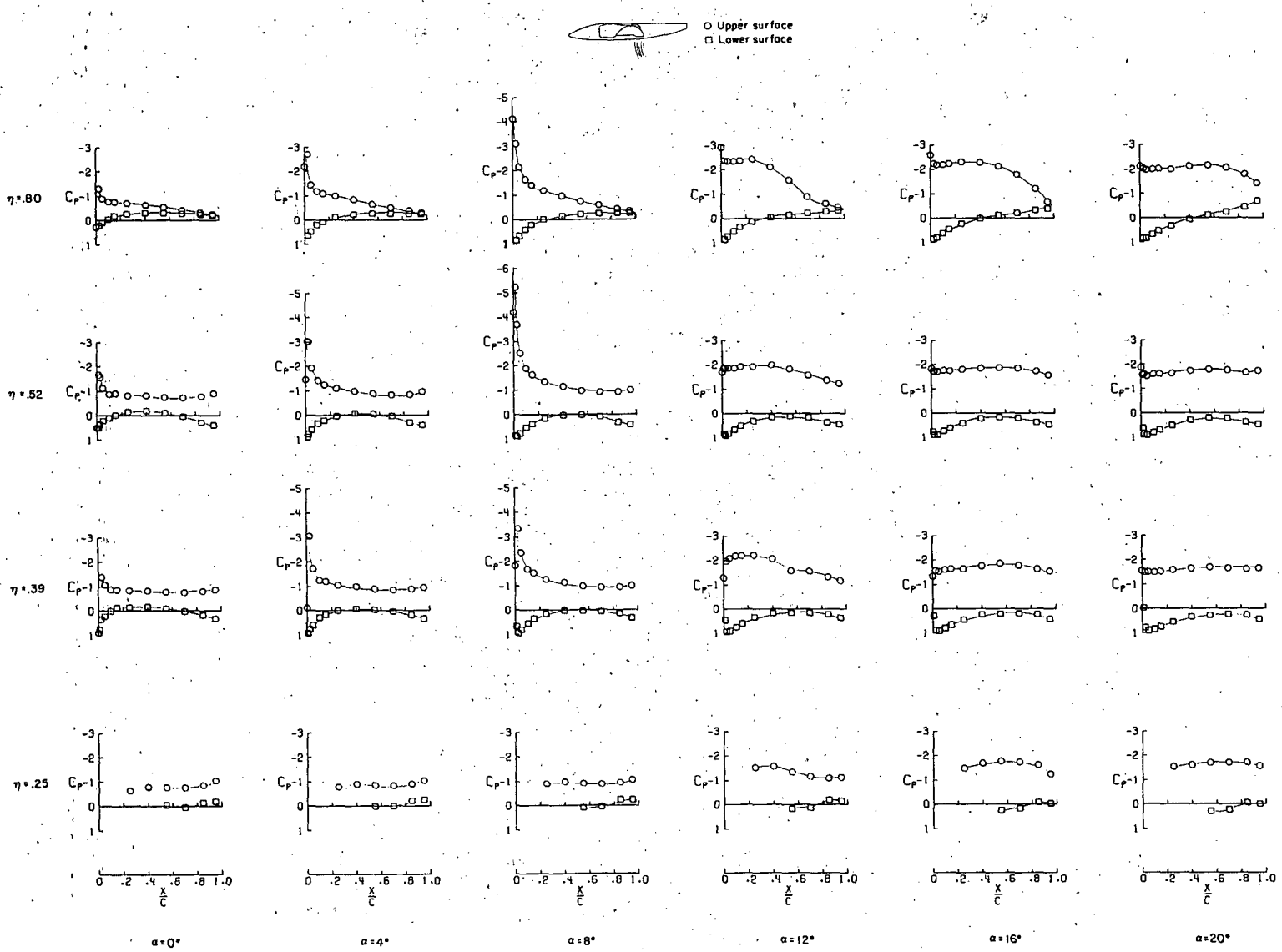
(a) $(V_\infty/V_j)_e \approx \infty$.

Figure 13.- Pressure distributions on the wing with flaps for the rear 90° nozzle configuration.



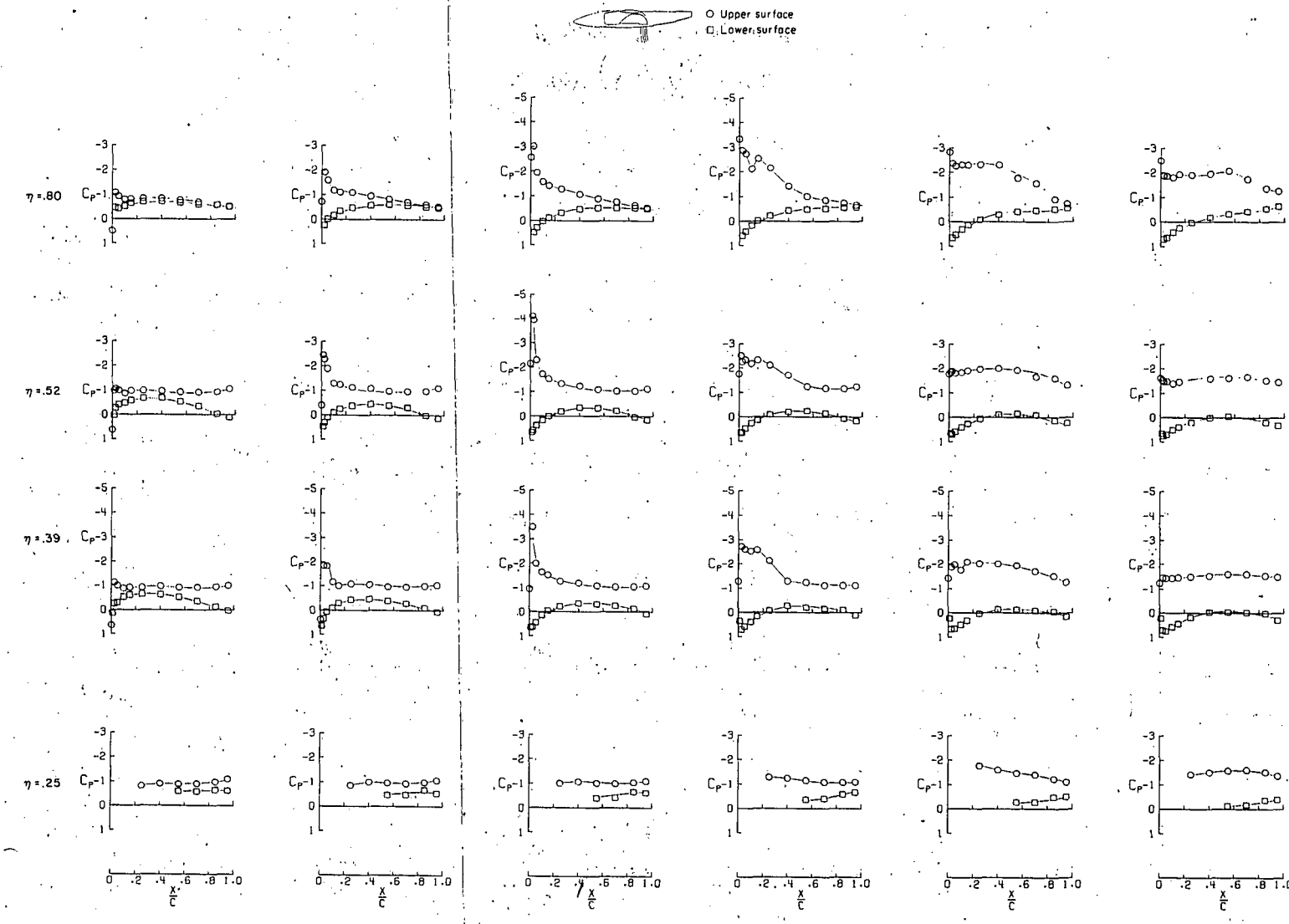
(b) $(V_\infty/V_j)_e \approx 0.3$.

Figure 13.- Continued.



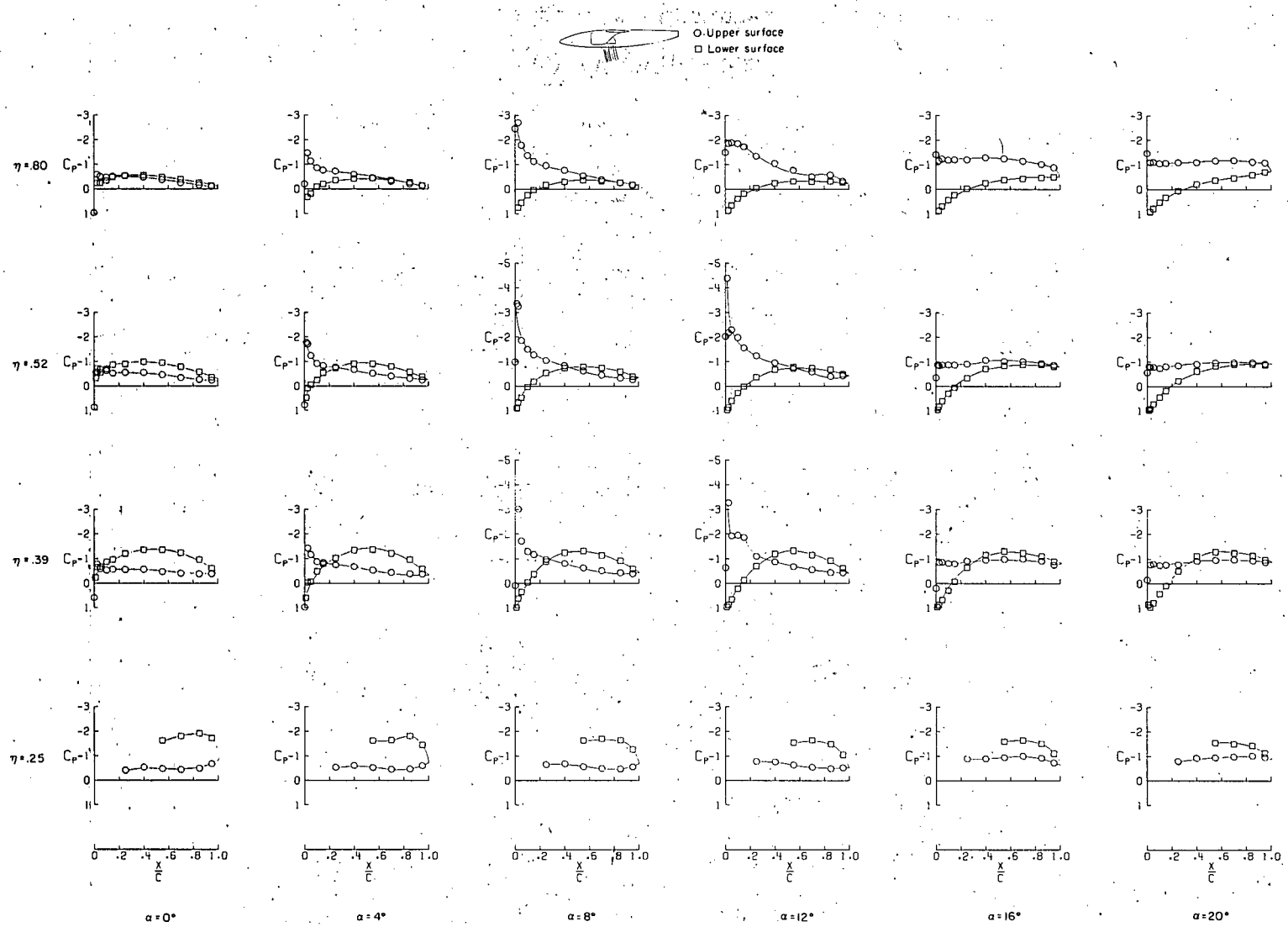
(c) $\left(V_\infty/V_j\right)_e \approx 0.2$.

Figure 13.- Continued.



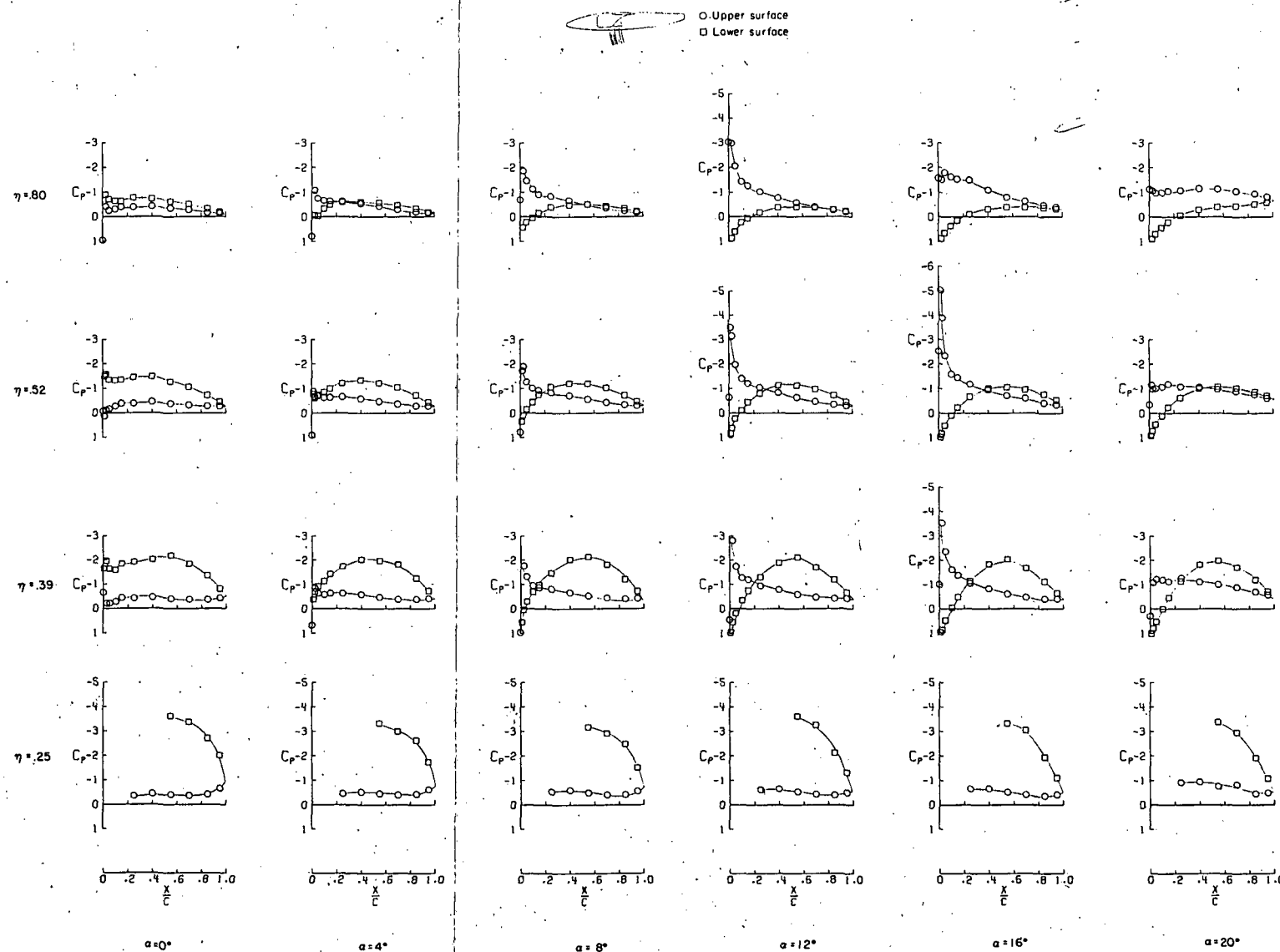
(d) $\left(V_\infty/V_j\right)_e \approx 0.1$.

Figure 13.- Concluded.



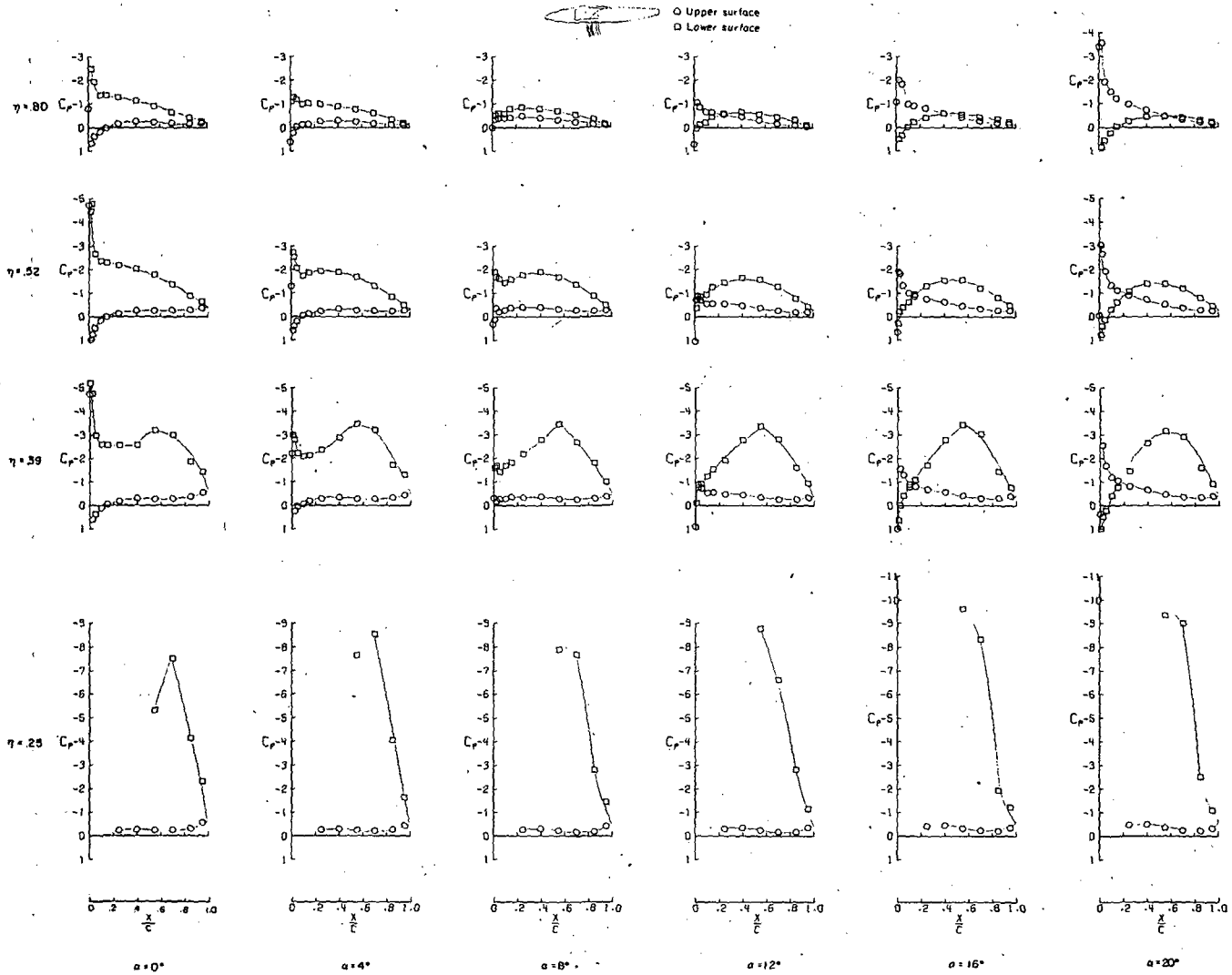
(a) $\left(V_\infty/V_j\right)_e \approx 0.3$.

Figure 14.- Pressure distributions on the wing for the lift-jet and forward 90° nozzle configuration.



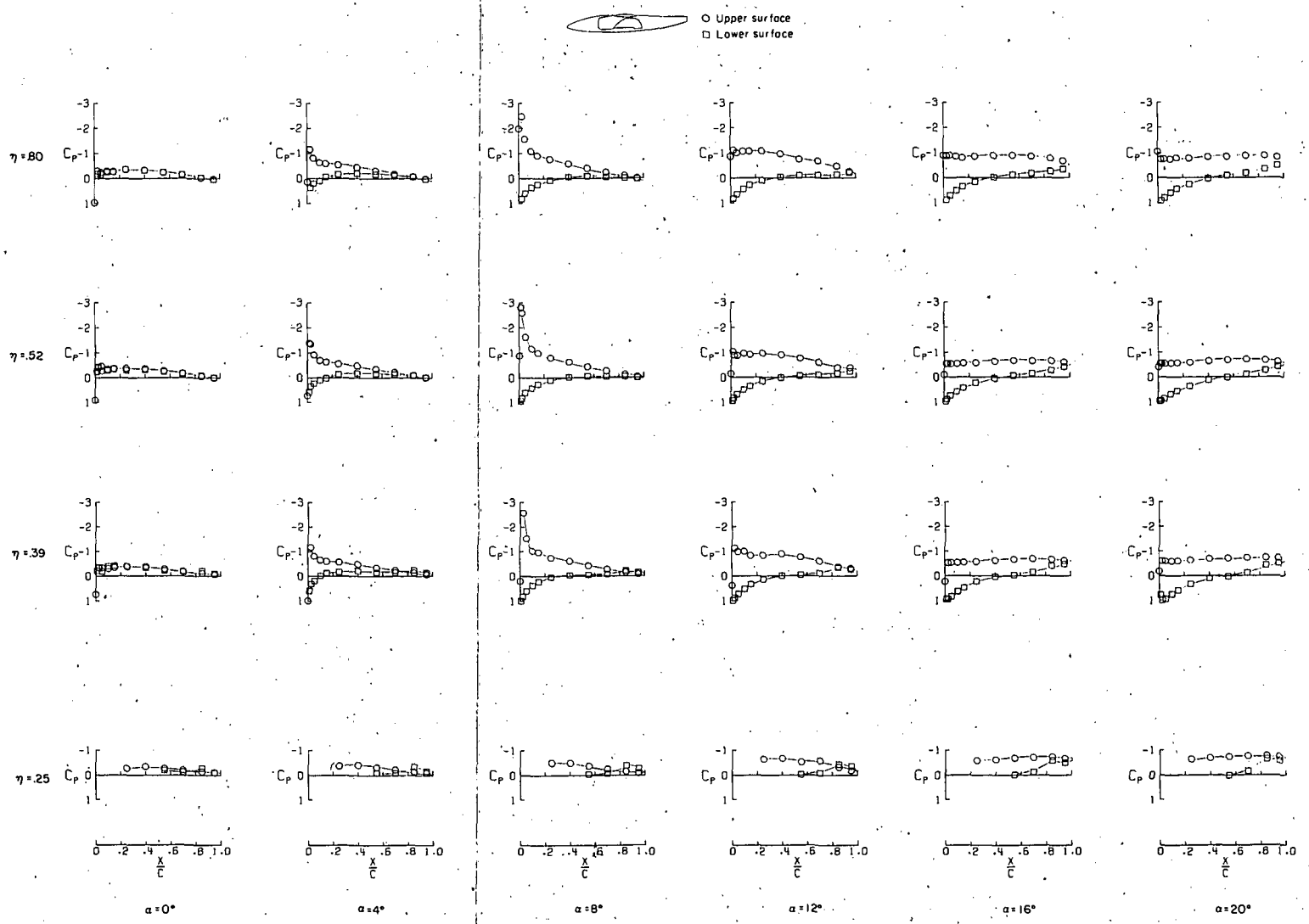
(b) $\left(V_\infty/V_j\right)_e \approx 0.2$.

Figure 14.- Continued.



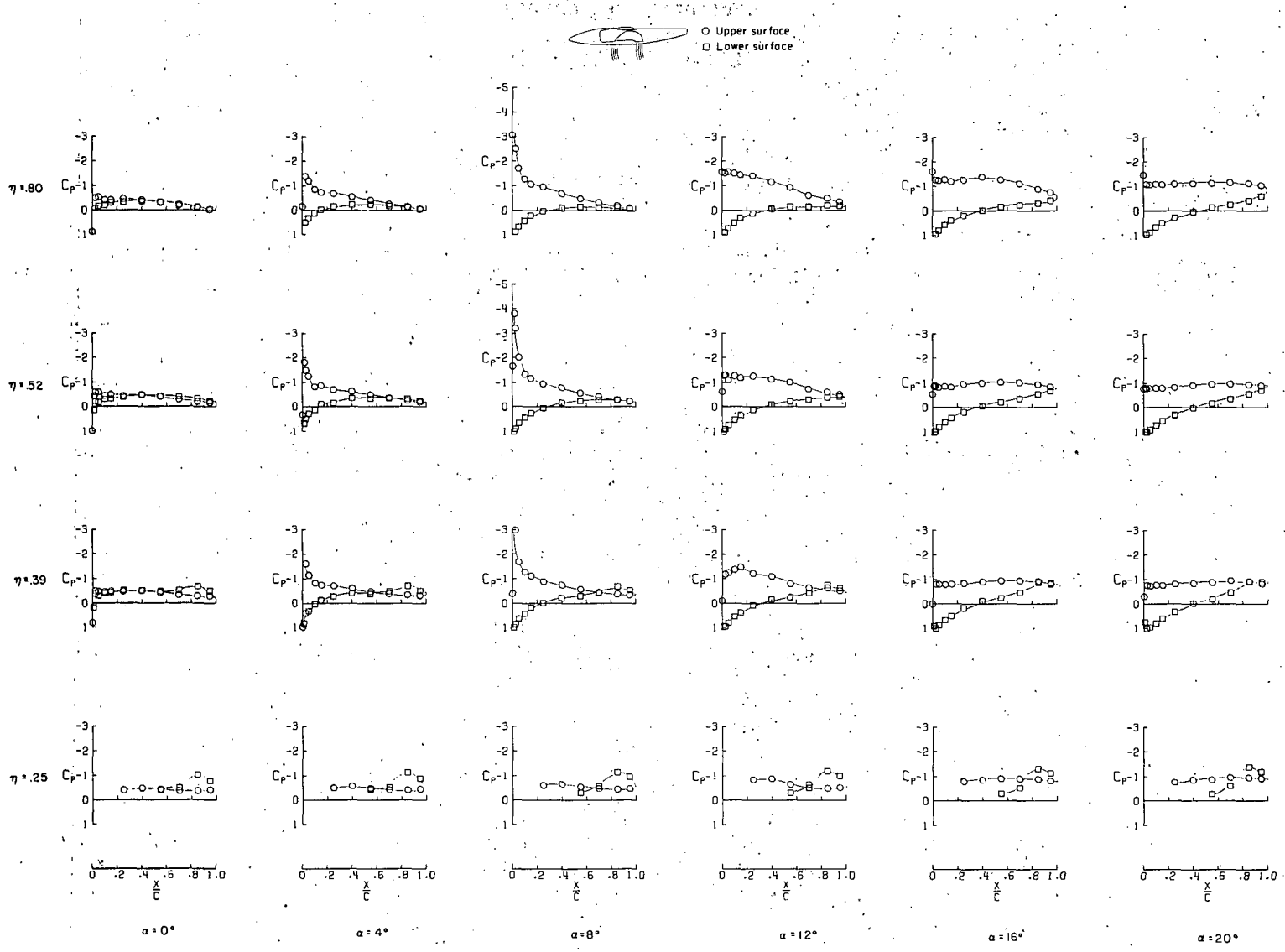
(c) $(V_\infty/V_j)_e \approx 0.1$.

Figure 14.- Concluded.



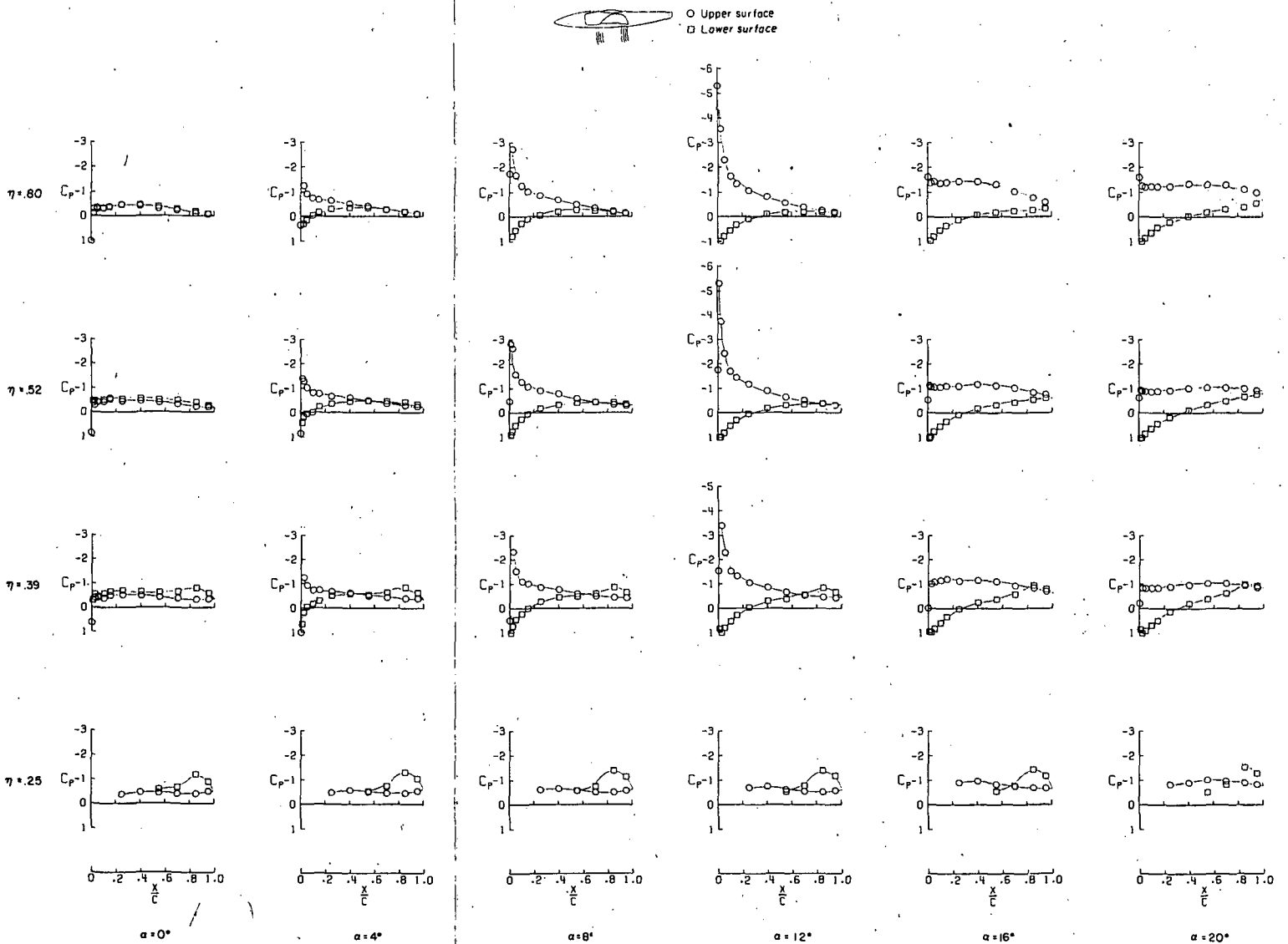
(a) $\left(V_\infty/V_j\right)_e \approx \infty$.

Figure 15.- Pressure distributions on the wing for the lift-jet and rear 90° nozzle configuration.



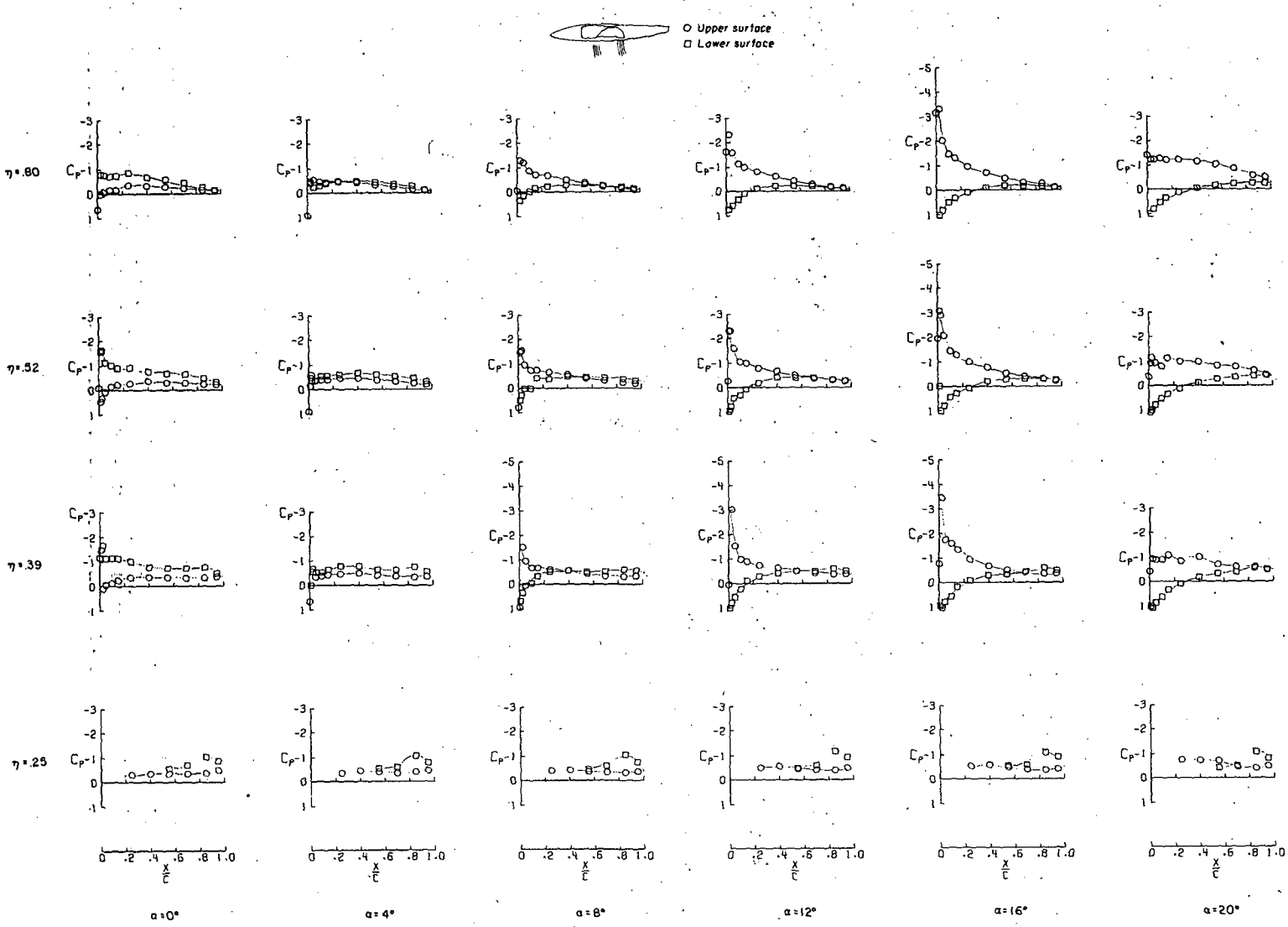
(b) $\left(V_\infty/V_j\right)_e \approx 0.3$.

Figure 15.- Continued.



(c) $\left(V_\infty/V_j\right)_e \approx 0.2$.

Figure 15.- Continued.



(d) $\left(V_\infty/V_j\right)_e \approx 0.1$.

Figure 15.- Concluded.

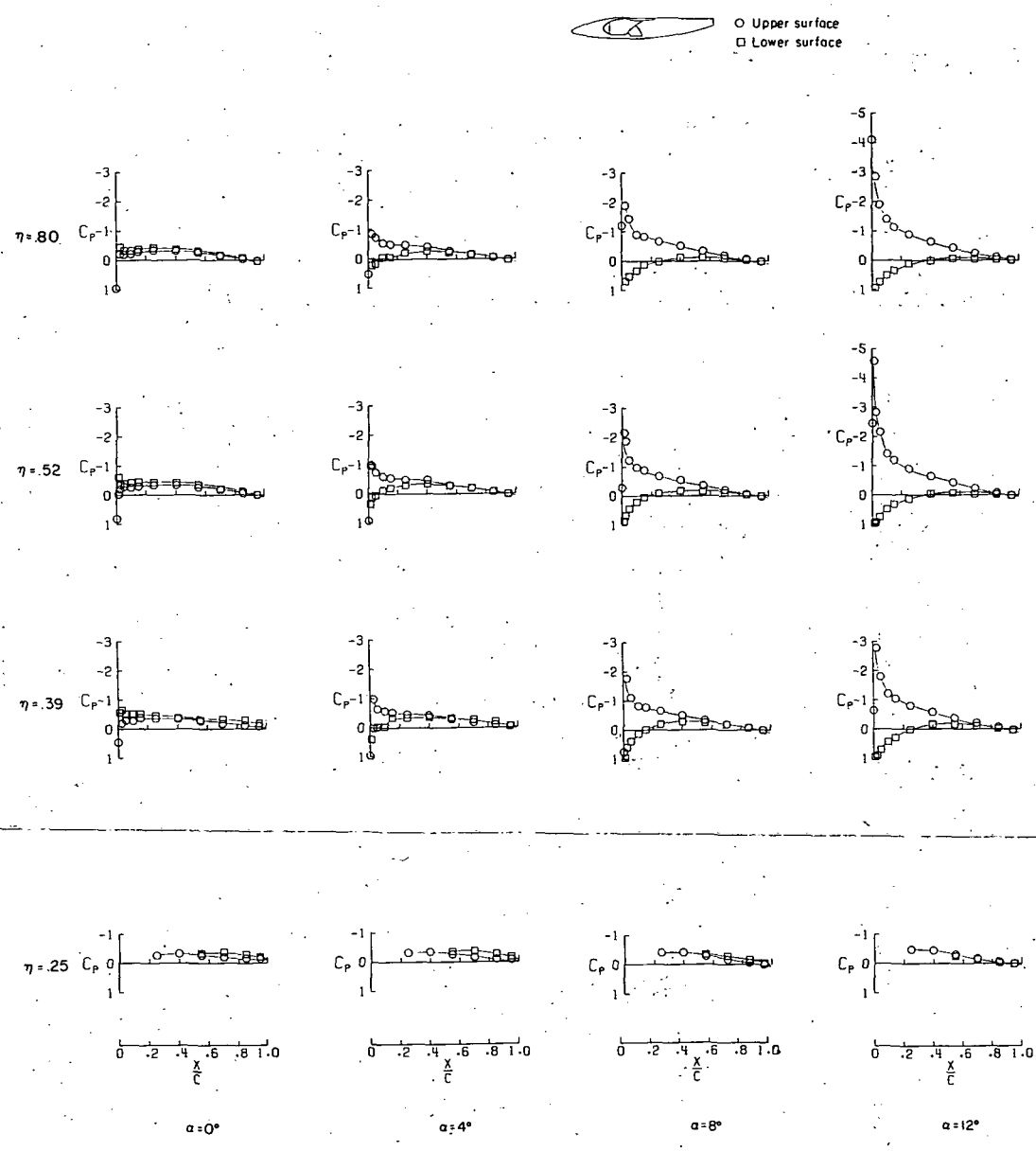
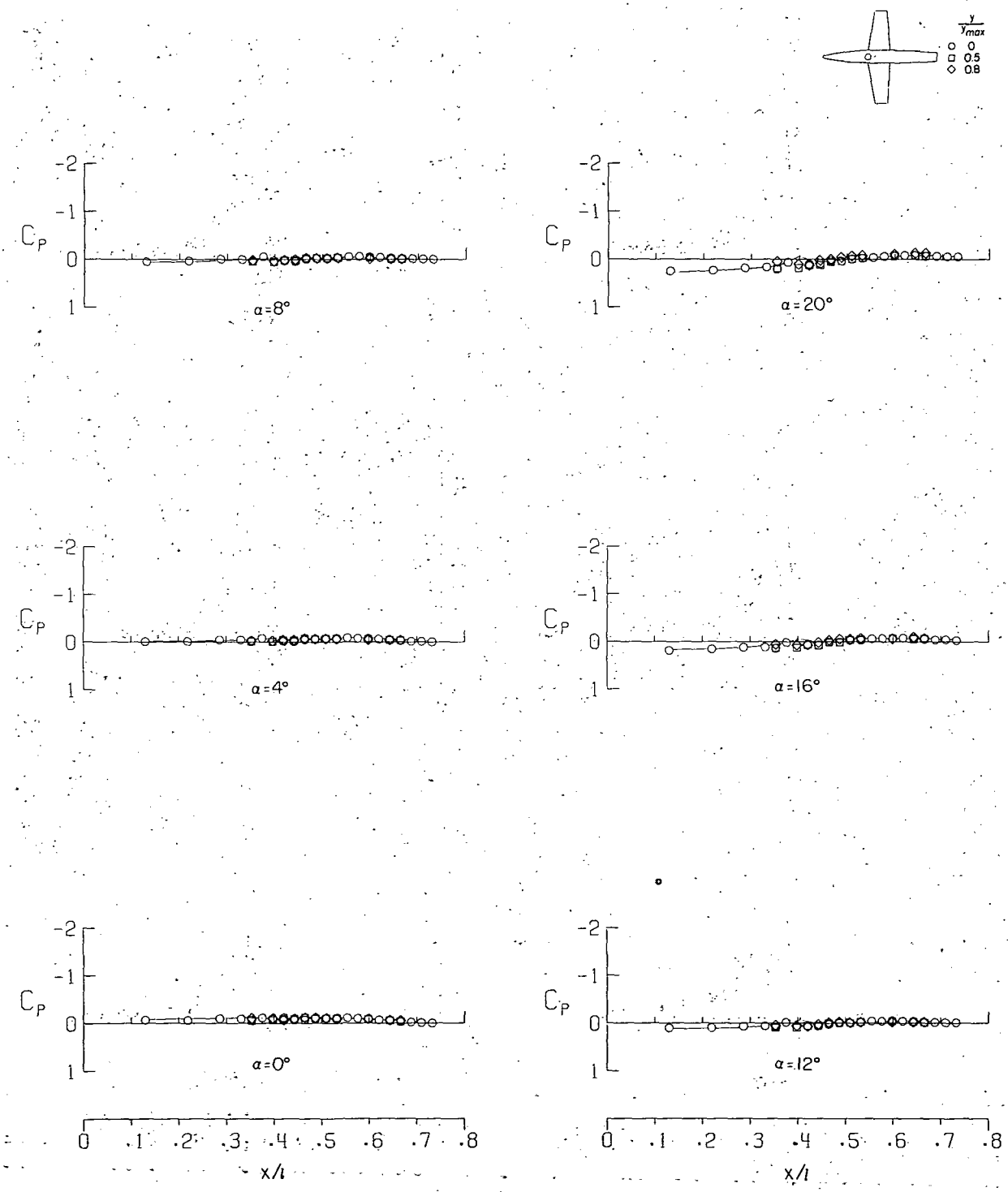
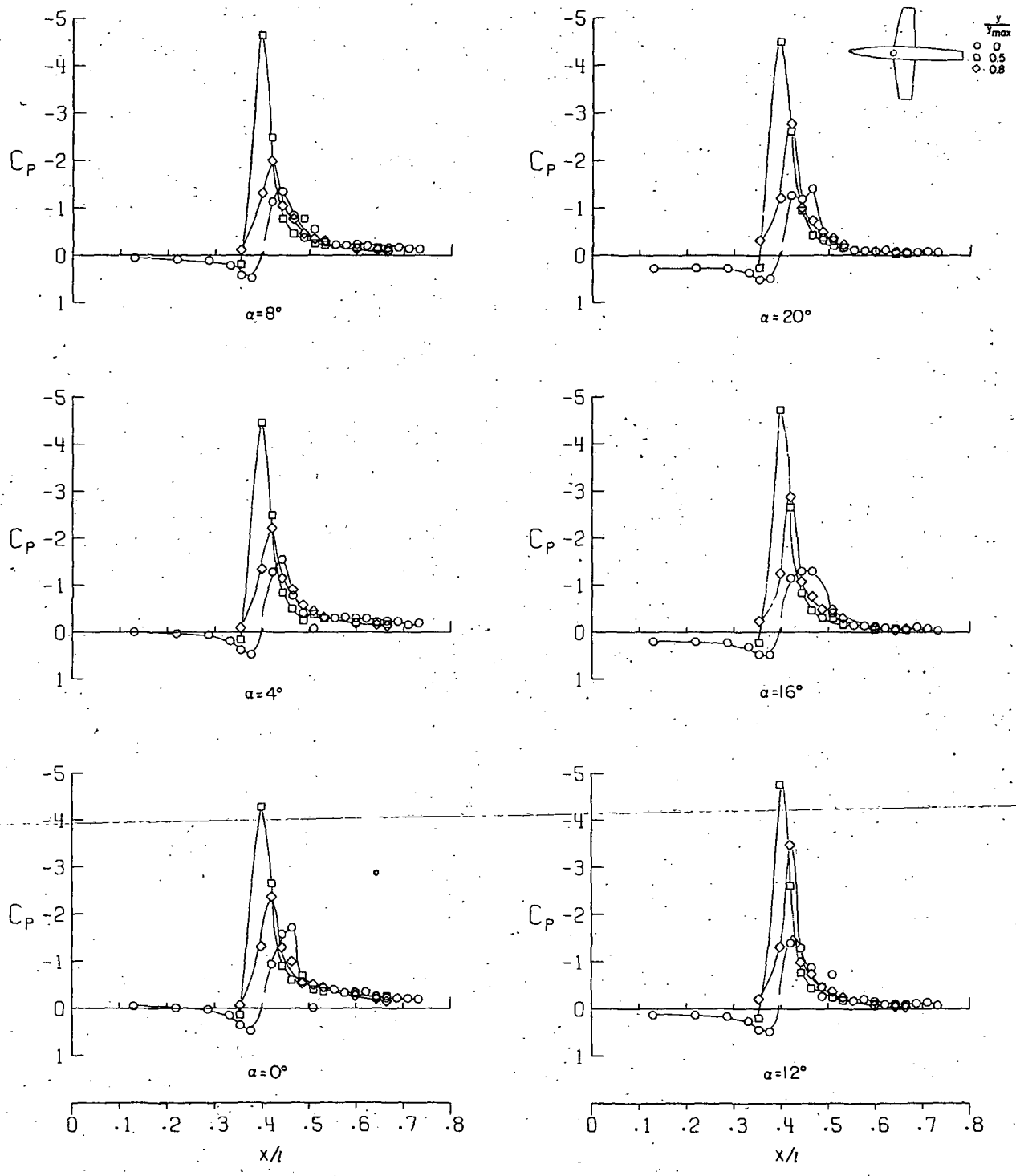


Figure 16.- Pressure distributions on the wing for the forward 90° nozzle configuration with inlets closed. $\left(V_\infty/V_j\right)_e \approx \infty$.



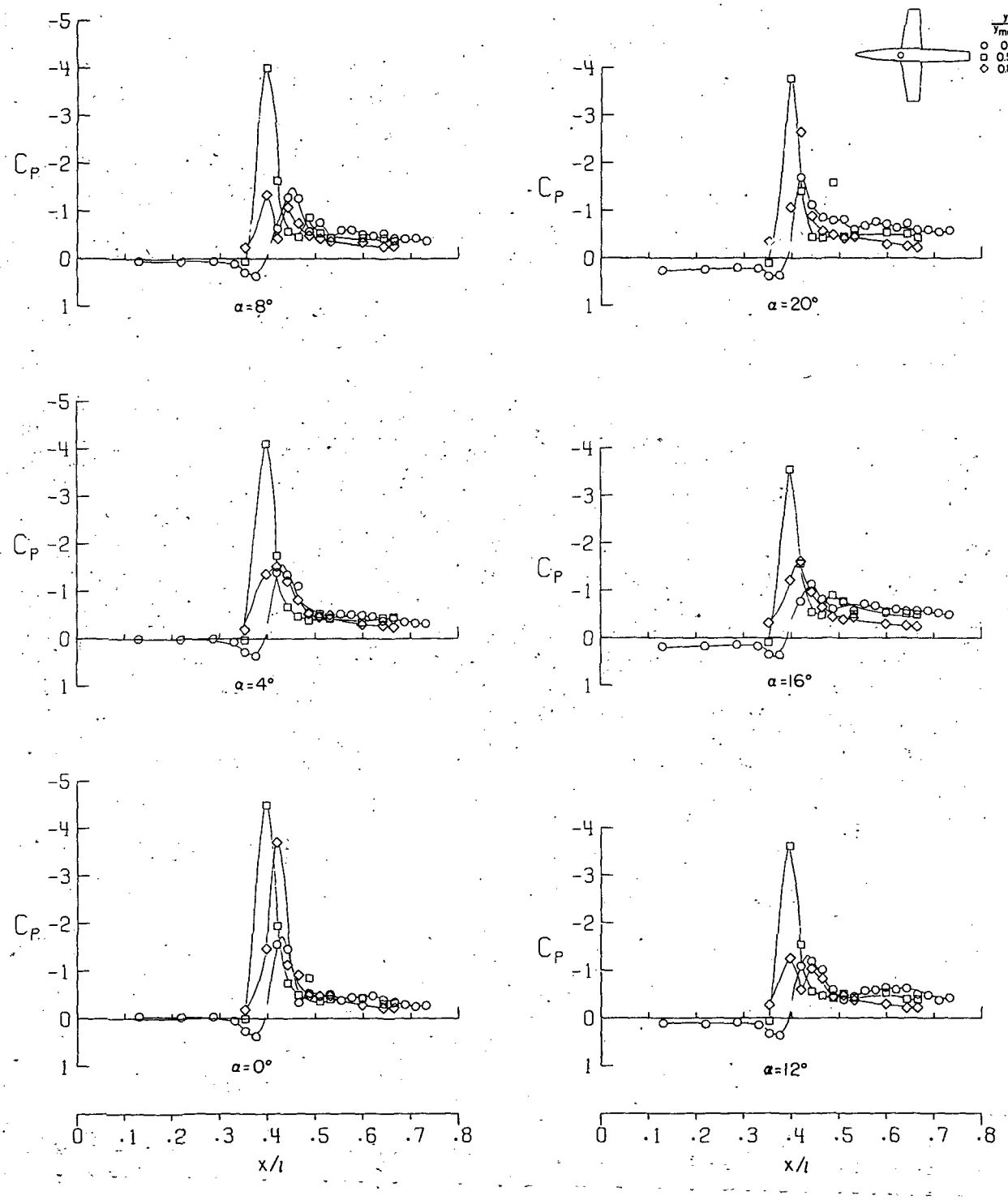
(a) $(V_\infty/V_j)_e \approx \infty$.

Figure 17.- Pressure distributions on the fuselage for the lift-jet configuration.



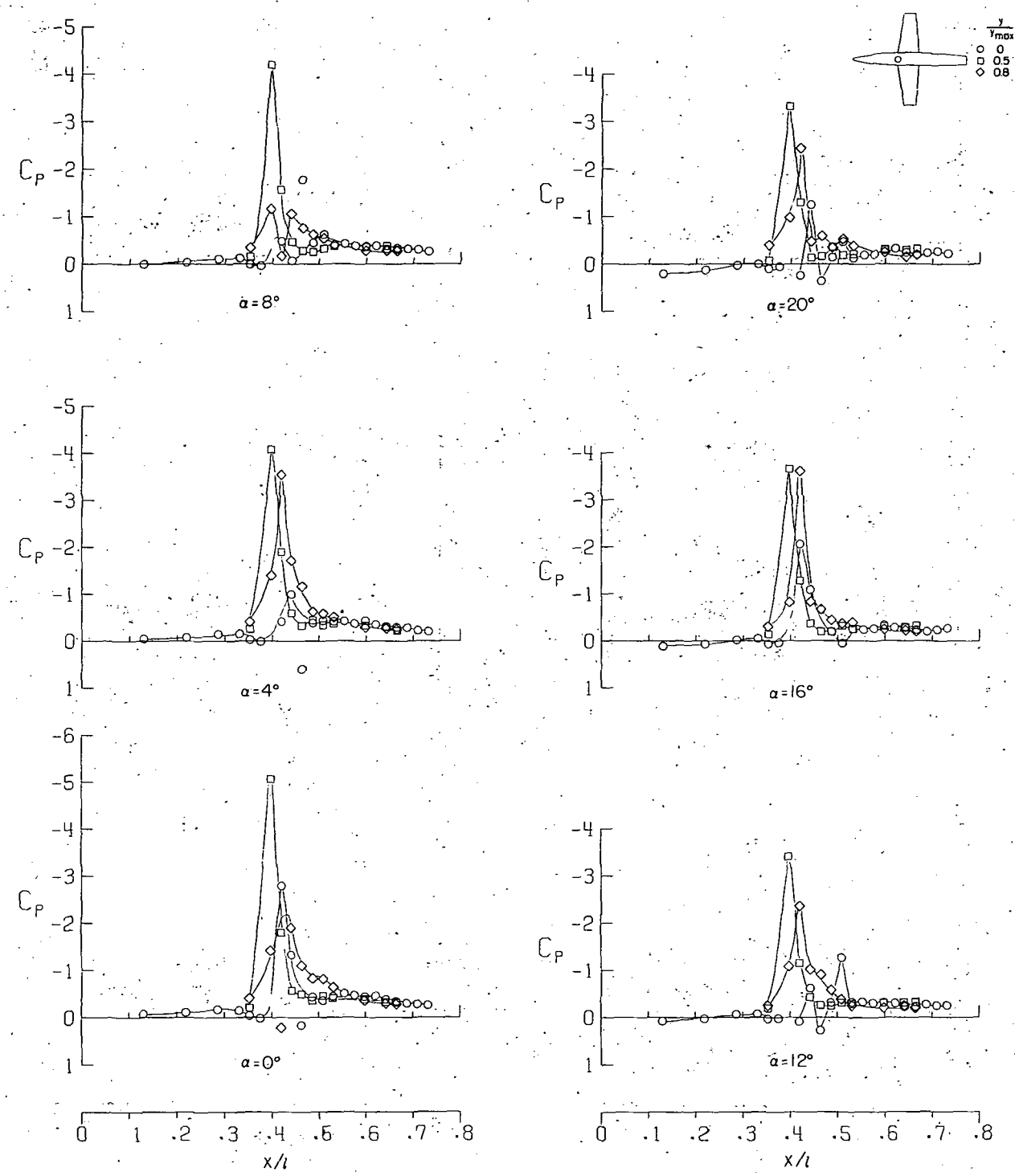
(b) $\left(V_\infty/V_j\right)_e \approx 0.3$.

Figure 17.- Continued.



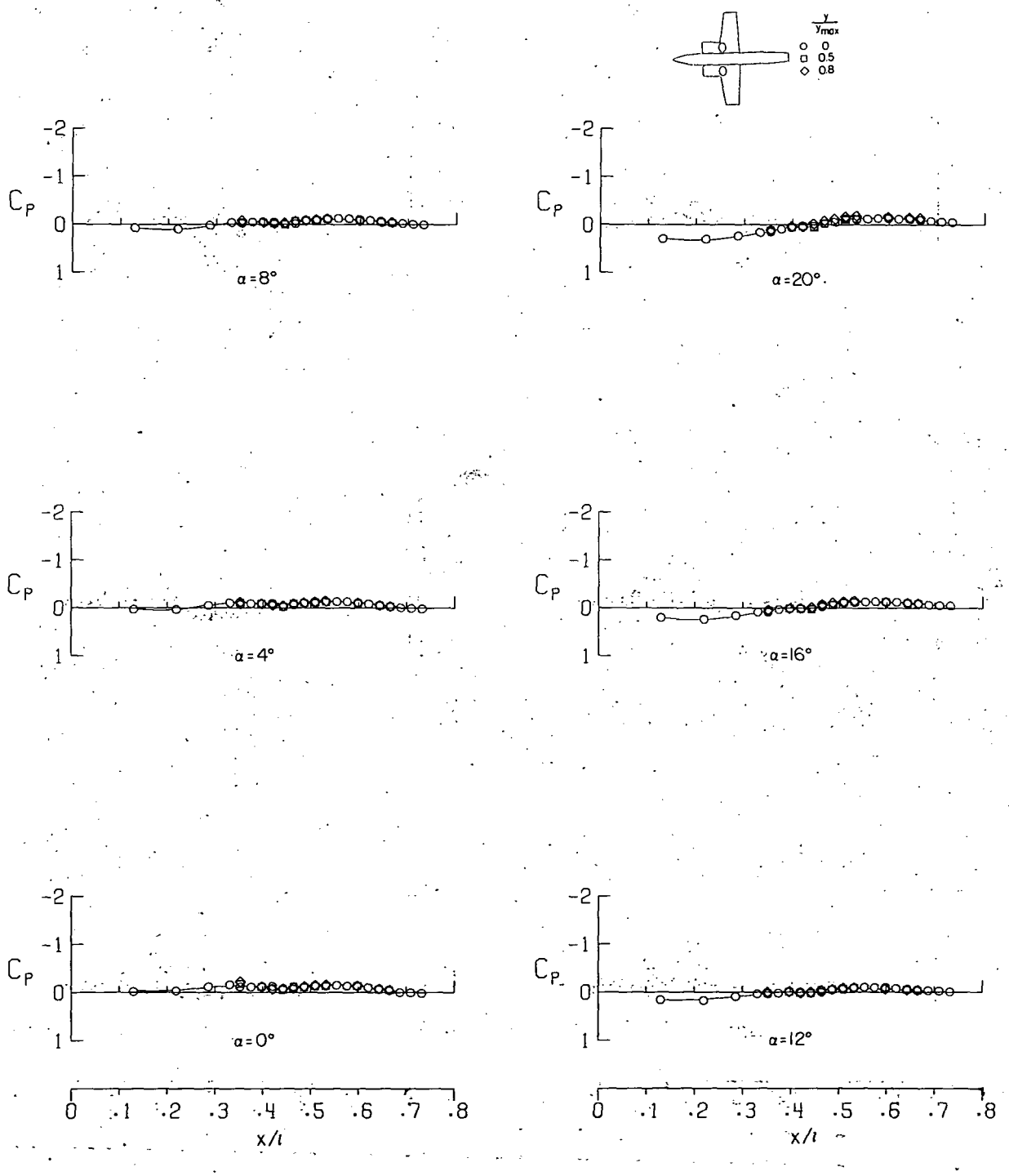
(c) $(V_\infty/V_j)_e \approx 0.2$.

Figure 17.- Continued.



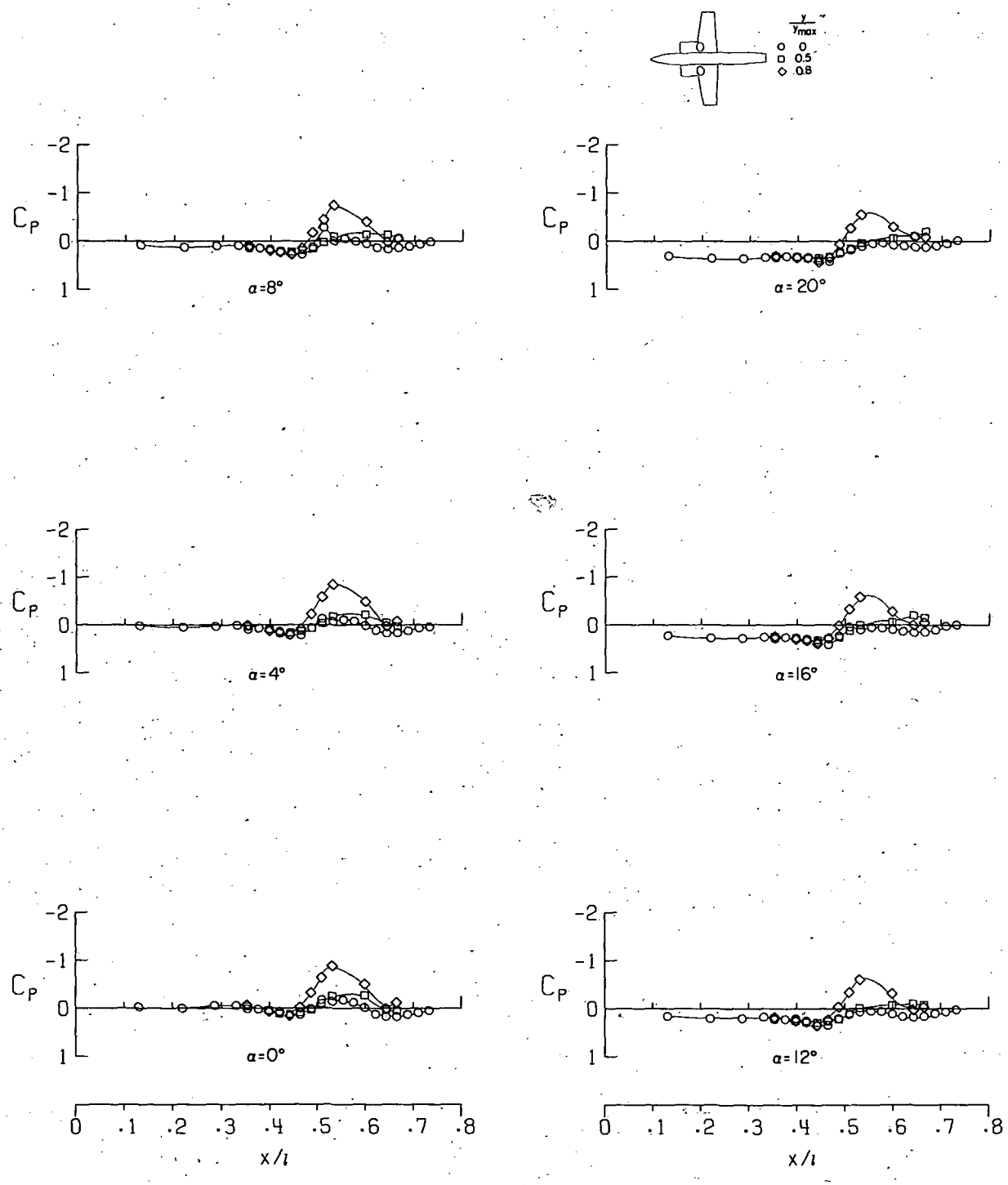
(d) $\left(V_\infty/V_j\right)_e \approx 0.1$.

Figure 17.- Concluded.



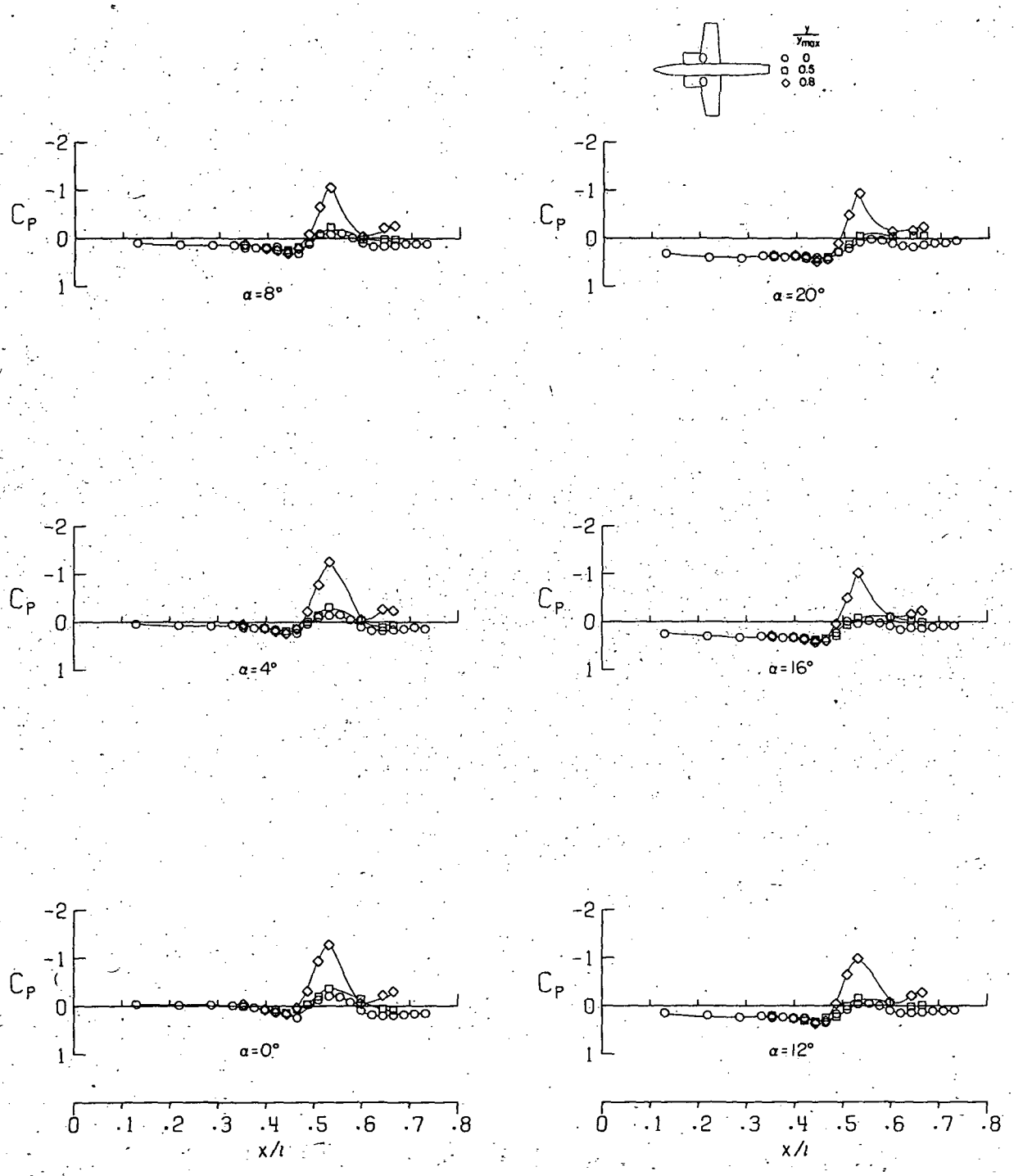
(a) $(V_\infty/V_j)_e \approx \infty$.

Figure 18.- Pressure distributions on the fuselage for the forward 45° nozzle configuration.



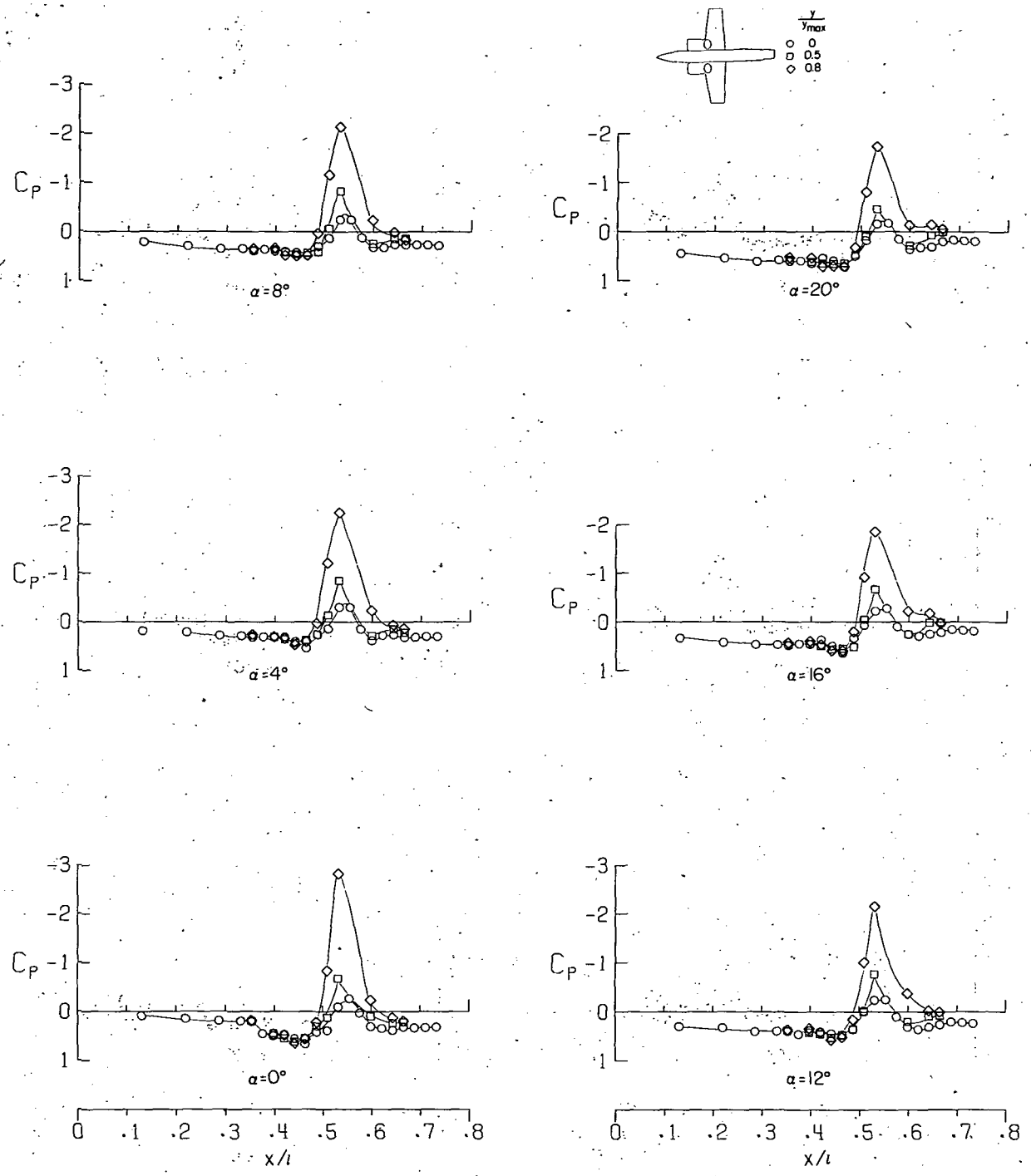
(b) $\left(V_\infty/V_j\right)_e \approx 0.3$.

Figure 18.- Continued.



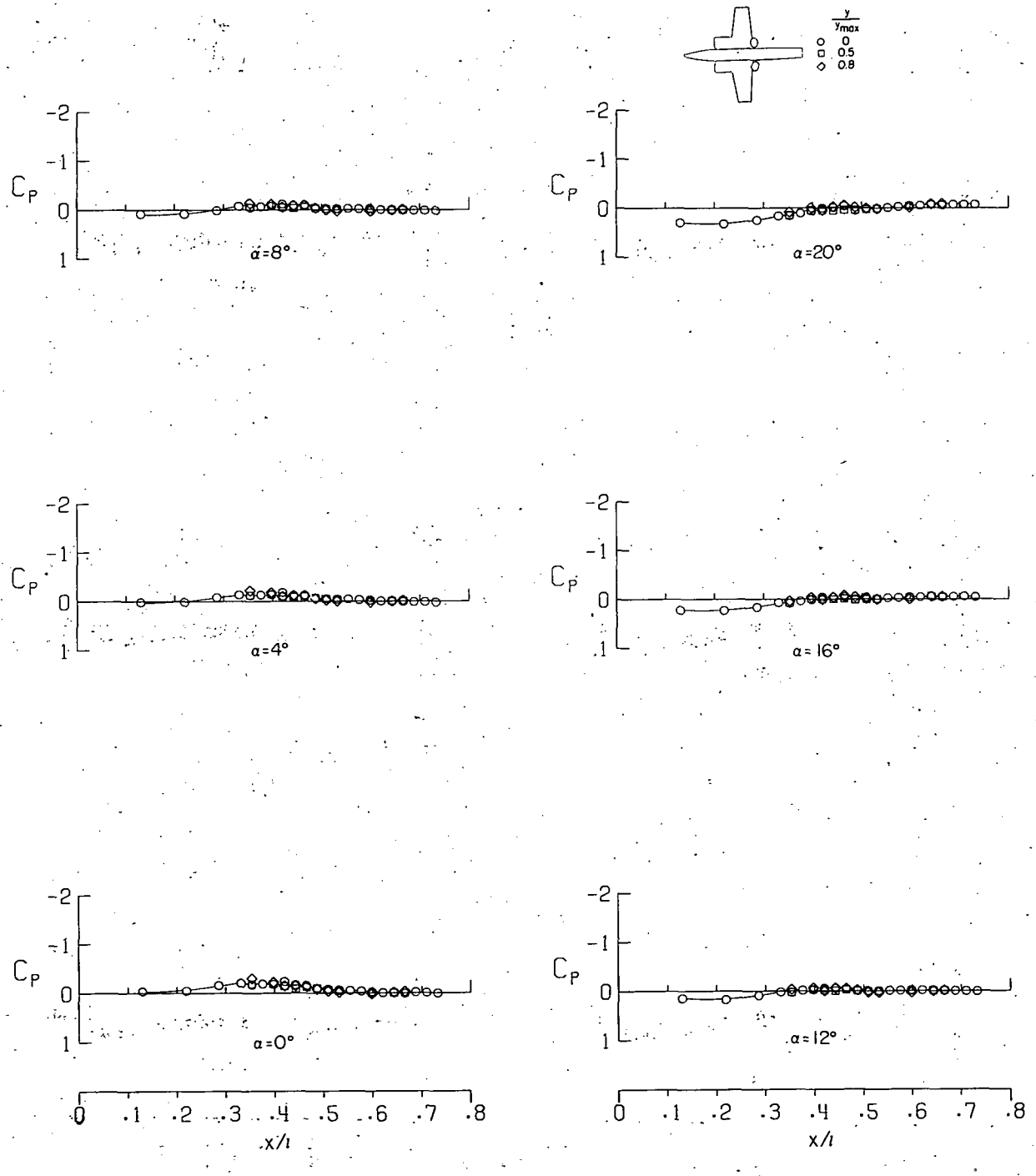
(c) $\left(V_\infty/V_j\right)_e \approx 0.2$.

Figure 18.- Continued.



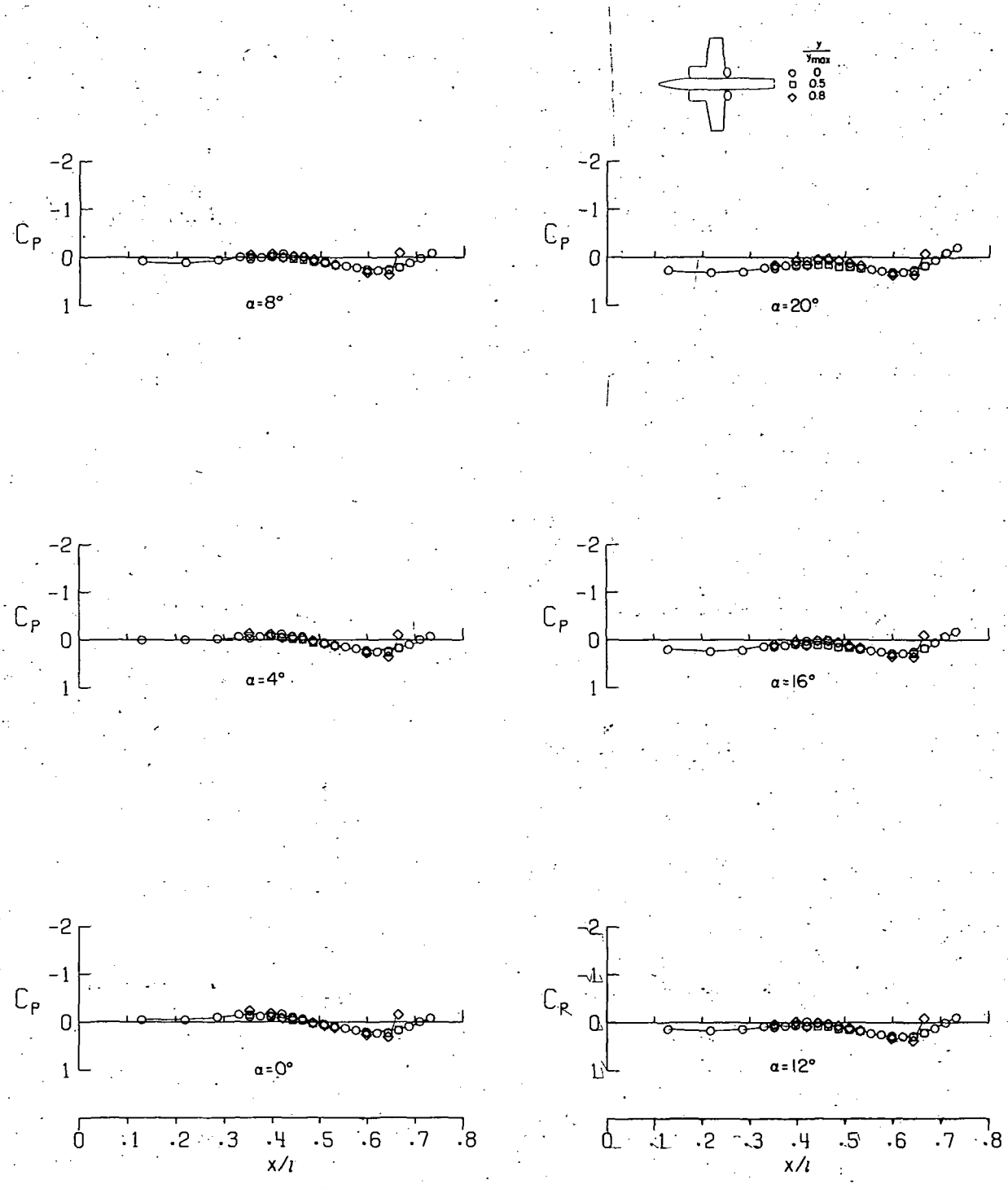
(d) $\left(V_\infty/V_j\right)_e \approx 0.1$.

Figure 18.- Concluded.



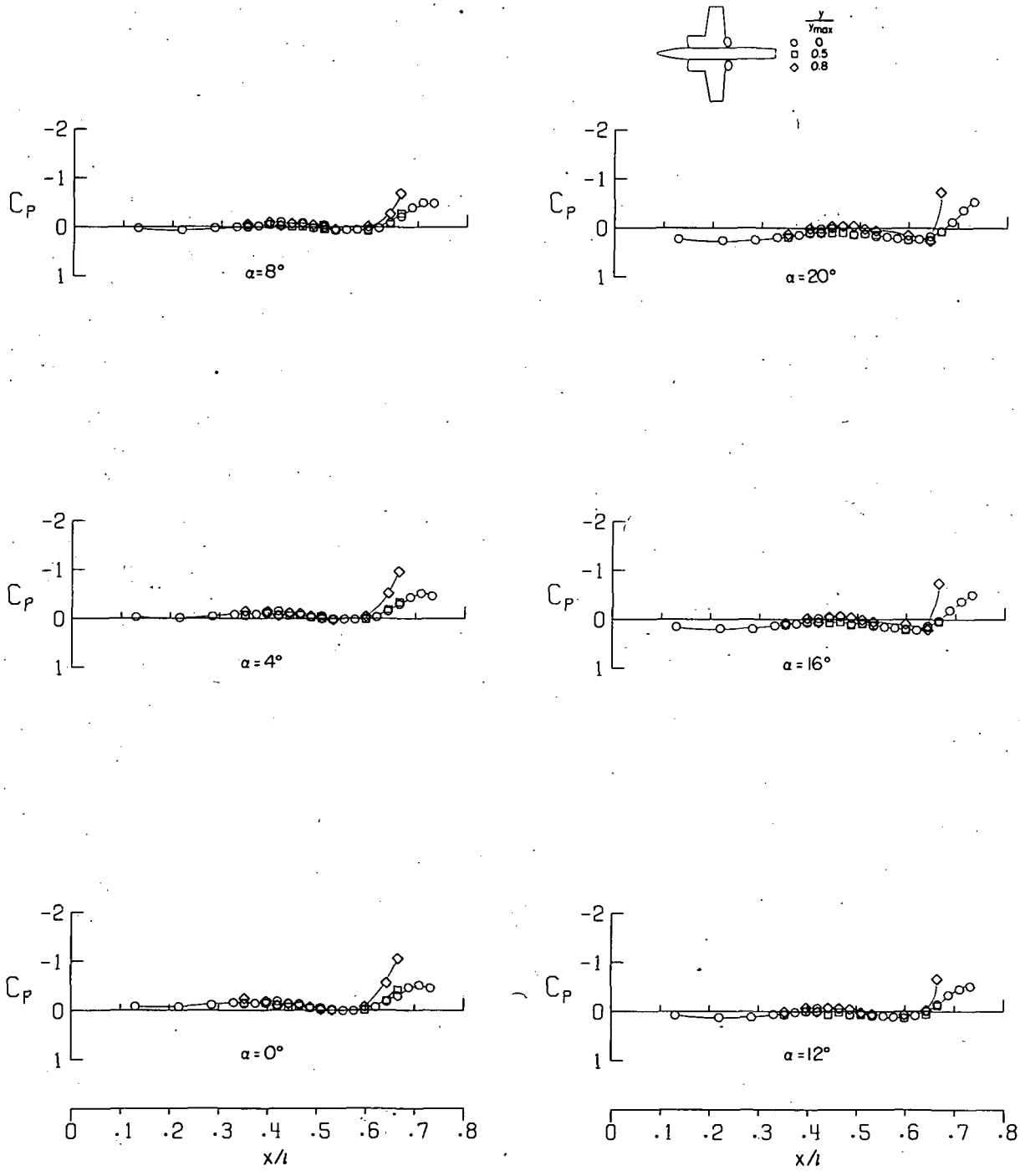
(a) $(V_\infty/V_j)_e \approx \infty$.

Figure 19.- Pressure distributions on the fuselage for the rear 45° nozzle configuration.



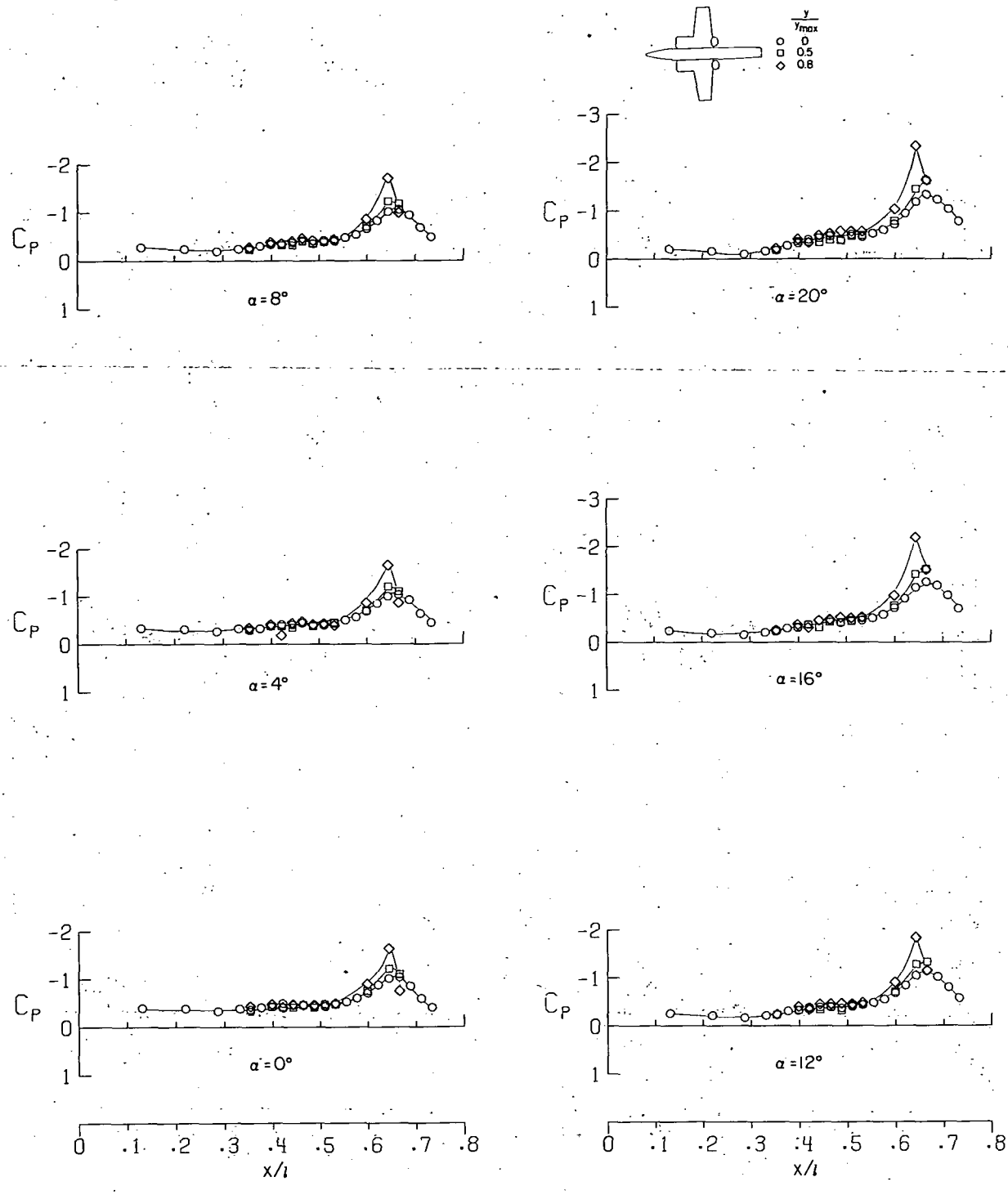
(b) $\left(V_\infty/V_j\right)_e \approx 0.3$.

Figure 19.- Continued.



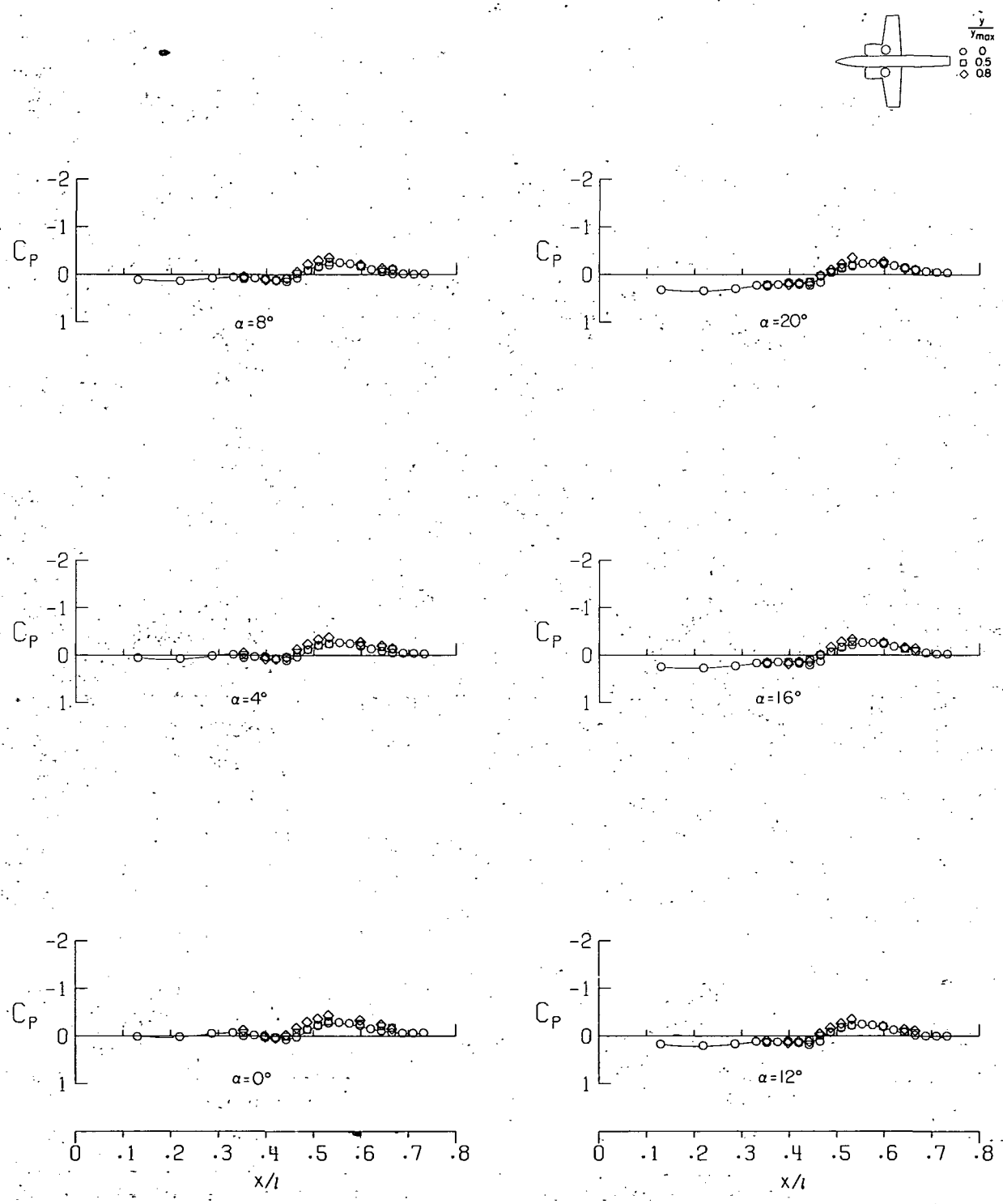
(c) $\left(V_\infty/V_j\right)_e \approx 0.2$.

Figure 19.- Continued.



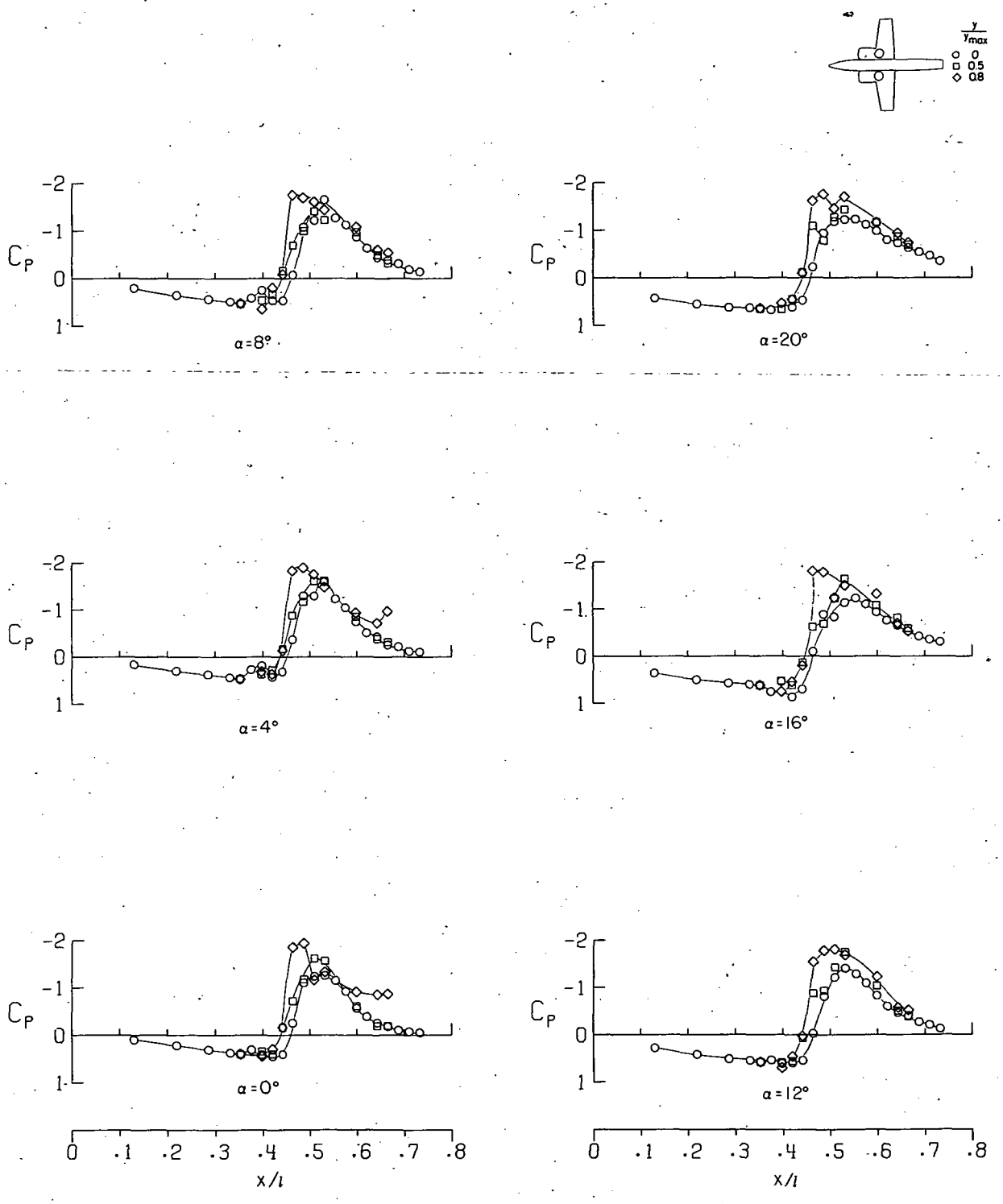
(d) $\left(V_\infty/V_j\right)_e \approx 0.1$.

Figure 19.- Concluded.



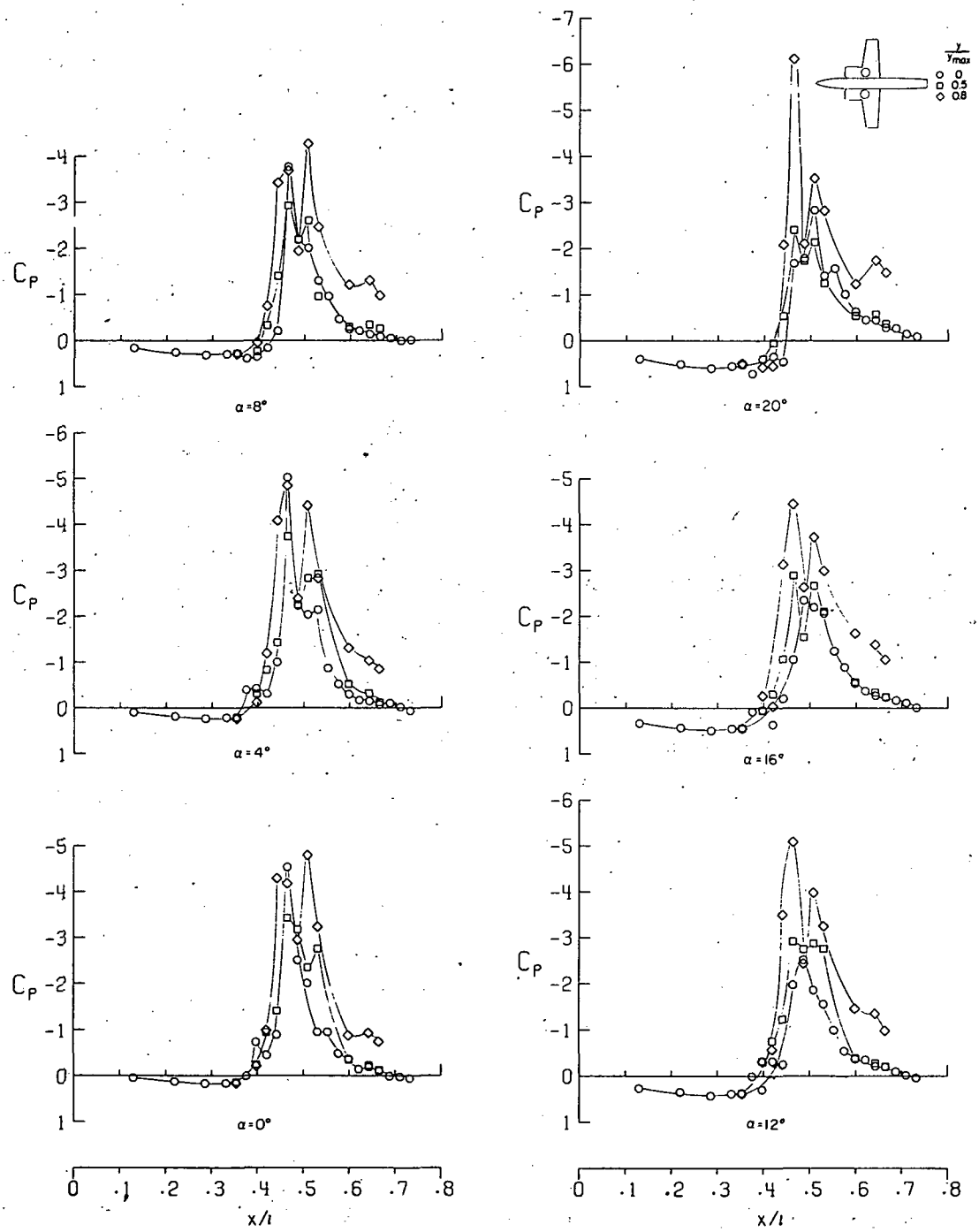
(a) $\left(V_\infty/V_j\right)_e \approx \infty$.

Figure 20.- Pressure distributions on the fuselage for the forward 90° nozzle configuration.



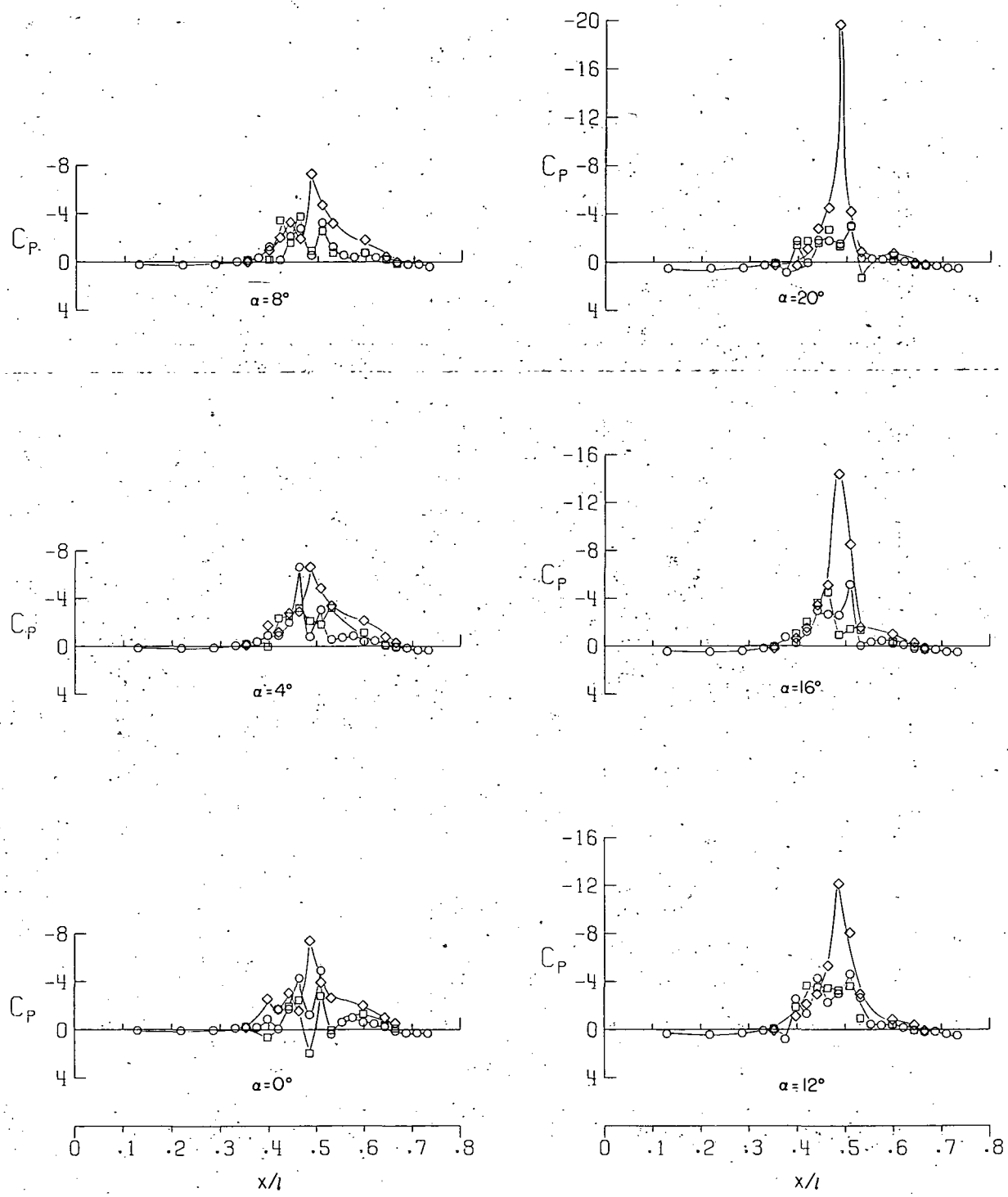
(b) $(V_\infty/V_j)_e \approx 0.3$.

Figure 20.- Continued.



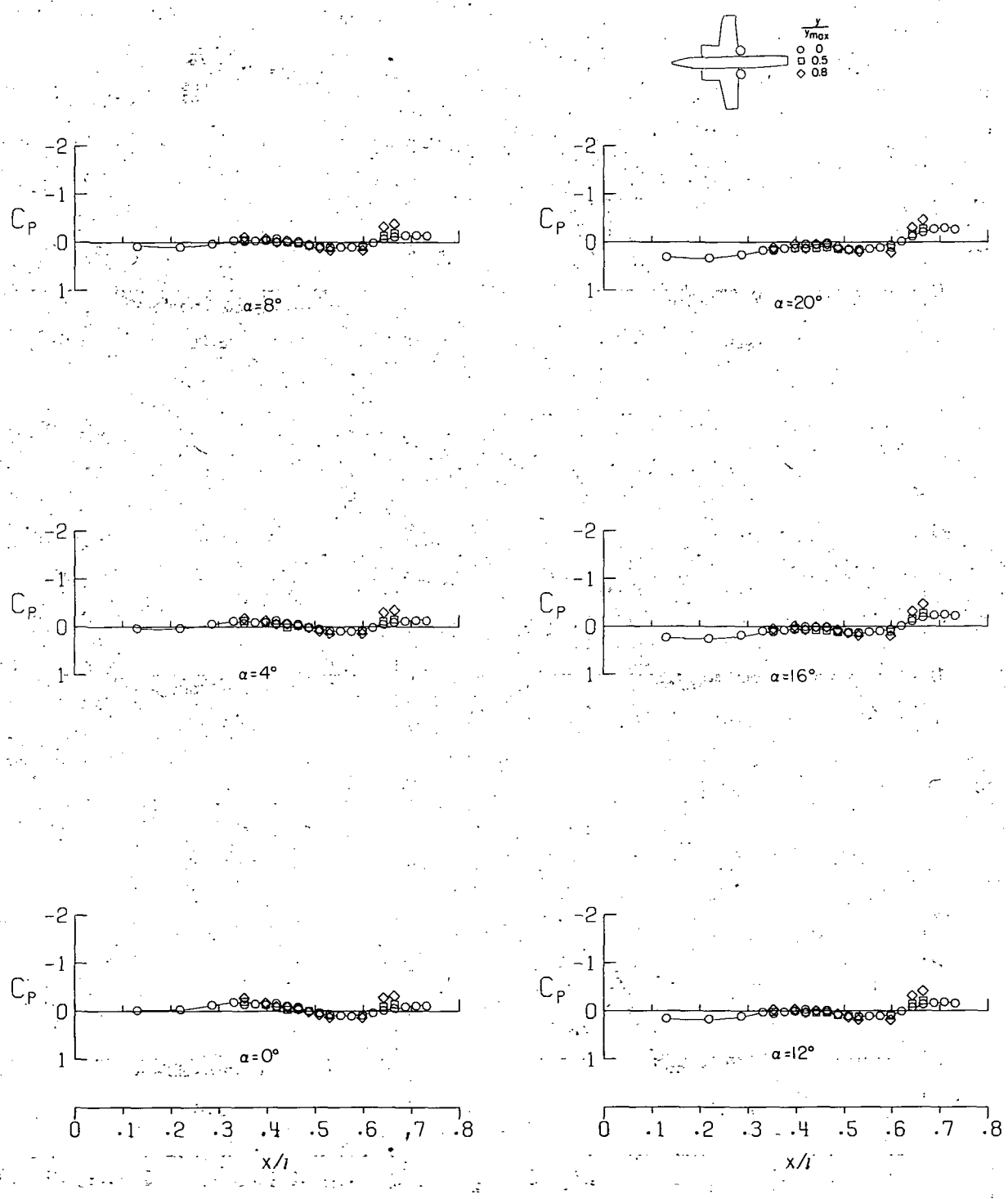
(c) $(V_\infty/V_j)_e \approx 0.2$.

Figure 20.- Continued.



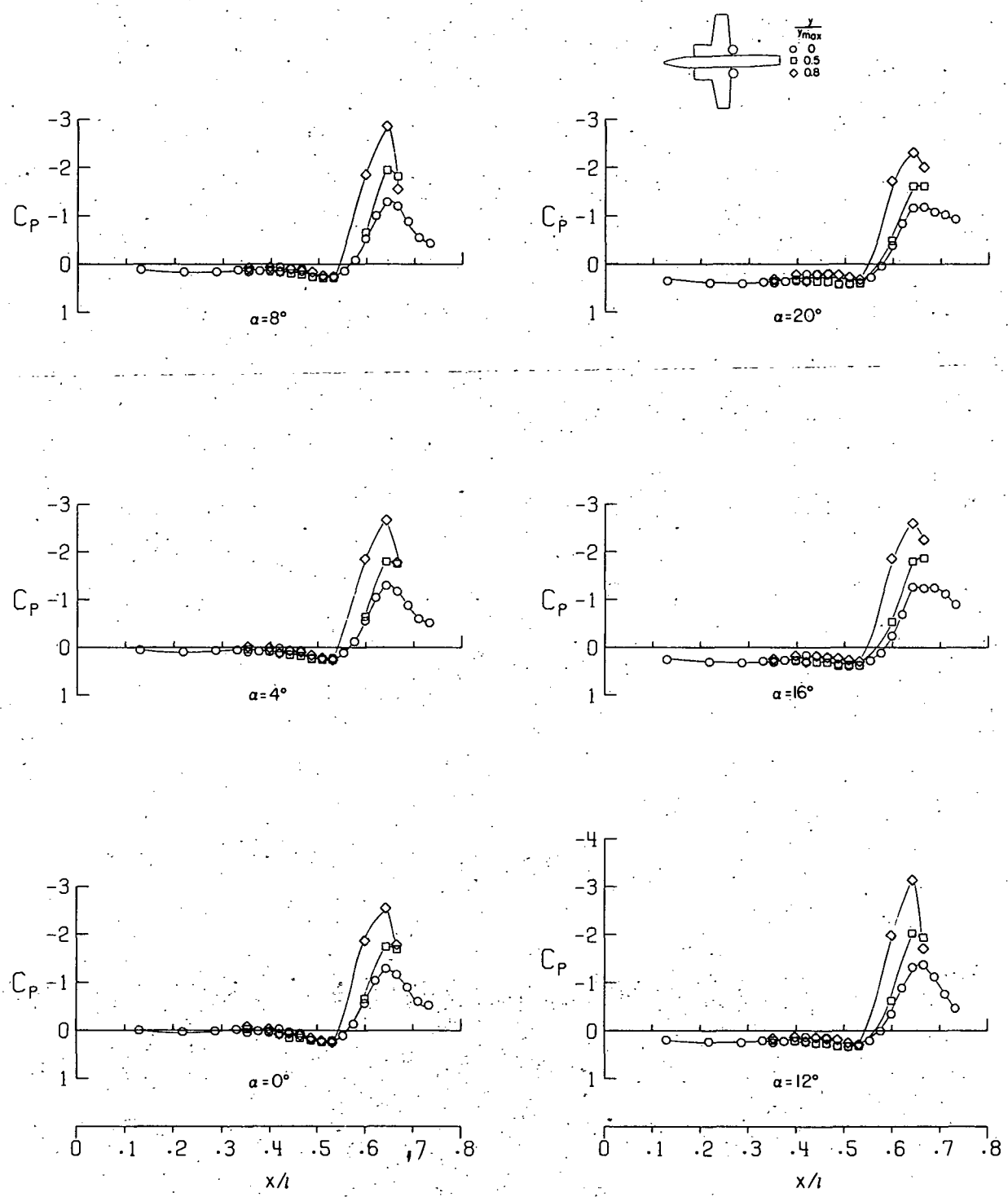
(d) $\left(V_\infty/V_j\right)_e \approx 0.1$.

Figure 20.- Concluded.



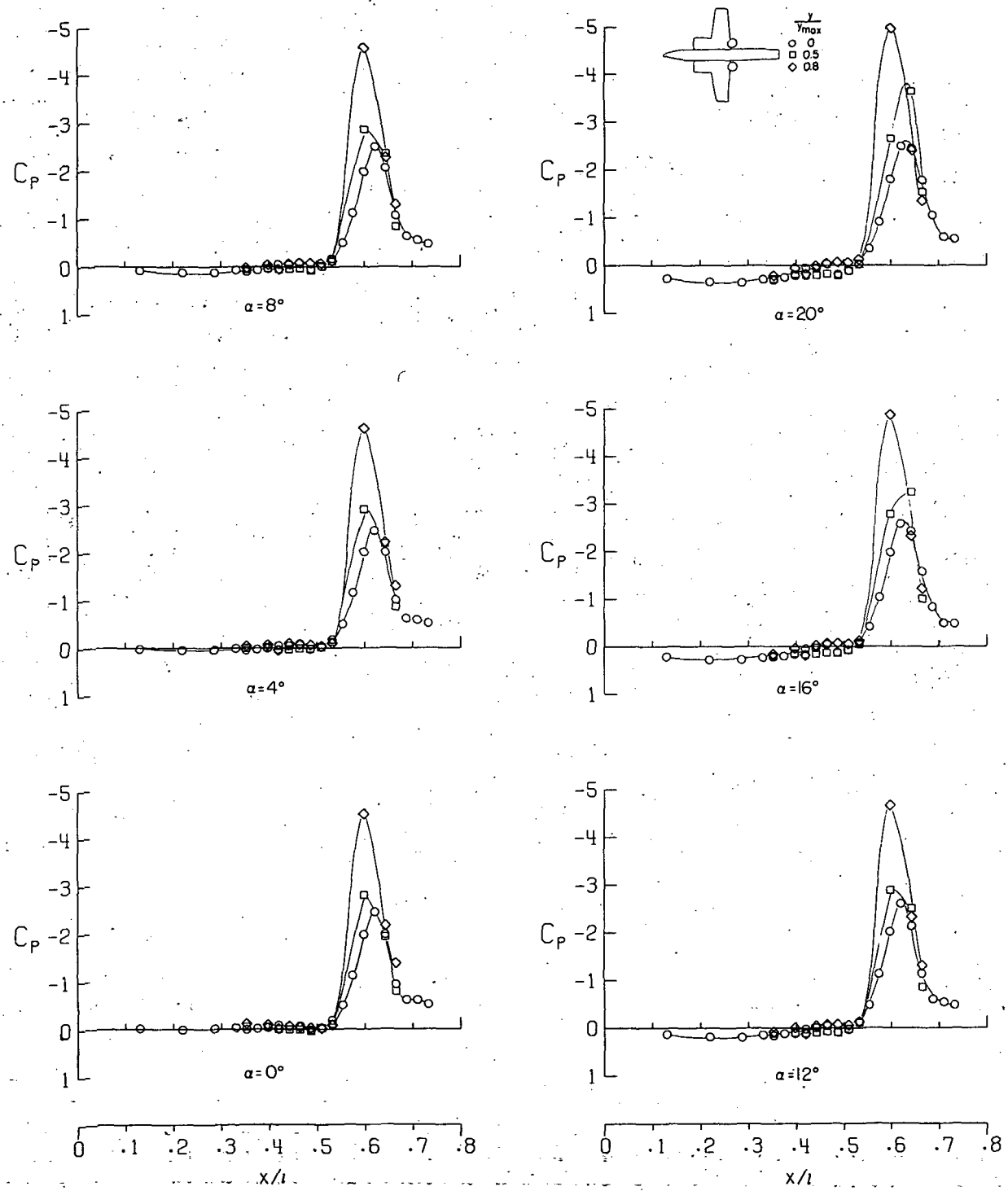
(a) $(V_\infty/V_j)_e \approx \infty$.

Figure 21.- Pressure distributions on the fuselage for the rear 90° nozzle configuration.



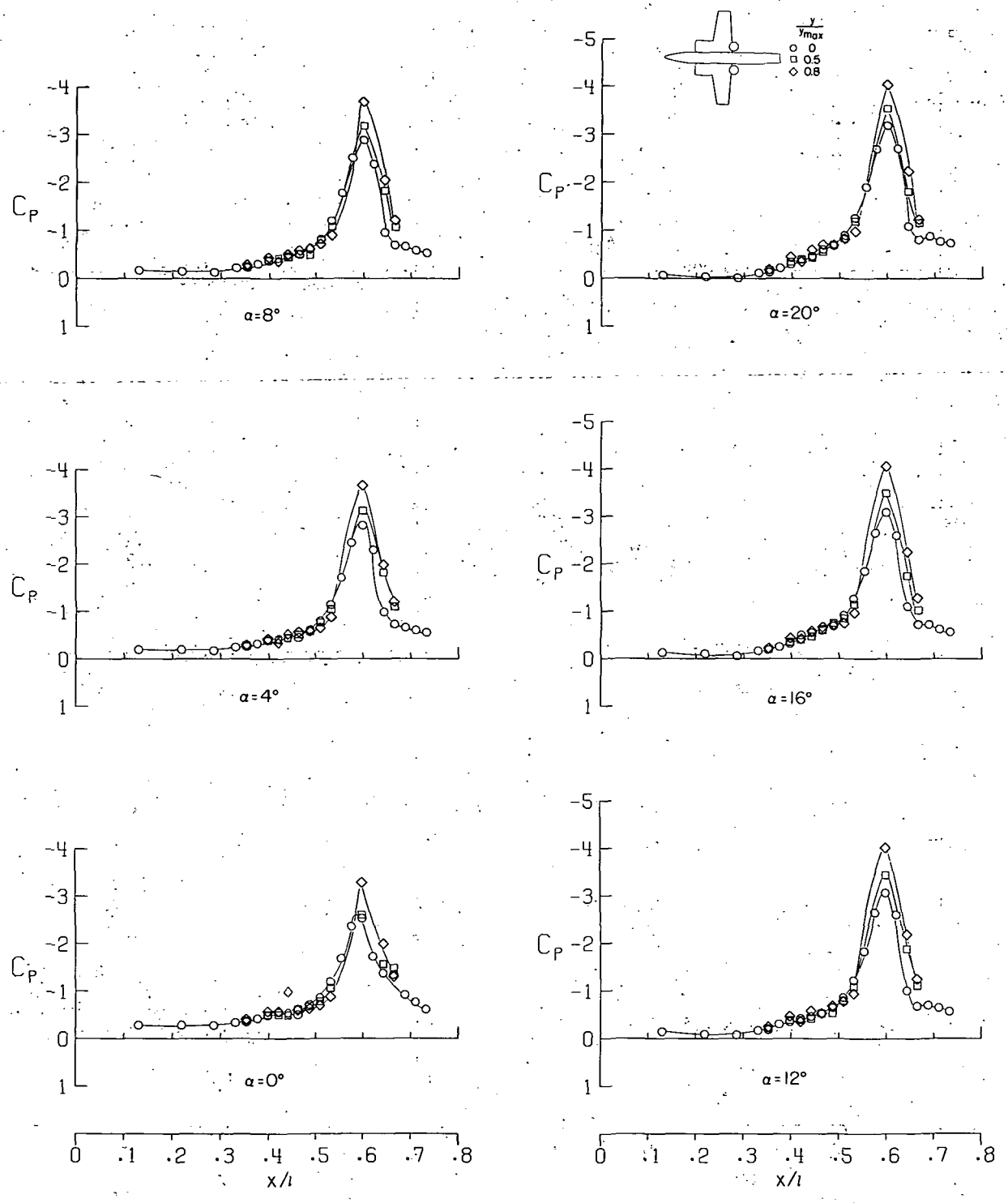
(b) $(V_\infty/V_j)_e \approx 0.3$.

Figure 21.- Continued.



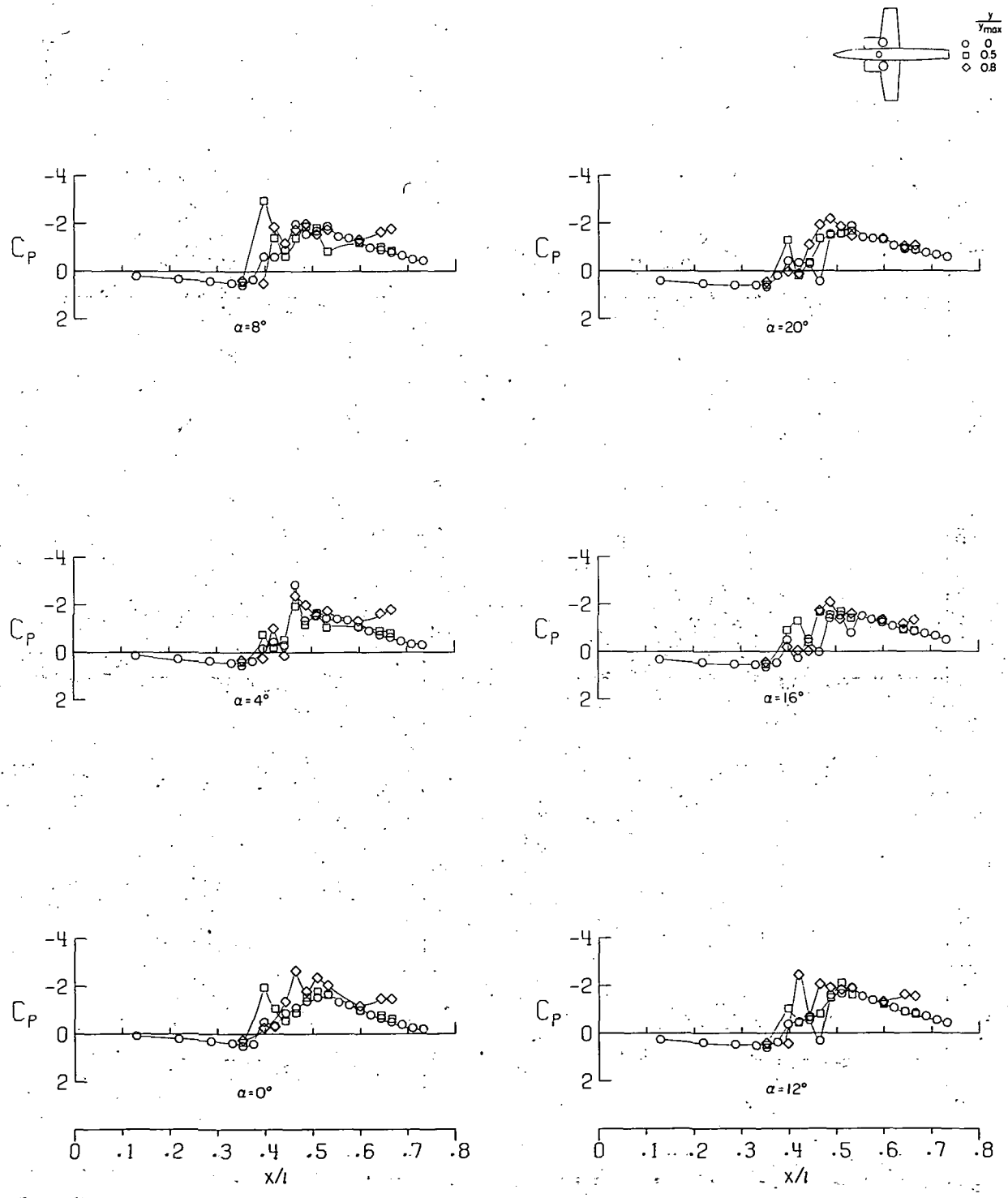
(c) $(V_\infty/V_j)_e \approx 0.2$.

Figure 21.- Continued.



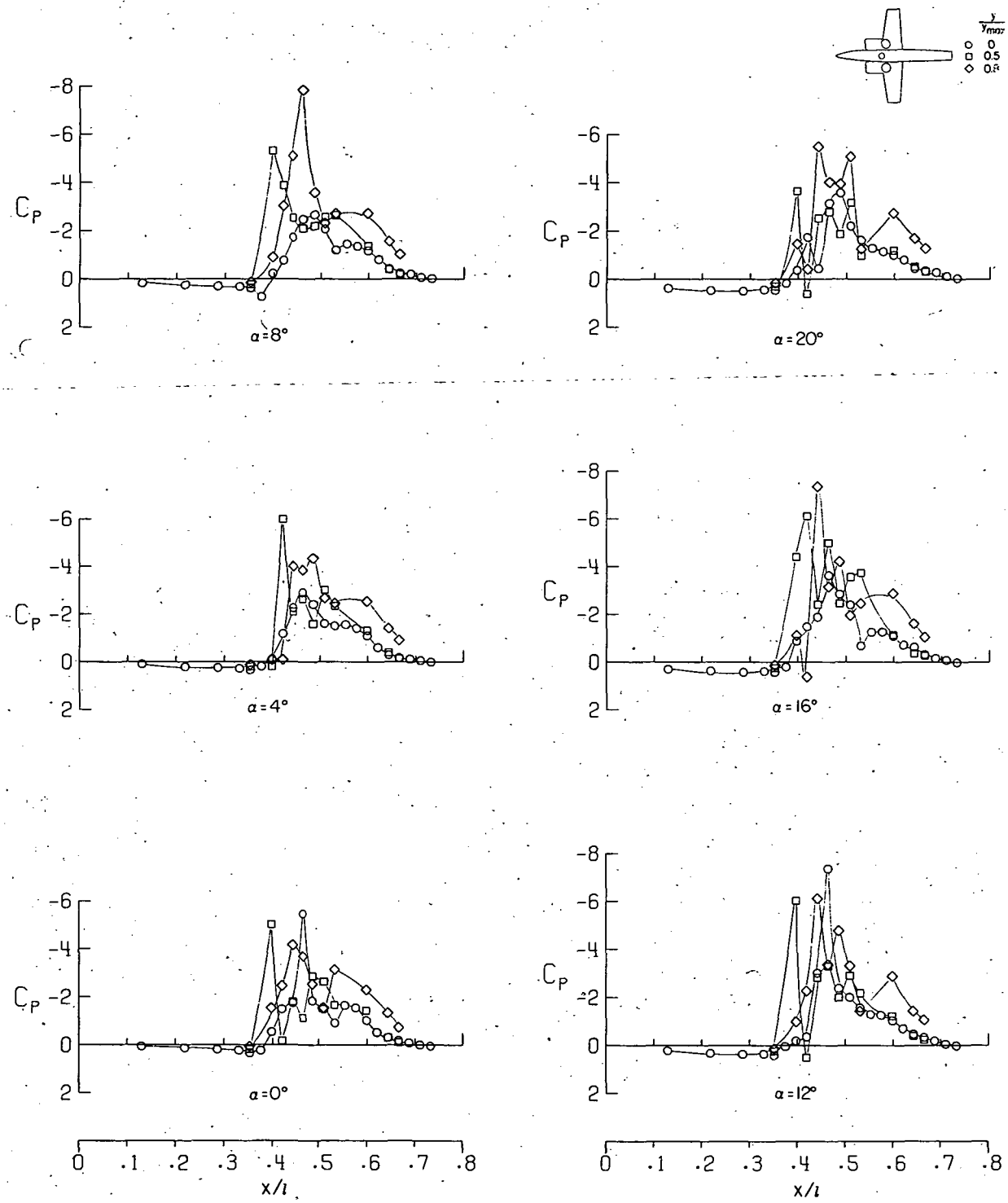
(d) $(V_\infty/V_j)_e \approx 0.1$.

Figure 21.- Concluded.



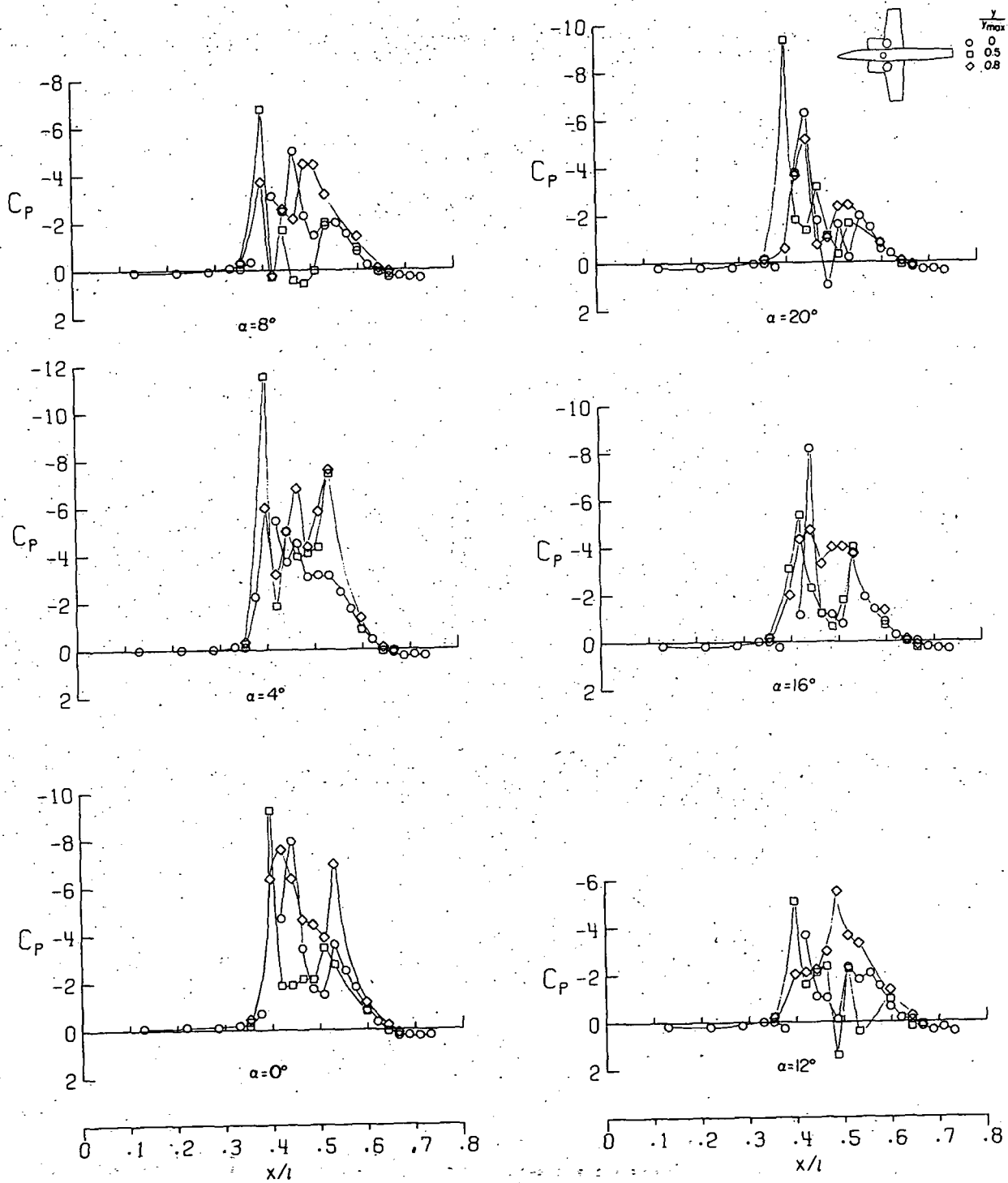
(a) $(V_\infty/V_j)_e \approx 0.3$.

Figure 22.- Pressure distributions on the fuselage for the lift-jet and forward 90° nozzle configuration.



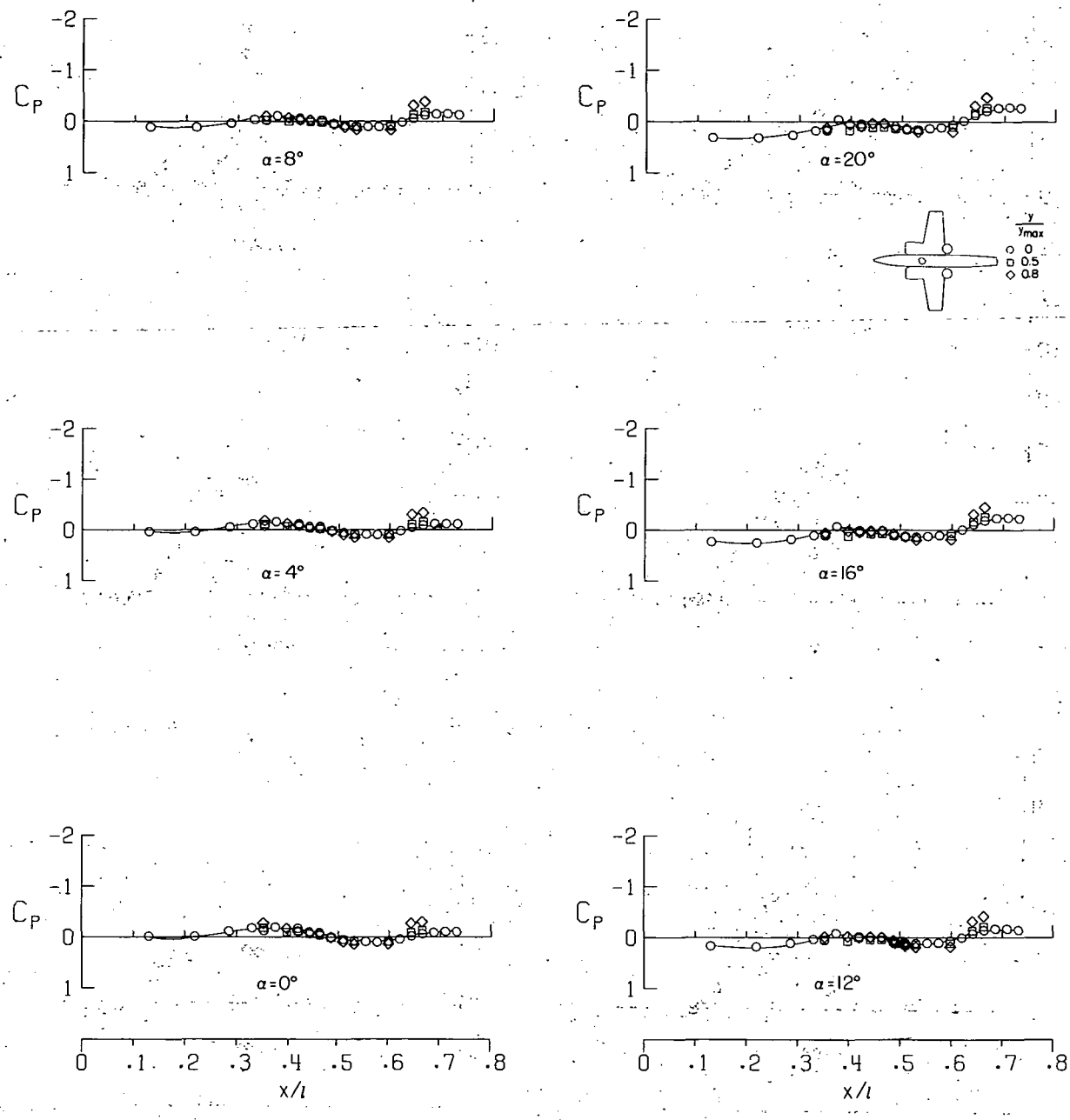
(b) $(V_\infty/V_j)_e \approx 0.2$.

Figure 22.- Continued.



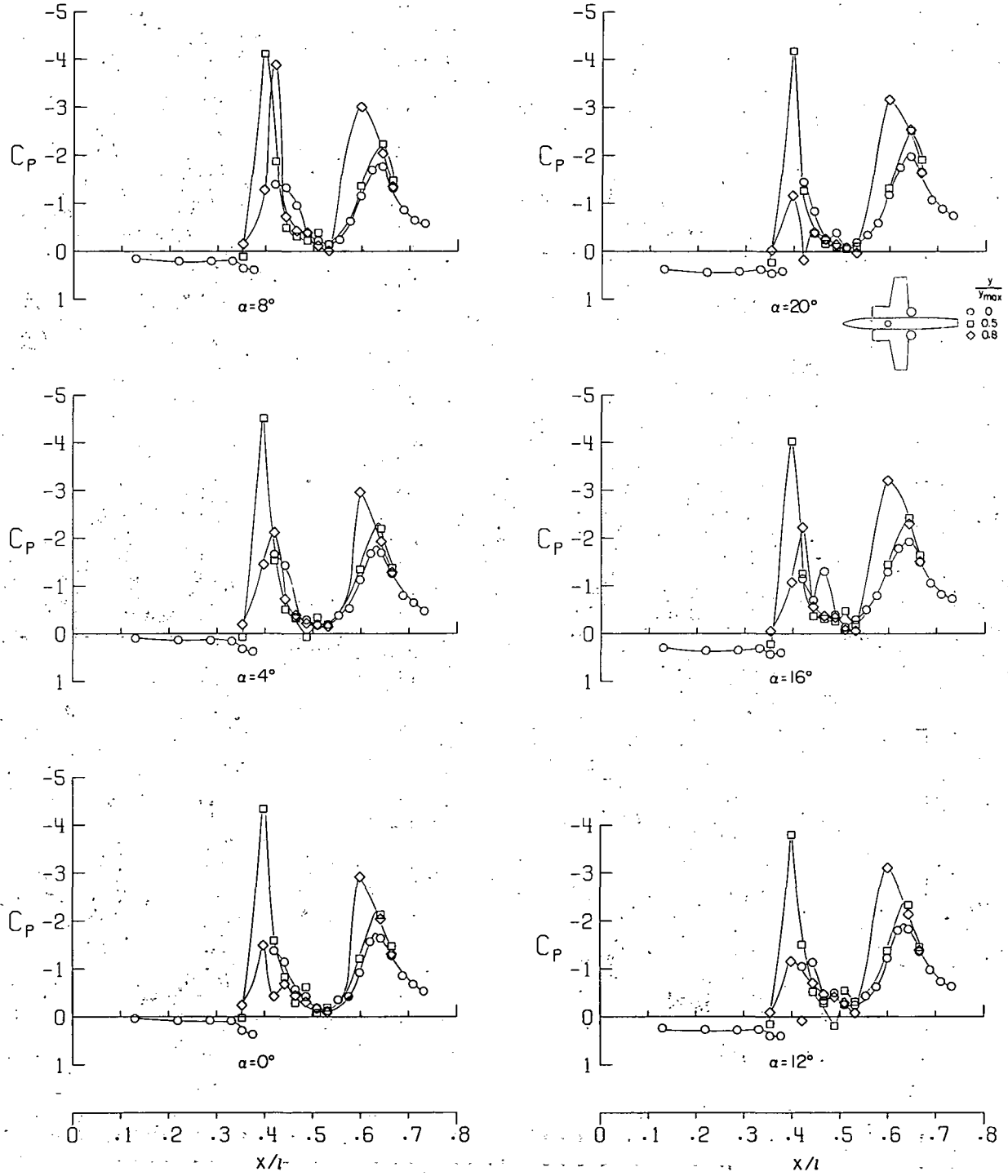
(c) $\left(V_\infty/V_j\right)_e \approx 0.1$.

Figure 22.- Concluded.



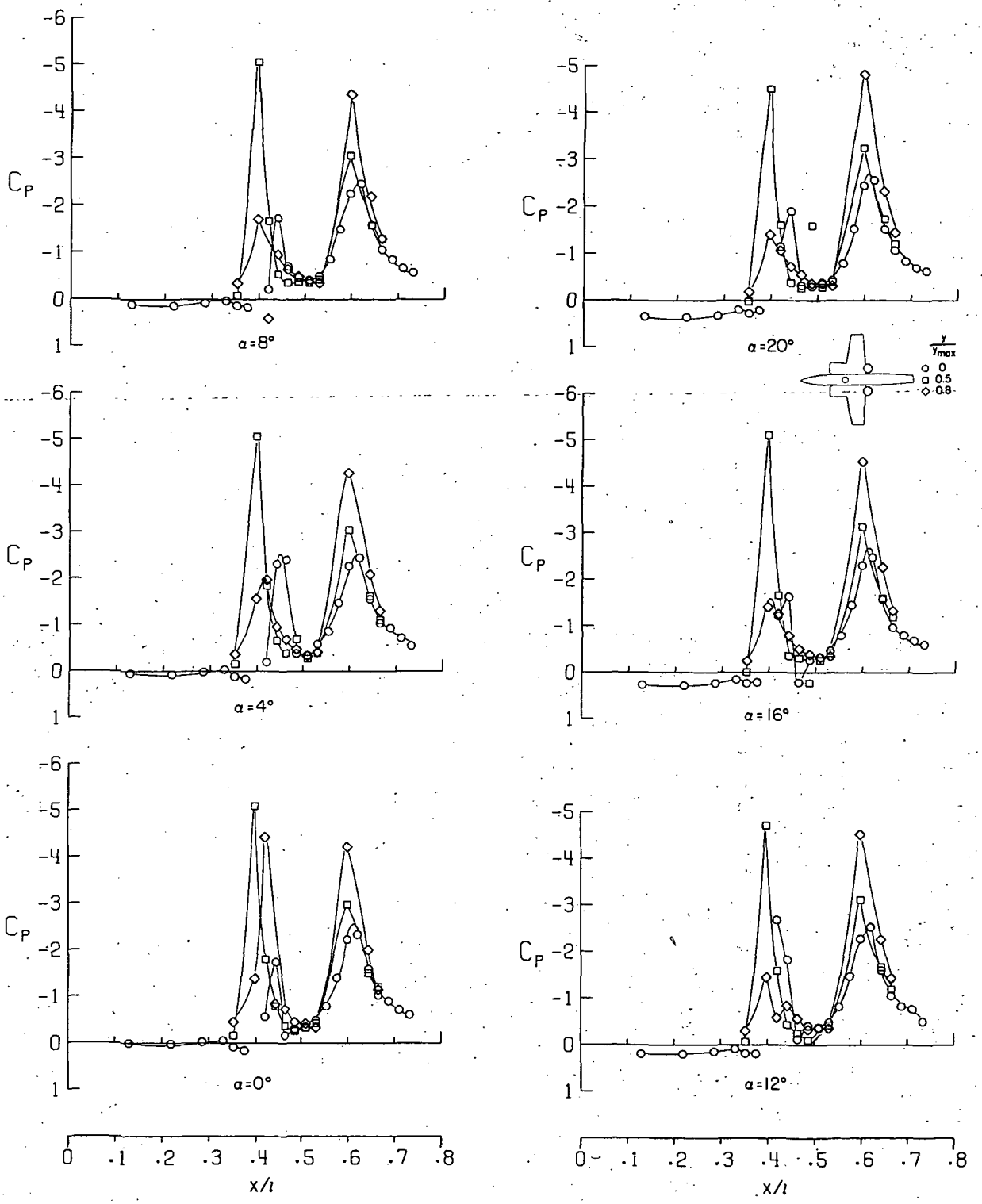
(a) $\left(V_\infty/V_j\right)_e \approx \infty$.

Figure 23.- Pressure distributions on the fuselage for the lift-jet and rear 90° nozzle configuration.



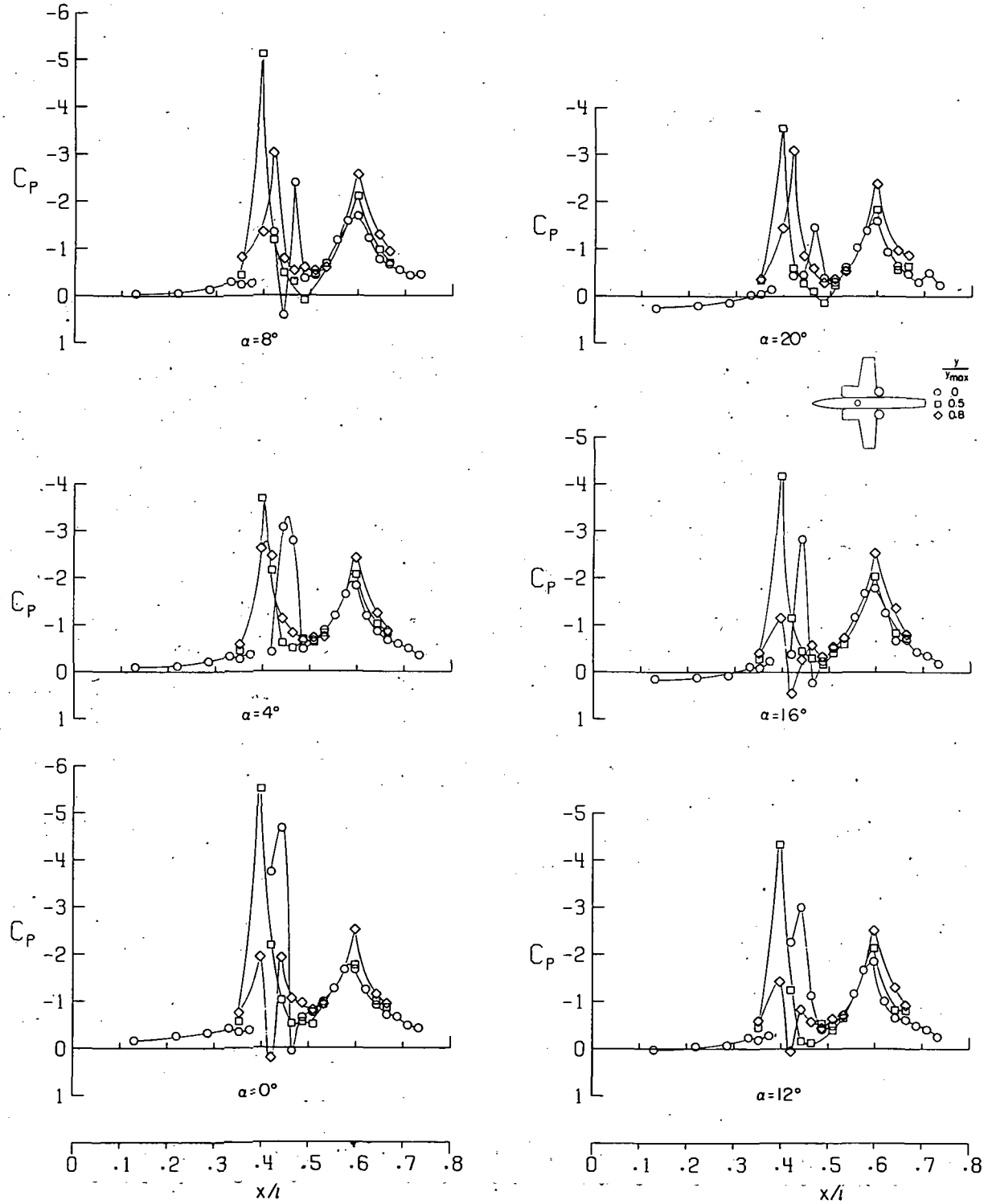
(b) $\left(V_\infty/V_j\right)_e \approx 0.3$.

Figure 23.- Continued.



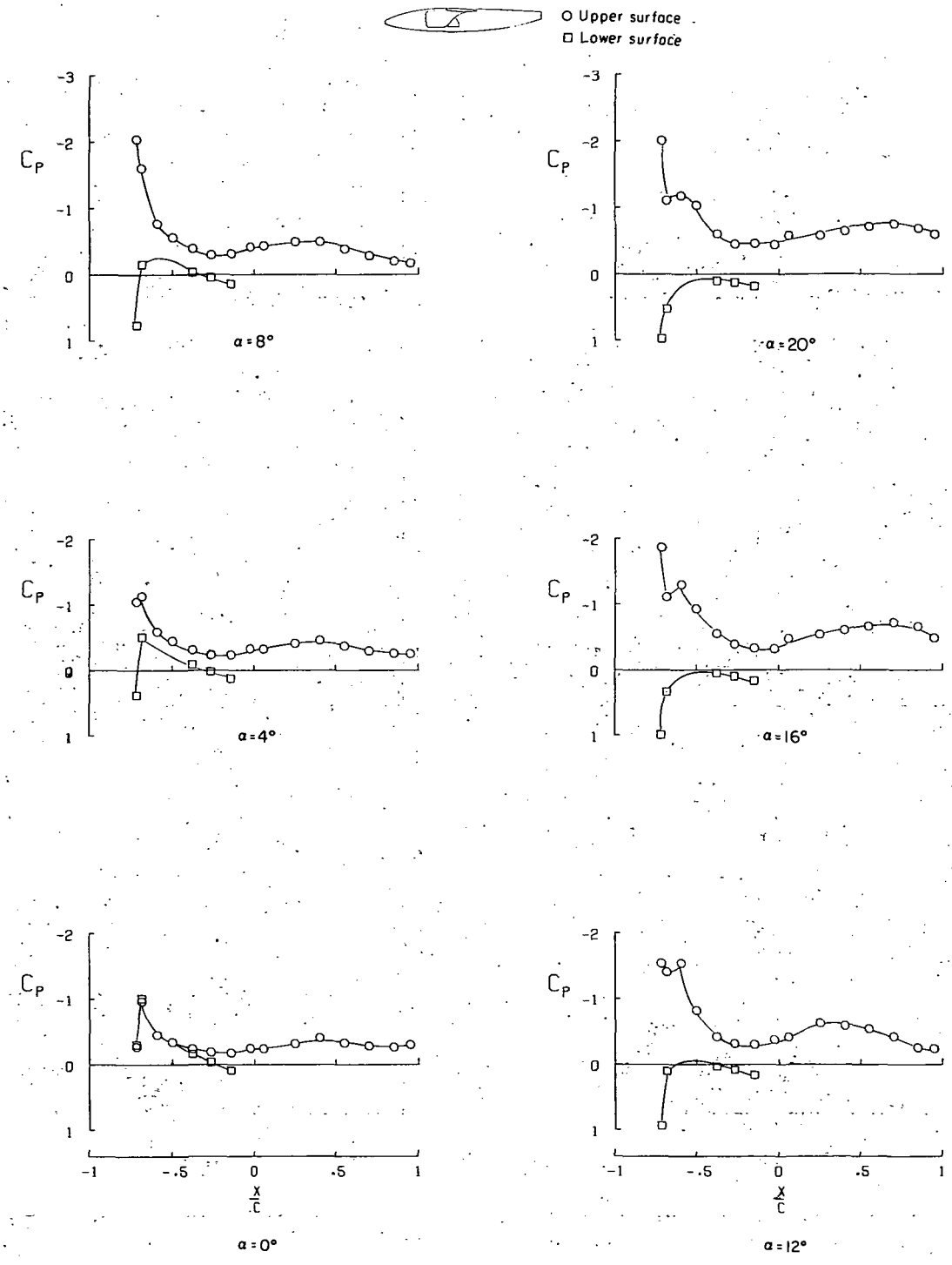
(c) $\left(V_\infty/V_j\right)_e \approx 0.2$.

Figure 23.- Continued.



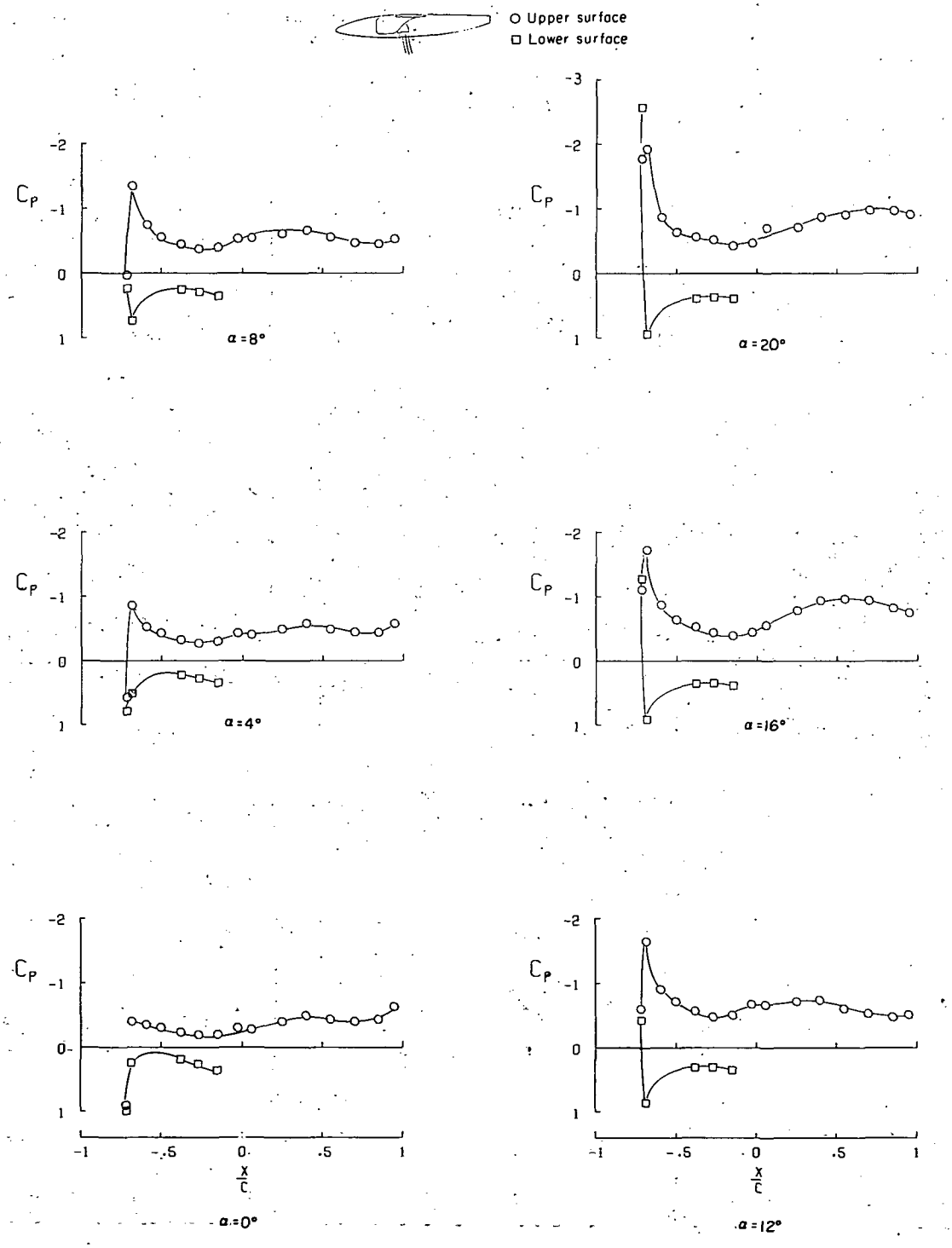
(d) $\left(V_\infty/V_j\right)_e \approx 0.1$.

Figure 23.- Concluded.



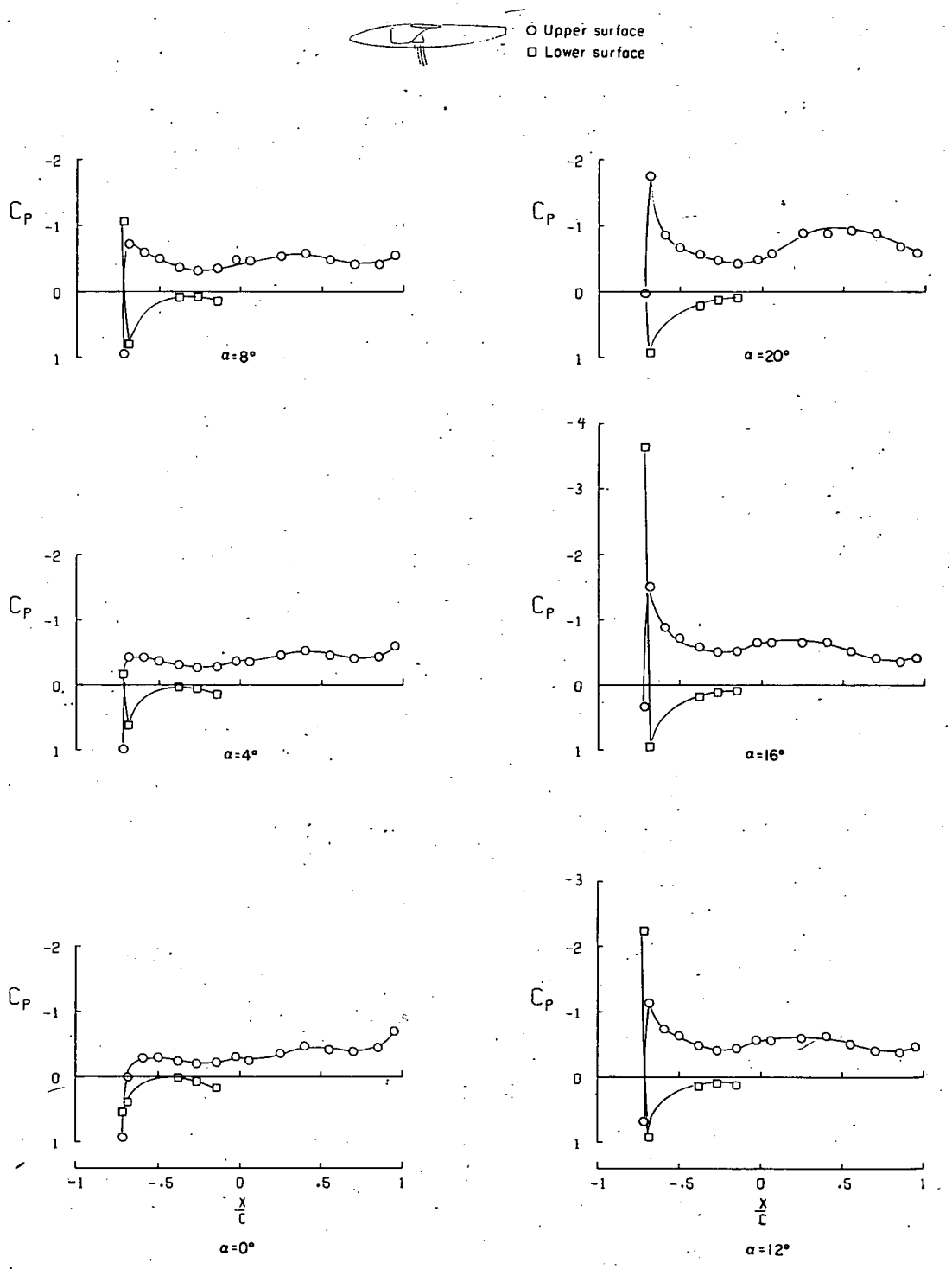
(a) $(V_\infty/V_j)_e \approx \infty$.

Figure 24.- Pressure distributions on the nacelle for the forward 90° nozzle configuration.



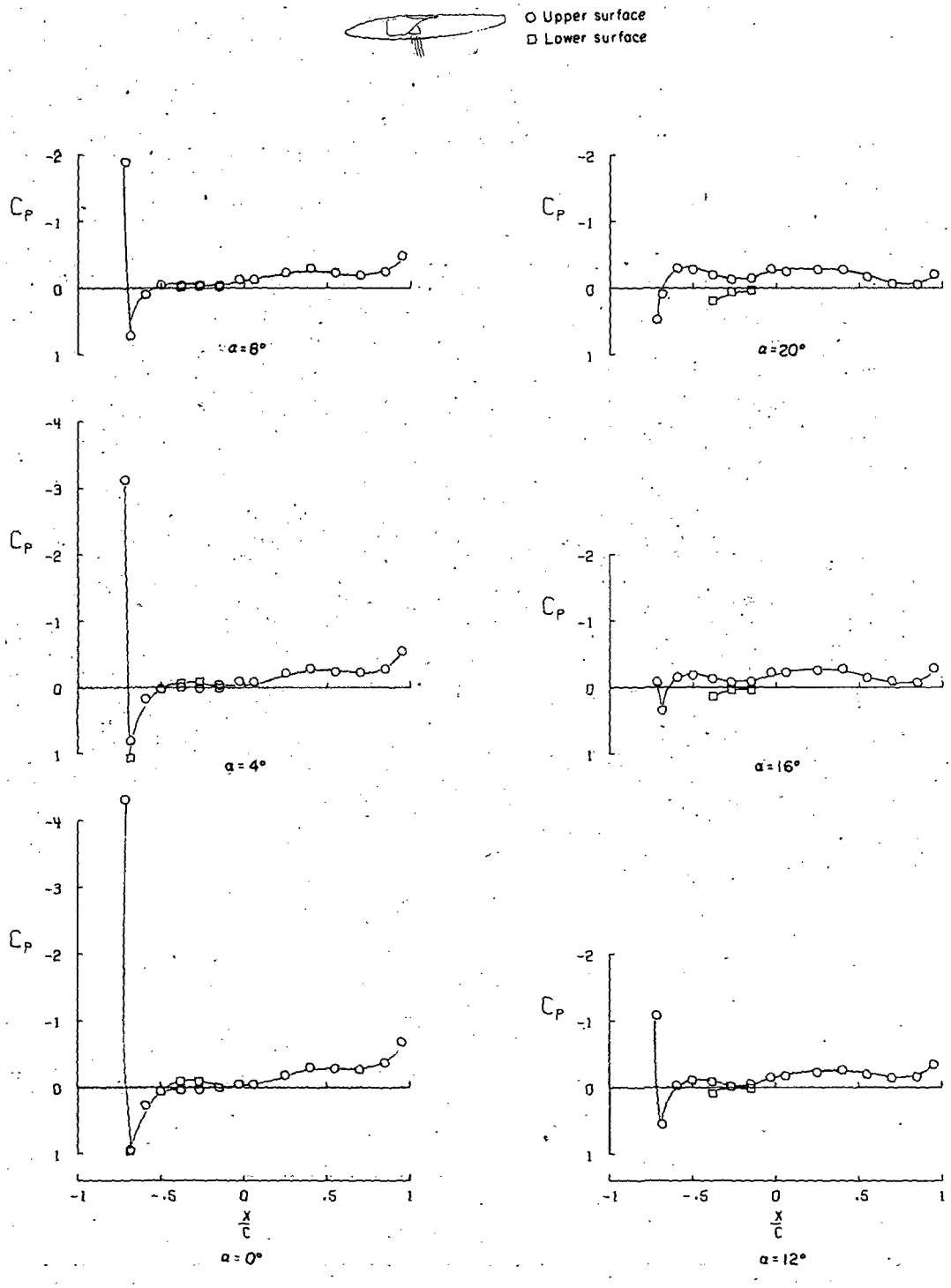
(b) $\left(V_\infty/V_j\right)_e \approx 0.3$.

Figure 24.- Continued.



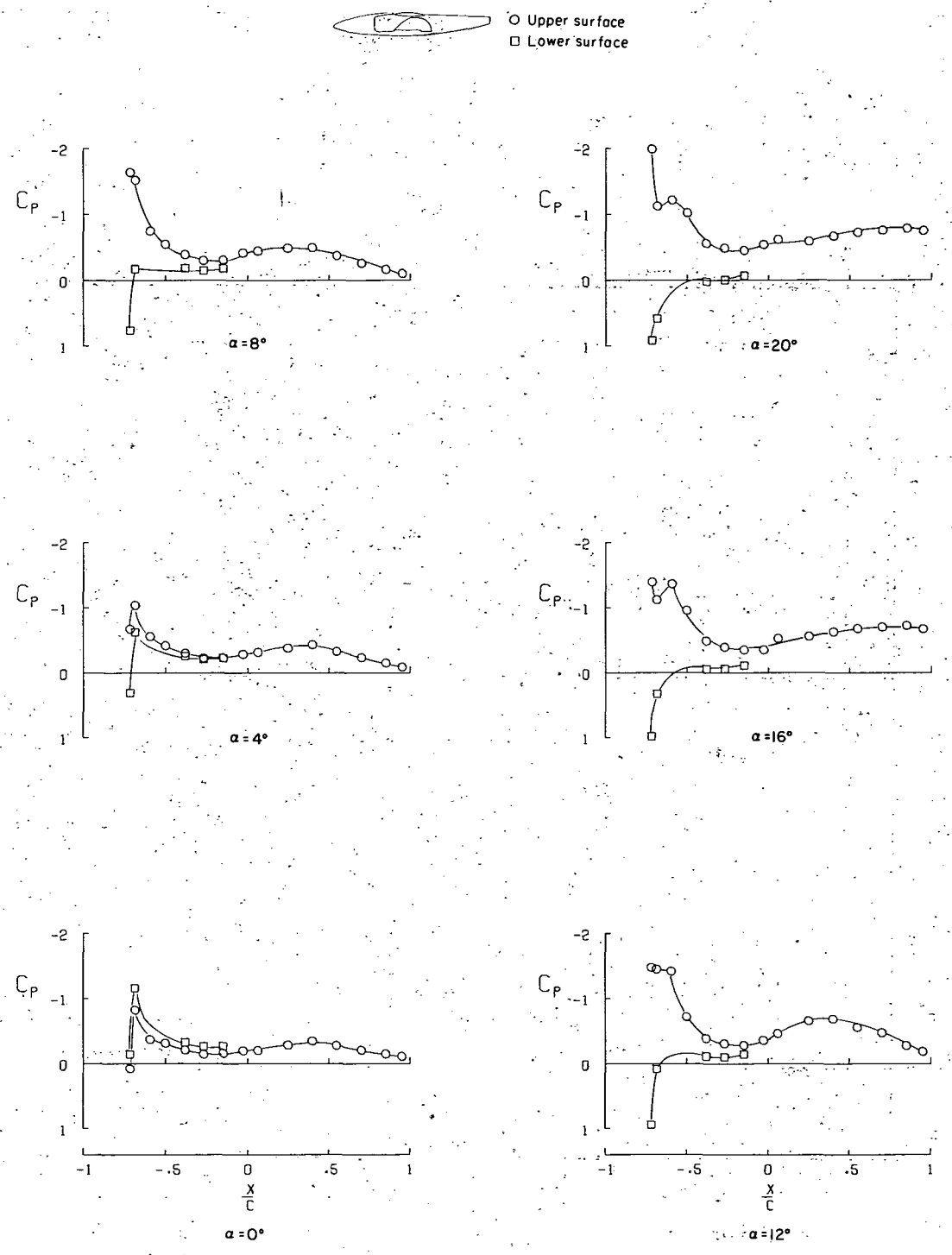
(c) $(V_\infty/V_j)_e \approx 0.2$.

Figure 24.- Continued.



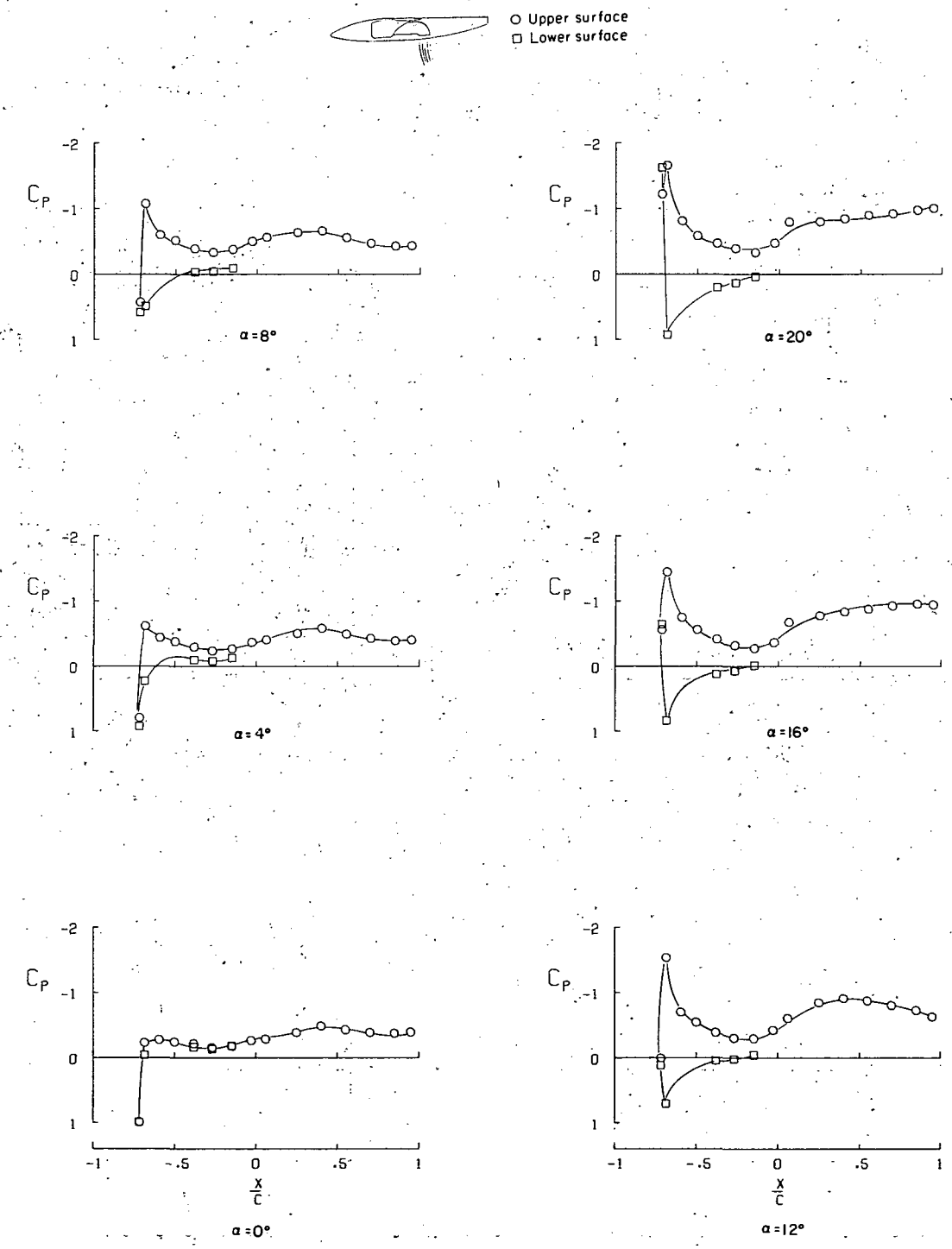
(d) $\left(V_\infty/V_j\right)_e \approx 0.1$.

Figure 24.- Concluded.



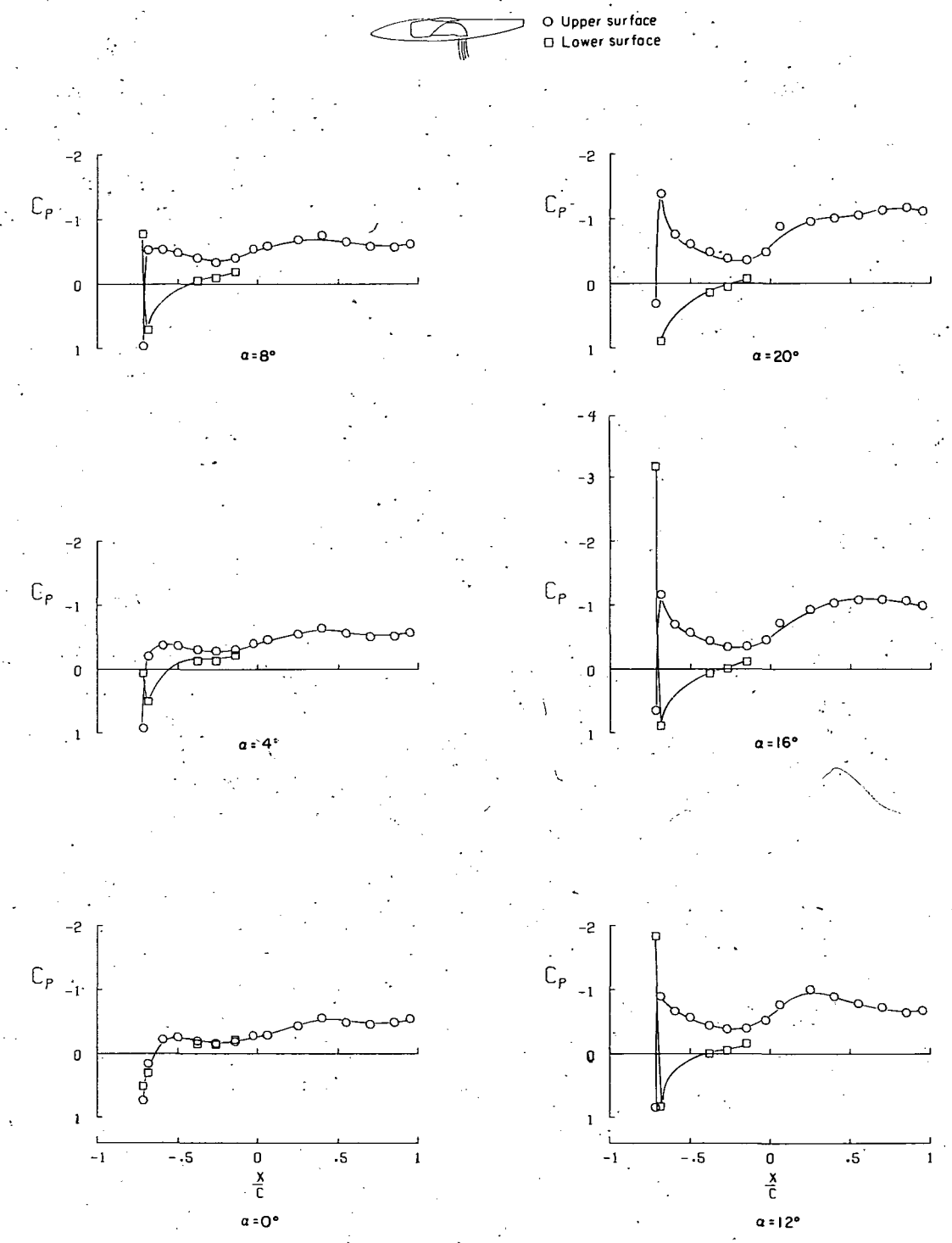
(a) $\left(V_\infty / V_j\right)_e = \infty$.

Figure 25.- Pressure distributions on the nacelle for the rear 90° nozzle configuration.



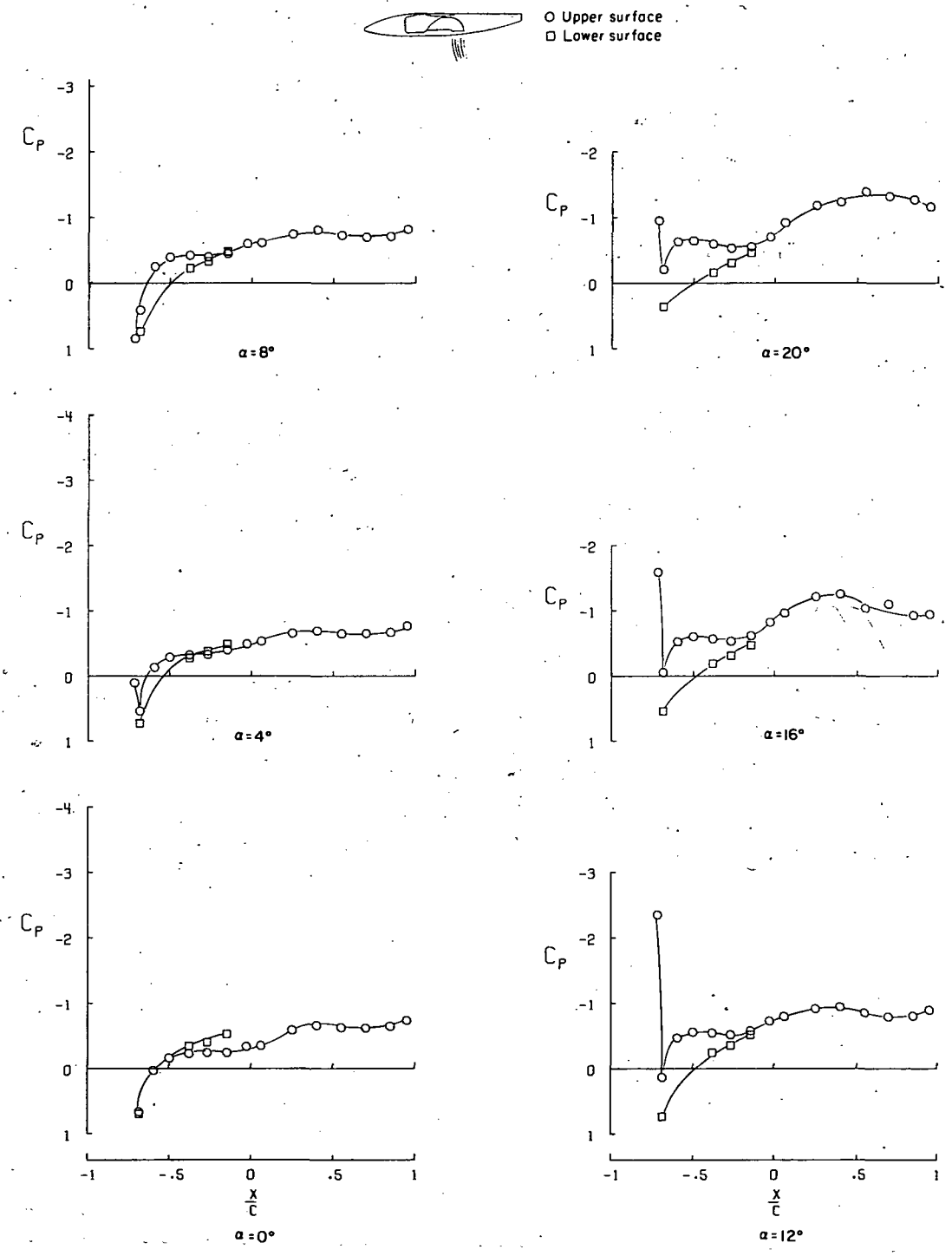
(b) $(V_\infty/V_j)_e \approx 0.3$.

Figure 25.- Continued.



(c) $\left(V_\infty/V_j\right)_e \approx 0.2$.

Figure 25.- Continued.



(d) $\left(V_\infty/V_j\right)_e \approx 0.1$.

Figure 25.- Concluded.

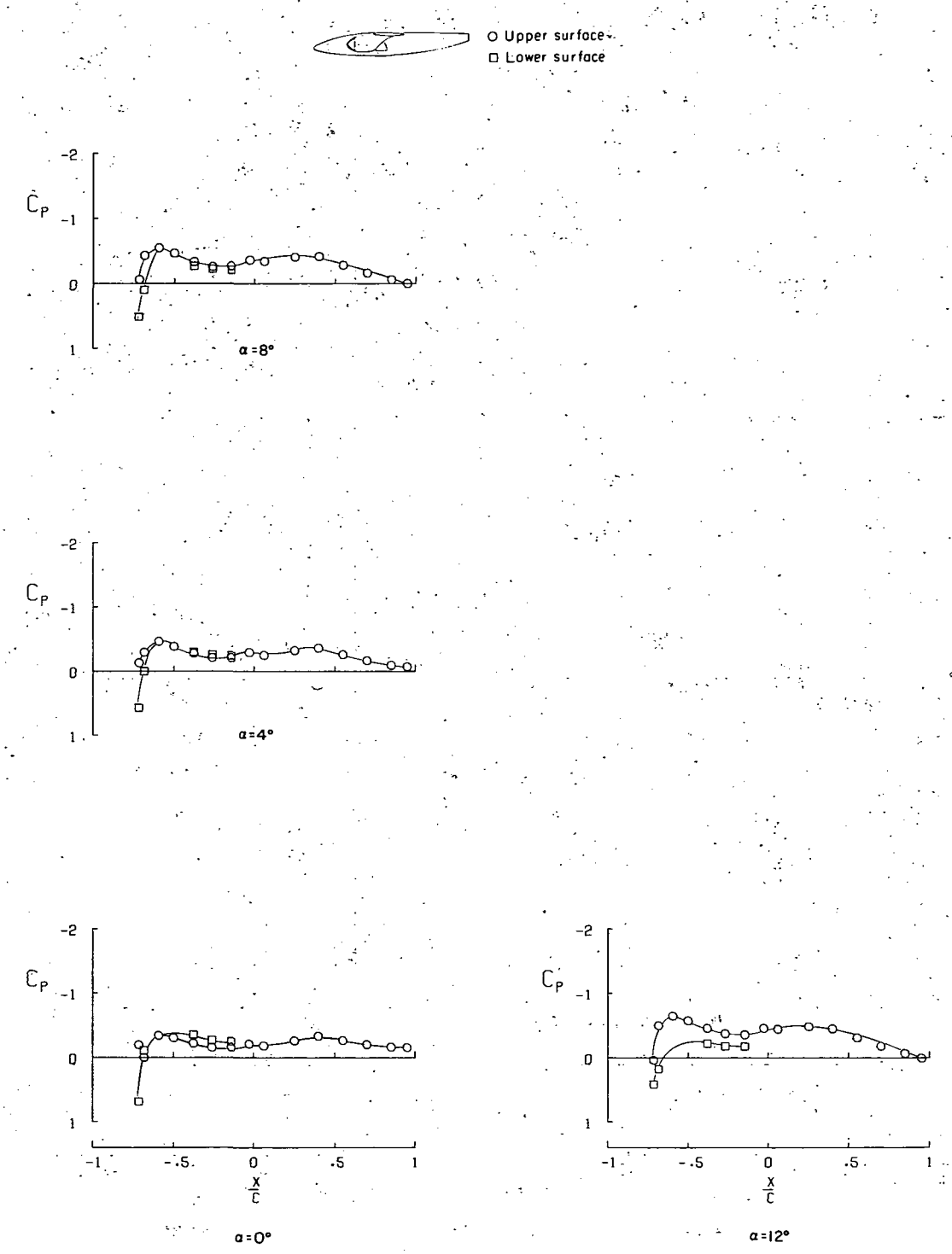
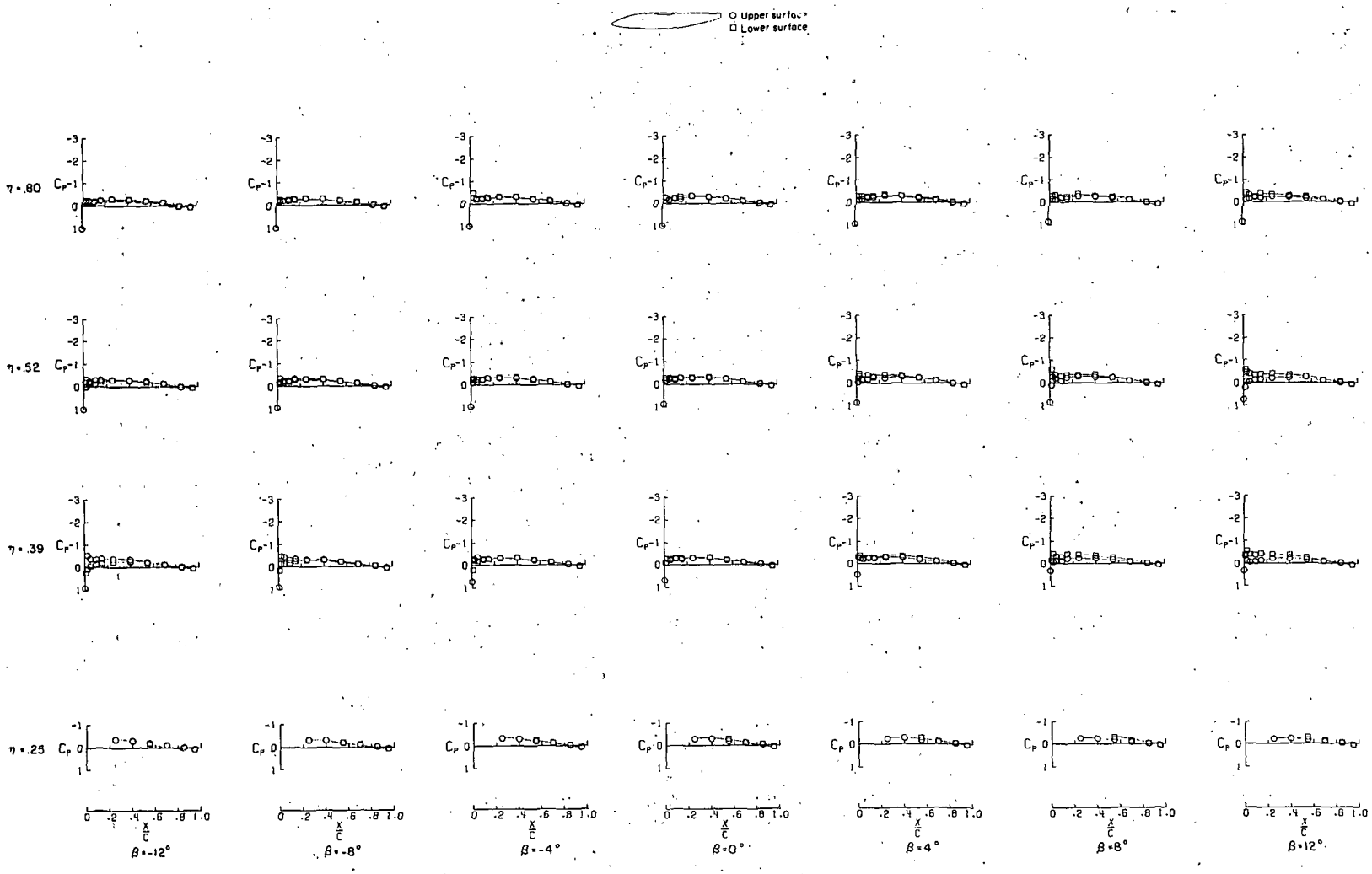
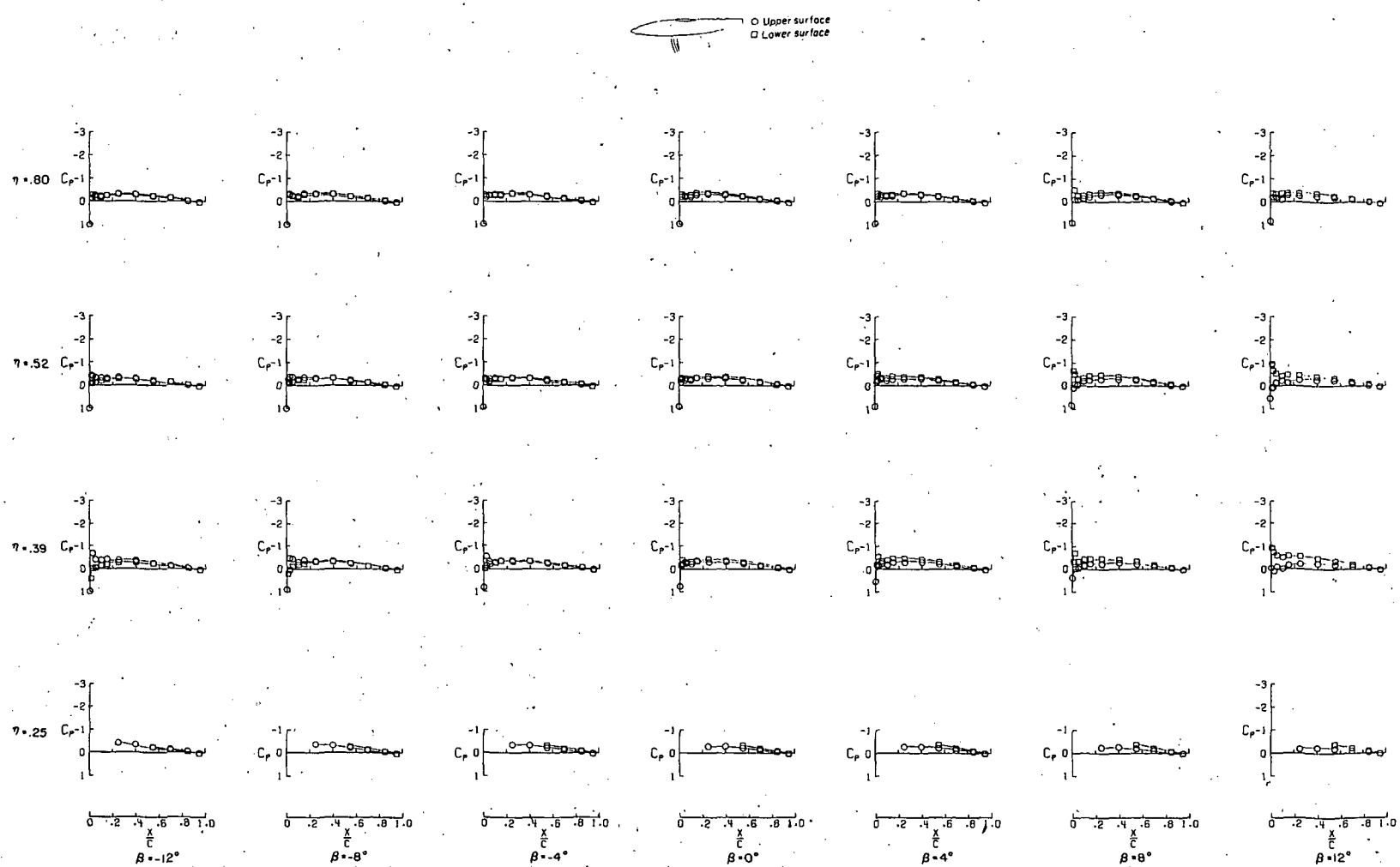


Figure 26.- Pressure distributions on the nacelle for the forward 90° nozzle configuration with inlets closed. $\left(\frac{V_\infty}{V_j}\right)_e = \infty$.



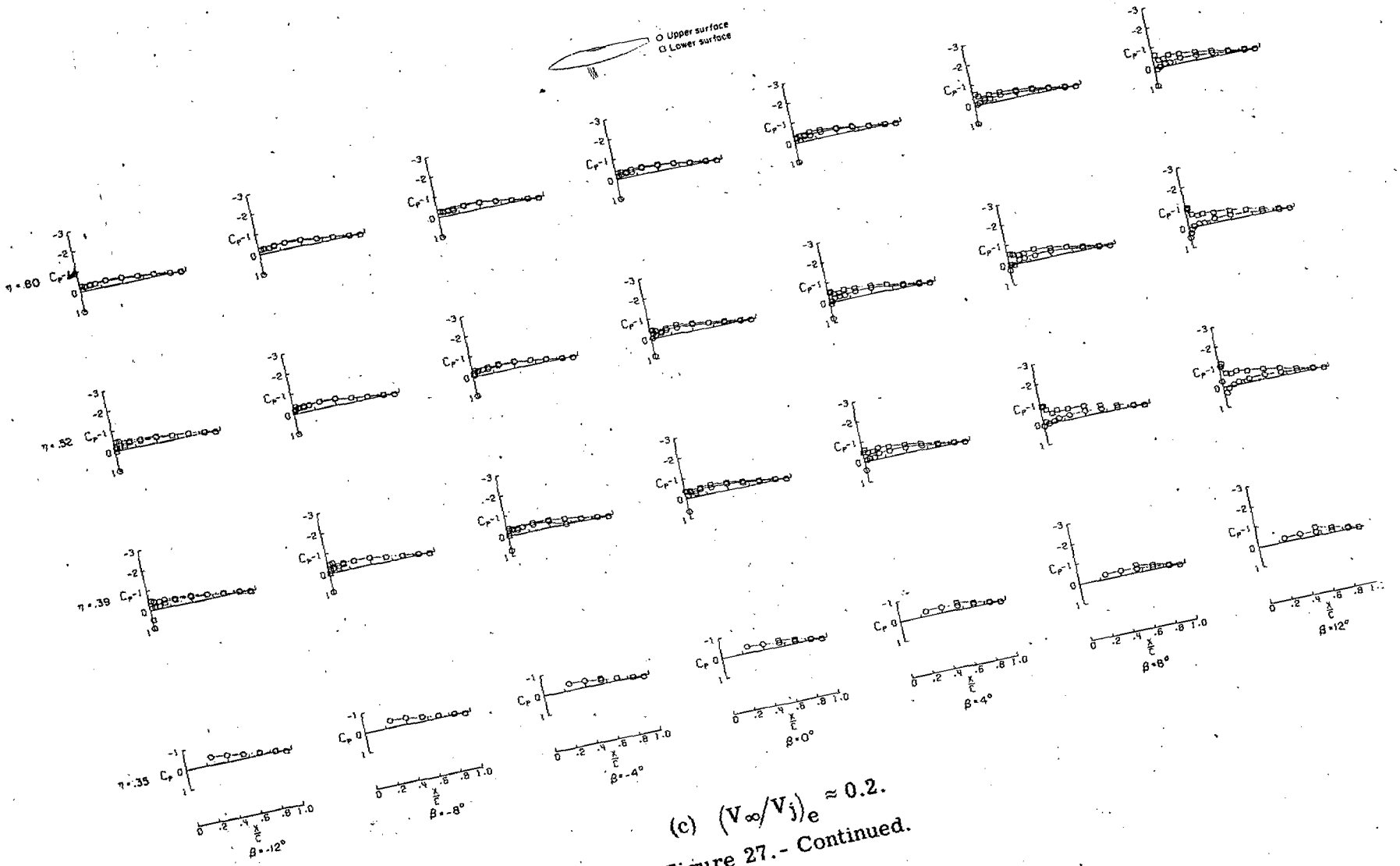
(a) $(V_\infty/V_j)_e \approx \infty$.

Figure 27.- Pressure distributions on the wing for the lift-jet configuration at an angle of attack of 0^o.



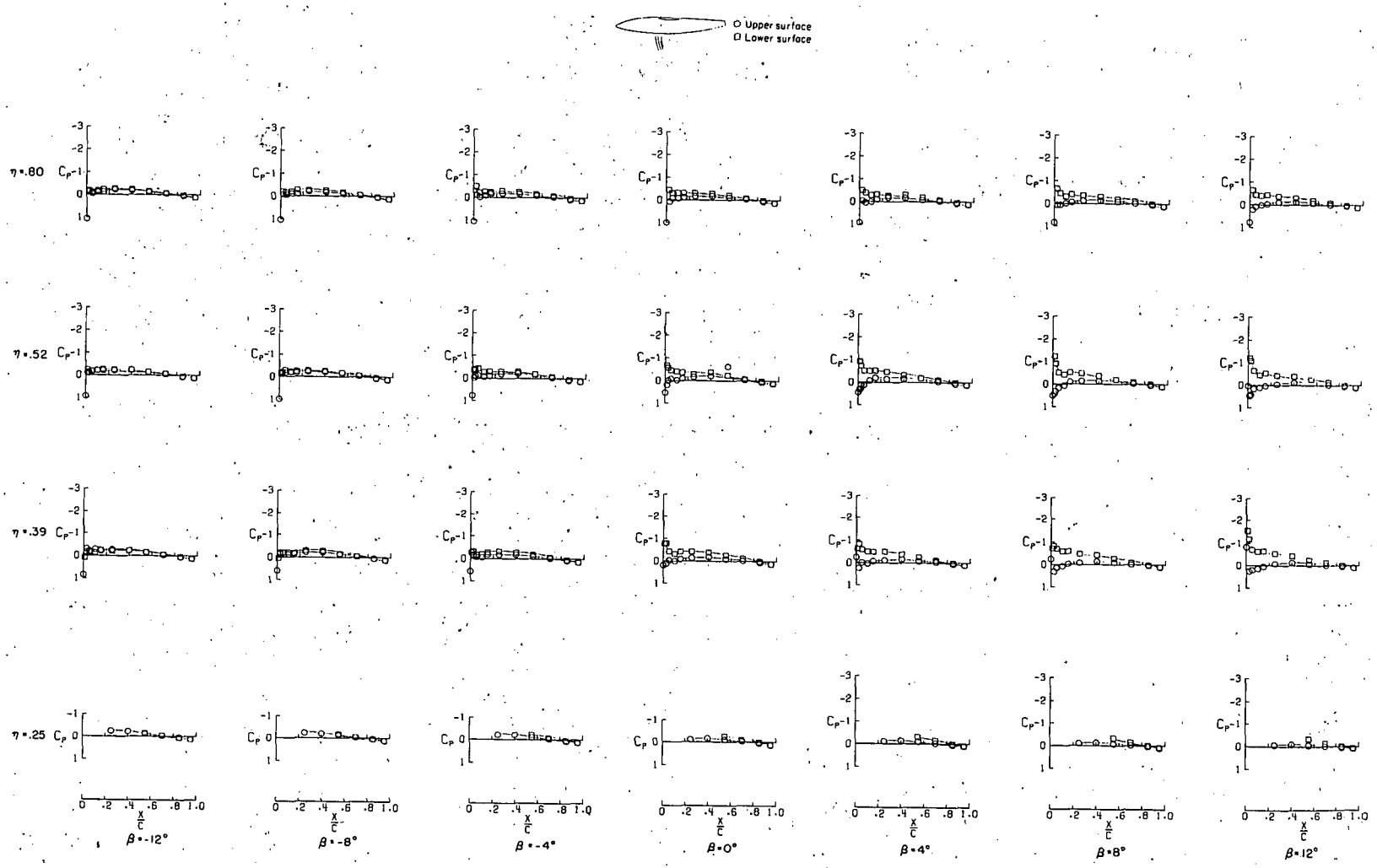
(b) $\left(V_\infty/V_j\right)_e \approx 0.3$.

Figure 27.- Continued.



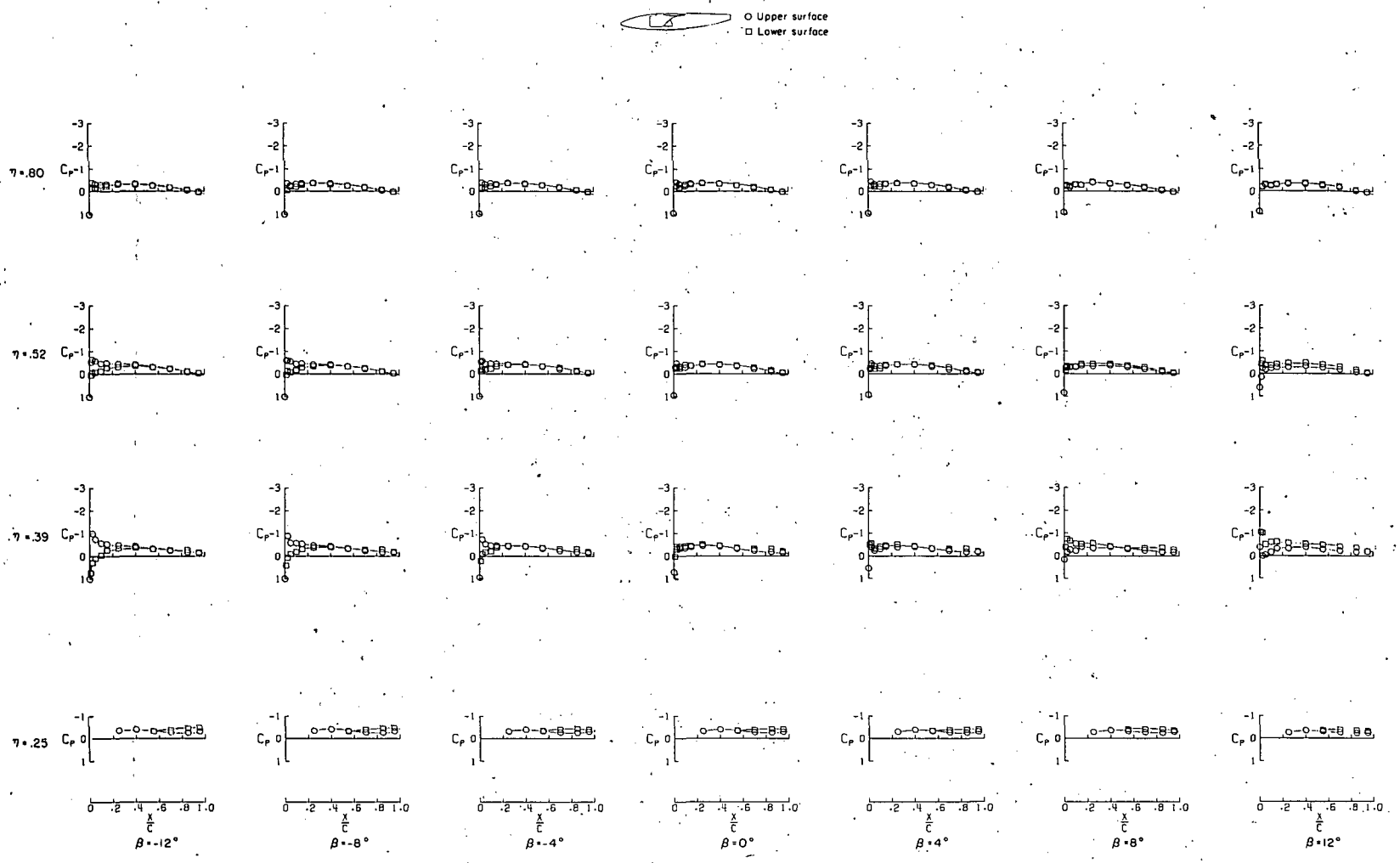
(c) $(V_\infty/V_j)_e \approx 0.2$.

Figure 27.- Continued.



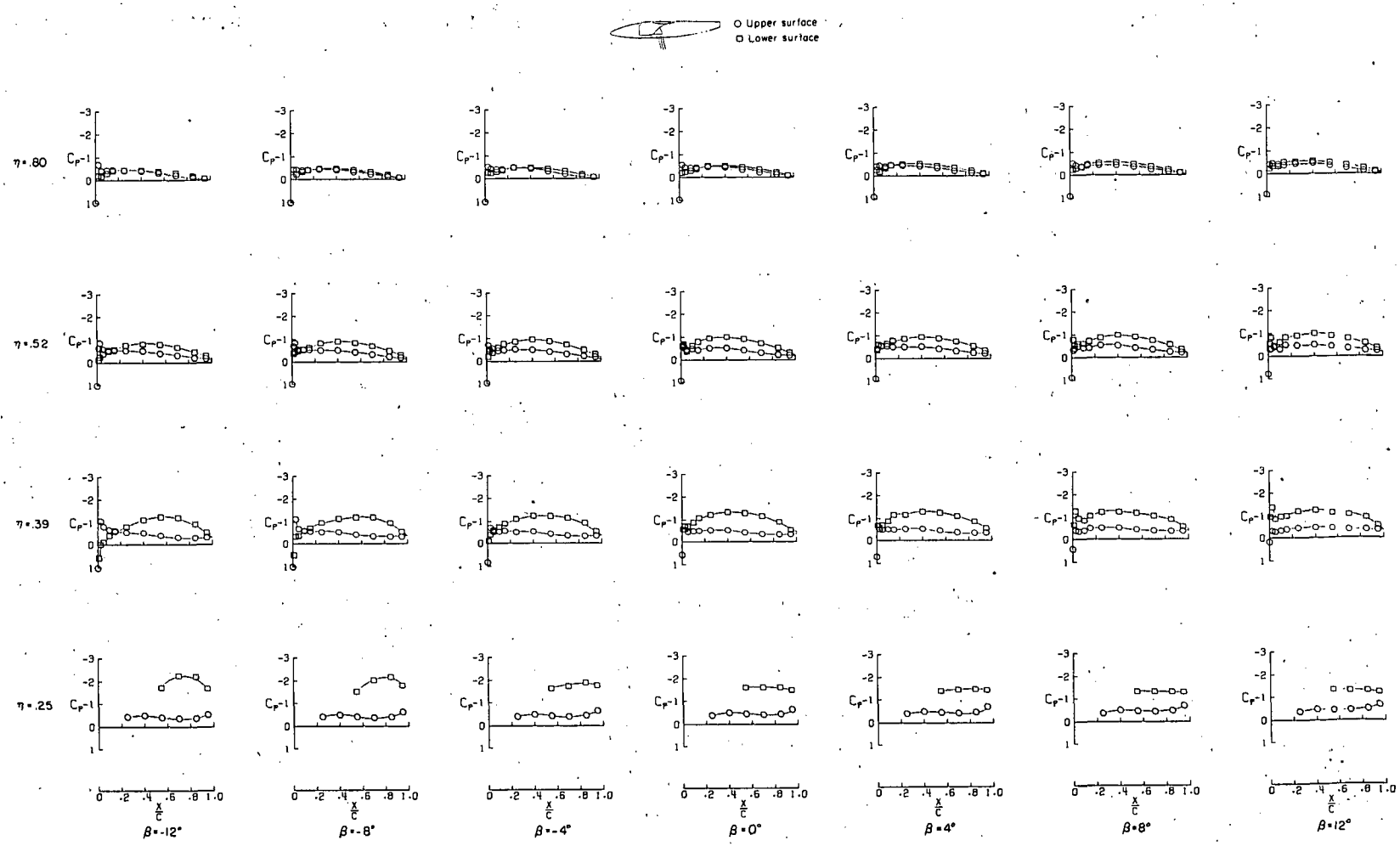
(d) $\left(V_\infty/V_j\right)_e \approx 0.1$.

Figure 27.- Concluded.



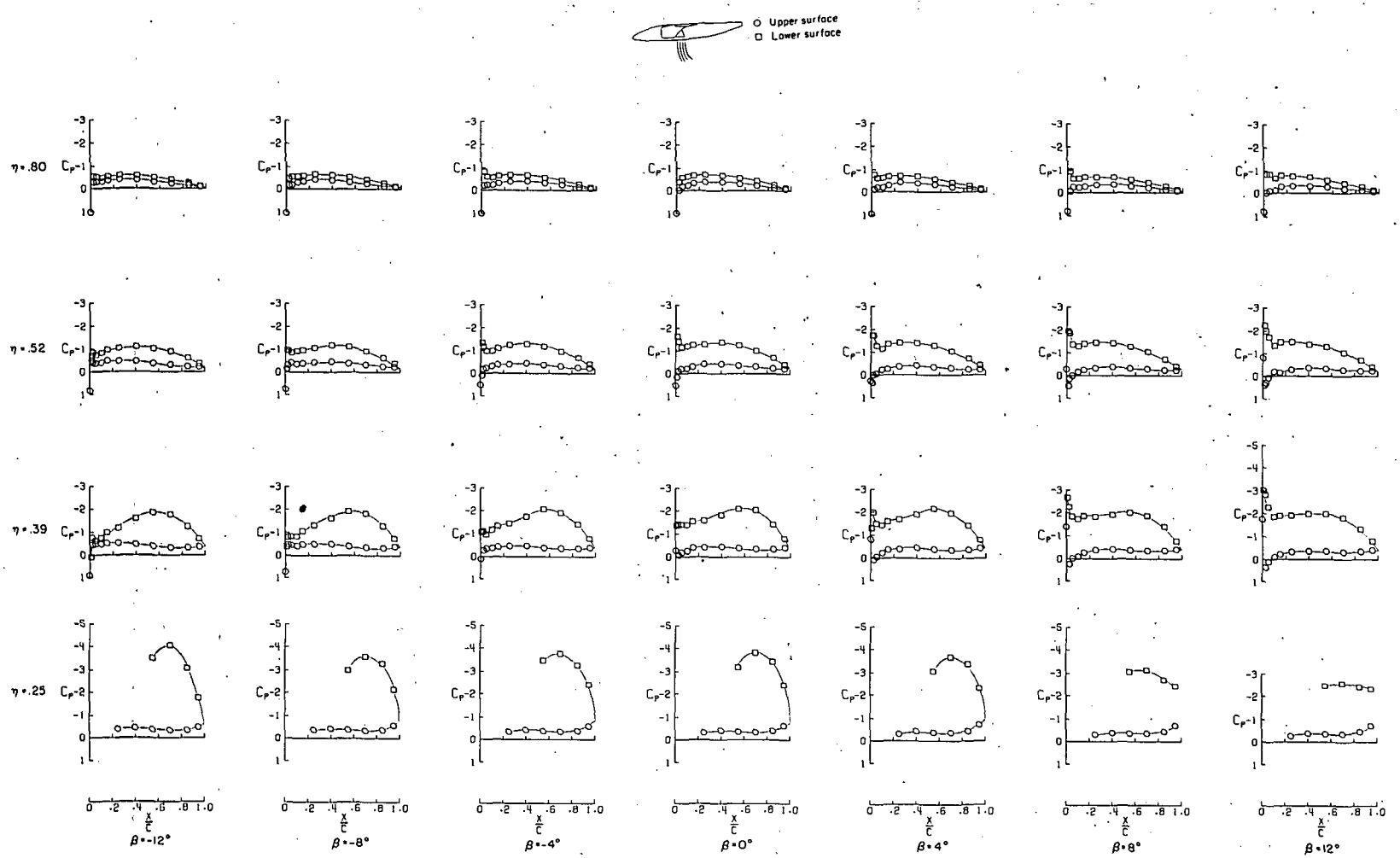
(a) $\left(V_\infty/V_j\right)_e = \infty$.

Figure 28.- Pressure distributions on the wing for the forward 90° nozzle configuration at an angle of attack of 0°.



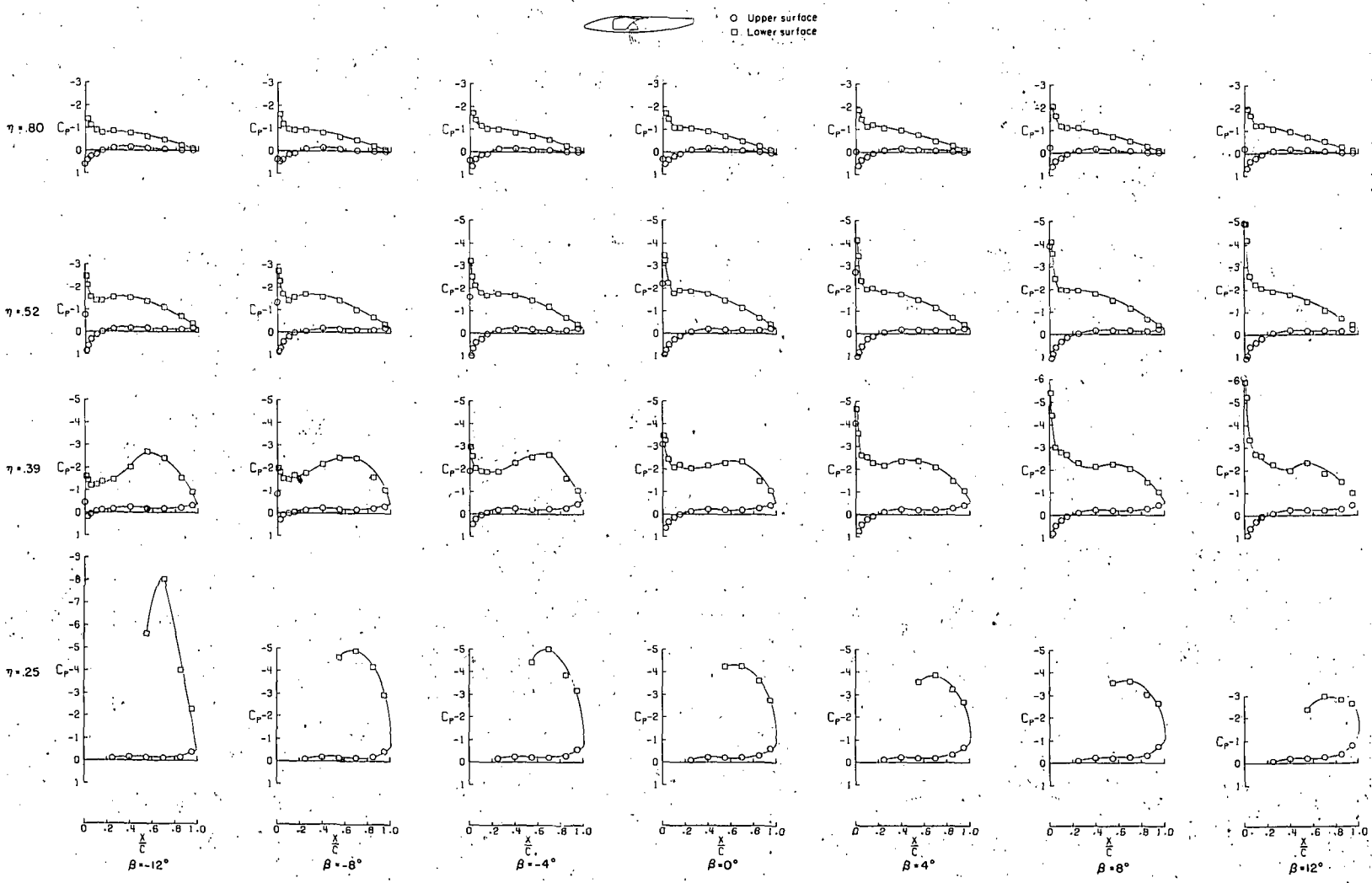
(b) $\left(V_\infty/V_j\right)_e \approx 0.3$.

Figure 28.- Continued.



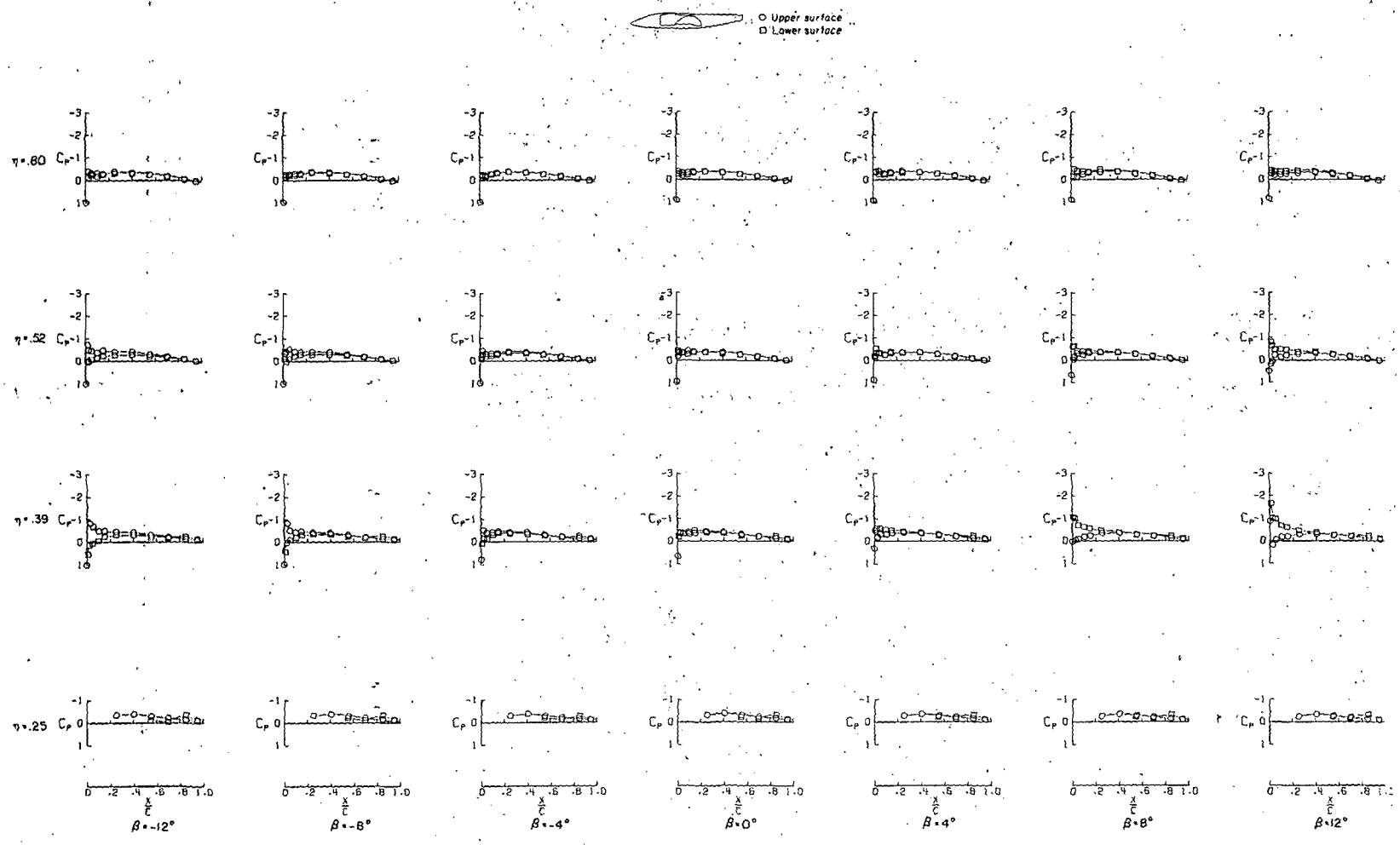
(c) $\left(V_\infty/V_j\right)_e \approx 0.2$.

Figure 28.- Continued.



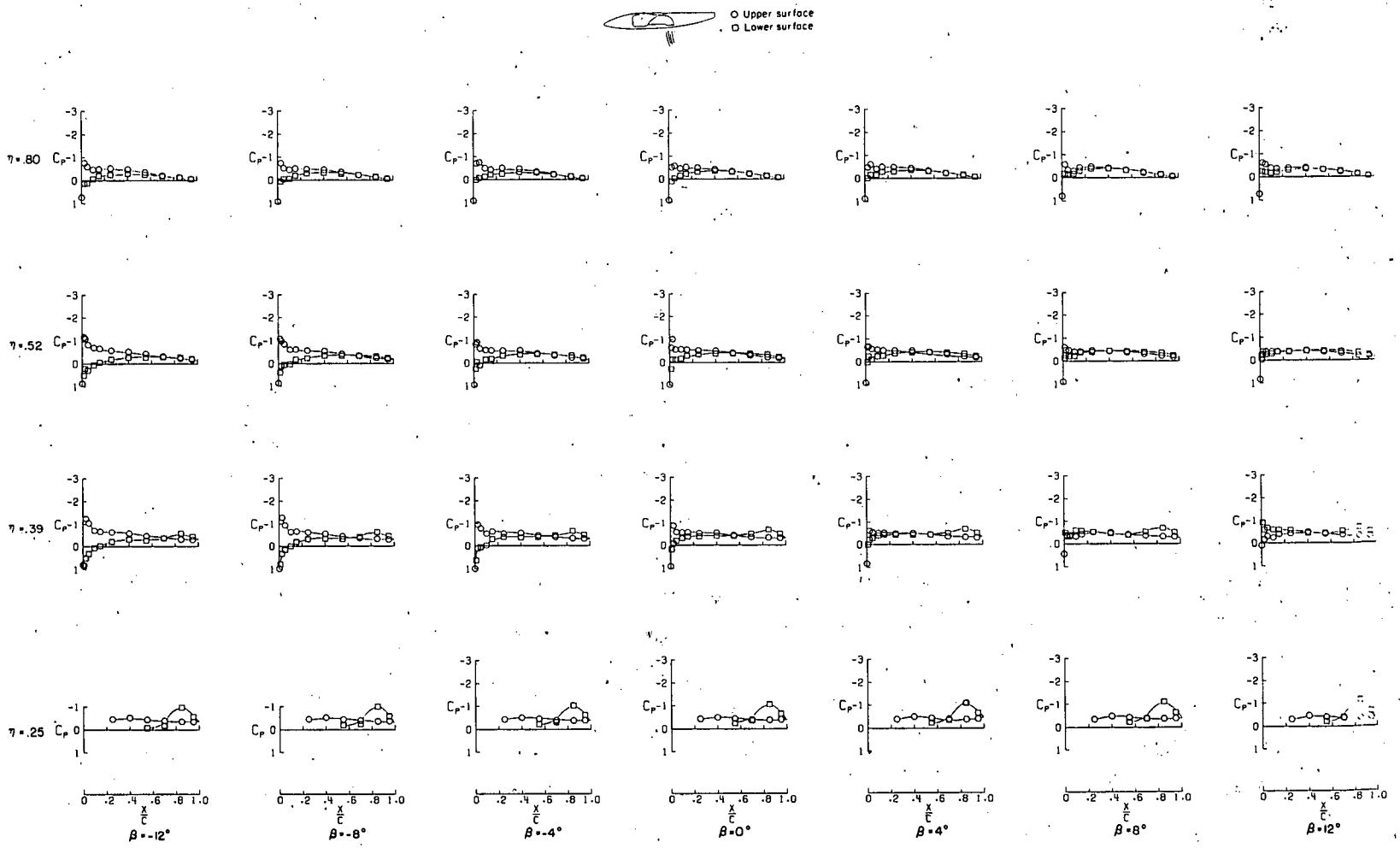
(d) $\left(V_\infty/V_j\right)_e \approx 0.1$.

Figure 28.- Concluded.



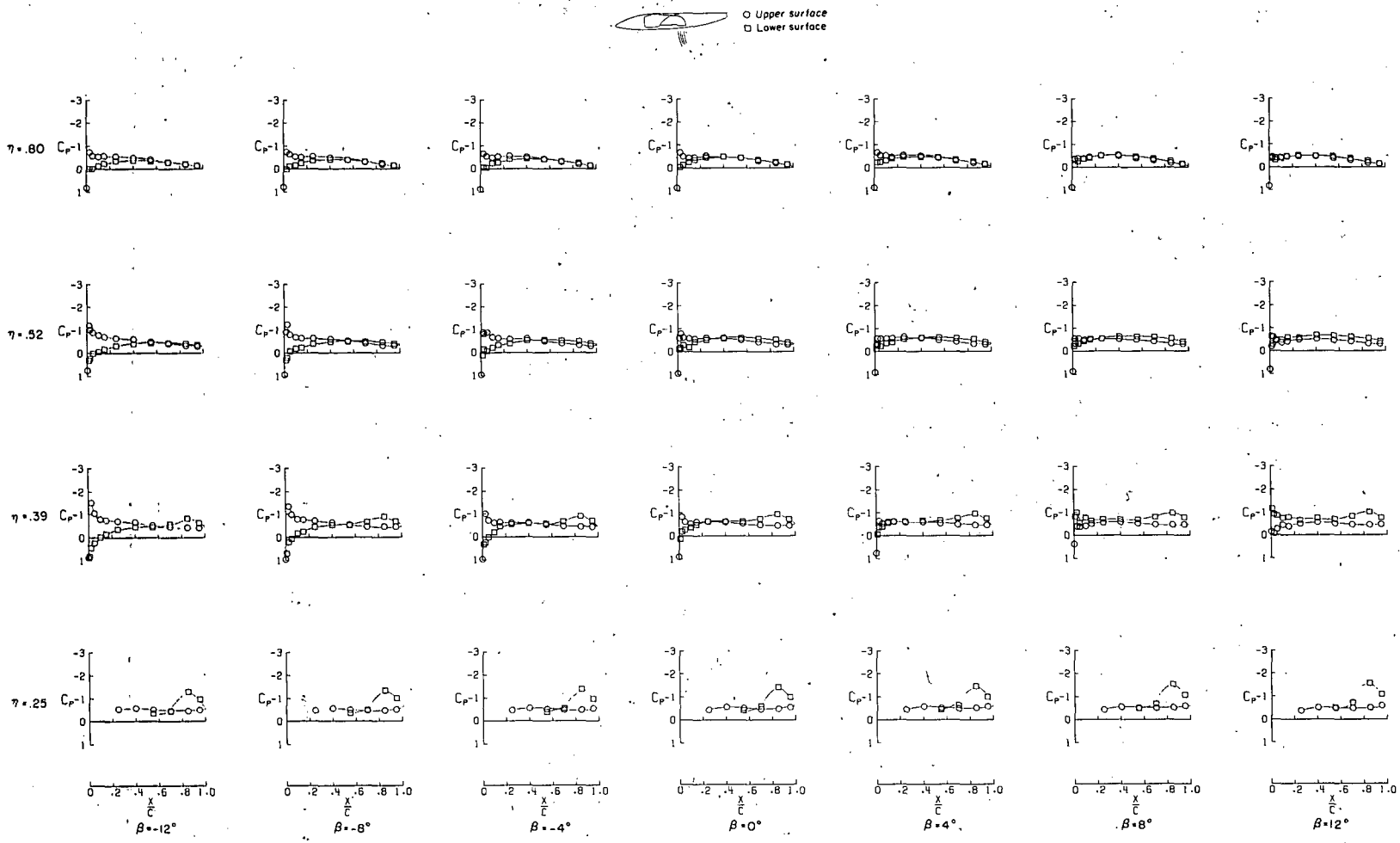
(a) $(V_\infty/V_j)_e \approx \infty$.

Figure 29.- Pressure distributions on the wing for the rear 90° nozzle configuration at an angle of attack of 0°.



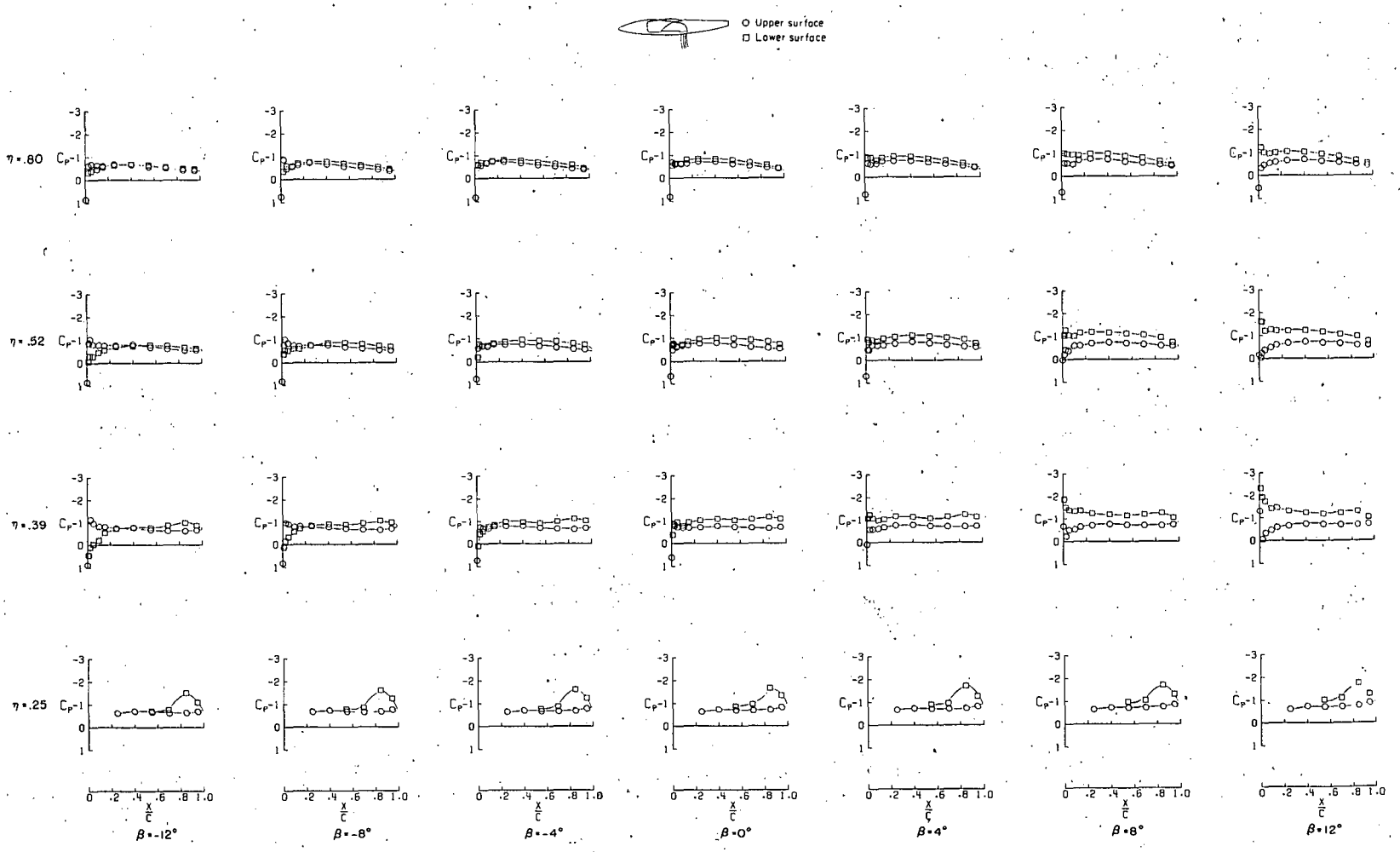
(b) $(V_\infty/V_j)_e \approx 0.3$.

Figure 29.- Continued.



(c) $(V_\infty/V_j)_e \approx 0.2$.

Figure 29.- Continued.



(d) $(V_\infty/V_j)_e \approx 0.1$.

Figure 29.- Concluded.

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